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# **RESEARCH MEMORANDUM**

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LOW-SPEED PRESSURE-DISTRIBUTION INVESTIGATION OF A  
THIN-DELTA-WING—FUSELAGE MODEL HAVING  
DOUBLE SLOTTED FLAPS AND SPOILERS

By Delwin R. Croom and Jarrett K. Huffman

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Langley Field, Va.

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**NATIONAL ADVISORY COMMITTEE  
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## RESEARCH MEMORANDUM

LOW-SPEED PRESSURE-DISTRIBUTION INVESTIGATION OF A  
THIN-DELTA-WING—FUSELAGE MODEL HAVING  
DOUBLE SLOTTED FLAPS AND SPOILERS

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## SUMMARY

An investigation was made in the Langley 300 MPH 7- by 10-foot tunnel to determine the chordwise and spanwise load distribution on a thin  $60^{\circ}$  delta-wing—fuselage model equipped with double slotted flaps having spoilers mounted on the flap. The wing had an aspect ratio of 2.31, taper ratio of 0, sweep of  $60^{\circ}$  at the leading edge, and NACA 65A003 airfoil sections parallel to the free stream.

Results of the investigation, without discussion, are presented in the form of lateral aerodynamic characteristics of the plain-wing and double-slotted-flap configurations and in the form of normal-force coefficients and pitching-moment coefficients of the wing, vane, flap, and spoiler; sample pressure plots; sample span-load distributions; tabulated pressure coefficients; and tabulated section normal-force coefficients and section pitching-moment coefficients.

## INTRODUCTION

Interest is being shown in the use of delta wings for high-speed airplanes because this plan form has some desirable aerodynamic and structural characteristics. Results of previous investigations (for example, see refs. 1 and 2) indicate that, by employing double slotted flaps on a  $60^{\circ}$  delta wing, the angle of attack necessary to obtain a given lift coefficient is considerably reduced, thereby making the use of double slotted flaps desirable for the landing configuration. Reference 3 indicates that spoiler-type controls located on the flap of a  $60^{\circ}$  delta wing having double slotted flaps gave rolling-moment coefficients that varied fairly linearly with spoiler projection and were about the same magnitude for the flap-retracted or the flap-deflected condition. The present investigation was made in the Langley 300 MPH 7- by 10-foot



tunnel to determine the loads at several spanwise stations on a  $60^{\circ}$  delta-wing model having double slotted flaps with spoilers located on the flaps. This investigation is an extension of the investigation reported on in reference 4.

The present paper presents the results, without discussion, of the investigation in the form of lateral aerodynamic characteristics of the complete model with flaps extended and retracted, and in the form of normal-force and pitching-moment coefficients of the wing, vane, flap, and spoiler. Included are sample pressure plots and span-load distributions, tabulated pressure coefficients, and tabulated section normal-force and section pitching-moment coefficients.

#### SYMBOLS

b	wing span (based on theoretical tip), ft (fig. 1)
$b_f$	flap span (exposed span of one flap), ft
$b_v$	vane span (exposed span of one vane), ft
$b_s$	spoiler span (exposed span of spoiler), ft
c	chord, ft
$\bar{c}$	wing mean aerodynamic chord (based on theoretical tip), $\frac{2}{S} \int_0^{b/2} c_w^2 dy, \text{ ft (fig. 1)}$
$c_{av}$	average wing chord, ft
$c_v$	vane chord, ft
$c_f$	flap chord, ft
$c_w$	plain-wing chord, ft
$c_s$	spoiler chord, ft
$c_F$	fuselage length, ft
S	wing area (based on theoretical tip), sq ft (fig. 1)

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$S_v$	vane area, sq ft
$S_f$	flap area, sq ft
$S_s$	spoiler area, sq ft
$C_L$	lift coefficient, $\frac{\text{Lift of model}}{q_0 S}$
$C_D$	drag coefficient, $\frac{\text{Drag of model}}{q_0 S}$
$\Delta C_D$	Jet-boundary correction applied to drag coefficient
$C_m$	pitching-moment coefficient of model referred to quarter mean aerodynamic chord, $\frac{\text{Pitching moment of model}}{q_0 S \bar{c}}$
$C_l$	rolling-moment coefficient due to spoilers
$C_n$	yawing-moment coefficient due to spoilers
$C_p$	pressure coefficient, $\frac{H_0 - p}{q_0}$
$h_s$	spoiler projection, ft
$l_v$	distance from wing quarter chord to vane nose, measured parallel to vane chord, ft
$l_f$	distance from wing quarter chord to flap nose measured parallel to flap chord, ft
$l_s$	distance from wing quarter chord to spoiler hinge line measured parallel to spoiler chord, ft
$x$	longitudinal distance, ft
$x_v$	distance from vane nose to center of load on vane, ft
$x_f$	distance from flap nose to center of load on flap, ft
$x_s$	distance from spoiler hinge line to center of load on spoiler, ft

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$y$	lateral distance, ft
$z$	vertical distance, ft
$H_0$	free-stream total pressure, lb/sq ft
$p$	local static pressure, lb/sq ft
$q_0$	free-stream dynamic pressure, $\frac{\rho V_0^2}{2}$ , lb/sq ft
$\rho$	mass density of air, slugs/cu ft
$V_0$	free-stream velocity, ft/sec
$\delta_f$	flap deflection (positive direction, trailing edge down), deg
$\delta_v$	vane deflection (angle between vane chord line and wing chord line; positive direction, trailing edge down), deg (fig. 2)
$\delta_s$	spoiler deflection (angle between spoiler chord and wing chord line; positive direction, trailing edge down), deg
$\alpha$	angle of attack set in tunnel
$\Delta\alpha$	jet-boundary correction applied to angle of attack
$\alpha_c$	corrected angle of attack
$c_{n,v}$	vane section normal-force coefficient, based on vane chord
$c_{n,f}$	flap section normal-force coefficient, based on flap chord
$c_{n,s}$	spoiler section normal-force coefficient, based on spoiler chord
$c_{n,WF}$	section normal-force coefficient of wing forward of slot lip, based on plain-wing chord
$c_{n,w}$	wing section normal-force coefficient, based on plain-wing chord (chord force of vane, flap, and spoiler neglected), $c_{n,WF} + c_{n,v}\left(\frac{c_v}{c_w}\right)\cos \delta_v + c_{n,f}\left(\frac{c_f}{c_w}\right)\cos \delta_f + c_{n,s}\left(\frac{c_s}{c_w}\right)\cos \delta_s$
$c_{m,v}$	vane section pitching-moment coefficient, based on vane chord (moments taken about vane nose)

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$c_{m,f}$	flap section pitching-moment coefficient, based on flap chord (moments taken about flap nose)
$c_{m,s}$	spoiler section pitching-moment coefficient, based on spoiler chord (moments taken about spoiler hinge line)
$c_{m,WF}$	section pitching-moment coefficient of wing forward of slot lip, based on plain-wing chord - (moments taken about wing quarter chord)
$c_{m,w}$	wing section pitching-moment coefficient, based on plain-wing chord (moments taken about wing quarter chord; chord force of vane, flap, and spoiler neglected), $c_{m,WF} = \frac{c_{n,v}(l_v + x_v)c_v}{c_w^2} - \frac{c_{n,f}(l_f + x_f)c_f}{c_w^2} - \frac{c_{n,s}(l_s + x_s)c_s}{c_w^2}$
$C_{N,w}$	wing normal-force coefficient (chord force of vane, flap, and spoiler neglected), $\frac{\text{Wing normal force}}{q_0 S}$
$C_{N,v}$	vane normal-force coefficient, $\frac{\text{Vane normal force}}{q_0 S_v}$
$C_{N,f}$	flap normal-force coefficient, $\frac{\text{Flap normal force}}{q_0 S_f}$
$C_{N,s}$	spoiler normal-force coefficient, $\frac{\text{Spoiler normal force}}{q_0 S_s}$
$C_{m,w}$	pitching-moment coefficient of wing, referred to quarter mean aerodynamic chord (chord force of vane, flap, and spoiler neglected), $\frac{\text{Pitching moment of wing}}{q_0 S_c}$
$C_{m,v}$	pitching-moment coefficient of vane, referred to vane nose, $\frac{\text{Pitching moment of vane}}{q_0 S_v c_v}$
$C_{m,f}$	pitching-moment coefficient of flap, referred to flap nose, $\frac{\text{Pitching moment of flap}}{q_0 S_f c_f}$
$C_{m,s}$	pitching-moment coefficient of spoiler, referred to spoiler hinge line, $\frac{\text{Pitching moment of spoiler}}{q_0 S_s c_s}$

## MODEL AND APPARATUS

The model was tested on the single-support-strut system in the Langley 300 MPH 7- by 10-foot tunnel. The geometric and physical characteristics of the wing-fuselage configuration are given in figure 1 and table I.

The wing of the model had a  $60^{\circ}$  apex angle, an aspect ratio of 2.31 (based on the theoretical tip), a taper ratio of 0, and an NACA 65A003 airfoil section parallel to the free stream.

The double-slotted-flap configuration used for this investigation is shown in figure 2. The general arrangement, that is, relation of flap to vane to wing, were obtained from preliminary explorative tests based on the information of the systematic investigations of references 2 and 5.

The flap which extended from the fuselage to  $0.67b/2$  had a constant chord of 6.86 inches and exposed area equal to 12.78 percent of the total wing area. The flap leading edge was constructed to the ordinates given in table II. The vane had a constant chord of 1.768 inches and was constructed to the ordinates given in table III. The vane and flap were deflected as a unit about the pivot point shown in figure 2. The spoiler which extended from the fuselage to  $0.67b/2$  had a constant chord of 0.10 $\bar{c}$  and was mounted with its hinge line perpendicular to the plane of symmetry. (See fig. 2.)

The wing, vane, flap, and spoiler were constructed with flush surface pressure orifices located on the right semispan at the 21-, 30-, 43-, and 55-percent-semispan stations. Orifices were also located on the wing at the 72-percent-semispan station and along the fuselage at the plane of symmetry.

TESTS

The tests were performed at a dynamic pressure of approximately 25 pounds per square foot which corresponds to a Mach number of approximately 0.13. Reynolds number based on the mean aerodynamic chord of the model was approximately  $2.7 \times 10^6$ . The tests were made through an angle-of-attack range of approximately  $-40^{\circ}$  through the stall. Flap deflection for tests with the double-slotted-flap configuration was  $60^{\circ}$ . Force data were obtained on the plain-wing configuration with the spoiler hinged at the 70.0-, 89.4-, and 93.3-percent-root-chord location for several spoiler projections and on the double-slotted-flap configuration at the 89.4- and 93.3-percent-root-chord location for a spoiler projection of -0.05 $\bar{c}$ . Pressure distributions were obtained for the plain-wing

configuration and the double-slotted-flap configuration with the spoiler located at the 93.3-percent-root-chord location at spoiler deflections of -0.5-, -1.0-, -3.5-, -5.0-, and -10.0-percent  $\bar{c}$ .

#### CORRECTIONS

The following jet-boundary corrections applied to the data of this paper were obtained by the method outlined in reference 6:

$$\Delta\alpha = 1.028C_L$$

$$\Delta C_D = 0.0179C_L^2$$

The blockage correction as applied to the dynamic pressure was obtained by the method outlined in reference 7. The buoyancy correction due to the longitudinal static pressure gradient in the tunnel as applied to the data increased the drag coefficient by 0.001.

#### RESULTS

The results of this investigation are presented without discussion. Sample data figures are presented in order to give the reader a general idea of the chordwise and spanwise loadings to expect over a delta wing that is equipped with double slotted flaps when spoilers are used on the flap for lateral control.

The model aerodynamic characteristics in pitch for the plain-wing and double-slotted-flap configuration have been presented in reference 1. The pressure-distribution data of the plain-wing and double-slotted-flap configurations without spoilers have been presented in reference 4. The data in the form of figures and tables are as listed below.

	Figures
Lateral aerodynamic data of plain-wing configuration . . . . .	3
Lateral aerodynamic data of double-slotted-flap	
configuration . . . . .	4
Sample chordwise pressure distribution . . . . .	5
Sample span-load distribution on wing, vane, flap, and	
spoiler at several spoiler projections . . . . .	6 to 9
Normal-force and pitching-moment coefficients of wing,	
vane, flap, and spoiler . . . . .	10 to 15

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The pressure coefficients are presented in tables IV to XIII.

The section data are presented in tables XIV and XV.

Langley Aeronautical Laboratory,  
National Advisory Committee for Aeronautics,  
Langley Field, Va., November 23, 1956.

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1. Croom, Delwin R.: A Low-Speed Investigation of a Thin 60° Delta Wing Equipped With a Double Slotted Flap To Determine the Chordwise Pressure Distribution and the Effect of Vane Size. NACA RM L54I03a, 1955.
2. Riebe, John M., and MacLeod, Richard G.: Low-Speed Wind-Tunnel Investigation of a Thin 60° Delta Wing With Double Slotted, Single Slotted, Plain, and Split Flaps. NACA RM L52J29, 1953.
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7. Herriot, John G.: Blockage Corrections for Three-Dimensional-Flow Closed-Throat Wind Tunnels, With Consideration of the Effect of Compressibility. NACA Rep. 995, 1950. (Supersedes NACA RM A7B28.)

TABLE I.- PHYSICAL CHARACTERISTICS OF TEST MODEL

## Wing:

Section parallel to free stream . . . . .	NACA 65A003
Span, ft . . . . .	5.00
Aspect ratio (based on theoretical tip) . . . . .	2.31
Leading-edge sweep, deg . . . . .	60.00
Trailing-edge sweep, deg . . . . .	0
Area (based on theoretical tip), sq ft . . . . .	10.83
Mean aerodynamic chord, ft . . . . .	2.89
Root chord, ft . . . . .	4.33

## Vane:

Span, ft . . . . .	3.33
Chord, ft . . . . .	0.15
Chord, percent wing root chord . . . . .	3.40
Chord, percent flap chord . . . . .	25.77

## Flap:

Span, ft . . . . .	3.33
Chord, ft . . . . .	0.57
Chord, percent wing root chord . . . . .	13.20
Exposed area, sq ft . . . . .	1.38
Exposed area, percent wing area . . . . .	12.78

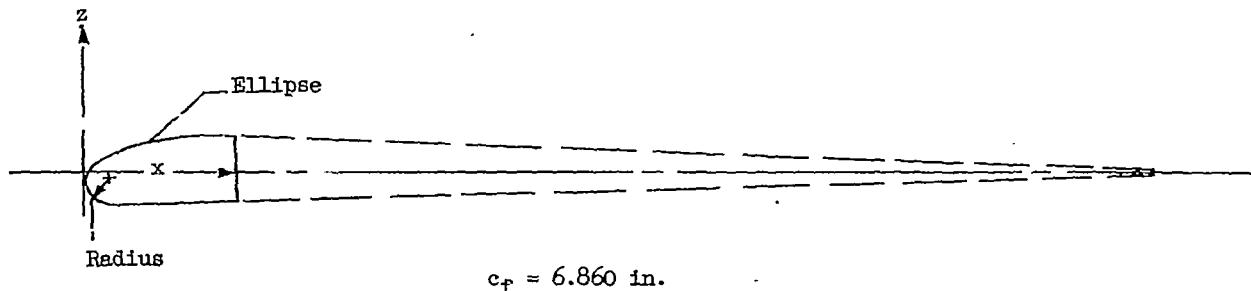
## Spoiler:

Span, ft . . . . .	3.33
Chord, ft . . . . .	0.29
Chord, percent mean aerodynamic chord . . . . .	10.0

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TABLE II.- ORDINATES OF THE LEADING EDGE OF THE TRAILING-EDGE FLAP

[All dimensions in inches]

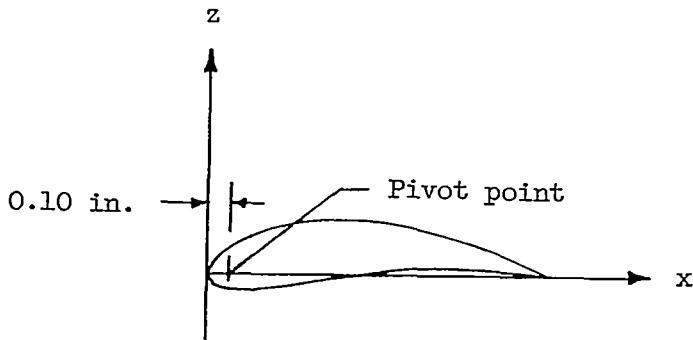


Station, in. x	Lower		Upper
	x	z	z
0		-0.107	-0.107
.010		Radius	-.059
.030			-.030
.050			-.005
.075			.022
.100			.041
.143		-.245	-----
.200		Straight taper	.096
.300			.133
.400			.161
.500			.182
.600			.198
.700			.208
.800			.215
.900			.217
1.000		-.216	.216

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TABLE III.- ORDINATES OF THE VANE

[All dimensions in inches]



$$c_v = 1.768 \text{ in.}$$

Station, in. x	Lower z	Upper z
0	0	0
.022	-.047	.067
.044	-.060	.092
.088	-.072	.131
.133	-.079	.160
.177	-.077	.185
.265	-.072	.224
.354	-.053	.255
.530	-.025	.288
.707	.002	.294
.884	.032	.283
1.061	.053	.255
1.238	.057	.207
1.414	.053	.147
1.591	.032	.080
1.680	.019	.046
1.768	0	0

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TABLE IV.- PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE

(a) Plain wing configuration with spoiler;  $h_s = -0.005\bar{c}$ ;  $\frac{V}{b/2} = 0$

x/c	Upper surface										Lower surface											
	$C_p$ for -											$C_p$ for -										
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 25^\circ$	Fuselage	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 25^\circ$	
Fuselage																						
.0000	.003	.000	.000	.000	.000	.019	.067	.099	.105	.122	.0500	.742	.837	.883	.953	.978	.104	.1051	.1081	.1095	.1095	
.0250	.024	.028	.063	.1025	.1034	.1094	.1113	.1102	.1151	.1153	.0500	.742	.837	.883	.953	.978	.104	.1051	.1081	.1095	.1095	
.1250	.934	1.000	1.015	1.075	1.083	1.125	1.143	1.135	1.157	1.156	.0500	.742	.837	.883	.953	.978	.104	.1051	.1081	.1095	.1095	
.2500	.972	1.041	1.061	1.110	1.099	1.138	1.134	1.123	1.142	1.144	.0500	.742	.837	.883	.953	.978	.104	.1051	.1081	.1095	.1095	
.3500	1.016	1.082	1.092	1.122	1.111	1.159	1.146	1.132	1.154	1.159	.0500	.742	.837	.883	.953	.978	.104	.1051	.1081	.1095	.1095	
.4500	1.050	1.092	1.092	1.122	1.105	1.134	1.113	1.093	1.136	1.131	.0500	.742	.837	.883	.953	.978	.104	.1051	.1081	.1095	.1095	
.5500	1.050	1.088	1.073	1.104	1.091	1.109	1.101	1.090	1.120	1.123	.0500	.742	.837	.883	.953	.978	.104	.1051	.1081	.1095	.1095	
.6500	1.022	1.046	1.037	1.063	1.062	1.086	1.080	1.070	1.082	1.086	.0500	.742	.837	.883	.953	.978	.104	.1051	.1081	.1095	.1095	
.7500	1.003	1.047	1.047	1.075	1.065	1.086	1.080	1.070	1.082	1.086	.0500	.742	.837	.883	.953	.978	.104	.1051	.1081	.1095	.1095	
.8500	1.028	1.075	1.061	1.097	1.102	1.133	1.156	1.186	1.207	1.250	.0500	.742	.837	.883	.953	.978	.104	.1051	.1081	.1095	.1095	
.9000	1.035	1.050	1.043	1.066	1.058	1.093	1.109	1.125	1.135	1.190	.0500	.742	.837	.883	.953	.978	.104	.1051	.1081	.1095	.1095	
.9300	1.028	1.072	1.055	1.084	1.065	1.097	1.101	1.105	1.151	1.147	.0500	.742	.837	.883	.953	.978	.104	.1051	.1081	.1095	.1095	
.9400	1.006	1.069	1.095	1.151	1.173	1.222	1.223	1.216	1.256	1.346	.0500	.742	.837	.883	.953	.978	.104	.1051	.1081	.1095	.1095	
.7500	1.016	1.066	1.086	1.135	1.148	1.191	1.207	1.213	1.253	1.315	.0500	.742	.837	.883	.953	.978	.104	.1051	.1081	.1095	.1095	
.8000	1.022	1.069	1.073	1.107	1.120	1.175	1.198	1.219	1.274	1.294	.0500	.742	.837	.883	.953	.978	.104	.1051	.1081	.1095	.1095	
.8500	1.028	1.075	1.061	1.097	1.102	1.133	1.156	1.186	1.207	1.250	.0500	.742	.837	.883	.953	.978	.104	.1051	.1081	.1095	.1095	
.9000	1.035	1.050	1.043	1.066	1.058	1.093	1.109	1.125	1.135	1.193	.0500	.742	.837	.883	.953	.978	.104	.1051	.1081	.1095	.1095	
.9300	1.028	1.072	1.055	1.084	1.065	1.097	1.101	1.105	1.151	1.147	.0500	.742	.837	.883	.953	.978	.104	.1051	.1081	.1095	.1095	
.9400	1.010	1.119	1.104	1.119	1.096	1.109	1.104	1.099	1.117	1.104	.0500	.742	.837	.883	.953	.978	.104	.1051	.1081	.1095	.1095	

TABLE IV.- PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(b) Plain wing configuration with spoiler;  $h_s = -0.005\bar{c}$ ;  $\frac{V}{b/2} = 0.21$

x/c	Upper surface										Lower surface											
	$C_p$ for -											$C_p$ for -										
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 25^\circ$	Wing	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 25^\circ$	
Wing																						
.0000	1.322	.827	1.230	2.592	4.490	6.828	9.400	10.402	8.604	4.103	.0125	1.209	.938	.806	.649	.586	.621	.697	.792	.704	.906	
.0250	.030	1.055	1.411	2.722	5.965	9.624	12.325	12.089	8.348	3.742	.0250	.782	.837	.883	.953	.978	.104	.1051	.1081	.1095	.1095	
.0500	.087	1.074	1.270	1.564	1.797	2.420	3.397	4.549	5.376	5.351	.0500	.782	.837	.883	.953	.978	.104	.1051	.1081	.1095	.1095	
.0750	.902	1.030	1.232	1.462	1.665	2.003	2.578	3.153	4.117	3.218	.0750	.782	.837	.883	.953	.978	.104	.1051	.1081	.1095	.1095	
.1000	.924	1.065	1.235	1.418	1.598	1.842	2.347	2.793	3.571	3.124	.1000	.782	.837	.883	.953	.978	.104	.1051	.1081	.1095	.1095	
.1250	.947	1.062	1.210	1.370	1.491	1.687	2.056	2.314	2.808	2.927	.1250	.782	.837	.883	.953	.978	.104	.1051	.1081	.1095	.1095	
.1500	.965	1.086	1.204	1.326	1.451	1.593	1.891	2.112	2.442	2.748	.1500	.782	.837	.883	.953	.978	.104	.1051	.1081	.1095	.1095	
.1750	.988	1.083	1.191	1.298	1.356	1.483	1.728	1.814	1.984	2.297	.1750	.782	.837	.883	.953	.978	.104	.1051	.1081	.1095	.1095	
.2000	.988	1.086	1.185	1.257	1.316	1.439	1.597	1.708	1.849	2.121	.2000	.782	.837	.883	.953	.978	.104	.1051	.1081	.1095	.1095	
.2500	1.020	1.086	1.166	1.250	1.324	1.417	1.518	1.621	1.751	1.985	.2500	.782	.837	.883	.953	.978	.104	.1051	.1081	.1095	.1095	
.3000	1.020	1.086	1.166	1.250	1.324	1.417	1.518	1.621	1.751	1.985	.3000	.782	.837	.883	.953	.978	.104	.1051	.1081	.1095	.1095	
.3500	1.029	1.099	1.176	1.258	1.329	1.414	1.515	1.616	1.757	1.987	.3500	.782	.837	.883	.953	.978	.104	.1051	.1081	.1095	.1095	
.4000	1.028	1.086	1.154	1.248	1.324	1.404	1.505	1.606	1.747	1.987	.4000	.782	.837	.883	.953	.978	.104	.1051	.1081	.1095	.1095	
.4500	1.028	1.086	1.154	1.248	1.324	1.404	1.505	1.606	1.747	1.987	.4500	.782	.837	.883	.953	.978	.104	.1051	.1081	.1095	.1095	
.5000	1.028	1.086	1.154	1.248	1.324	1.404	1.505	1.606	1.747	1.987	.5000	.782	.837	.883	.953	.978	.104	.1051	.1081	.1095	.1095	
.5500	1.028	1.086	1.154	1.248	1.324	1.404	1.505	1.606	1.747	1.987	.5500	.782	.837	.883	.953	.978	.104	.1051	.1081	.1095	.1095	
.6000	1.029	1.080	1.144	1.177	1.205	1.267	1.381	1.466	1.559	1.663	.6000	.782	.837	.883	.953	.978	.104	.1051	.1081	.1095	.1095	
.6500	1.013	1.062	1.139	1.158	1.184	1.226	1.349	1.457	1.546	1.612	.6500	.782	.837	.883	.953	.978	.104	.1051	.1081	.1095	.1095	
.7000	1.006	1.037	1.107	1.133	1.159	1.201	1.393	1.423	1.483	1.542	.7000	.782	.837	.883	.953	.978</						

TABLE IV.- PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(c) Plain wing configuration with spoiler;  $h_S = -0.005\bar{c}$ ;  $\frac{V}{b/2} = 0.30$

x/c	Upper surface										Lower surface										
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$	
<b>Wing</b>																					
.0000	1.855	.895	1.571	2.418	3.125	4.088	4.553	4.944	4.664	3.104	.0125	1.249	.957	.796	.772	.874	1.078	1.259	1.336	1.385	1.285
.0250	.808	1.126	1.770	2.450	3.064	3.740	4.034	4.140	3.975	3.105	.0375	1.224	1.015	.840	.705	.771	.841	.873	.905	.970	.970
.0500	.855	1.111	1.761	2.545	3.174	3.891	4.150	4.134	3.975	3.100	.0750	1.224	1.015	.854	.737	.647	.827	.606	.596	.599	.594
.0750	.909	1.108	1.671	2.282	3.490	3.809	4.006	4.153	3.975	3.100	.1000	1.182	1.049	.862	.756	.850	.831	.814	.814	.815	.815
.1500	.928	1.092	1.301	2.763	4.542	5.126	4.634	4.776	4.076	3.115	.1750	1.177	1.080	.944	.807	.719	.824	.744	.744	.825	.825
.2000	.947	1.089	1.393	1.206	1.748	2.862	3.878	3.904	3.710	3.042	.2000	1.193	1.105	1.035	.908	.773	.674	.594	.593	.517	.594
.2500	.956	1.083	1.398	1.250	1.388	2.047	2.994	3.236	3.297	2.921	.3000	1.224	1.148	1.097	.975	.843	.756	.670	.631	.596	.543
.3000	.981	1.089	1.398	1.250	1.294	1.568	2.184	2.451	2.701	2.679	.3500	1.237	1.170	1.116	1.003	.880	.796	.716	.671	.631	.600
.3500	.994	1.088	1.285	1.223	1.287	1.644	1.934	2.177	2.439	2.515	.4000	1.243	1.173	1.125	1.029	.914	.825	.753	.711	.665	.640
.4000	.998	1.082	1.281	1.223	1.287	1.645	1.933	2.179	2.437	2.490	.4500	1.234	1.182	1.144	1.044	.932	.859	.783	.745	.710	.682
.4500	1.016	1.099	1.393	1.323	1.321	1.739	2.052	2.322	2.482	2.516	.5000	1.224	1.173	1.144	1.051	.931	.850	.809	.777	.735	.706
.5000	1.029	1.092	1.369	1.206	1.239	1.576	1.666	1.811	2.010	2.176	.5500	1.202	1.179	1.154	1.076	.969	.919	.850	.811	.776	.745
.5500	1.003	1.077	1.166	1.190	1.205	1.551	1.600	1.721	1.912	2.069	.6000	1.192	1.179	1.154	1.076	.963	.919	.859	.823	.792	.764
.6000	1.005	1.080	1.154	1.177	1.190	1.529	1.544	1.652	1.830	1.976	.6500	1.182	1.173	1.104	1.076	.975	.924	.857	.817	.791	.759
.6500	1.013	1.074	1.147	1.168	1.178	1.504	1.497	1.590	1.760	1.888	.7500	1.050	1.088	1.051	.972	.946	.903	.859	.839	.835	.835
.7500	.991	1.031	1.091	1.101	1.107	1.209	1.369	1.408	1.609	1.727	.8000	1.019	1.072	1.041	.972	.956	.913	.891	.853	.857	.857
.8000	.947	1.012	1.054	1.076	1.083	1.169	1.328	1.441	1.536	1.639	.8500	1.060	1.080	1.075	1.054	.991	.972	.941	.926	.924	.912
.8500	.937	.984	1.035	1.041	1.043	1.113	1.284	1.376	1.451	1.539	.9000	1.041	1.074	1.075	1.060	1.009	.991	.973	.964	.945	.947
.9000	.849	.901	.956	.956	.988	1.009	1.150	1.239	1.284	1.403	.9500	1.079	1.096	1.100	1.092	1.040	1.035	1.030	1.044	1.054	1.071
.9500	1.068	1.099	1.138	1.196	1.236	1.304	1.363	1.388	1.417	1.430											
<b>Spoiler</b>																					
.2475	.871	1.025	1.024	1.060	1.058	1.106	1.189	1.249	1.292	1.352	.2500	.629	.762	.783	.811	.855	.834	.854	.928	.979	1.038
.4909	.884	1.041	1.037	1.056	1.049	1.091	1.171	1.222	1.259	1.327	.5254	.711	.850	.885	.904	.924	.975	1.021	1.098		
.7394	.896	1.050	1.046	1.097	1.055	1.091	1.156	1.177	1.214	1.263	.7351	1.040	1.220	1.196	1.210	1.240	1.255	1.287	1.325	1.382	
.9794	1.019	1.213	1.202	1.216	1.241	1.262	1.273	1.307	1.332	1.397	.9940	1.079	1.219	1.203	1.226	1.240	1.272	1.274	1.283	1.337	1.388

TABLE IV.- PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(d) Plain wing configuration with spoiler;  $h_S = -0.005\bar{c}$ ;  $\frac{V}{b/2} = 0.43$

x/c	Upper surface										Lower surface										
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$	
<b>Wing</b>																					
.0000	1.508	.907	1.668	1.994	2.497	2.712	2.681	2.468	2.716	2.831	.0125	1.587	1.062	.831	.842	.957	1.141	1.288	1.342	1.422	
.0250	.776	1.099	1.049	1.584	2.423	2.687	2.763	2.780	2.789	2.843	.0375	1.514	1.092	.893	.798	.840	.906	.954	.997	.983	
.0500	.871	1.086	1.039	1.539	2.073	2.490	2.822	2.811	2.836	2.651	.0750	1.404	1.117	.937	.829	.730	.699	.719	.721	.748	.730
.0750	.926	1.092	1.083	1.583	2.130	2.492	2.706	2.803	2.849	2.645	.1000	1.331	1.129	.939	.851	.733	.677	.666	.649	.659	.639
.1000	.915	1.083	1.011	1.511	2.047	2.447	2.768	2.870	2.868	2.657	.1250	1.278	1.133	1.016	.859	.773	.705	.644	.609	.594	
.2000	.947	1.077	1.201	1.785	2.187	2.301	2.903	2.976	2.874	2.645	.2000	1.265	1.148	1.035	.931	.860	.718	.666	.615	.582	
.2500	.984	1.082	1.201	1.789	2.180	2.301	2.909	2.976	2.874	2.645	.3000	1.224	1.173	1.082	.974	.854	.778	.686	.637	.606	
.3500	.984	1.102	1.201	1.882	2.183	2.333	2.954	2.997	2.872	2.645	.4000	1.233	1.176	1.113	1.022	.911	.846	.778	.745	.681	
.4000	.997	1.102	1.163	1.145	1.308	2.147	2.453	2.503	2.540	2.418	.4500	1.230	1.173	1.100	1.038	.917	.853	.800	.752	.735	.688
.4500	1.016	1.108	1.169	1.171	1.303	1.944	2.313	2.373	2.451	2.360	.5000	1.212	1.170	1.113	1.074	.929	.868	.828	.767	.719	
.5000	.965	1.086	1.135	1.152	1.199	1.712	2.116	2.190	2.303	2.260	.5500	1.205	1.160	1.100	1.021	.942	.890	.834	.814	.792	
.6000	1.011	1.086	1.135	1.152	1.199	1.712	2.116	2.190	2.303	2.260	.6000	1.183	1.151	1.094	1.054	.951	.903	.859	.826	.820	
.7000	.984	1.049	1.088	1.114	1.116	1.376	1.728	1.826	1.981	2.009	.7500	1.145	1.123	1.097	1.073	.988	.966	.947	.913	.879	
.7500	.994	1.046	1.075	1.089	1.098	1.320	1.665	1.761	1.902	1.970	.8000	1.107	1.111	1.088	1.063	.985	.972	.966	.938	.959	.918
.8000	.981	1.025	1.031	1.051	1.067	1.254	1.578	1.690	1.824	1.918	.8500	1.101	1.076	1.078	1.060	.997	.994	.975	.997	.982	
.8500	.981	1.025	1.031	1.051	1.067	1.254	1.578	1.691</													

TABLE IV.- PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(e) Plain wing configuration with spoiler;  $h_S = -0.005\bar{c}$ ;  $\frac{V}{b/2} = 0.55$

Upper surface													Lower surface																																																																																																																																																																																																																																					
x/c	C <sub>p</sub> for -												x/c	C <sub>p</sub> for -																																																																																																																																																																																																																																				
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$																																																																																																																																																																																																																				
Wing																																																																																																																																																																																																																																																		
.0000	1.243	.930	1.649	1.795	2.073	2.204	2.180	2.199	2.284	2.239	.0125	1.735	1.136	.862	.915	1.077	1.251	1.394	1.463	1.600	.0250	1.704	1.148	.912	.848	.859	.934	1.003	1.031	1.126	1.130																																																																																																																																																																																																																			
.0125	.836	1.151	1.633	1.839	2.110	2.213	2.231	2.277	2.341	2.269	.0375	1.521	1.139	.969	.864	.748	.721	.706	.699	.748	.718	.0500	.858	1.099	1.542	1.879	2.064	2.210	2.284	2.302	2.354	2.251	.0750	.902	1.096	1.433	1.861	2.070	2.245	2.264	2.314	2.340	2.251	.1000	.915	1.105	1.364	1.848	2.110	2.267	2.276	2.330	2.360	2.239	.1250	.920	1.102	1.314	1.905	2.082	2.169	2.303	2.393	2.354	2.215	.1500	.922	1.105	1.314	1.905	2.082	2.169	2.303	2.393	2.354	2.215	.1750	.922	1.105	1.314	1.905	2.082	2.169	2.303	2.393	2.354	2.215	.2000	.922	1.105	1.314	1.905	2.082	2.169	2.303	2.393	2.354	2.215	.2500	.972	1.145	1.436	1.724	2.073	2.218	2.218	2.342	2.354	2.227	.3000	.997	1.123	1.273	1.285	1.365	2.132	2.155	2.157	2.157	2.159	2.227	.3500	1.006	1.111	1.245	2.098	2.527	2.336	2.356	2.345	2.354	2.320	.4000	1.016	1.117	1.235	1.829	2.454	2.355	2.300	2.308	2.319	2.200	.4500	1.019	1.111	1.213	1.589	2.300	2.333	2.259	2.267	2.291	2.185	.5000	1.032	1.114	1.210	1.380	2.110	2.267	2.218	2.224	2.256	2.154	.6000	1.006	1.077	1.154	1.433	1.721	2.041	2.069	2.087	2.152	2.091	.6500	1.019	1.074	1.141	1.089	1.576	1.950	2.003	2.041	2.104	2.063	.7000	1.006	1.065	1.125	1.057	1.454	1.850	1.991	1.991	2.057	2.030	.7500	1.006	1.049	1.097	1.032	1.362	1.762	1.878	1.926	2.010	1.997	.8500	.915	.941	.949	.893	1.098	1.514	1.884	1.724	1.855	1.924	.9000	1.099	1.114	1.122	1.317	1.445	1.524	1.609	1.681	1.710	1.727	.9500	1.104	1.114	1.141	1.298	1.417	1.489	1.584	1.600	1.682	1.691
Spoiler																																																																																																																																																																																																																																																		
.2743	.837	1.019	1.049	1.031	1.173	1.431	1.610	1.682	1.762	1.889	.2284	.616	.732	.701	.743	.884	1.225	1.424	1.474	1.566	1.700	.3059	.777	.954	.994	.997	1.111	1.335	1.487	1.504	1.541	1.651	.3716	.859	1.221	1.220	1.224	1.270	1.324	1.425	1.494	1.541	1.623	1.743	.4716	.859	1.031	1.058	1.053	1.086	1.325	1.482	1.550	1.660	1.777	.5936	.991	1.204	1.221	1.263	1.281	1.409	1.506	1.571	1.664	1.792	.6940	.991	1.232	1.232	1.254	1.292	1.335	1.450	1.509	1.559	1.651	1.774																																																																																																																																																																				

TABLE IV.- PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Concluded

(f) Plain wing configuration with spoiler;  $h_S = -0.005\bar{c}$ ;  $\frac{y}{b/2} = 0.72$

Upper surface												Lower surface											
x/c	C <sub>p</sub> for -											x/c	C <sub>p</sub> for -										
	a = -4°	a = 0°	a = 4°	a = 8°	a = 12°	a = 16°	a = 20°	a = 22°	a = 24°	a = 26°	Wing	a = -4°	a = 0°	a = 4°	a = 8°	a = 12°	a = 16°	a = 20°	a = 22°	a = 24°	a = 26°		
.0000	1.4585	.897	1.635	1.665	1.734	1.813	1.793	1.793	1.864	1.871	Wing	.0125	1.673	1.201	.813	.828	.892	1.009	1.076	1.129	1.202	1.280	
.0125	.764	1.063	1.610	1.533	1.685	1.772	1.738	1.766	1.846	1.845		.0250	1.676	1.191	.834	.793	.784	.849	.906	.937	.958	1.024	
.0250	.780	1.085	1.610	1.543	1.643	1.697	1.775	1.768	1.857	1.871		.0375	1.652	1.163	.877	.790	.734	.756	.753	.781	.814	1.024	
.0375	.833	1.100	1.644	1.652	1.697	1.784	1.790	1.823	1.883	1.887		.0500	1.585	1.201	.917	.821	.734	.810	.850	.894	.911	.933	
.0750	.865	1.100	1.665	1.674	1.694	1.809	1.811	1.829	1.878	1.887		.1000	1.554	1.191	.943	.828	.735	.725	.699	.751	.771	.793	
.1000	.855	1.082	1.546	1.543	1.644	1.797	1.802	1.820	1.883	1.881		.1500	1.406	1.157	.948	.846	.744	.764	.671	.711	.737	.767	
.1500	.893	1.088	1.423	1.646	1.670	1.806	1.811	1.826	1.901	1.893		.2000	1.362	1.154	.949	.854	.784	.753	.695	.732	.767	.797	
.2000	.940	1.100	1.362	1.664	1.691	1.831	1.832	1.850	1.907	1.881		.2500	1.325	1.154	.950	.854	.784	.753	.695	.732	.767	.797	
.2500	.956	1.110	1.294	1.665	1.691	1.805	1.829	1.844	1.895	1.888		.3000	1.283	1.144	.994	.929	.827	.800	.738	.721	.717	.703	
.3000	.959	1.113	1.251	1.655	1.676	1.797	1.826	1.838	1.886	1.882		.3500	1.258	1.141	.997	.937	.855	.823	.756	.745	.741	.719	
.3500	.994	1.113	1.236	1.648	1.660	1.794	1.826	1.841	1.879	1.859		.4000	1.167	1.129	1.006	.962	.873	.844	.781	.757	.762	.740	
.4000	.994	1.122	1.221	1.653	1.667	1.781	1.817	1.844	1.881	1.841		.4500	1.154	1.129	1.012	.978	.892	.866	.802	.781	.783	.774	
.4500	1.000	1.119	1.217	1.645	1.664	1.779	1.805	1.836	1.885	1.850		.5000	1.142	1.125	1.018	.988	.913	.878	.826	.802	.807	.777	
.5000	1.013	1.125	1.184	1.749	1.648	1.759	1.805	1.796	1.849	1.832		.5500	1.101	1.125	1.024	.994	.923	.900	.845	.820	.810	.793	
.5500	1.017	1.159	1.179	1.765	1.645	1.744	1.784	1.790	1.828	1.813		.6000	1.091	1.122	1.031	.022	.947	.931	.864	.847	.846	.829	
.6000	.993	1.104	1.141	1.752	1.646	1.731	1.765	1.760	1.813	1.807		.6500	1.085	1.116	1.031	.016	.964	.953	.876	.877	.873	.866	
.6500	.994	1.094	1.132	1.737	1.637	1.711	1.750	1.748	1.795	1.804		.7000	1.076	1.091	1.021	.016	.991	.966	.918	.901	.913	.896	
.7000	1.006	1.088	1.126	1.749	1.666	1.722	1.750	1.733	1.786	1.792		.7500	1.069	1.082	1.015	.029	.997	1.006	.980	.943	.952	.942	
.7500	1.019	1.091	1.107	1.715	1.665	1.713	1.729	1.721	1.771	1.777		.8000	1.041	1.069	1.012	.041	1.028	1.031	1.003	.985	1.000	1.001	
.8000	1.013	1.085	1.098	1.665	1.646	1.700	1.701	1.700	1.756	1.777		.8500	1.016	1.069	1.015	.053	1.058	1.091	1.084	1.045	1.061	1.080	
.8500	1.016	1.063	1.080	1.602	1.639	1.670	1.677	1.670	1.726	1.761		.9000	.997	1.054	1.078	1.108	1.147	1.125	1.123	1.154	1.180	1.200	
.9000	.997	1.056	1.061	1.483	1.584	1.644	1.628	1.631	1.693	1.743		.9500	.973	1.028	.994	1.113	1.185	1.250	1.223	1.245	1.262	1.300	

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TABLE V.- PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE

(a) Plain wing configuration with spoiler;  $h_s = -0.010\bar{c}$ ;  $\frac{V}{b/2} = 0$

x/c	Upper surface										Lower surface										
	$\alpha = -10^\circ$	$\alpha = 0^\circ$	$\alpha = 40^\circ$	$\alpha = 80^\circ$	$\alpha = 120^\circ$	$\alpha = 160^\circ$	$\alpha = 200^\circ$	$\alpha = 230^\circ$	$\alpha = 240^\circ$	$\alpha = 260^\circ$	$\alpha = -10^\circ$	$\alpha = 0^\circ$	$\alpha = 40^\circ$	$\alpha = 80^\circ$	$\alpha = 120^\circ$	$\alpha = 160^\circ$	$\alpha = 200^\circ$	$\alpha = 230^\circ$	$\alpha = 240^\circ$	$\alpha = 260^\circ$	
<b>Fuselage</b>																					
.0000	.000	.006	.000	.015	.000	.015	.077	.107	.107	.142	.0500	.793	.850	.802	.716	.626	.575	.500	.482	.430	.390
.0500	.701	.841	.856	.944	.969	1.043	1.073	1.092	1.088	1.091	.0500	.854	.963	.899	.842	.745	.705	.628	.586	.543	.478
.1000	.774	.925	.978	1.006	1.024	1.089	1.122	1.129	1.146	1.148	.0500	.949	1.019	.959	.913	.816	.782	.701	.672	.622	.571
.1500	.901	1.006	1.038	1.059	1.070	1.117	1.143	1.156	1.159	1.154	.0500	.973	1.031	1.003	.941	.865	.837	.771	.739	.686	.634
.2000	.933	1.050	1.079	1.099	1.080	1.126	1.149	1.147	1.143	1.148	.0500	.978	1.073	1.041	.994	.917	.880	.823	.794	.744	.698
.2500	.981	1.087	1.110	1.111	1.101	1.151	1.162	1.165	1.153	1.148	.0500	1.038	1.081	1.040	1.003	.931	.932	.866	.834	.787	.737
.3000	1.035	1.109	1.110	1.105	1.099	1.129	1.134	1.132	1.125	1.142	.0500	1.016	1.058	1.050	1.003	.960	.973	.887	.883	.811	.752
.3500	1.035	1.084	1.103	1.096	1.073	1.102	1.119	1.126	1.119	1.145	.0500	.943	1.006	.994	.957	.911	.889	.835	.804	.759	.710
.4000	.984	1.037	1.060	1.056	1.047	1.074	1.129	1.183	1.204	1.217	.0500	.991	.974	.978	.959	.876	.828	.745	.717	.665	.624
.4500	.974	1.037	1.050	1.056	1.047	1.074	1.129	1.183	1.204	1.217	.0500	.984	.984	.981	.873	.785	.748	.668	.629	.582	.523
.5000	.971	1.034	1.049	1.052	1.048	1.126	1.148	1.159	1.152	1.150	.0500	.970	.955	.887	.824	.739	.696	.622	.577	.540	.498
.5500	.987	1.059	1.113	1.180	1.193	1.225	1.320	1.346	1.349	1.625	.0500	1.008	1.081	1.092	1.071	1.037	1.023	.951	.926	.876	.837
.6000	.994	1.068	1.120	1.152	1.163	1.220	1.262	1.273	1.296	1.477	.0500	1.009	1.081	1.092	1.071	1.037	1.023	.951	.926	.876	.837
.6500	.997	1.045	1.111	1.156	1.164	1.203	1.222	1.221	1.247	1.393	.0500	1.036	1.091	1.096	1.098	1.076	1.061	1.031	1.020	1.014	1.072
.7000	.975	1.050	1.098	1.103	1.129	1.172	1.210	1.236	1.253	1.350	.0500	1.050	1.096	1.098	1.076	1.037	1.023	.951	.926	.876	.837
.7500	1.000	1.050	1.098	1.103	1.129	1.172	1.210	1.236	1.253	1.350	.0500	1.050	1.096	1.098	1.076	1.037	1.023	.951	.926	.876	.837
.8000	1.010	1.056	1.091	1.096	1.099	1.148	1.207	1.242	1.253	1.302	.0500	1.019	1.068	1.076	1.087	1.035	1.021	1.021	1.023	1.026	1.026
.8500	1.019	1.056	1.076	1.087	1.077	1.135	1.201	1.221	1.235	1.260	.0500	1.013	1.056	1.076	1.045	1.092	1.143	1.172	1.168	1.195	1.195
.9000	1.006	1.081	1.076	1.068	1.052	1.089	1.129	1.147	1.143	1.139	.0500	1.077	1.124	1.111	1.080	1.098	1.107	1.138	1.119	1.118	1.118
.9500	1.077	1.124	1.111	1.111	1.111	1.111	1.111	1.111	1.111	1.111	.0500	1.048	1.101	1.096	1.083	1.111	1.098	1.104	1.089	1.072	1.072

TABLE V.- PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(b) Plain wing configuration with spoiler;  $h_s = -0.010\bar{c}$ ;  $\frac{V}{b/2} = 0.21$

x/c	Upper surface										Lower surface										
	$\alpha = -10^\circ$	$\alpha = 0^\circ$	$\alpha = 40^\circ$	$\alpha = 80^\circ$	$\alpha = 120^\circ$	$\alpha = 160^\circ$	$\alpha = 200^\circ$	$\alpha = 220^\circ$	$\alpha = 240^\circ$	$\alpha = 260^\circ$	$\alpha = -10^\circ$	$\alpha = 0^\circ$	$\alpha = 40^\circ$	$\alpha = 80^\circ$	$\alpha = 120^\circ$	$\alpha = 160^\circ$	$\alpha = 200^\circ$	$\alpha = 220^\circ$	$\alpha = 240^\circ$	$\alpha = 260^\circ$	
<b>Wing</b>																					
.0000	1.344	.830	1.275	2.743	4.551	6.963	9.340	10.225	8.031	4.222	.0125	1.232	.991	.768	.627	.586	.622	.680	.727	.686	.509
.0500	.817	1.062	1.410	2.339	6.174	9.906	12.326	11.750	7.662	3.638	.0500	1.173	.798	.840	.822	.822	.837	.844	.817	.751	.624
.1000	.870	1.062	1.313	1.906	3.055	5.234	7.954	9.078	7.061	3.814	.0500	1.142	1.000	.979	.931	.931	.936	.976	.942	.823	.624
.1500	.904	1.077	1.254	1.583	1.788	2.448	3.410	4.610	5.275	5.985	.0500	1.142	1.000	.979	.931	.931	.936	.976	.942	.823	.624
.2000	.920	1.056	1.222	1.470	1.653	2.029	2.578	3.177	4.103	5.296	.0500	1.127	1.019	1.083	1.043	1.064	1.064	1.082	1.082	1.082	1.082
.2500	.975	1.056	1.213	1.453	1.592	1.792	2.472	3.177	4.103	5.296	.0500	1.115	1.019	1.019	1.019	1.019	1.019	1.019	1.019	1.019	1.019
.3000	.950	1.056	1.213	1.453	1.592	1.792	2.472	3.177	4.103	5.296	.0500	1.115	1.019	1.019	1.019	1.019	1.019	1.019	1.019	1.019	1.019
.3500	.950	1.056	1.213	1.453	1.592	1.792	2.472	3.177	4.103	5.296	.0500	1.115	1.019	1.019	1.019	1.019	1.019	1.019	1.019	1.019	1.019
.4000	.975	1.077	1.222	1.453	1.592	1.792	2.472	3.177	4.103	5.296	.0500	1.115	1.019	1.019	1.019	1.019	1.019	1.019	1.019	1.019	1.019
.4500	.981	1.087	1.226	1.453	1.592	1.792	2.472	3.177	4.103	5.296	.0500	1.115	1.019	1.019	1.019	1.019	1.019	1.019	1.019	1.019	1.019
.5000	.994	1.081	1.179	1.285	1.334	1.459	1.699	1.785	1.970	2.293	.0500	1.115	1.019	1.019	1.019	1.019	1.019	1.019	1.019	1.019	1.019
.5500	.991	1.087	1.179	1.285	1.334	1.459	1.699	1.785	1.970	2.293	.0500	1.115	1.019	1.019	1.019	1.019	1.019	1.019	1.019	1.019	1.019
.6000	.991	1.087	1.179	1.285	1.334	1.459	1.699	1.785	1.970	2.293	.0500	1.115	1.019	1.019	1.019	1.019	1.019	1.019	1.019	1.019	1.019
.6500	.991	1.087	1.179	1.285	1.334	1.459	1.699	1.785	1.970	2.293	.0500	1.115	1.019	1.019	1.019	1.019	1.019	1.019	1.019	1.019	1.019
.7000	.991	1.087	1.179	1.285	1.334	1.459	1.699	1.785	1.970	2.293	.0500	1.115	1.019	1.019	1.019	1.019	1.019	1.019	1.019	1.019	1.019
.7500	.991	1.087	1.179	1.285	1.334	1.459	1.699	1.785	1.970	2.293	.0500	1.115	1.019	1.019	1.019	1.019	1.019	1.019	1.019	1.019	1.019
.8000	.991	1.087	1.179	1.285	1.334	1.459	1.699	1.785	1.970	2.293	.0500	1.115	1.019	1.019	1.019	1.019	1.019	1.019	1.019	1.019	1.019
.8500	.991	1.087	1.179	1.285	1.334	1.459	1.699	1.785	1.970	2.293	.0500	1.115	1.019	1.019	1.019	1.019	1.019	1.019	1.019	1.019	1.019
.9000	.991	1.087	1.179	1.285	1.334	1.459	1.699	1.785	1.970	2.293	.0500	1.115	1.019	1.							

**THE INDEPENDENT**

TABLE V.- PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(c) Plain wing configuration with spoiler;  $h_S = -0.010\bar{c}$ ;  $\frac{V}{b/2} = 0.30$

TABLE V.- PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(d) Plain wing configuration with spoiler;  $h_S = -0.010\bar{c}$ ;  $\frac{V}{b/2} = 0.43$

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TABLE V.- PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(e) Plain wing configuration with spoiler;  $h_s = -0.010\bar{c}$ ;  $\frac{V}{b/2} = 0.55$

z/c	Upper surface										Lower surface										
	$C_p$ for -										$C_p$ for -										
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$	
<b>Wing</b>																					
.0000	1.229	.926	1.608	1.798	2.070	2.229	2.152	2.175	2.211	2.234	.0125	1.731	1.133	.858	.922	1.086	1.289	1.382	1.445	1.543	1.599
.0125	.833	1.127	1.602	1.670	2.055	2.245	2.202	2.262	2.251	2.251	.0250	1.703	1.158	.901	.850	.890	.959	.994	1.021	1.088	1.126
.0250	.831	1.127	1.617	1.618	2.057	2.219	2.211	2.248	2.265	2.257	.0375	1.687	1.161	.935	.859	.816	.797	.789	.797	.823	.832
.0500	.845	1.090	1.540	1.625	2.073	2.245	2.236	2.279	2.272	2.231	.0750	1.511	1.146	.954	.843	.758	.737	.714	.702	.701	.695
.1000	.904	1.084	1.333	1.684	2.104	2.296	2.239	2.300	2.287	2.243	.1250	1.443	1.155	.972	.865	.776	.721	.680	.669	.681	.653
.1500	.926	1.084	1.284	1.694	2.104	2.296	2.239	2.300	2.287	2.271	.2000	1.362	1.155	.997	.900	.797	.770	.724	.690	.670	.654
.2500	.926	1.084	1.284	1.694	2.104	2.296	2.239	2.300	2.287	2.271	.3000	1.307	1.146	1.053	.975	.853	.797	.733	.684	.668	.636
.3500	.966	1.096	1.234	2.207	2.057	2.185	2.302	2.322	2.287	2.219	.4000	1.276	1.173	1.074	.991	.877	.816	.758	.724	.701	.662
.4000	.981	1.105	1.241	2.307	2.355	2.267	2.330	2.316	2.287	2.243	.4500	1.260	1.173	1.059	1.022	1.008	.883	.798	.761	.738	.704
.5000	.973	1.090	1.219	2.132	2.500	2.359	2.321	2.300	2.275	2.217	.5500	1.245	1.173	1.092	1.031	1.029	.892	.826	.770	.771	.746
.6000	.991	1.093	1.182	1.652	2.227	2.343	2.318	2.307	2.287	2.211	.6500	1.217	1.164	1.102	1.047	.975	.937	.873	.840	.826	.805
.7000	.965	1.093	1.182	1.652	2.201	2.343	2.318	2.307	2.287	2.211	.7000	1.176	1.159	1.086	1.056	.988	.987	.926	.911	.890	.814
.7500	.963	1.066	1.055	1.009	1.342	1.778	1.848	1.889	1.933	1.988	.7500	1.173	1.156	1.082	1.052	1.029	1.009	.978	.979	.976	
.8500	.914	.861	.895	.818	1.043	1.505	1.612	1.645	1.784	1.878	.9000	1.294	1.350	1.423	1.583	1.632	1.711	1.718	1.805	1.846	1.894
.9500	.950	1.294	1.350	1.315	1.401	1.583	1.632	1.711	1.718	1.805											
<b>Spoiler</b>																					
.2441	.704	.931	1.005	.987	1.037	1.323	1.592	1.695	1.731	1.819	.4581	.959	1.268	1.302	1.322	1.411	1.511	1.595	1.655	1.674	1.782
.4281	.768	1.041	1.497	1.034	1.326	1.595	1.635	1.701	1.701	1.819	.5039	.927	1.252	1.302	1.325	1.402	1.514	1.601	1.675	1.723	1.864
.7416	.787	1.044	1.079	1.050	1.040	1.295	1.518	1.615	1.662	1.801	.9854	.949	1.253	1.293	1.303	1.340	1.483	1.604	1.675	1.747	1.870

TABLE V.- PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Concluded

(f) Plain wing configuration with spoiler;  $h_s = -0.010\bar{c}$ ;  $\frac{V}{b/2} = 0.72$

z/c	Upper surface										Lower surface										
	$C_p$ for -										$C_p$ for -										
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$	
<b>Wing</b>																					
.0000	1.542	.944	1.642	1.628	1.681	1.797	1.809	1.834	1.860	1.834	.0125	1.970	1.258	.874	.817	.842	.985	1.082	1.147	1.186	1.245
.0125	.733	1.037	1.404	1.601	1.650	1.751	1.753	1.800	1.835	1.834	.0250	1.586	1.230	.896	.789	.779	.855	.915	.949	.979	1.033
.0250	.752	1.062	1.569	1.604	1.653	1.760	1.771	1.825	1.860	1.837	.0500	1.542	1.206	.921	.783	.718	.735	.781	.791	.805	.834
.0500	.796	1.078	1.566	1.610	1.650	1.760	1.800	1.849	1.866	1.846	.0750	1.459	1.234	.966	.820	.736	.729	.759	.745	.758	
.0750	.825	1.087	1.585	1.616	1.650	1.791	1.823	1.868	1.881	1.849	.1000	1.459	1.215	.978	.830	.736	.729	.713	.705	.704	.710
.1000	.806	1.088	1.494	1.598	1.616	1.769	1.808	1.852	1.872	1.849	.1250	1.411	1.190	.997	.834	.758	.729	.704	.690	.688	.671
.1500	.841	1.078	1.418	1.594	1.626	1.785	1.826	1.865	1.881	1.851	.2000	1.389	1.171	1.003	.882	.782	.748	.717	.705	.677	.671
.2000	.905	1.103	1.371	1.616	1.641	1.800	1.834	1.886	1.890	1.828	.2500	1.328	1.149	1.016	.878	.797	.778	.721	.689	.692	
.2500	.917	1.100	1.311	1.628	1.650	1.779	1.845	1.877	1.887	1.822	.3000	1.280	1.153	1.028	.923	.828	.800	.742	.717	.710	
.3000	.914	1.098	1.258	1.607	1.628	1.763	1.842	1.888	1.893	1.813	.3500	1.233	1.140	1.035	.944	.846	.822	.787	.738	.728	
.4000	.963	1.120	1.217	1.619	1.601	1.820	1.849	1.884	1.894	1.827	.4500	1.131	1.187	1.047	.969	.892	.877	.832	.804	.777	.747
.4500	.971	1.125	1.208	1.632	1.592	1.735	1.820	1.843	1.859	1.831	.5000	1.108	1.140	1.057	.981	.908	.892	.848	.828	.796	.779
.5000	.985	1.112	1.198	1.687	1.410	1.739	1.820	1.834	1.823	1.779	.5500	1.057	1.131	1.063	.988	.926	.908	.872	.853	.820	.798
.5500	.959	1.100	1.170	1.709	1.607	1.714	1.799	1.813	1.811	1.779	.6000	1.038	1.140	1.076	1.009	.951	.942	.903	.880	.851	.837
.6000	.930	1.093	1.154	1.690	1.604	1.702	1.775	1.797	1.796	1.773	.6500	1.047	1.131	1.072	1.012	.969	.960	.927	.905	.872	.864
.6500	.943	1.084	1.132	1.687	1.604	1.689	1.759	1.770	1.778	1.755	.7000	1.051	1.109	1.057	1.012	.981	.975	.945	.935	.912	.897
.7000	.965	1.075	1.126	1.694	1.613	1.692	1.756	1.767	1.771	1.767	.7500	1.037	1.090	1.060	1.025	.994	1.006	.979	.975	.951	.943
.7500	.991	1.081	1.116	1.663	1.632	1.695	1.744	1.760	1.753	1.755	.8000	1.019	1.084	1.044	1.031	1.043	1.019				

TABLE VI.- PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE

(a) Plain wing configuration with spoiler;  $h_s = -0.035\bar{c}$ ;  $\frac{V}{b/2} = 0$

z/c	Upper surface										Lower surface											
	$C_p$ for -											$C_p$ for -										
	$\alpha = -1^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 28^\circ$	$\alpha = -1^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 28^\circ$		
Fuselage																						
.0000	.012	.000	.000	.000	.021	.050	.085	.098	.149		.0500	.853	.842	.783	.715	.669	.570	.509	.453	.412	.367	
.0500	.758	.853	.910	.941	.1003	.1028	.1048	.1079	.1082	.1084	.1000	.929	.950	.882	.820	.768	.684	.624	.562	.527	.479	
.1000	.856	.926	.978	.1.012	.1.043	.1.054	.1.067	.1.079	.1.082	.1.084	.1.000	.1.052	.1.012	.994	.954	.890	.842	.747	.693	.641	.592	
.1500	.932	.988	1.034	1.042	1.064	1.104	1.137	1.151	1.159	1.161	1.000	.1.037	1.037	.997	.947	.894	.820	.764	.708	.677	.615	
.2000	.976	1.037	1.077	1.084	1.127	1.140	1.157	1.171	1.178	1.180	1.000	1.080	1.080	1.059	1.025	.985	.947	.872	.811	.772	.735	.681
.2500	1.024	1.080	1.105	1.105	1.133	1.134	1.146	1.146	1.134	1.140	1.000	1.109	1.110	1.113	1.116	1.131	1.131	1.131	1.131	1.131	1.131	1.131
.3000	1.049	1.092	1.099	1.102	1.133	1.116	1.131	1.128	1.119	1.122	1.000	1.108	1.110	1.109	1.113	1.116	1.116	1.116	1.116	1.116	1.116	1.116
.3500	1.052	1.080	1.087	1.087	1.108	1.110	1.110	1.109	1.113	1.116	1.000	1.105	1.110	1.116	1.116	1.116	1.116	1.116	1.116	1.116	1.116	1.116
.4000	1.018	1.037	1.040	1.050	1.068	1.070	1.100	1.116	1.131	1.143	1.000	1.053	1.053	1.059	1.125	1.219	1.229	1.257	1.257	1.257	1.257	1.257
.4500	1.003	1.031	1.050	1.059	1.059	1.125	1.171	1.152	1.198	1.267	1.322	1.348	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478
.5000	.981	1.028	1.065	1.102	1.152	1.152	1.198	1.267	1.322	1.348	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478
.5500	.981	1.043	1.081	1.137	1.211	1.250	1.336	1.362	1.412	1.719	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478
.6000	.988	1.040	1.093	1.149	1.214	1.241	1.292	1.335	1.478	1.791	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478
.6500	.978	1.040	1.094	1.142	1.207	1.249	1.252	1.274	1.561	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478
.7000	.951	1.052	1.087	1.115	1.164	1.208	1.220	1.226	1.226	1.242	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478
.7500	.953	1.056	1.095	1.136	1.195	1.236	1.256	1.274	1.274	1.274	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478
.8000	.981	1.037	1.059	1.074	1.105	1.119	1.129	1.134	1.134	1.134	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478
.8500	1.003	1.043	1.059	1.062	1.099	1.104	1.174	1.204	1.228	1.250	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478
.9000	1.015	1.049	1.053	1.056	1.074	1.074	1.067	1.134	1.152	1.165	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478
.9500	1.034	1.065	1.065	1.062	1.071	1.076	1.109	1.128	1.128	1.128	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478
.9940	1.132	1.157	1.142	1.127	1.133	1.104	1.118	1.119	1.122	1.122	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478	1.478

TABLE VI.- PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(b) Plain wing configuration with spoiler;  $h_s = -0.035\bar{c}$ ;  $\frac{V}{b/2} = 0.21$

z/c	Upper surface										Lower surface											
	$C_p$ for -											$C_p$ for -										
	$\alpha = -1^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 28^\circ$	$\alpha = -1^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 28^\circ$		
Wing																						
.0125	1.235	1.977	.736	.622	.574	.612	.687	.684	.621	.489	.0125	1.235	1.977	.736	.622	.574	.612	.687	.684	.621	.489	
.0250	1.185	.988	.779	.625	.500	.428	.362	.334	.292	.243	.0250	1.185	.988	.779	.625	.500	.428	.362	.334	.292	.243	
.0500	1.135	.997	.816	.682	.524	.394	.282	.243	.224	.218	.0500	1.135	.997	.816	.682	.524	.394	.282	.243	.224	.218	
.0750	1.122	1.006	.846	.722	.561	.431	.331	.304	.283	.299	.0750	1.122	1.006	.846	.722	.561	.431	.331	.304	.283	.299	
.1000	1.141	1.012	.871	.738	.598	.468	.399	.380	.363	.361	.1000	1.141	1.012	.871	.738	.598	.468	.399	.380	.363	.361	
.1500	1.129	1.015	.896	.781	.655	.557	.488	.471	.441	.439	.1500	1.129	1.015	.896	.781	.655	.557	.488	.471	.441	.439	
.2000	1.132	1.046	.932	.838	.708	.618	.549	.529	.506	.489	.2000	1.132	1.046	.932	.838	.708	.618	.549	.529	.506	.489	
.2500	1.164	1.081	1.344	1.463	2.021	2.274	2.844	2.772	2.652	2.592	.2500	1.164	1.081	1.344	1.463	2.021	2.274	2.844	2.772	2.652	2.592	
.3000	1.153	1.058	1.319	1.408	1.546	1.884	2.070	2.469	2.592	2.652	.3000	1.153	1.058	1.319	1.408	1.546	1.884	2.070	2.469	2.592	2.652	
.3500	1.153	1.058	1.291	1.346	1.502	1.767	2.149	2.387	2.469	2.592	.3500	1.153	1.058	1.291	1.346	1.502	1.767	2.149	2.387	2.469	2.592	
.4000	1.153	1.064	1.256	1.316	1.438	1.699	2.082	2.326	2.469	2.592	.4000	1.153	1.064	1.256	1.316	1.438	1.699	2.082	2.326	2.469	2.592	
.4500	1.153	1.064	1.219	1.271	1.406	1.670	2.050	2.322	2.469	2.592	.4500	1.153	1.064	1.219	1.271	1.406	1.670	2.050	2.322	2.469	2.592	
.5000	1.153	1.064	1.194	1.236	1.348	1.441	1.529	1.665	1.922	2.050	.5000	1.153	1.064	1.194	1.236	1.348	1.441	1.529	1.665	1.922	2.050	
.5500	1.153	1.034	1.081	1.147	1.165	1.244	1.365	1.417	1.556	1.751	.5500	1.153	1.034	1.081	1.147	1.165	1.244	1.365	1.417	1.556	1.751	
.6000	1.169	1.010	1.073	1.125	1.153	1.																

TABLE VI - PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(c) Plain wing configuration with spoiler;  $h_S = -0.035\bar{c}$ ;  $\frac{V}{b/2} = 0.30$

Upper surface												Lower surface																														
x/c	C <sub>p</sub> for -											x/c	C <sub>p</sub> for -																													
	a = -40°	a = 0°	a = 40°	a = 80°	a = 120°	a = 160°	a = 200°	a = 220°	a = 240°	a = 260°	a = -40°	a = 0°	a = 40°	a = 80°	a = 120°	a = 160°	a = 200°	a = 220°	a = 240°	a = 260°	a = -40°	a = 0°	a = 40°	a = 80°	a = 120°	a = 160°	a = 200°	a = 220°	a = 240°	a = 260°												
Wing																																										
.0000	2.006	.918	1.593	2.441	3.057	3.935	4.367	4.256	3.648	3.025	.0125	1.793	1.764	2.450	3.006	3.594	3.883	3.906	3.625	3.028	.0250	.850	1.079	1.803	2.547	3.095	3.714	3.969	3.916	3.637	3.028											
.0500	.900	1.070	1.656	3.263	3.417	3.619	3.852	3.955	3.650	3.028	.0780	.909	1.073	1.267	2.875	4.423	4.840	4.392	4.147	3.668	3.053	.1000	.915	1.052	1.172	1.750	3.266	4.382	4.331	4.034	3.578	3.034										
.1500	.937	1.049	1.150	1.181	1.702	2.854	3.711	3.627	3.392	2.950	.2000	.975	1.061	1.169	1.225	1.359	2.024	2.917	3.095	3.100	2.625	.2500	.978	1.043	1.144	1.225	1.265	1.605	2.447	2.660	2.645	2.041										
.3000	.984	1.049	1.129	1.200	1.273	1.440	1.902	2.131	2.401	2.436	.3500	.981	1.049	1.132	1.191	1.250	1.397	1.767	1.982	2.243	2.336	.4000	.978	1.049	1.132	1.191	1.250	1.397	1.767	1.982	2.243	2.336										
.4500	.988	1.061	1.128	1.181	1.221	1.378	1.730	1.921	2.171	2.274	.5000	.988	1.043	1.104	1.159	1.190	1.323	1.613	1.754	1.985	2.118	.5500	.988	1.012	1.070	1.134	1.159	1.293	1.558	1.687	1.901	2.037										
.6000	.956	1.003	1.055	1.113	1.126	1.268	1.484	1.620	1.817	1.956	.6500	.944	1.037	1.071	1.113	1.231	1.452	1.529	1.736	1.863	.7000	.930	1.026	1.068	1.113	1.126	1.268	1.484	1.620	1.817	1.956											
.7500	.872	.908	.942	.981	.999	1.091	1.270	1.398	1.565	1.688	.8000	.784	.833	.874	.913	1.025	1.184	1.319	1.429	1.547	.8500	.850	.894	.934	1.037	1.071	1.113	1.231	1.359	1.478												
.8500	.693	.736	.740	.813	.843	.849	.967	1.186	1.318	1.323	.9000	.534	.556	.620	.647	.770	.840	.848	.945	1.084	.921	.736	.794	.874	.934	1.026	1.068	1.113	1.126	1.268												
.9500	1.903	1.926	1.943	1.644	1.656	1.729	1.791	1.803	1.845	1.844																																
Spoiler																																										
.2475	.598	.679	.721	.740	.783	.765	.876	.988	1.055	1.227	.4909	.733	.809	.861	.876	.904	.887	1.025	1.094	1.125	1.266	.7396	.889	.994	1.022	1.037	1.071	1.061	1.177	1.204	1.229	1.337	.9796	1.365	1.354	1.585	1.616	1.618	1.677	1.783	1.787	1.788
.5267	1.365	1.334	1.337	1.334	1.337	1.337	1.337	1.337	1.337	1.337	.5024	1.399	1.337	1.337	1.337	1.337	1.337	1.337	1.337	1.337	1.337	.7381	1.420	1.549	1.576	1.632	1.697	1.717	1.792	1.829	1.862	1.861										
.5940	1.433	1.568	1.598	1.653	1.775	1.778	1.873	1.879	1.877	1.878	.9500	1.301	1.265	1.279	1.284	1.284	1.289	1.243	1.215	1.210	1.211	1.215																				
Spoiler																																										
.5267	1.365	1.334	1.337	1.334	1.337	1.337	1.337	1.337	1.337	1.337	.5024	1.399	1.337	1.337	1.337	1.337	1.337	1.337	1.337	1.337	1.337	.7381	1.420	1.549	1.576	1.632	1.697	1.717	1.792	1.829	1.862	1.861										

TABLE VI - PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(d) Plain wing configuration with spoiler;  $h_s = -0.035\bar{c}$ ;  $\frac{V}{b/2} = 0.43$

TABLE VI - PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(e) Plain wing configuration with spoiler;  $h_s = -0.035\bar{c}$ ;  $\frac{V}{b/2} = 0.55$

x/c	Upper surface										Lower surface										
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 23^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 23^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$	
<b>Wing</b>																					
.0000	1.914	.891	1.849	1.744	1.763	2.065	2.002	2.101	2.160	2.221	.0128	1.793	1.176	.859	.894	1.012	1.212	1.340	1.395	1.528	1.610
.0125	.691	1.035	1.805	1.763	1.951	2.074	2.126	2.158	2.227	2.249	.0120	1.821	1.189	.905	.938	.925	1.026	.969	.974	1.076	1.128
.0250	.615	1.031	1.840	1.763	1.941	2.062	2.135	2.161	2.233	2.249	.0125	1.825	1.189	.946	.863	.751	.779	.708	.787	.845	.841
.0375	.845	1.031	1.830	1.775	1.941	2.089	2.150	2.186	2.239	2.233	.0130	1.630	1.149	.943	.844	.745	.711	.699	.693	.650	.650
.0500	.868	1.043	1.825	1.794	1.944	2.123	2.156	2.198	2.230	2.240	.0135	1.552	1.158	.988	.872	.742	.711	.684	.680	.650	.652
.1000	.893	1.052	1.825	1.822	1.973	2.117	2.168	2.210	2.227	2.218	.0140	1.458	1.164	1.012	.906	.773	.723	.678	.641	.671	.626
.1500	.919	1.049	1.827	1.825	1.944	2.055	2.193	2.210	2.218	2.212	.0145	1.433	1.173	1.043	.941	.800	.745	.705	.660	.674	.635
.2000	.937	1.046	1.819	1.913	1.902	2.034	2.205	2.204	2.224	2.209	.0150	1.398	1.176	1.067	.969	.828	.763	.721	.672	.696	.642
.2500	.931	1.043	1.818	1.813	1.917	2.025	2.217	2.219	2.233	2.212	.0155	1.379	1.183	1.083	.975	.856	.797	.745	.696	.714	.654
.3000	.962	1.061	1.815	2.197	2.125	2.083	2.233	2.198	2.224	2.221	.0160	1.347	1.189	1.107	1.016	.889	.822	.776	.733	.745	.685
.3500	.962	1.049	1.815	2.054	2.245	2.135	2.211	2.174	2.211	2.209	.0165	1.354	1.195	1.119	1.031	.908	.849	.800	.760	.767	.723
.4000	.975	1.037	1.815	1.808	2.026	2.142	2.168	2.137	2.193	2.181	.0170	1.326	1.195	1.123	1.050	.929	.862	.816	.775	.795	.741
.4500	.975	1.037	1.814	1.804	2.011	2.138	2.169	2.140	2.185	2.159	.0175	1.323	1.197	1.135	1.059	.991	.892	.850	.803	.823	.776
.5000	.966	1.018	1.811	1.804	1.941	2.059	2.072	2.149	2.153	2.141	.0180	1.301	1.195	1.150	1.084	.991	.892	.851	.888	.832	
.5500	.925	.970	1.814	1.807	1.919	1.948	1.953	2.050	2.059	2.055	.0185	1.281	1.195	1.165	1.101	1.012	.978	.945	.877	.919	.882
.6000	.853	.949	1.809	1.809	1.919	1.949	1.975	1.985	2.054	2.015	.0190	1.261	1.195	1.165	1.113	1.034	1.000	.957	.921	.950	.906
.6500	.855	.949	1.809	1.809	1.919	1.949	1.975	1.985	2.054	2.015	.0195	1.279	1.198	1.165	1.138	1.065	1.015	.967	.937	.966	.915
.7000	.865	.900	1.843	1.808	1.959	1.980	1.981	1.983	2.056	2.015	.0200	1.273	1.201	1.184	1.166	1.066	1.077	1.061	1.024	1.044	.985
.7500	.866	.824	.908	.866	1.233	1.569	1.727	1.769	1.892	1.956	.0205	1.282	1.231	1.199	1.200	1.129	1.132	1.138	1.079	1.109	.919
.8000	.858	.590	1.850	1.850	1.819	1.805	1.820	1.820	1.850	1.832	.0210	1.326	1.265	1.242	1.250	1.199	1.194	1.215	1.183	1.215	.911
.8500	1.455	1.555	1.552	1.697	1.714	1.791	1.834	1.809	1.910	1.931	.0215	1.339	1.547	1.546	1.659	1.681	1.739	1.775	1.864	1.860	
.9000	1.539	1.547	1.546	1.659	1.681	1.739	1.785	1.775	1.864	1.860	.0220	1.338	1.534	1.534	1.634	1.620	1.734	1.747	1.843	1.841	
<b>Spoiler</b>																					
.2443	.561	.685	.743	.893	.763	1.107	1.405	1.405	1.643	1.779	.0225	1.509	1.522	1.560	1.585	1.712	1.720	1.798	1.789	1.796	1.806
.5716	.560	1.004	1.043	.966	1.006	1.232	1.910	1.408	1.699	1.788	.0230	1.503	1.531	1.582	1.591	1.706	1.723	1.798	1.789	1.815	
.8786	1.121	1.149	1.495	1.498	1.498	1.640	1.774	1.774	1.820	1.863	.0235	1.505	1.531	1.594	1.616	1.728	1.736	1.766	1.802	1.830	

TABLE VI - PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Concluded

(f) Plain wing configuration with spoiler;  $h_s = -0.035\bar{c}$ ;  $\frac{V}{b/2} = 0.72$

x/c	Upper surface										Lower surface										
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 23^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 23^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$	
<b>Wing</b>																					
.0000	1.653	.926	1.529	1.520	1.619	1.659	1.718	1.760	1.767	1.785	.0125	1.714	1.321	.848	.802	.858	.930	1.041	1.119	1.183	1.227
.0125	.770	.954	1.495	1.492	1.585	1.616	1.683	1.733	1.784	1.788	.0130	1.711	1.281	.879	.786	.805	.879	.930	.985	1.009	
.0250	.773	.994	1.483	1.492	1.591	1.625	1.702	1.751	1.790	1.791	.0135	1.699	1.234	.904	.784	.746	.723	.755	.778	.805	.824
.0375	.800	1.028	1.477	1.492	1.585	1.640	1.730	1.769	1.805	1.794	.0140	1.672	1.236	.950	.817	.771	.724	.730	.741	.752	
.0500	.831	1.031	1.502	1.511	1.601	1.659	1.742	1.781	1.805	1.785	.0145	1.690	1.237	.957	.842	.777	.710	.705	.699	.710	.707
.1000	.834	1.012	1.480	1.480	1.560	1.637	1.724	1.772	1.799	1.782	.0150	1.690	1.222	.991	.857	.799	.713	.696	.687	.687	.665
.1500	.871	1.025	1.427	1.483	1.563	1.646	1.746	1.788	1.802	1.781	.0155	1.688	1.206	1.024	.912	.844	.764	.704	.683	.683	.658
.2000	.889	1.040	1.337	1.480	1.573	1.644	1.746	1.788	1.802	1.784	.0160	1.686	1.207	.930	.848	.765	.723	.708	.701	.687	
.2500	.911	1.037	1.260	1.404	1.573	1.631	1.742	1.778	1.788	1.775	.0165	1.686	1.206	.935	.882	.790	.767	.736	.726	.707	
.3000	.902	1.028	1.242	1.435	1.573	1.631	1.742	1.778	1.788	1.775	.0170	1.686	1.208	.938	.882	.790	.767	.736	.726	.707	
.3500	.935	1.032	1.270	1.444	1.542	1.628	1.733	1.772	1.786	1.776	.0175	1.686	1.210	.939	.884	.791	.768	.737	.728	.708	
.4000	.935	1.049	1.146	1.449	1.536	1.601	1.721	1.751	1.750	1.737	.0180	1.686	1.215	.936	.884	.792	.764	.738	.729	.709	
.4500	.942	1.037	1.121	1.452	1.517	1.585	1.705	1.736	1.744	1.731	.0185	1.686	1.217	.937	.885	.793	.765	.739	.729	.709	
.5000	.951	1.037	1.118	1.493	1.523	1.579	1.699	1.730	1.735	1.731	.0190	1.686	1.218	.938	.885	.794	.766	.739	.729	.709	
.5500	.948	1.006	1.071	1.511	1.505	1.525	1.646	1.678	1.701	1.707	.0195	1.686	1.221	.939	.886	.795	.767</				

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TABLE VII - PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE

(a) Plain wing configuration with spoiler;  $h_S = -0.050\bar{c}$ ;  $\frac{V}{b/2} = 0$

x/c	Upper surface										Lower surface										
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$	
<i>Fuselage</i>																					
.0000	.003	.006	.000	.000	.006	.015	.041	.101	.097	.139	.0500	.837	.859	.784	.713	.632	.584	.503	.457	.432	.375
.0300	.740	.880	.722	.994	1.022	1.037	1.071	1.036	1.079	1.085	.0800	.919	.791	.881	.851	.745	.703	.616	.570	.526	.480
.1000	.919	1.006	1.041	1.081	1.100	1.147	1.129	1.109	1.149	1.151	.1000	1.004	1.047	1.077	1.050	.972	.764	.705	.675	.631	.574
.1500	.989	1.044	1.072	1.100	1.098	1.122	1.144	1.119	1.131	1.130	.1500	1.041	1.072	1.034	1.000	.929	.872	.800	.760	.733	.687
.2000	.984	1.044	1.072	1.100	1.098	1.122	1.144	1.119	1.131	1.130	.2000	1.043	1.082	1.053	1.031	.943	.842	.744	.692	.651	.591
.2500	.919	1.091	1.106	1.131	1.107	1.128	1.144	1.116	1.137	1.148	.2500	1.047	1.075	1.041	1.025	.966	.939	.886	.834	.806	.755
.3000	.953	1.094	1.109	1.119	1.101	1.131	1.126	1.101	1.128	1.130	.3000	1.053	1.084	1.080	1.092	1.116	1.089	1.113	1.130	1.127	1.127
.3500	.956	1.094	1.109	1.119	1.101	1.131	1.126	1.101	1.128	1.130	.3500	1.016	1.044	1.056	1.069	1.079	1.104	1.098	1.122	1.157	1.157
.4000	1.006	1.047	1.047	1.084	1.086	1.110	1.181	1.175	1.219	1.266	.4000	1.086	1.104	1.116	1.121	1.131	1.141	1.151	1.161	1.171	1.180
.4500	.984	1.041	1.063	1.119	1.152	1.190	1.279	1.338	1.462	1.500	.4500	.937	.978	.934	.900	.882	.774	.715	.647	.620	.583
.5000	.984	1.044	1.064	1.108	1.130	1.178	1.245	1.325	1.392	1.695	.5000	.912	.969	.928	.872	.785	.743	.666	.599	.572	.532
.5500	.984	1.044	1.064	1.108	1.130	1.162	1.245	1.300	1.291	1.758	.5500	.934	.928	.884	.823	.739	.688	.626	.564	.529	.489
.6000	.993	1.054	1.064	1.108	1.130	1.162	1.205	1.251	1.208	1.544	.6000	1.210	1.216	1.159	1.144	1.064	1.021	.960	.893	.891	.858
.6500	.993	1.054	1.064	1.108	1.130	1.162	1.205	1.251	1.208	1.544	.6500	1.000	1.038	1.025	1.025	1.000	.979	.963	.923	.927	.906
.7000	.994	1.044	1.059	1.109	1.110	1.138	1.175	1.159	1.169	1.235	.7000	.997	1.100	1.116	1.131	1.104	1.113	1.126	1.101	1.110	1.089
.7500	.991	1.031	1.050	1.088	1.082	1.119	1.149	1.172	1.221	1.287	.7500	.991	1.031	1.050	1.088	1.082	1.119	1.149	1.172	1.221	1.287
.8000	.991	1.031	1.050	1.088	1.082	1.119	1.149	1.172	1.221	1.287	.8000	.991	1.031	1.050	1.088	1.082	1.119	1.149	1.172	1.221	1.287
.8500	.993	1.053	1.050	1.084	1.080	1.116	1.159	1.178	1.237	1.289	.8500	.993	1.053	1.050	1.084	1.080	1.116	1.159	1.178	1.237	1.289
.9000	.996	1.046	1.059	1.078	1.055	1.076	1.132	1.124	1.175	1.181	.9000	.996	1.046	1.059	1.078	1.055	1.076	1.132	1.124	1.175	1.181
.9500	.994	1.075	1.072	1.078	1.055	1.070	1.113	1.101	1.131	1.142	.9500	.994	1.075	1.072	1.078	1.055	1.070	1.113	1.101	1.131	1.142
.9940	1.154	1.166	1.147	1.113	1.128	1.123	1.101	1.131	1.127	1.127	.9940	1.154	1.166	1.147	1.113	1.128	1.123	1.101	1.131	1.127	1.127

TABLE VII - PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(b) Plain wing configuration with spoiler;  $h_S = -0.050\bar{c}$ ;  $\frac{V}{b/2} = 0.21$

x/c	Upper surface										Lower surface										
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$	
<i>Wing</i>																					
.0000	1.328	.841	1.217	2.435	4.337	6.566	8.944	9.352	6.640	3.603	.0125	1.187	1.019	.748	.608	.561	.604	.677	.684	.620	.448
.0125	.745	1.063	1.404	2.234	5.677	9.314	11.640	10.464	6.297	3.301	.0250	1.141	1.019	.793	.626	.518	.442	.561	.534	.597	.245
.0250	.785	1.060	1.291	1.808	2.957	4.834	7.574	8.315	9.916	3.322	.0375	1.093	1.038	.842	.657	.534	.414	.523	.233	.227	
.0500	.868	1.098	1.248	1.524	1.739	2.327	3.521	4.413	5.640	2.904	.0750	1.093	1.073	.873	.697	.570	.452	.526	.322	.301	.207
.0750	.871	1.076	1.214	1.432	1.622	1.950	2.561	3.189	3.846	2.791	.1000	1.119	1.051	.898	.741	.607	.505	.589	.371	.360	.267
.1000	.876	1.095	1.195	1.389	1.584	1.800	2.019	2.289	2.616	2.689	.1250	1.104	1.046	.901	.784	.642	.579	.683	.445	.448	
.1250	.926	1.084	1.186	1.315	1.454	1.651	1.819	2.019	2.289	2.493	.1500	1.084	1.084	.947	.784	.642	.579	.683	.445	.448	
.1500	.948	1.089	1.173	1.399	1.564	1.875	2.079	2.447	2.611	2.361	.1750	1.107	1.071	.949	.712	.659	.549	.626	.415	.416	
.2000	.954	1.089	1.173	1.281	1.399	1.564	1.875	2.079	2.447	2.361	.2000	1.107	1.071	.949	.712	.659	.549	.626	.415	.416	
.2500	.959	1.098	1.184	1.281	1.402	1.651	1.875	2.079	2.447	2.361	.2500	1.107	1.071	.949	.712	.659	.549	.626	.415	.416	
.3000	.963	1.092	1.173	1.276	1.402	1.654	1.874	2.079	2.447	2.361	.3000	1.107	1.071	.949	.712	.659	.549	.626	.415	.416	
.3500	.963	1.092	1.173	1.276	1.402	1.654	1.874	2.079	2.447	2.361	.3500	1.107	1.071	.949	.712	.659	.549	.626	.415	.416	
.4000	.960	1.095	1.178	1.273	1.408	1.658	1.876	2.079	2.447	2.361	.4000	1.107	1.071	.949	.712	.659	.549	.626	.415	.416	
.4500	.966	1.079	1.118	1.154	1.230	1.344	1.458	1.514	1.659	1.925	.4500	1.107	1.071	.949	.712	.659	.549	.626	.415	.416	
.5000	.975	1.076	1.109	1.145	1.292	1.311	1.426	1.477	1.601	1.875	.5000	1.107	1.071	.949	.712	.659	.549	.626	.415	.416	
.5500	.951	1.054	1.059	1.111	1.159	1.292	1.331	1.414	1.456	1.776	.5500	1.107	1.071	.949	.712	.659	.549	.626	.415	.416	
.6000	.957	1.051	1.050	1.080	1.132	1.212	1.337	1.389	1.530	1.725	.6000	1.107	1.071	.949	.712	.659	.549	.626	.415	.416	
.6500	.908	1.000	1.034	1.037	1.092	1.165	1.320	1.371	1.503	1.660	.6500	1.107	1.071	.949	.712	.659	.549	.626	.415	.416	
.7000	.877	.973	.997	1.035	1.115	1.263	1.333	1.463	1.543	1.660	.7000	1.107	1.071	.949	.712	.659	.				

TABLE VII.- PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(c) Plain wing configuration with spoiler;  $h_S = -0.050\bar{c}$ ;  $\frac{V}{b/2} = 0.30$

x/c	Upper surface										Lower surface											
	$C_p$ for -										$C_p$ for -											
x/c	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 28^\circ$	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 28^\circ$		
<b>Wing</b>																						
.0000	1.938	.924	1.548	2.364	3.033	3.906	4.276	4.168	3.637	2.657	.0125	1.296	1.013	.783	.756	.856	1.053	1.194	1.252	1.294	1.134	
.0125	.770	1.127	1.731	2.370	2.972	3.582	3.822	3.849	3.585	2.648	.0250	1.270	1.048	.824	.716	.715	.760	.806	.836	.862	.788	
.0500	.811	1.130	1.757	2.463	3.079	3.710	3.900	3.864	3.594	2.654	.0750	1.162	1.082	.904	.775	.669	.617	.533	.533	.528	.490	
.0750	.892	1.108	1.536	3.169	3.414	3.579	3.819	3.894	3.610	2.636	.1000	1.165	1.092	.938	.796	.696	.620	.536	.517	.494	.469	
.1000	.806	1.098	1.180	1.623	3.190	3.452	4.207	3.946	3.552	2.666	.1250	1.178	1.143	.988	.839	.733	.670	.571	.541	.524	.475	
.1500	.920	1.095	1.149	1.145	1.675	2.847	3.618	3.557	3.337	2.591	.2000	1.162	1.158	1.022	.895	.782	.704	.589	.565	.534	.487	
.2000	.935	1.101	1.167	1.197	1.356	2.053	2.900	3.046	3.049	2.522	.2500	1.175	1.198	1.059	.944	.828	.744	.649	.605	.547	.513	
.3000	.899	1.092	1.152	1.208	1.276	1.695	2.386	2.633	2.794	2.466	.3500	1.230	1.234	1.078	1.026	.979	.860	.884	.833	.800		
.3500	.932	1.095	1.146	1.194	1.257	1.564	2.147	2.383	2.589	2.573	.4000	1.248	1.248	1.124	1.025	.939	.860	.874	.863	.833		
.4000	.935	1.089	1.130	1.173	1.224	1.458	1.906	2.128	2.359	2.268	.5000	1.227	1.263	1.146	1.065	.981	.822	.834	.803	.767	.698	
.5500	.940	1.082	1.129	1.170	1.236	1.484	1.936	2.121	2.353	2.285	.6000	1.193	1.260	1.167	1.086	1.009	.959	.872	.845	.804	.737	
.6000	.984	1.063	1.090	1.117	1.184	1.336	1.615	1.763	1.951	2.033	.6500	1.191	1.225	1.146	1.066	1.044	.956	.878	.851	.810	.764	
.6500	.911	1.038	1.087	1.096	1.144	1.305	1.533	1.678	1.874	1.940	.6500	1.159	1.228	1.139	1.083	1.024	.987	.909	.885	.856	.794	
.7000	.902	1.025	1.037	1.065	1.113	1.274	1.477	1.602	1.800	1.863	.7500	1.070	1.209	1.142	1.099	1.043	1.025	.950	.933	.917	.854	
.7500	.889	1.003	1.012	1.037	1.083	1.230	1.408	1.520	1.703	1.791	.8000	1.031	1.209	1.142	1.117	1.064	1.050	.984	.961	.948	.890	
.7794	.794	.893	.889	.917	.951	.972	.972	.975	.975	.983	.8500	1.129	1.247	1.176	1.151	1.098	1.100	1.035	1.024	1.012	.955	
.8000	.684	.804	.805	.830	.874	.975	1.135	1.244	1.417	1.522	.8500	1.156	1.291	1.223	1.191	1.165	1.159	1.110	1.100	1.089	1.033	
.9000	.648	.535	.576	.586	.712	.579	.705	.851	1.034	1.230	.9500	1.549	1.703	1.644	1.703	1.844	1.856	1.891	1.806	1.809		
<b>Spoiler</b>																						
.2475	.464	.533	.584	.591	.662	.648	.754	.825	.921	1.118	.5000	1.287	1.383	1.583	1.616	1.703	1.718	1.784	1.849	1.792	1.845	
.4500	.506	.680	.719	.731	.761	.777	.823	.923	.912	1.181	.5500	1.411	1.599	1.626	1.725	1.742	1.753	1.804	1.807	1.825		
.7394	.784	.897	.931	.944	.948	.966	1.119	1.095	1.131	1.263	.7794	1.448	1.599	1.622	1.719	1.727	1.792	1.852	1.857	1.852		
.9795	1.326	1.527	1.563	1.641	1.647	1.716	1.803	1.727	1.769	1.767		.7500	1.477	1.621	1.653	1.731	1.739	1.835	1.905	1.863	1.949	

TABLE VII.- PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(d) Plain wing configuration with spoiler;  $h_S = -0.050\bar{c}$ ;  $\frac{V}{b/2} = 0.43$

x/c	Upper surface										Lower surface										
	$C_p$ for -										$C_p$ for -										
x/c	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 28^\circ$	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 28^\circ$	
<b>Wing</b>																					
.0000	1.550	.996	1.619	1.910	2.362	2.579	2.511	2.532	2.555	2.328	.0125	1.647	1.117	.824	.809	.942	1.103	1.185	1.274	1.356	1.295
.0125	.758	1.105	1.598	1.899	2.319	2.551	2.584	2.627	2.610	2.349	.0250	1.561	1.168	.882	.781	.785	.832	.850	.854	.942	.904
.0500	.788	1.114	1.542	1.929	2.319	2.567	2.684	2.625	2.398		.0750	1.588	1.171	.938	.812	.753	.723	.690	.693	.721	.693
.0750	.825	1.101	1.511	1.984	2.386	2.660	2.646	2.675	2.638	2.367	.1000	1.414	1.174	.957	.821	.739	.692	.640	.641	.659	.618
.0750	.856	1.108	1.559	2.031	2.368	2.557	2.645	2.700	2.635	2.349	.1250	1.274	1.177	.985	.855	.758	.692	.630	.626	.582	
.1000	.856	1.095	1.474	2.210	2.316	2.492	2.637	2.715	2.656	2.364	.1500	1.242	1.190	1.015	.895	.788	.716	.627	.607	.561	
.1500	.847	1.088	1.424	2.444	2.680	2.818	2.765	2.735	2.631	2.398	.2000	1.225	1.200	1.082	.914	.822	.745	.645	.635	.587	
.2000	.807	1.072	1.318	1.426	2.414	2.610	2.665	2.599	2.555	2.329	.2500	1.239	1.222	1.094	.963	.883	.804	.709	.672	.659	.588
.3000	.920	1.086	1.316	1.497	2.003	2.629	2.583	2.563	2.546	2.281	.3500	1.233	1.238	1.105	1.009	.920	.841	.746	.724	.702	.645
.3500	.923	1.076	1.099	1.089	1.395	2.296	2.379	2.389	2.426	2.233	.4000	1.267	1.241	1.120	1.028	.942	.866	.784	.748	.730	.663
.4000	.902	1.070	1.096	1.074	1.387	2.078	2.260	2.298	2.365	2.188	.5000	1.224	1.241	1.127	1.040	.957	.894	.803	.769	.761	.696
.5000	.888	1.063	1.087	1.071	1.294	1.909	2.154	2.219	2.285	2.146	.5500	1.221	1.231	1.153	1.057	.966	.910	.818	.780	.716	
.6000	.851	1.011	1.054	1.052	1.172	1.670	1.963	2.057	2.171	2.056	.6000	1.205	1.228	1.156	1.065	1.009	.933	.878	.863	.843	.782
.6500	.865	.994	1.006	1.015	1.095	1.480	1.787	1.891	2.021	1.985	.7000	1.224	1.250	1.152	1.012	1.051	1.022	.962	.924	.923	.848
.7000	.776	.889	.904	.910	.994	1.283	1.589	1.708	1.862	1.863	.7500	1.221	1.244	1.183	1.083	1.047	.984	.973	.963	.896	</td

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TABLE VII.- PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(e) Plain wing configuration with spoiler;  $h_s = -0.050\bar{c}$ ;  $\frac{V}{b/2} = 0.55$

x/c	Upper surface										Lower surface										
	$C_p$ for -										$C_p$ for -										
x/c	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$	x/c	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
<b>Wing</b>																					
.0000	1.267	1.889	1.619	1.670	1.932	2.056	2.038	2.067	2.147	2.081	.0125	1.751	1.234	.882	.864	1.012	1.199	1.317	1.374	1.491	1.496
.0125	.880	1.082	1.594	1.700	1.966	2.075	2.085	2.137	2.184	2.104	.0250	1.776	1.238	.929	.810	.831	.910	.956	.991	1.030	1.075
.0250	.782	1.063	1.604	1.659	1.938	2.053	2.088	2.143	2.187	2.101	.0375	1.586	1.206	.978	.839	.788	.763	.765	.773	.804	.812
.0500	.791	1.067	1.497	1.694	1.926	2.103	2.110	2.171	2.193	2.090	.0750	.834	1.070	1.279	1.377	2.180	2.193	2.084			
.1000	.853	1.076	1.238	1.751	1.954	2.112	2.132	2.189	2.190	2.073	.1250	.868	1.080	1.244	2.177	2.184	2.054				
.2000	.880	1.073	1.193	1.830	1.986	2.040	2.144	2.171	2.178	2.060	.3000	.858	1.067	1.219	2.120	2.137	2.104	2.066			
.4000	.880	1.073	1.180	2.022	1.993	2.029	2.137	2.171	2.178	2.066	.5000	.858	1.067	1.219	2.120	2.137	2.104	2.066			
.6000	.880	1.073	1.180	2.022	1.993	2.029	2.137	2.171	2.178	2.066	.7000	.858	1.067	1.219	2.120	2.137	2.104	2.066			
.8000	.880	1.073	1.180	2.022	1.993	2.029	2.137	2.171	2.178	2.066	.9000	.880	1.073	1.180	2.022	1.993	2.029	2.137	2.171	2.066	
.9300	1.091	1.091	1.190	1.694	2.153	2.100	2.100	2.101	2.144	2.030											
.4500	.917	1.041	1.115	1.459	2.044	2.075	2.057	2.055	2.113	2.006	.5000	.924	1.041	1.115	1.459	2.044	2.075	2.057	2.113	2.006	
.5000	.905	1.022	1.087	1.273	1.923	2.022	2.022	2.034	2.082	1.994	.6000	1.288	1.260	1.161	1.074	1.022	.963	.922	.869	.877	.821
.6000	.902	.953	1.009	1.018	1.598	1.838	1.890	1.912	1.979	1.934	.6500	1.267	1.263	1.170	1.096	1.043	1.003	.956	.904	.898	.848
.6500	.919	.903	.963	.938	1.469	1.744	1.828	1.864	1.938	1.898	.7000	1.254	1.253	1.164	1.099	1.053	1.003	.975	.939	.929	.893
.7000	.770	.848	.898	.864	1.343	1.648	1.749	1.781	1.865	1.812	.7500	1.261	1.269	1.189	1.126	1.093	1.059	1.016	.982	.901	.928
.7500	.684	.756	.827	.762	1.193	1.533	1.655	1.696	1.822	1.845	.8000	.849	.841	.863	1.010	1.141	1.176	1.201	1.231	1.251	1.266
.8000	.649	.841	.863	.820	1.199	1.530	1.652	1.691	1.820	1.845	.8500	1.261	1.269	1.189	1.126	1.093	1.059	1.016	1.043	.997	
.9000	1.016	1.050	1.089	1.079	1.797	1.832	1.843	1.884	1.920	1.857	.9300	1.092	1.090	1.094	1.092	1.092	1.092	1.092	1.092	1.092	
.9300	1.090	1.090	1.093	1.094	1.721	1.804	1.808	1.842	1.889	1.877											
<b>Spoiler</b>																					
.2443	.423	.493	.613	.474	.533	.978	1.721	1.353	1.311	1.677	.4881	.583	.656	.765	1.031	1.431	1.593	1.638	1.673	1.657	1.616
.57416	.759	.919	.983	.878	.871	1.156	1.435	1.501	1.632	1.731	.9354	1.013	1.242	1.441	1.503	1.618	1.700	1.750	1.784	1.812	1.822
.9354	1.267	1.499	1.538	1.534	1.481	1.627	1.767	1.785	1.820	1.854											

TABLE VII.- PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Concluded

(f) Plain wing configuration with spoiler;  $h_s = -0.050\bar{c}$ ;  $\frac{V}{b/2} = 0.72$

x/c	Upper surface										Lower surface										
	$C_p$ for -										$C_p$ for -										
x/c	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$	x/c	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
<b>Wing</b>																					
.0000	1.731	1.031	1.431	1.481	1.546	1.633	1.699	1.682	1.751	1.770	.0125	1.743	1.411	.866	.840	.920	1.034	1.056	1.155	1.202	
.0125	.771	.900	1.082	1.594	1.700	1.966	2.075	2.085	2.137	2.184	.0250	1.743	1.361	.897	.788	.767	.804	.871	.884	.955	1.000
.0250	.759	.929	1.444	1.459	1.506	1.596	1.684	1.667	1.751	1.776	.0375	1.737	1.288	.931	.791	.724	.758	.784	.810		
.0500	.784	.978	1.425	1.456	1.509	1.605	1.696	1.691	1.760	1.777	.0750	1.707	1.323	.975	.839	.748	.722	.727	.703	.724	.740
.0750	.809	.994	1.441	1.463	1.509	1.621	1.711	1.701	1.765	1.770	.1000	1.710	1.295	.991	.865	.798	.722	.712	.685	.699	.692
.1000	.787	.978	1.403	1.491	1.599	1.697	1.891	1.757	1.757	1.764	.1250	1.729	1.323	.907	.878	.875	.875	.875	.875	.875	.875
.1250	.805	.994	1.422	1.448	1.504	1.622	1.718	1.703	1.767	1.775	.1500	1.748	1.348	.914	.816	.758	.725	.735	.745	.755	.765
.2000	.865	1.013	1.242	1.441	1.503	1.618	1.700	1.703	1.774	1.784	.2500	1.734	1.210	.934	.828	.828	.786	.751	.703	.696	.689
.2500	.878	1.019	1.189	1.428	1.497	1.503	1.727	1.700	1.754	1.784	.3000	1.574	1.194	1.050	.959	.862	.826	.770	.724	.720	.710
.3000	.884	1.006	1.116	1.416	1.466	1.593	1.711	1.688	1.733	1.726	.3500	1.417	1.223	.979	.894	.876	.850	.797	.749	.745	.737
.3500	.903	1.016	1.116	1.419	1.466	1.590	1.696	1.682	1.730	1.731	.4000	1.248	1.198	1.078	1.000	.908	.862	.825	.774	.775	.753
.4000	.906	1.006	1.091	1.400	1.454	1.563	1.675	1.659	1.715	1.716	.5000	1.141	1.191	1.106	1.047	.954	.917	.868	.819	.816	.792
.4500	.900	1.003	1.078	1.404	1.453	1.538	1.662	1.641	1.699	1.713	.6000	1.129	1.191	1.119	1.081	1.003	.972	.924	.872	.856	.855
.5000	.893	.981	1.041	1.434	1.599	1.692	1.599	1.649	1.698		.6500	1.151	1.242	1.138	1.103	1.021	.997	.974	.899	.872	.862
.6000	.884	.966	1.022	1.447	1.580	1.604	1.584	1.648	1.677		.7000	.972	1.172	1.128	1.119	1.070	1.034	.973	.979	.910	
.7000	.97																				

TABLE VIII.- PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE

(a) Plain wing configuration with spoiler;  $h_s = -0.100\bar{c}$ ;  $\frac{V}{b/2} = 0$

$x/c$	Upper surface										Lower surface												
	$C_p$ for -											$C_p$ for -											
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 24^\circ$	$\alpha = 28^\circ$	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 24^\circ$	$\alpha = 28^\circ$	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$		
Fuselage																							
.0000	.009	.000	.009	.000	.000	.027	.046	.082	.111	.134	.009	.000	.000	.000	.000	.000	.000	.000	.000	.019	.002		
.0500	.799	.831	.808	.946	1.003	.994	1.043	1.073	1.066	1.095	.799	.831	.808	.946	1.003	.994	1.043	1.073	1.066	.588	.521	.500	
.1000	.873	.920	.925	1.049	1.059	1.048	1.086	1.125	1.120	1.149	.873	.920	.925	1.049	1.059	1.048	1.086	1.125	1.120	.684	.605	.592	
.1500	.957	.988	1.045	1.073	1.082	1.082	1.119	1.140	1.158	1.168	.957	.988	1.045	1.073	1.082	1.082	1.119	1.140	1.158	.684	.605	.592	
.2000	.991	1.028	1.028	1.105	1.109	1.105	1.144	1.128	1.123	1.143	.991	1.028	1.028	1.105	1.109	1.105	1.144	1.128	1.123	.684	.605	.592	
.2500	1.040	1.058	1.098	1.118	1.127	1.100	1.123	1.100	1.100	1.118	1.040	1.058	1.098	1.118	1.127	1.100	1.123	1.100	1.100	.684	.605	.592	
.3000	1.058	1.074	1.098	1.108	1.112	1.091	1.095	1.116	1.111	1.127	1.058	1.074	1.098	1.108	1.112	1.091	1.095	1.116	1.111	1.127	.684	.605	.592
.3500	1.068	1.052	1.093	1.102	1.103	1.066	1.095	1.107	1.090	1.146	1.068	1.052	1.093	1.102	1.103	1.066	1.095	1.107	1.090	1.146	.684	.605	.592
.4000	1.021	1.018	1.052	1.048	1.062	1.054	1.083	1.097	1.105	1.149	1.021	1.018	1.052	1.048	1.062	1.054	1.083	1.097	1.105	1.149	.684	.605	.592
.4500	.997	1.022	1.043	1.073	1.100	1.094	1.153	1.170	1.193	1.256	.997	1.022	1.043	1.073	1.100	1.094	1.153	1.170	1.193	1.256	.684	.605	.592
.5000	.984	1.012	1.061	1.105	1.143	1.251	1.280	1.313	1.485	1.485	.984	1.012	1.061	1.105	1.143	1.251	1.280	1.313	1.485	.684	.605	.592	
.5500	.975	1.018	1.073	1.133	1.183	1.208	1.291	1.319	1.410	1.741	.975	1.018	1.073	1.133	1.183	1.208	1.291	1.319	1.410	1.741	.684	.605	.592
.6000	.978	1.022	1.059	1.130	1.180	1.208	1.270	1.298	1.386	1.835	.978	1.022	1.059	1.130	1.180	1.208	1.270	1.298	1.386	1.835	.684	.605	.592
.6500	.954	.991	1.046	1.092	1.137	1.136	1.173	1.171	1.171	1.479	.954	.991	1.046	1.092	1.137	1.136	1.173	1.171	1.171	1.479	.684	.605	.592
.7000	.957	.987	1.042	1.091	1.136	1.141	1.167	1.167	1.167	1.390	.957	.987	1.042	1.091	1.136	1.141	1.167	1.167	1.167	1.390	.684	.605	.592
.7500	.957	.987	1.043	1.092	1.136	1.146	1.167	1.167	1.167	1.395	.957	.987	1.043	1.092	1.136	1.146	1.167	1.167	1.167	1.395	.684	.605	.592
.8000	.957	.987	1.043	1.092	1.136	1.146	1.167	1.167	1.167	1.395	.957	.987	1.043	1.092	1.136	1.146	1.167	1.167	1.167	1.395	.684	.605	.592
.8500	1.037	1.086	1.083	1.069	1.028	1.121	1.172	1.210	1.241	1.262	1.037	1.086	1.083	1.069	1.028	1.121	1.172	1.210	1.241	1.262	.684	.605	.592
.9000	1.154	1.178	1.208	1.210	1.205	1.159	1.196	1.210	1.217	1.223	1.154	1.178	1.208	1.210	1.205	1.159	1.196	1.210	1.217	1.223	.684	.605	.592
.9500	1.126	1.169	1.193	1.181	1.168	1.097	1.147	1.155	1.169	1.177	1.126	1.169	1.193	1.181	1.168	1.097	1.147	1.155	1.169	1.177	.684	.605	.592
.9940	1.194	1.203	1.202	1.207	1.183	1.142	1.192	1.158	1.163	1.163	1.194	1.203	1.202	1.207	1.183	1.142	1.192	1.158	1.163	.684	.605	.592	

TABLE VIII.- PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(b) Plain wing configuration with spoiler;  $h_s = -0.100\bar{c}$ ;  $\frac{V}{b/2} = 0.21$

$x/c$	Upper surface										Lower surface												
	$C_p$ for -											$C_p$ for -											
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 24^\circ$	$\alpha = 28^\circ$	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 24^\circ$	$\alpha = 28^\circ$	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$		
Wing																							
.0000	1.386	.830	1.236	2.469	4.163	6.359	8.193	8.675	4.766	2.938	.0000	1.190	.994	.778	.627	.565	.584	.635	.658	.521	.405		
.0500	1.028	1.134	2.219	2.644	5.991	10.354	14.124	14.773	2.853	1.781	1.028	1.142	.997	.816	.651	.509	.419	.339	.326	.277	.230		
.1000	.704	.743	1.043	1.213	1.749	2.735	4.716	7.237	7.312	7.347	.704	.743	1.043	1.213	1.749	2.735	4.716	7.237	7.312	7.347	.326	.277	.230
.1500	.841	.905	1.055	1.289	1.505	1.685	2.244	3.857	4.712	3.650	.841	.905	1.055	1.289	1.505	1.685	2.244	3.857	4.712	3.650	.326	.277	.230
.2000	.757	.787	1.055	1.289	1.505	1.685	2.244	3.857	4.712	3.650	.757	.787	1.055	1.289	1.505	1.685	2.244	3.857	4.712	3.650	.326	.277	.230
.2500	.757	.787	1.055	1.289	1.505	1.685	2.244	3.857	4.712	3.650	.757	.787	1.055	1.289	1.505	1.685	2.244	3.857	4.712	3.650	.326	.277	.230
.3000	.907	.981	1.046	1.071	1.159	1.254	1.415	1.815	2.103	2.476	.907	.981	1.046	1.071	1.159	1.254	1.415	1.815	2.103	2.476	.326	.277	.230
.3500	.926	1.043	1.144	1.223	1.302	1.459	1.721	1.932	2.165	2.405	.926	1.043	1.144	1.223	1.302	1.459	1.721	1.932	2.165	2.405	.326	.277	.230
.4000	.925	1.031	1.128	1.193	1.265	1.416	1.653	1.774	1.967	2.139	.925	1.031	1.128	1.193	1.265	1.416	1.653	1.774	1.967	2.139	.326	.277	.230
.4500	.913	1.018	1.088	1.156	1.231	1.369	1.515	1.662	1.823	2.084	.913	1.018	1.088	1.156	1.231	1.369	1.515	1.662	1.823	2.084	.326	.277	.230
.5000	.900	.991	1.053	1.098	1.154	1.310	1.385	1.516	1.640	1.982	.900	.991	1.053	1.098	1.154	1.310	1.385	1.516	1.640	1.982	.326	.277	.230
.5500	.891	1.028	1.055	1.114	1.259	1.357	1.454	1.585	1.640	1.927	.891	1.028	1.055	1.114	1.259	1.357	1.454	1.585	1.640	1.927	.326	.277	.230
.6000	.813	.864	.916	.957	1.021	1.131	1.242	1.356	1.457	1.587	.813	.864	.916	.957	1.021	1.131	1.242	1.356	1.457	1.587	.326	.277	.230
.6500	.744	.787	.828	.842	1.026	1.053	1.076	1.107	1.135	1.160	.744	.787	.828	.842	1.026	1.053	1.076	1.107	1.135	1.160	.326	.277	.230
.7000	.657	.682	.731	.756	.889	.956	1.																

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TABLE VIII.- PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(c) Plain wing configuration with spoiler;  $h_S = -0.100\bar{c}$ ;  $\frac{V}{b/2} = 0.30$

x/c	Upper surface										Lower surface										
	$C_p$ for -										$C_p$ for -										
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$		$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
<b>Wing</b>																					
.0000	2.078	.883	1.938	2.266	2.882	3.566	3.875	3.814	3.110	2.444	.0125	1.293	.991	.791	.746	.827	.991	1.109	1.199	1.189	1.094
.0125	.741	1.083	1.722	2.278	2.845	3.306	3.527	3.609	3.104	2.423	.0250	1.293	1.025	.838	.722	.685	.725	.731	.804	.811	.773
.0370	.79	1.074	1.705	2.367	2.944	3.406	3.544	3.640	3.150	2.423	.0500	1.293	1.025	.838	.742	.685	.725	.731	.804	.811	.773
.0500	.847	1.083	1.256	3.027	3.247	3.416	3.540	3.650	3.125	2.423	.0750	1.277	1.058	.913	.713	.657	.730	.758	.806	.849	.755
.0750	.850	1.092	1.256	3.027	3.247	3.416	3.540	3.650	3.125	2.423	.1000	1.177	1.071	.944	.801	.713	.713	.724	.742	.761	.766
.1000	.850	1.092	1.272	3.023	3.015	3.963	3.648	3.546	3.073	2.447	.1250	1.190	1.126	.997	.847	.728	.647	.641	.653	.612	.501
.1250	.850	1.092	1.272	3.023	3.015	3.963	3.648	3.546	3.073	2.447	.1500	1.190	1.170	.978	.948	.824	.738	.639	.609	.561	.344
.1500	.872	1.040	1.241	1.125	1.602	2.866	3.224	3.255	2.951	2.393	.2000	1.168	1.126	1.034	.902	.775	.688	.588	.586	.527	.508
.2000	.891	1.037	1.241	1.165	1.318	2.103	2.682	2.867	2.772	2.362	.2500	1.190	1.170	1.078	.948	.824	.738	.639	.609	.561	.344
.2500	.841	1.021	1.111	1.159	1.241	1.719	2.300	2.553	2.589	2.365	.3000	1.227	1.188	1.100	.976	.861	.775	.673	.658	.601	.577
.3000	.869	1.009	1.103	1.138	1.216	1.569	2.094	2.345	2.444	2.284	.3500	1.249	1.207	1.144	.912	.819	.718	.699	.643	.616	.516
.3500	.869	.991	1.069	1.107	1.173	1.450	1.863	2.093	2.282	2.220	.4000	1.262	1.222	1.159	1.034	.938	.863	.754	.745	.683	.693
.4000	.880	.966	1.047	1.089	1.142	1.388	1.727	1.975	2.140	2.154	.4500	1.277	1.241	1.178	1.064	.972	.894	.806	.780	.717	.695
.4500	.869	.950	1.025	1.058	1.117	1.344	1.667	1.895	2.067	2.103	.5000	1.257	1.237	1.181	1.083	.991	.922	.836	.814	.756	.731
.5000	.844	.913	.975	1.009	1.077	1.278	1.548	1.742	1.888	2.003	.5500	1.282	1.242	1.192	1.093	.991	.922	.836	.814	.756	.731
.5500	.773	.861	.913	.948	1.024	1.247	1.437	1.648	1.815	1.948	.6000	1.314	1.262	1.191	1.093	.991	.922	.836	.814	.756	.731
.6000	.794	.894	.944	.974	1.024	1.266	1.463	1.663	1.833	1.963	.6500	1.314	1.222	1.197	1.119	.934	.938	.909	.904	.881	.793
.6500	.645	.707	.741	.774	.800	1.059	1.279	1.432	1.604	1.822	.7500	1.094	1.216	1.200	1.131	1.053	1.023	.981	.966	.906	.882
.7500	.427	.485	.575	.572	.738	.750	.967	1.146	1.345	1.656	.8000	1.050	1.216	1.200	1.141	1.045	1.036	.991	.939	.912	.877
.8000	.402	.497	.553	.603	.753	.751	.981	1.186	1.386	1.652	.9000	1.193	1.305	1.284	1.179	1.116	1.048	1.050	.991	.939	.912
.8500	.592	.707	.750	.875	1.102	.769	.954	1.236	1.421	1.680	.9500	1.361	1.404	1.369	1.327	1.312	1.325	1.282	1.301	1.244	1.170
.9000	.190	.225	.334	.260	.133	.100	.339	.553	.774	1.103	.9500	1.611	1.572	1.547	1.602	1.684	1.651	1.614	1.619	1.572	1.479
.9500	1.329	1.611	1.572	1.547	1.602	1.684	1.766	1.851	1.814	1.819											
<b>Spoiler</b>																					
.2475	.420	.446	.518	.518	.298	.290	.371	.590	.625	1.034	.5000	1.257	1.293	1.188	1.083	.991	.922	.836	.814	.756	.731
.4900	.290	.327	.423	.410	.205	.211	.357	.547	.598	1.055	.7300	1.257	1.293	1.188	1.083	.991	.922	.836	.814	.756	.731
.7300	.105	.114	.218	.171	.052	.100	.440	.608	.774	1.043	.9500	1.257	1.293	1.188	1.083	.991	.922	.836	.814	.756	.731
.9750	.867	.840	.822	.806	.798	.776	.1119	1.164	1.208	1.378											

TABLE VIII.- PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(d) Plain wing configuration with spoiler;  $h_S = -0.100\bar{c}$ ;  $\frac{V}{b/2} = 0.43$

x/c	Upper surface										Lower surface										
	$C_p$ for -										$C_p$ for -										
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$		$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
<b>Wing</b>																					
.0000	1.607	.842	1.346	1.801	2.139	2.313	2.273	2.104	2.415	2.448	.0125	1.760	1.124	.828	.795	.870	1.022	1.127	1.218	1.287	1.293
.0125	.716	1.006	1.541	1.783	2.102	2.309	2.369	2.485	2.436	2.260	.0250	1.679	1.157	.897	.781	.818	.879	.906	.924	.944	.954
.0370	.741	1.040	1.503	1.810	2.105	2.356	2.419	2.510	2.422	2.279	.0500	1.776	1.160	.947	.813	.707	.694	.661	.693	.707	.713
.0500	.776	1.040	1.469	1.856	2.142	2.394	2.427	2.531	2.421	2.272	.0750	1.545	1.142	.981	.838	.725	.678	.621	.634	.646	.646
.0750	.819	1.037	1.475	1.887	2.123	2.338	2.424	2.541	2.424	2.260	.1000	1.333	1.151	1.003	.862	.730	.668	.624	.618	.613	.607
.1000	.801	1.028	1.409	2.037	2.065	2.300	2.421	2.541	2.424	2.272	.1250	1.262	1.179	1.065	.920	.821	.758	.661	.634	.619	.598
.1250	.801	1.012	1.447	1.804	2.438	2.500	2.421	2.494	2.409	2.272	.1500	1.221	1.194	1.060	.966	.846	.784	.671	.640	.640	.640
.1500	.810	1.012	1.447	1.804	2.438	2.500	2.421	2.494	2.409	2.272	.1750	1.237	1.193	1.063	.967	.846	.784	.672	.645	.645	.645
.1750	.838	.978	1.078	1.122	.907	1.281	1.254	1.293	1.322	1.229	.2000	1.282	1.235	1.150	1.027	.917	.844	.756	.701	.677	.677
.2000	.825	.947	1.025	1.009	1.405	2.053	2.115	2.205	2.238	2.181	.2250	1.286	1.225	1.189	1.049	.947	.889	.794	.767	.730	.693
.2500	.791	.917	1.091	1.977	1.438	1.913	2.030	2.149	2.183	2.166	.3000	1.257</									

TABLE VIII.- PRESSURE COEFFICIENT  $C_p$ . AT THE SIX SPANWISE STATIONS THROUGH THE  
ANGLE-OF-ATTACK RANGE - Continued

(e) Plain wing configuration with spoiler;  $h_S = -0.100\bar{c}$ ;  $\frac{y}{b/2} = 0.55$

TABLE VIII.- PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Concluded

(f) Plain wing configuration with spoiler;  $h_S = -0.100\bar{c}$ ;  $\frac{V}{b/\bar{c}} = 0.72$

Upper surface												Lower surface											
z/c	C <sub>p</sub> for -											C <sub>p</sub> for -											
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$			
Wing												Wing											
.0000	1.929	1.579	1.340	1.172	1.277	1.308	1.463	1.544	1.614	1.726	.0125	1.944	1.628	.954	.813	.792	.795	.923	1.006	1.069	1.135		
.0125	.833	.757	1.264	1.153	1.242	1.272	1.441	1.535	1.605	1.723	.0250	1.944	1.732	.978	.858	.761	.722	.804	.857	.895	.967		
.0250	.756	.705	1.172	1.133	1.242	1.272	1.449	1.535	1.599	1.723	.0375	1.973	1.605	1.003	.707	.745	.674	.712	.745	.750	.793		
.0500	.728	.809	1.104	1.118	1.236	1.269	1.448	1.544	1.608	1.723	.0750	1.973	1.605	1.002	.905	.783	.683	.693	.705	.711	.732		
.0750	.744	.828	1.067	1.108	1.236	1.272	1.463	1.544	1.599	1.710	.1000	1.973	1.605	1.002	.907	.787	.692	.693	.704	.711	.731		
.1000	.741	.815	1.015	1.089	1.214	1.245	1.438	1.535	1.593	1.695	.1500	.753	.828	.981	1.054	1.205	1.445	1.538	1.587	1.689	1.767		
.2000	.759	.828	.963	1.064	1.199	1.230	1.454	1.532	1.578	1.671	.2500	.762	.842	.917	1.034	1.174	1.429	1.577	1.652	1.689	1.769		
.3000	.753	.791	.860	1.015	1.143	1.157	1.374	1.474	1.524	1.650	.3500	.753	.772	.860	1.015	1.153	1.399	1.487	1.536	1.640	1.733		
.4000	.731	.717	.819	1.015	1.085	1.100	1.346	1.435	1.515	1.616	.4500	.711	.780	.943	1.080	1.159	1.300	1.435	1.497	1.613	1.700		
.5000	.682	.686	.800	.997	1.105	1.121	1.321	1.414	1.482	1.604	.6000	.682	.686	.800	.997	1.105	1.121	1.325	1.414	1.482	1.604	1.694	
.6000	.713	.711	.840	1.080	1.159	1.169	1.303	1.377	1.457	1.598	.6500	.930	.951	.978	1.454	1.484	1.592	1.629	1.686	1.722	1.760		
.7000	1.197	1.206	1.199	1.403	1.454	1.387	1.451	1.511	1.527	1.628	.7500	1.252	1.218	1.218	1.218	1.218	1.218	1.218	1.218	1.218	1.218	1.218	
.7500	1.657	1.646	1.558	1.330	1.370	1.372	1.420	1.465	1.521	1.634	.8000	1.222	1.268	1.239	1.249	1.186	1.048	1.056	1.070	1.027	1.061		
.8000	1.342	1.326	1.395	1.556	1.596	1.526	1.564	1.581	1.584	1.674	.8500	1.256	1.302	1.279	1.289	1.258	1.232	1.219	1.193	1.163	1.220		
.8500	1.429	1.349	1.457	1.689	1.587	1.516	1.549	1.566	1.581	1.674	.9000	1.324	1.358	1.322	1.346	1.308	1.248	1.208	1.204	1.181	1.220		
.9000	1.626	1.569	1.607	1.721	1.615	1.526	1.558	1.572	1.578	1.662	.9300	1.389	1.403	1.345	1.376	1.321	1.288	1.293	1.274	1.274	1.274		

TABLE IX.- PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE

(a) Double slotted flap configuration with spoiler;  $h_s = -0.005\bar{c}$ ;  $\frac{y}{b/2} = 0$

z/c	Upper surface										Lower surface										
	$C_p$ for -										$C_p$ for -										
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 28^\circ$	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 28^\circ$	
Fuselage											Fuselage										
.0200	.919	.902	.897	.899	.899	.899	.900	.909	.905	.903	.0500	.875	.826	.755	.696	.637	.540	.481	.445	.431	.379
.0500	.792	.823	.873	.917	.979	.915	.935	.945	.945	.945	.075	.906	.904	.864	.826	.741	.647	.570	.502	.512	.491
.1000	.884	.911	.945	.971	1.037	1.066	1.077	1.103	1.127	1.145	.103	.924	.966	.993	.993	.966	.914	.727	.655	.630	.596
.1500	.981	.972	1.009	1.042	1.095	1.075	1.106	1.125	1.127	1.145	.127	.942	.975	.975	.963	.977	.726	.676	.652	.624	
.2000	1.050	1.021	1.037	1.068	1.124	1.143	1.153	1.172	1.181	1.196	.142	1.072	1.021	1.021	1.021	1.021	.740	.705	.691	.670	
.2500	1.043	1.024	1.042	1.075	1.105	1.125	1.132	1.152	1.167	1.182	.142	1.072	1.024	1.024	1.024	1.024	.740	.705	.691	.670	
.3000	1.073	1.076	1.082	1.095	1.111	1.125	1.133	1.150	1.165	1.181	.142	1.072	1.024	1.024	1.024	1.024	.740	.705	.691	.670	
.3500	1.057	1.064	1.072	1.074	1.127	1.125	1.125	1.142	1.150	1.167	.142	1.072	1.024	1.024	1.024	1.024	.740	.705	.691	.670	
.4000	1.042	1.051	1.045	1.048	1.082	1.077	1.036	1.097	1.115	1.127	.142	1.072	1.024	1.024	1.024	1.024	.740	.705	.691	.670	
.4500	1.042	1.049	1.060	1.068	1.121	1.127	1.187	1.205	1.233	1.249	.142	1.072	1.024	1.024	1.024	1.024	.740	.705	.691	.670	
.5000	1.046	1.064	1.103	1.158	1.210	1.249	1.275	1.331	1.327	1.333	.142	1.072	1.024	1.024	1.024	1.024	.740	.705	.691	.670	
.5500	1.070	1.092	1.154	1.220	1.287	1.325	1.266	1.406	1.540	1.267	.142	1.072	1.024	1.024	1.024	1.024	.740	.705	.691	.670	
.6000	1.104	1.141	1.199	1.266	1.308	1.379	1.362	1.377	1.572	1.923	.142	1.072	1.024	1.024	1.024	1.024	.740	.705	.691	.670	
.6500	1.156	1.193	1.239	1.292	1.339	1.322	1.322	1.316	1.454	1.781	.142	1.072	1.024	1.024	1.024	1.024	.740	.705	.691	.670	
.7000	1.190	1.230	1.270	1.295	1.351	1.326	1.322	1.292	1.604	1.731	.142	1.072	1.024	1.024	1.024	1.024	.740	.705	.691	.670	
.7500	1.262	1.281	1.328	1.321	1.372	1.353	1.353	1.373	1.434	1.627	.142	1.072	1.024	1.024	1.024	1.024	.740	.705	.691	.670	
.8000	1.312	1.324	1.325	1.354	1.453	1.392	1.417	1.442	1.434	1.583	.142	1.072	1.024	1.024	1.024	1.024	.740	.705	.691	.670	
.8500	1.322	1.328	1.327	1.344	1.422	1.363	1.425	1.442	1.492	1.536	.142	1.072	1.024	1.024	1.024	1.024	.740	.705	.691	.670	
.9000	1.322	1.332	1.333	1.336	1.382	1.400	1.442	1.492	1.525	1.536	.142	1.072	1.024	1.024	1.024	1.024	.740	.705	.691	.670	
.9500	1.371	1.426	1.427	1.444	1.422	1.363	1.425	1.482	1.483	1.485	.142	1.072	1.024	1.024	1.024	1.024	.740	.705	.691	.670	
.9940	1.210	1.137	1.166	1.173	1.210	1.246	1.292	1.337	1.404	1.417	.142	1.072	1.024	1.024	1.024	1.024	.740	.705	.691	.670	

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TABLE IX.- PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(b) Double slotted flap configuration with spoiler;  $h_s = -0.005c$ ;  $\frac{V}{b/2} = 0.21$

x/c	Upper surface								Lower surface												
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$	
<i>Wing</i>																					
.0000	.750	.971	1.270	3.352	5.441	7.605	10.098	7.702	3.559	2.409	.0125	.937	.791	.649	.581	.583	.447	.376	.629	.443	.370
.0125	.928	1.277	1.724	3.464	7.629	10.365	12.050	7.413	3.207	2.459	.0250	.969	.831	.774	.645	.643	.592	.539	.599	.551	.408
.0375	.947	1.223	1.571	2.622	3.719	6.222	8.863	6.778	3.224	2.433	.0500	.956	.854	.712	.572	.572	.529	.456	.523	.551	.408
.0750	.927	1.201	1.464	1.617	2.046	2.701	4.287	4.876	2.941	2.366	.1000	.950	.863	.727	.611	.611	.565	.493	.524	.492	.490
.1250	.977	1.195	1.408	1.545	1.842	2.162	2.979	3.763	2.738	2.376	.1500	.956	.870	.752	.623	.622	.543	.413	.348	.327	.354
.1875	1.025	1.191	1.389	1.530	1.765	2.012	2.665	3.125	2.676	2.357	.2000	.959	.889	.787	.687	.687	.611	.524	.494	.455	.464
.2500	1.047	1.201	1.364	1.476	1.642	1.790	2.244	2.605	2.555	2.300	.2500	.966	.901	.821	.726	.654	.578	.531	.492	.502	.502
.2500	1.104	1.239	1.357	1.434	1.534	1.677	1.942	2.079	2.285	2.243	.3000	.950	.905	.821	.744	.679	.608	.561	.519	.525	.540
.3500	1.117	1.252	1.367	1.431	1.512	1.614	1.860	2.122	2.171	2.218	.4000	.962	.924	.853	.771	.719	.644	.601	.554	.561	.571
.3500	1.144	1.258	1.367	1.416	1.491	1.551	1.732	1.800	2.065	2.190	.4500	.958	.930	.859	.792	.741	.665	.616	.562	.578	.587
.4000	1.172	1.284	1.387	1.416	1.466	1.566	1.674	1.720	1.980	2.174	.5000	.956	.917	.856	.792	.744	.671	.620	.595	.609	.609
.4200	1.207	1.309	1.407	1.420	1.475	1.487	1.606	1.667	1.917	2.158	.5500	.952	.903	.852	.788	.734	.672	.624	.597	.616	.596
.5000	1.245	1.340	1.430	1.457	1.487	1.527	1.616	1.655	1.882	2.112	.6000	.950	.895	.827	.762	.736	.656	.595	.636	.571	.571
.5500	1.276	1.373	1.432	1.443	1.484	1.491	1.835	1.897	1.055	2.098	.6500	.958	.877	.737	.687	.660	.593	.547	.578	.557	.558
.6000	1.314	1.398	1.455	1.500	1.503	1.595	1.600	1.767	2.079	.6500	.696	.694	.658	.617	.602	.542	.509	.476	.493	.498	
.6500	1.370	1.449	1.489	1.497	1.521	1.515	1.634	1.621	1.711	2.041	.7000	.677	.608	.564	.545	.512	.479	.457	.432	.448	.457
.7000	1.430	1.526	1.548	1.545	1.571	1.557	1.660	1.647	1.702	1.994	.7515	.587	.503	.443	.407	.392	.356	.335	.322	.325	.331
.7515	1.621	1.670	1.705	1.693	1.725	1.723	1.762	1.710	1.741	1.978	.8179	.492	.427	.398	.367	.361	.335	.311	.301	.307	.306
.8179	1.023	1.063	1.053	1.049	1.073	1.083	1.064	1.026	1.086	2.003	.8525	.470	.427	.389	.370	.364	.341	.323	.313	.319	.319
.8525	1.212	1.254	1.254	1.254	1.287	1.297	1.277	1.207	1.282	2.032	.9232	.448	.436	.414	.398	.398	.392	.381	.374	.375	.385
.9232	1.923	2.064	2.028	2.006	2.114	2.210	2.253	2.200	2.077	2.117											
<i>Vane</i>																					
.3000	2.325	2.319	2.332	2.292	2.413	2.503	2.506	2.458	2.295	2.246	.3500	1.310	1.306	1.276	1.175	1.142	1.078	1.049	1.000	.945	.940
.3500	3.766	3.772	3.953	3.876	4.191	4.653	4.790	4.766	4.177	3.795	.4000	.505	.459	.459	.364	.395	.346	.351	.348	.338	
.4000	3.699	3.733	3.706	3.599	3.901	4.383	4.519	4.493	3.906	3.461	.4500	.505	.459	.459	.364	.395	.346	.351	.348	.338	
.4200	3.615	3.647	3.577	3.446	3.753	4.272	4.473	4.491	3.803	3.205	.4800	.596	.473	.473	.394	.424	.394	.324	.312	.318	
.5000	3.403	3.523	3.427	3.310	3.655	4.144	4.415	4.462	3.788	3.164	.5500	.596	.504	.504	.424	.454	.424	.354	.342	.348	
.6000	3.354	3.402	3.278	3.196	3.567	4.042	4.381	4.473	3.832	3.221	.6000	2.347	2.293	1.911	.081	.015	.012	.003	.012	.015	
.7000	2.337	2.798	2.890	2.834	3.220	3.647	4.083	4.250	3.761	3.215	.7000	.260	.229	.141	.048	.039	.024	.014	.030	.019	
.8000	2.354	2.329	2.349	2.700	2.898	3.363	3.627	3.516	2.903	2.493	.8000	.204	.185	.116	.045	.031	.042	.040	.035	.047	.041
.8500	2.332	2.354	2.329	2.349	2.700	2.898	3.363	3.627	3.516	2.903	.8500	.172	.166	.113	.060	.055	.059	.070	.067	.074	.057
.9000	2.104	2.153	2.179	2.114	2.370	2.551	2.654	3.085	2.916	2.650	.9000	.154	.159	.110	.081	.083	.076	.104	.096	.100	.088
.7000	2.061	2.093	2.063	2.045	2.277	2.476	2.732	2.905	2.717	2.511	.7000	.179	.175	.135	.136	.142	.156	.171	.168	.159	.159
.8000	2.072	2.070	2.036	2.059	2.262	2.454	2.720	2.911	2.651	2.579	.8000	.235	.245	.223	.232	.250	.287	.299	.307	.292	.265
.9000	2.044	2.051	2.010	2.015	2.216	2.383	2.683	2.896	2.543	2.626	.9000	.683	.694	.680	.705	.759	.838	.921	.971	.850	.773
<i>Flap</i>																					
.0125	.480	.303	.176	.093	.083	.099	.119	.136	.121	.085	.0250	.169	.064	.016	.024	.012	.021	.027	.017	.032	.019
.0250	.408	.303	.176	.093	.083	.099	.119	.136	.121	.085	.0500	.107	.061	.034	.054	.049	.051	.046	.043	.062	.063
.0500	.408	.303	.176	.093	.083	.099	.119	.136	.121	.085	.0750	.094	.076	.047	.073	.074	.072	.067	.064	.096	.085
.1000	.110	.099	.091	.099	.105	.105	.104	.096	.096	.096	.1500	.110	.134	.122	.136	.139	.141	.134	.125	.142	.142
.2000	.100	.100	.100	.100	.100	.100	.100	.100	.100	.100	.2000	.130	.134	.122	.136	.139	.141	.134	.125	.142	.142
.4000	.342	.363	.339	.343	.352	.341	.329	.307	.322	.322	.4000	.342	.363	.339	.343	.352	.341	.329	.307	.322	.331
.6000	.625	.550	.570	.575	.685	.650	.613	.567	.567	.567	.6000	.625	.651	.605	.590	.574	.551	.548	.540	.581	.593
.8000	.630	.651	.605	.590	.574	.551	.548	.540	.540	.540	.8000	.630	.651	.605	.605	.656	.625	.582	.583	.625	
.9000	.715	.726	.712	.681	.685	.685	.685	.685	.685	.685	.9000	.715	.726	.712	.681	.685	.685	.685	.685	.685	
<i>Spoiler</i>																					
.2573	1.826	1.792	1.707	1.693	1.744	1.866	1.918	1.994	2.036	2.015	.4984	1.796	1.755	1.680	1.673	1.726	1.848	1.885	1.973	2.003	1.985
.7485	1.379	1.547	1.498	1.500	1.655	1.655	1.682	1.681	1.829	1.805	.7485	1.379	1.547	1.498	1.500	1.655	1.653	1.649	1.633	1.528	1.515
.9940	1.619	1.590	1.559	1.574	1.759	1.764	1.776	1.785	1.785	1.785	.9940	1.619	1.590	1.559	1.574	1.625	1.756	1.785	1.797	1.657	1.598

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NACA RM L56107

TABLE IX.- PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(c) Double slotted flap configuration with spoiler;  $h_s = -0.005\bar{c}$ ;  $\frac{V}{b/2} = 0.30$

x/c	Upper surface									Lower surface											
	$C_p$ for -									$C_p$ for -											
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 24^\circ$	$\alpha = 28^\circ$		$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 24^\circ$	$\alpha = 28^\circ$		
<b>Wing</b>																					
.0000	.857	.743	2.242	2.918	3.541	4.183	4.262	3.938	2.982	2.277	.0125	.979	.771	.734	.823	.979	1.153	1.289	1.301	1.176	1.048
.125	1.021	1.254	2.341	2.945	3.480	3.916	3.947	3.844	2.955	2.265	.0250	1.000	.810	.712	.698	.737	.778	.810	.844	.806	.735
.250	1.052	1.480	2.449	3.040	3.607	4.059	3.988	3.856	2.982	2.274	.0375	.994	.847	.724	.659	.616	.578	.554	.552	.524	.493
.375	1.070	1.379	3.084	3.942	3.936	4.021	3.944	3.858	2.904	2.253	.0500	.997	.862	.746	.671	.607	.542	.490	.475	.484	.453
.500	1.079	1.266	1.932	4.029	5.254	5.302	4.384	3.956	2.925	2.250	.0750	.994	.872	.768	.677	.607	.530	.461	.448	.454	.429
.625	1.075	1.229	1.282	2.509	4.042	4.802	4.320	3.850	2.934	2.271	.1000	.994	.872	.768	.677	.607	.530	.461	.448	.454	.429
.750	1.04	1.211	1.288	1.439	3.245	3.685	3.505	2.791	2.244	.1250	.928	.857	.793	.707	.634	.542	.461	.448	.439	.417	
.875	1.129	1.211	1.344	1.378	1.613	2.371	3.026	3.118	2.695	2.241	.2000	.920	.814	.732	.646	.554	.472	.451	.442	.417	
.2500	1.159	1.233	1.363	1.396	1.674	1.928	2.571	2.800	2.603	2.220	.3000	.988	.905	.835	.756	.677	.590	.504	.481	.465	.419
.3750	1.160	1.261	1.393	1.409	1.438	1.697	2.479	2.709	2.574	2.220	.4000	.989	.844	.768	.695	.618	.542	.493	.473	.437	
.5000	1.200	1.269	1.363	1.378	1.467	1.670	2.055	2.239	2.361	2.178	.5000	.997	.857	.793	.701	.608	.542	.493	.473	.453	
.6250	1.278	1.336	1.412	1.444	1.599	1.941	2.180	2.272	2.152	2.140	.7521	.971	.816	.771	.726	.671	.608	.542	.520	.503	
.8750	1.317	1.352	1.449	1.451	1.474	1.531	1.784	1.974	2.049	2.107	.8017	.930	.857	.793	.707	.634	.542	.516	.519	.488	
.9375	1.339	1.398	1.464	1.470	1.484	1.575	1.740	1.903	2.006	2.059	.8144	.979	.862	.814	.750	.671	.587	.520	.516	.498	
.0000	1.396	1.437	1.505	1.518	1.511	1.608	1.734	1.859	1.952	2.030	.8228	.927	.855	.793	.702	.634	.542	.472	.451	.442	
.1250	1.424	1.523	1.570	1.577	1.571	1.623	1.720	1.823	1.938	2.003	.8300	.997	.862	.814	.750	.671	.587	.520	.516	.498	
.2500	1.425	1.761	1.780	1.787	1.783	1.787	1.831	1.894	1.890	1.917	.7521	.971	.802	.739	.670	.589	.520	.493	.473	.437	
.3750	1.427	2.072	2.001	2.087	2.078	2.108	2.084	2.121	1.961	1.961	.7234	.972	.862	.793	.727	.654	.586	.520	.493	.473	
.5000	1.451	2.372	2.272	2.292	2.266	2.332	2.248	2.249	2.044	2.044	.8017	.930	.857	.847	.731	.602	.527	.474	.451	.429	
.6250	1.451	2.422	2.422	2.450	2.420	2.467	2.414	2.410	2.119	1.994	.8099	.924	.867	.834	.730	.605	.527	.474	.451	.429	
.8750	2.567	2.567	2.568	2.568	2.568	2.598	2.559	2.540	2.230	2.027	.8224	.924	.862	.834	.730	.605	.527	.474	.451	.429	
<b>Vane</b>																					
.0000	4.662	4.492	4.415	4.384	4.405	4.515	4.332	4.280	3.660	3.303	.0250	1.720	1.676	1.625	1.528	1.441	1.377	1.265	1.233	1.098	1.018
.0500	6.089	5.746	5.808	5.683	5.844	5.725	5.531	4.182	3.354	.0375	.756	.740	.749	.634	.526	.425	.367	.334	.313	.289	
.0750	6.147	5.833	5.777	5.793	5.710	5.922	5.739	4.205	3.330	.1000	.287	.287	.310	.210	.121	.057	.047	.032	.036	.018	
.1000	6.177	5.871	5.785	5.753	5.625	5.862	5.810	5.484	4.075	2.911	.1250	.149	.165	.189	.107	.036	.000	.000	.000	.000	
.1500	6.135	5.829	5.687	5.687	5.541	5.799	5.778	5.449	4.087	2.867	.2000	.076	.095	.118	.052	.024	.000	.000	.000	.000	
.2000	6.077	5.814	5.629	5.629	5.447	5.704	5.676	5.308	4.076	2.855	.3000	.037	.057	.050	.037	.024	.000	.000	.000	.000	
.3000	5.574	5.343	5.164	5.164	4.667	5.111	4.413	4.195	4.030	2.922	.4000	.040	.052	.024	.020	.030	.009	.017	.012	.030	.031
.4000	4.436	4.436	4.436	4.436	4.437	4.701	4.795	4.652	3.713	2.738	.4000	.043	.049	.059	.048	.039	.038	.035	.036	.039	
.5000	4.336	4.098	3.867	3.867	3.746	4.237	4.305	4.227	3.409	2.607	.5000	.079	.049	.050	.066	.060	.070	.077	.075	.060	
.6250	3.768	3.495	3.291	3.291	3.229	3.205	3.695	3.778	3.744	3.122	.7000	.140	.113	.111	.134	.142	.126	.131	.130	.128	.116
.8750	3.159	2.969	2.793	2.793	2.761	3.213	3.289	3.301	2.839	2.339	.8000	.268	.235	.226	.247	.245	.254	.271	.260	.236	.226
.9375	2.653	2.492	2.413	2.413	2.390	2.765	2.830	2.873	2.522	2.208	.9200	.811	.752	.749	.798	.722	.796	.816	.823	.767	.702
<b>Flap</b>																					
.0125	.232	.220	.229	.210	.172	.172	.171	.171	.192	.173	.0250	.232	.220	.219	.172	.171	.171	.192	.173	.155	
.0500	.059	.050	.050	.050	.050	.050	.050	.050	.050	.050	.050	.059	.057	.057	.057	.057	.057	.057	.057	.053	
.0750	.053	.053	.053	.053	.053	.053	.053	.053	.053	.053	.053	.053	.053	.053	.053	.053	.053	.053	.053	.053	
.1000	.037	.037	.037	.037	.037	.037	.037	.037	.037	.037	.037	.037	.037	.037	.037	.037	.037	.037	.037	.037	
.1500	.067	.067	.067	.067	.067	.067	.067	.067	.067	.067	.067	.067	.067	.067	.067	.067	.067	.067	.067	.067	
.2000	.101	.076	.087	.101	.091	.091	.063	.063	.058	.062	.062	.062	.062	.062	.062	.062	.062	.062	.062	.062	
.3000	.226	.214	.195	.223	.214	.214	.159	.159	.146	.139	.137	.137	.137	.137	.137	.137	.137	.137	.137	.137	
.4000	.366	.352	.337	.345	.326	.326	.281	.281	.259	.242	.242	.242	.242	.242	.242	.242	.242	.242	.242	.242	
.5000	.540	.536	.533	.512	.489	.489	.467	.467	.431	.410	.410	.410	.410	.410	.410	.410	.410	.410	.410	.410	
.6250	.729	.725	.712	.704	.686	.686	.665	.665	.621	.599	.599	.599	.599	.599	.599	.599	.599	.599	.599	.599	
<b>Spoiler</b>																					
.2475	1.686	1.667	1.616	1.640	1.717	1.622	1.601	1.601	1.614	1.602	.5000	1.842	1.801	1.719	1.693	1.747	1.654	1.600	1.602	1.602	
.4909	1.707	1.691	1.640	1.655	1.735	1.619	1.609	1.609	1.750	1.769	.5000	1.796	1.758	1.649	1.649	1.740	1.682	1.692	1.692	1.692	
.7375	1.677	1.664	1.601	1.628	1.677	1.759	1.746	1.687	1.687	1.687	.5000	1.622	1.593	1.544	1.544	1.577	1.651	1.519	1.475	1.562	
.9798	1.714	1.688	1.619	1.631	1.651	1.743	1.743	1.696	1.696	1.696	.5000	1.646	1.645	1.601	1.601						

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TABLE IX.- PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(d) Double slotted flap configuration with spoiler;  $h_s = -0.005\bar{c}$ ;  $\frac{V}{b/2} = 0.43$

x/c	Upper surface										Lower surface																		
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$									
<b>Wing</b>																													
.0000	.846	1.063	2.147	3.527	3.001	2.893	2.780	2.510	2.500	2.291	.0125	.947	.787	.812	.953	1.163	1.278	1.360	1.316	1.333	1.344								
.0125	1.078	1.822	2.119	2.509	2.953	2.692	2.680	2.589	2.407	2.216	.0250	.972	.809	.762	.777	.852	.948	.904	.923	.959	.959	.959							
.0250	1.122	1.758	2.184	2.551	2.954	2.949	2.909	2.589	2.407	2.297	.0375	.974	.841	.762	.714	.722	.686	.701	.657	.690	.752	.752							
.0500	1.066	1.726	2.223	2.629	3.089	2.997	2.909	2.635	2.422	2.347	.0750	.981	.860	.771	.708	.673	.626	.622	.591	.608	.647	.647							
.1000	1.122	1.542	2.072	2.925	2.987	2.859	2.921	2.638	2.413	2.322	.1250	.972	.873	.793	.755	.665	.599	.573	.555	.555	.592	.596							
.1500	1.122	1.290	2.423	3.696	3.704	3.078	3.012	2.644	2.413	2.322	.2000	.953	.876	.806	.734	.663	.575	.537	.504	.522	.528	.528							
.2000	1.169	1.303	1.605	2.949	3.999	3.335	2.994	2.603	2.395	2.303	.2500	.941	.876	.803	.714	.679	.594	.546	.510	.519	.543	.543							
.2500	1.182	1.319	1.276	1.964	3.200	3.189	2.872	2.529	2.372	2.281	.3000	.953	.880	.799	.703	.643	.594	.543	.504	.516	.536	.536							
.3000	1.226	1.370	1.314	1.545	2.571	3.000	2.768	2.458	2.345	2.281	.3500	.861	.838	.790	.714	.670	.594	.552	.513	.519	.546	.546							
.3500	1.251	1.355	1.345	1.345	2.080	2.665	2.598	2.351	2.289	2.259	.4000	.840	.799	.756	.694	.648	.572	.546	.513	.519	.540	.540							
.4000	1.292	1.417	1.395	1.343	1.035	2.452	2.488	2.298	2.248	2.250	.4500	.776	.761	.721	.663	.617	.554	.515	.493	.502	.521	.521							
.4500	1.329	1.459	1.352	1.358	1.128	2.278	2.384	2.224	2.207	2.224	.5000	.734	.710	.674	.605	.556	.527	.494	.461	.472	.505	.505							
.5000	1.348	1.437	1.410	1.406	1.130	2.206	2.414	2.116	2.136	2.212	.5500	.690	.634	.583	.539	.537	.452	.454	.417	.422	.448	.448							
.5500	1.349	1.557	1.555	1.451	1.611	1.975	2.146	2.052	2.077	2.180	.6000	.608	.567	.549	.494	.500	.428	.424	.391	.413	.429	.429							
.6000	1.324	1.640	1.633	1.536	1.632	1.907	2.076	2.000	2.041	2.188	.6500	.614	.595	.575	.525	.530	.431	.427	.400	.415	.457	.457							
.6500	1.662	1.921	1.915	1.834	1.821	1.865	1.994	1.919	1.974	2.101	.7000	.614	.595	.575	.525	.530	.431	.427	.400	.415	.457	.457							
.7000	2.100	2.344	2.351	2.262	2.188	2.057	2.110	1.983	1.977	2.108	.7500	.648	.611	.585	.516	.505	.406	.400	.381	.396	.457	.457							
.7500	2.383	2.701	2.690	2.608	2.534	2.272	2.262	2.070	2.012	2.120	.7700	.634	.582	.535	.516	.502	.257	.259	.249	.242	.259	.259							
.7700	2.464	2.784	2.784	2.714	2.626	2.344	2.299	2.102	2.033	2.123	.8000	.634	.582	.535	.516	.502	.257	.259	.249	.242	.259	.259							
.8000	2.590	2.924	2.925	2.834	2.750	2.428	2.357	2.134	2.047	2.133	.8500	.634	.582	.535	.516	.502	.257	.259	.249	.242	.259	.259							
<b>Vane</b>																													
.0000	.6750	4.160	4.027	3.922	3.706	3.344	3.137	2.787	2.561	2.425	.0250	.962	.994	1.063	.949	.873	.716	.649	.574	.572	.596	.596							
.0500	6.781	6.918	6.915	6.901	6.490	5.040	4.891	3.972	3.159	2.972	.1000	.938	.207	.317	.232	.179	.111	.079	.072	.080	.093	.093							
.1000	6.997	7.064	7.101	6.997	6.678	5.467	4.851	3.809	2.891	2.650	.1500	.905	.204	.254	.151	.114	.060	.042	.027	.029	.018	.044							
.1500	7.193	7.262	7.295	7.220	6.903	5.632	4.958	3.841	2.832	2.537	.2000	.900	.012	.097	.046	.039	.030	.020	.016	.033	.033	.033							
.2000	7.411	7.523	7.558	7.488	7.212	5.910	5.192	4.024	2.996	2.644	.3000	.019	.022	.075	.048	.031	.024	.017	.023	.025	.025	.025							
.3000	6.778	6.886	6.878	6.834	6.635	5.446	4.772	3.702	2.741	2.477	.4000	.016	.035	.063	.042	.040	.036	.032	.029	.046	.046	.046							
.4000	6.041	6.137	6.110	6.075	5.987	4.967	4.381	3.422	2.605	2.398	.5000	.069	.057	.069	.072	.093	.087	.085	.052	.089	.095	.095							
.5000	5.370	5.405	5.411	5.404	5.311	4.500	4.016	3.169	2.472	2.322	.6000	.088	.089	.094	.096	.105	.111	.110	.099	.112	.101	.101							
.6000	4.840	4.885	4.671	4.678	4.623	3.970	3.595	2.916	2.360	2.259	.7000	.147	.147	.141	.154	.167	.153	.152	.139	.136	.142	.142							
.7000	4.041	4.093	4.068	4.084	4.055	3.554	3.287	2.713	2.260	2.221	.8000	.314	.312	.307	.325	.327	.293	.293	.258	.248	.249	.249							
.8000	3.447	3.491	3.464	3.479	3.450	3.105	2.924	2.470	2.145	2.152	.9000	1.022	1.026	1.044	1.033	1.025	.919	.872	.762	.720	.773	.773							
<b>Flap</b>																													
.0125	.226	.261	.351	.340	.315	.272	.256	.226	.201	.215	.0250	.226	.261	.300	.340	.315	.272	.256	.201	.215	.215	.215							
.0500	.032	.030	.080	.089	.089	.027	.027	.052	.046	.071	.0500	.003	.013	.011	.021	.049	.021	.049	.012	.047	.054	.054							
.0750	.003	.003	.003	.003	.003	.003	.003	.003	.003	.003	.0750	.003	.003	.003	.003	.003	.003	.003	.003	.003	.003	.003							
.1000	.009	.025	.047	.033	.049	.018	.043	.025	.018	.025	.1500	.031	.067	.078	.084	.071	.084	.066	.058	.074	.059	.059							
.2000	.409	.469	.483	.475	.408	.081	.108	.087	.091	.075	.4000	.191	.169	.172	.166	.170	.159	.152	.148	.171	.190	.190							
.4000	.467	.449	.448	.443	.447	.447	.416	.399	.394	.443	.6000	.298	.299	.279	.286	.284	.263	.244	.238	.257	.290	.290							
.8000	.467	.449	.448	.443	.447	.447	.416	.399	.394	.443	.9000	.574	.602	.574	.584	.599	.569	.546	.539	.603	.669	.669							
.9000	1.226	1.248	1.254	1.303	1.424	1.347	1.351	1.398	1.470	1.610	<b>Spoiler</b>																		
.2443	1.518	1.501	1.483	1.518	1.625	1.724	1.764	1.806	1.808	1.852	.5287	1.869	1.844	1.785	1.678	1.698	1.729	1.791	1.854	1.876	1.906								
.4206	1.193	1.388	1.370	1.434	1.337	1.602	1.620	1.683	1.636	1.769	.5008	1.753	1.731	1.665	1.601	1.640	1.676	1.714	1.770	1.814	1.855								
.7370	1.270	1.278	1.284	1.348	1.460	1.486	1.481	1.519	1.605	1.690	.5954	1.152	1.212	1.193	1.255	1.297	1.294	1.288	1.334	1.437	1.596								
.9856	1.226	1.248	1.254	1.303	1.424	1.347	1.351	1.398	1.470	1.610	.5940	1.193	1.217	1.233	1.268	1.364	1.292	1.361	1.349	1.473	1.592								

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TABLE IX.- PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(e) Double slotted flap configuration with spoiler;  $h_s = -0.005\bar{c}$ ;  $\frac{V}{b/2} = 0.55$

x/c	Upper surface										Lower surface											
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$		
Wing																						
.0000	1.923	1.890	2.118	2.405	2.804	2.889	2.180	2.724	2.161	2.101	.0125	.903	.786	.954	1.189	1.359	1.455	1.501	1.584	1.582	1.598	
.0125	1.349	1.349	2.176	2.473	2.526	2.419	2.265	2.272	2.185	2.116	.0250	.951	.786	.817	.884	1.046	1.012	1.046	1.094	1.107	1.137	
.0375	1.229	1.847	2.136	2.430	2.480	2.410	2.265	2.266	2.188	2.113	.0500	.967	.823	.780	.750	.755	.755	.772	.817	.800	.823	
.0750	1.210	1.804	2.282	2.561	2.550	2.488	2.315	2.312	2.197	2.095	.1000	1.226	1.764	2.337	2.616	2.595	2.464	2.326	2.191	2.101		
.1500	1.241	1.593	2.740	2.659	2.526	2.431	2.352	2.331	2.173	2.101	.2000	1.271	1.407	2.972	2.926	2.477	2.437	2.307	2.173	2.104		
.2500	1.308	1.373	2.039	3.584	2.655	2.479	2.358	2.313	2.191	2.093	.3000	1.332	1.376	2.328	3.604	2.935	2.545	2.341	2.289	2.182	2.086	
.3500	1.301	1.410	1.824	3.137	3.000	2.545	2.303	2.274	2.170	2.086	.4000	1.430	1.452	1.450	2.561	2.376	2.505	2.259	2.143	2.083		
.4500	1.451	1.551	1.359	2.048	2.686	2.451	2.218	2.195	2.120	2.075	.5000	1.540	1.487	1.459	1.468	2.403	2.358	2.118	2.124	2.056		
.5500	1.540	1.507	1.459	1.458	2.403	2.358	2.118	2.124	2.124	2.075	.6021	1.726	1.616	1.771	1.659	1.659	1.656	1.656	1.659	1.659		
.6250	1.621	1.616	1.771	1.659	1.612	1.672	2.026	2.047	2.030	2.018	.6534	2.204	2.294	2.266	2.142	2.049	2.059	2.033	2.027			
.7087	2.543	2.683	2.598	2.357	2.272	2.162	2.084	2.024	2.054	2.030	.7151	2.570	2.679	2.632	2.415	2.290	2.177	2.084	2.054	2.042		
.7214	2.637	2.810	2.830	2.592	2.396	2.240	2.125	2.121	2.066	2.042	.7278	2.784	2.948	2.950	2.701	2.435	2.257	2.143	2.127	2.072	2.036	
Vane																						
.0000	3.354	3.339	3.214	3.845	2.482	2.287	2.160	2.162	2.075	2.042	.0250	6.543	6.526	6.508	6.808	4.643	3.946	3.571	3.522	3.140	2.743	
.0625	6.663	6.627	6.687	5.970	4.707	3.940	3.513	3.422	2.970	2.774	.1000	7.153	7.058	7.482	6.482	4.898	5.598	3.381	3.231	2.749	2.556	
.1500	7.205	7.098	7.273	6.577	6.823	3.814	3.288	3.154	2.615	2.464	.2000	7.229	7.113	7.334	6.720	4.905	3.832	3.291	3.142	2.576	2.417	
.2500	6.897	6.777	7.009	6.485	6.692	3.662	3.139	2.994	2.451	2.286	.3000	6.156	6.055	6.276	5.872	4.248	3.323	2.871	2.758	2.299	2.178	
.3500	6.055	6.276	5.872	4.248	3.323	2.632	2.179	2.020	1.777	1.656	.4000	5.583	5.486	5.721	5.404	3.948	3.117	2.731	2.634	2.245	2.131	
.4500	4.777	4.777	4.777	4.777	4.777	4.777	4.777	4.777	4.777	4.777	.5000	4.205	4.205	4.205	4.205	4.205	4.205	4.205	4.205	4.205	4.205	
.5500	3.944	3.101	3.189	3.061	2.544	2.440	2.245	2.221	2.221	2.038	.6000	3.648	3.627	3.724	3.583	2.884	2.440	2.245	2.221	2.103	2.018	
Flap																						
.0000	1.906	1.954	2.111	2.073	1.767	1.548	1.423	1.392	1.334	1.348	.0125	2.254	2.254	2.124	2.064	1.854	1.738	1.622	1.507	1.402	1.307	
.0250	2.254	3.015	3.124	3.046	2.631	2.311	2.143	2.112	2.045	2.077	.0375	2.455	2.455	2.312	2.178	1.945	1.783	1.622	1.507	1.402	1.307	
.0500	3.455	3.581	3.678	3.804	3.136	2.766	2.556	2.528	2.454	2.467	.0750	3.685	3.685	3.685	3.685	3.685	3.685	3.685	3.685	3.685	3.685	
.1000	3.580	3.685	3.786	3.738	3.269	2.895	2.667	2.631	2.519	2.461	.1500	3.574	3.700	3.790	3.317	2.934	2.685	2.628	2.495	2.396	2.306	
.2000	3.574	3.700	3.790	3.769	3.317	2.934	2.685	2.628	2.495	2.396	.2500	3.626	3.626	3.626	3.626	3.626	3.626	3.626	3.626	3.626	3.626	
.3000	3.626	3.626	3.626	3.626	3.626	3.626	3.626	3.626	3.626	3.626	.3500	3.626	3.626	3.626	3.626	3.626	3.626	3.626	3.626	3.626	3.626	
.4000	3.626	3.626	3.626	3.626	3.626	3.626	3.626	3.626	3.626	3.626	.4500	3.626	3.626	3.626	3.626	3.626	3.626	3.626	3.626	3.626	3.626	
.5000	3.626	3.626	3.626	3.626	3.626	3.626	3.626	3.626	3.626	3.626	.5500	3.626	3.626	3.626	3.626	3.626	3.626	3.626	3.626	3.626	3.626	
.6000	3.626	3.626	3.626	3.626	3.626	3.626	3.626	3.626	3.626	3.626	.6500	3.626	3.626	3.626	3.626	3.626	3.626	3.626	3.626	3.626	3.626	
.7087	3.626	3.626	3.626	3.626	3.626	3.626	3.626	3.626	3.626	3.626	.7151	3.626	3.626	3.626	3.626	3.626	3.626	3.626	3.626	3.626	3.626	
Spoiler																						
.2443	1.814	1.786	1.737	1.723	1.845	1.822	1.805	1.815	1.805	1.840	.4861	1.651	1.602	1.541	1.565	1.695	1.712	1.696	1.722	1.738	1.772	
.7416	1.421	1.410	1.350	1.372	1.540	1.596	1.587	1.621	1.661	1.716	.9856	1.268	1.272	1.217	1.226	1.381	1.472	1.475	1.510	1.578	1.654	

TABLE IX. - PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Concluded

(f) Double slotted flap configuration with spoiler;  $h_s = -0.005c$ ;  $\frac{V}{b/2} = 0.72$

x/c	Upper surface										Lower surface												
	$C_p$ for -											$C_p$ for -											
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$		$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$		
Wing																							
.0000	2.165	2.208	2.359	2.292	2.226	2.015	1.935	1.925	1.904	1.917		.0125	.892	.862	1.012	1.137	1.241	1.282	1.328	1.364	1.401	1.441	
.0125	2.101	2.162	2.269	2.202	2.159	1.955	1.923	1.910	1.900	1.914		.0250	.799	.783	.884	.937	.994	1.047	1.080	1.119	1.162	1.201	
.0250	2.174	2.196	2.308	2.244	2.180	1.997	1.950	1.940	1.900	1.920		.0375	.829	.771	.871	.914	.956	.974	.970	.974	.980	.983	.990
.0500	2.348	2.257	2.338	2.253	2.186	2.038	1.977	1.946	1.912	1.923		.0750	.758	.758	.780	.811	.846	.855	.879	.923	.950		
.0750	1.970	2.297	2.408	2.283	2.217	2.056	1.982	1.955	1.909	1.923		.1000	.829	.771	.771	.774	.776	.794	.790	.820	.843		
.1000	1.451	2.266	2.384	2.271	2.195	2.032	1.974	1.958	1.912	1.923		.1250	.854	.783	.740	.762	.892	.879	.875	.875	.875	.875	
.1500	1.390	2.388	2.459	2.300	2.212	2.044	2.003	1.970	1.918	1.923		.2000	.860	.807	.758	.717	.696	.674	.670	.678	.685	.694	
.2000	1.443	2.563	2.559	2.300	2.201	2.068	2.021	1.970	1.923	1.926		.2500	.881	.832	.792	.738	.713	.682	.661	.666	.673	.681	
.2500	1.500	2.344	2.643	2.274	2.186	2.068	2.021	1.970	1.920	1.935		.3000	.921	.875	.849	.786	.753	.706	.699	.698	.693	.710	
.3000	1.456	2.354	2.800	2.262	2.192	2.056	2.006	1.976	1.932	1.941		.3500	1.009	.979	.949	.881	.835	.780	.755	.779	.761	.775	
.3500	1.467	2.427	2.603	2.255	2.211	2.052	2.009	1.973	1.938	1.947		.4000	1.146	1.138	1.115	1.012	.954	.881	.861	.875	.886	.889	
.4000	1.610	1.927	3.023	2.072	2.226	2.059	2.003	1.984	1.941	1.953		.4500	1.363	1.370	1.353	1.184	1.098	1.015	.982	.979	.976	.979	
.4500	1.656	1.734	3.075	2.425	2.422	2.059	2.003	1.984	1.941	1.953		.5000	1.428	1.627	1.619	1.396	1.274	1.166	1.133	1.125	1.118	1.101	
.5000	1.720	1.673	3.030	2.530	2.275	2.043	1.977	1.966	1.944	1.956		.5500	1.878	1.887	1.870	1.613	1.483	1.353	1.325	1.334	1.319	1.296	
.5500	1.750	1.624	2.683	2.470	2.247	2.032	1.982	1.958	1.944	1.962		.6000	2.078	2.110	2.069	1.803	1.692	1.567	1.552	1.549	1.549	1.542	
.6000	1.790	1.627	2.580	2.381	2.232	2.015	1.971	1.955	1.950	1.974		.6500	2.259	2.294	2.190	1.912	1.832	1.742	1.746	1.764	1.773	1.731	
.6500	1.759	1.593	2.326	2.250	2.195	2.009	1.971	1.943	1.959	1.983		.7000	2.440	2.355	2.286	1.982	1.882	1.822	1.841	1.857	1.847	1.808	
.7000	1.686	1.523	2.145	2.199	2.174	1.988	1.956	1.940	1.965	1.994		.7500	2.329	2.327	2.142	1.922	1.875	1.819	1.833	1.830	1.826	1.799	
.7500	1.610	1.477	2.000	2.202	2.168	1.979	1.956	1.937	1.971	2.003		.8000	2.183	2.168	2.039	1.887	1.832	1.791	1.743	1.711	1.722	1.769	
.8000	1.497	1.383	1.855	2.137	2.122	1.976	1.953	1.943	1.977	2.003		.8500	2.012	2.006	1.936	1.848	1.802	1.718	1.702	1.719	1.720		
.8500	1.430	1.333	1.749	2.024	2.058	1.949	1.938	1.937	1.985	2.006		.9000	1.842	1.844	1.834	1.824	1.802	1.727	1.702	1.722	1.724	1.746	
.9000	1.384	1.312	1.656	1.961	2.031	1.940	1.932	1.937	1.985	2.009		.9500	1.640	1.646	1.710	1.806	1.808	1.733	1.726	1.737	1.739	1.759	

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TABLE X.- PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE

(a) Double slotted flap configuration with spoiler;  $h_s = -0.010\bar{c}$ ;  $\frac{V}{b/2} = 0$

x/c	Upper surface										Lower surface										
	$C_p$ for -										$C_p$ for -										
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$	
<b>Fuselage</b>																					
.0200	.000	.003	.000	.000	.000	.030	.053	.084	.105	.130	.0500	.650	.809	.749	.695	.619	.563	.472	.445	.409	.386
.0300	.585	.631	.690	.941	.973	.015	.018	.072	.047	.075	.1000	.709	.911	.847	.813	.713	.628	.534	.552	.518	.499
.1000	.641	.623	.965	1.009	1.024	1.066	1.071	1.131	1.120	1.133	.1500	.805	.960	.914	.872	.793	.654	.539	.599	.574	
.1500	.771	.982	1.024	1.065	1.069	1.096	1.094	1.140	1.126	1.135	.2000	.826	.997	.951	.875	.855	.805	.717	.701	.658	.628
.2000	.814	1.034	1.067	1.109	1.091	1.111	1.100	1.140	1.111	1.127	.2500	.868	1.034	1.004	1.002	1.002	1.002	1.002	1.002	1.002	1.002
.2500	.854	1.064	1.099	1.116	1.106	1.131	1.125	1.160	1.130	1.146	.3000	.930	1.080	1.101	1.115	1.120	1.125	1.128	1.130	1.132	1.134
.3000	.930	1.080	1.101	1.115	1.100	1.122	1.114	1.142	1.119	1.138	.3500	.910	1.031	1.012	.997	.921	.874	.808	.797	.737	.726
.4000	.910	1.040	1.089	1.096	1.082	1.094	1.089	1.119	1.099	1.121	.4000	.799	.963	.936	.919	.867	.832	.767	.746	.702	.690
.4500	.928	1.043	1.055	1.084	1.049	1.084	1.092	1.125	1.103	1.133	.4500	.747	.932	.893	.872	.819	.772	.699	.684	.643	.617
.5000	.947	1.053	1.070	1.109	1.109	1.133	1.171	1.221	1.222	1.231	.5000	.703	.880	.841	.816	.749	.698	.622	.609	.567	.539
.5500	.957	1.053	1.110	1.165	1.193	1.246	1.289	1.349	1.386	1.422	.5500	.632	.843	.798	.763	.695	.638	.555	.531	.494	.481
.6000	.957	1.111	1.165	1.237	1.269	1.320	1.363	1.418	1.344	1.393	.6000	.628	.774	.731	.685	.616	.563	.499	.478	.444	.438
.6500	1.015	1.163	1.217	1.286	1.293	1.329	1.357	1.404	1.353	1.393	.6500	1.022	1.003	.978	.927	.886	.817	.769	.751	.741	
.7000	1.062	1.203	1.269	1.315	1.302	1.329	1.319	1.334	1.447	1.755	.7000	.901	1.098	1.122	1.156	1.157	1.168	1.153	1.103	1.116	1.142
.7500	1.182	1.292	1.300	1.330	1.320	1.329	1.310	1.322	1.404	1.680	.7500	.774	.908	.859	.841	.788	.766	.690	.690	.658	.646
.8000	1.266	1.358	1.377	1.384	1.382	1.401	1.442	1.486	1.468	1.556	.8000	1.192	1.252	1.259	1.264	1.261	1.268	1.269	1.269	1.269	
.8500	1.192	1.324	1.350	1.382	1.359	1.384	1.380	1.440	1.414	1.529	.8500	1.172	1.251	1.274	1.286	1.317	1.357	1.424	1.453	1.432	
.9000	1.192	1.324	1.350	1.382	1.359	1.384	1.380	1.440	1.414	1.529	.9000	1.075	1.175	1.168	1.190	1.234	1.286	1.361	1.374	1.398	
.9500	1.046	1.203	1.190	1.193	1.240	1.292	1.352	1.380	1.392		.9500	.879	1.148	1.183	1.218	1.224	1.278	1.313	1.370	1.386	1.389

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TABLE X.- PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(b) Double slotted flap configuration with spoiler;  $h_s = -0.010\bar{c}$ ;  $\frac{y}{b/2} = 0.21$

		Upper surface										Lower surface									
		$C_p$ for -										$C_p$ for -									
$x/c$		$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
<b>Wing<sup>1</sup></b>																					
-0.0000	.820	.988	1.938	3.581	5.423	7.439	9.589	8.334	3.482	2.616											
.0125	.953	1.294	1.722	3.806	7.646	10.289	11.679	8.027	3.123	2.396											
.0250	.938	1.245	1.556	2.824	3.733	5.932	8.345	7.307	3.168	2.405											
.0500	1.019	1.211	1.456	1.678	2.027	2.648	4.087	5.204	2.889	2.315											
.0750	1.029	1.201	1.400	1.596	1.814	2.131	2.859	3.691	2.835	2.294											
.1000	1.038	1.208	1.378	1.576	1.736	1.967	2.568	3.342	2.799	2.277											
.1500	1.060	1.211	1.353	1.514	1.607	1.782	2.201	2.626	2.703	2.226											
.2000	1.098	1.220	1.356	1.473	1.553	1.707	2.030	2.322	2.575	2.196											
.2500	1.117	1.233	1.338	1.473	1.508	1.645	1.910	2.103	2.366	2.149											
.3000	1.132	1.261	1.336	1.453	1.477	1.600	1.826	1.959	2.207	2.113											
.3500	1.164	1.271	1.355	1.432	1.447	1.570	1.708	1.852	2.075	2.089											
.4000	1.187	1.293	1.356	1.432	1.438	1.555	1.747	1.876	2.057	2.056											
.4500	1.212	1.303	1.356	1.432	1.438	1.515	1.613	1.711	1.877	2.033											
.5000	1.226	1.365	1.406	1.453	1.488	1.516	1.592	1.676	1.029	2.012											
.5500	1.229	1.381	1.425	1.473	1.444	1.484	1.559	1.623	1.769	1.976											
.6000	1.331	1.422	1.450	1.494	1.445	1.484	1.565	1.640	1.728	1.949											
.6500	1.308	1.466	1.488	1.514	1.489	1.504	1.601	1.655	1.686	1.922											
.7000	1.451	1.522	1.531	1.596	1.550	1.543	1.643	1.670	1.688	1.881											
.7815	1.634	1.686	1.678	1.698	1.685	1.713	1.733	1.729	1.725	1.869											
.8179	1.824	1.846	1.831	1.842	1.829	1.922	1.919	1.900	1.853	1.902											
.8325	1.934	1.947	1.919	1.944	1.925	2.042	2.051	2.027	1.943	1.940											
.8362	1.959	1.978	1.956	1.964	1.973	2.084	2.099	2.083	1.997	1.955											
.8394	1.928	1.931	1.909	1.944	1.928	2.030	2.051	2.044	1.970	1.949											
.8434	2.016	2.044	2.016	2.046	2.024	2.155	2.198	2.183	2.078	1.994											
<b>Vane</b>																					
-0.0000	2.328	2.305	2.297	2.312	2.321	2.454	2.462	2.431	2.302	2.128											
.0250	3.988	3.931	3.691	3.929	4.012	4.531	4.664	4.664	4.207	3.601											
.0500	3.767	3.689	3.628	3.642	3.736	4.280	4.390	4.390	3.943	3.315											
.1000	5.717	5.604	5.525	5.479	5.492	6.149	6.336	6.357	5.829	3.041											
.1500	3.590	3.475	3.388	3.315	3.495	4.021	4.255	4.513	3.811	2.997											
.2000	3.480	3.359	3.269	3.233	3.417	3.913	4.201	4.319	3.847	3.047											
.3000	3.035	2.950	2.859	2.865	3.087	3.531	3.877	4.083	3.740	3.009											
.4000	2.663	2.588	2.525	2.578	2.781	3.104	3.492	3.752	3.509	2.859											
.5000	2.391	2.350	2.303	2.394	2.589	2.830	3.204	3.416	3.296	2.738											
.6000	2.171	2.129	2.106	2.149	2.273	2.484	2.742	2.909	2.685	2.482											
.7000	2.123	2.073	2.057	2.087	2.183	2.409	2.646	2.774	2.667	2.560											
.8000	2.011	2.060	2.004	2.057	2.180	2.397	2.688	2.782	2.652	2.477											
.9000	2.076	2.025	1.975	2.046	2.132	2.346	2.613	2.770	2.572	2.178											
<b>Flap</b>																					
.0000	1.208	1.525	1.528	1.412	1.267	1.337	1.456	1.590	1.518	1.375											
.0125	2.205	2.233	2.141	2.005	2.078	2.316	2.541	2.679	2.398	2.045											
.0250	2.391	2.396	2.341	2.415	2.417	2.692	2.976	3.130	2.829	2.422											
.0500	2.489	2.485	2.428	2.497	2.568	2.827	3.099	3.266	3.024	2.696											
.0750	2.335	2.318	2.253	2.292	2.372	2.609	2.844	2.971	2.787	2.544											
.1000	2.130	2.120	2.050	2.087	2.162	2.352	2.556	2.637	2.548	2.363											
.1500	1.799	1.729	1.698	1.808	1.964	2.126	2.213	2.186	2.065	1.865											
.2000	1.767	1.745	1.694	1.750	1.901	2.063	2.171	2.129	2.006	1.800											
.4000	1.814	1.805	1.750	1.821	1.778	1.916	2.087	2.186	2.159	2.042											
.6000	1.622	1.626	1.597	1.678	1.634	1.687	1.652	1.596	1.593	1.708											
.8000	1.628	1.629	1.600	1.678	1.637	1.681	1.658	1.603	1.590	1.708											
<b>Spoiler</b>																					
.2429	1.028	1.612	1.657	1.663	1.674	1.778	1.879	2.012	1.959	1.911											
.4852	.966	1.560	1.621	1.651	1.652	1.740	1.841	1.955	1.871	1.833											
.7367	.938	1.520	1.575	1.632	1.616	1.704	1.791	1.901	1.795	1.772											
.9796	1.009	1.551	1.584	1.620	1.598	1.614	1.729	1.806	1.681	1.654											

TABLE X.- PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(c) Double slotted flap configuration with spoiler;  $h_s = -0.010\bar{c}$ ;  $\frac{V}{b/2} = 0.30$

x/c	Upper surface									Lower surface												
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 24^\circ$	$\alpha = 28^\circ$	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 24^\circ$	$\alpha = 28^\circ$				
<b>Wing</b>																						
.0000	.070	1.303	2.157	2.917	3.501	4.297	4.291	3.900	3.092	2.320	.0125	.979	.772	.742	.816	.958	1.170	1.259	1.271	1.198	1.042	
.0125	1.012	1.526	2.283	2.929	3.448	3.957	3.947	3.781	3.064	2.299	.0250	1.000	.818	.792	.702	.722	.785	.826	.031	.011	.737	
.0250	1.042	1.409	2.373	3.033	3.564	4.115	3.979	3.793	3.092	2.311	.0375	.988	.858	.742	.662	.597	.563	.551	.547	.521	.521	
.0500	1.060	1.292	2.966	3.917	3.893	3.984	3.959	3.816	3.015	2.281	.0750	1.072	1.255	1.840	4.036	5.194	5.212	5.000	4.407	3.900	3.035	2.290
.1000	1.070	1.249	2.263	2.497	4.012	4.878	4.343	3.611	3.048	2.302	.1250	1.070	1.249	1.426	2.025	2.351	2.647	2.245	2.025	1.745	1.426	
.2000	1.139	1.255	1.932	1.589	1.582	2.421	2.351	2.061	1.705	1.407	.2500	1.140	1.374	1.457	1.997	2.571	2.752	2.639	2.507	2.322	2.044	
.3000	1.181	1.280	1.354	1.386	1.415	1.794	2.326	2.542	2.512	2.172	.3500	1.193	1.274	1.366	1.670	2.052	2.274	2.373	2.142	2.050	1.745	
.4000	1.226	1.308	1.391	1.408	1.424	1.636	1.941	2.140	2.302	2.119	.4500	1.271	1.342	1.415	1.429	1.427	1.621	1.893	2.044	2.194	2.089	
.5000	1.298	1.373	1.449	1.445	1.448	1.594	1.804	1.938	2.077	2.039	.5500	1.343	1.397	1.446	1.457	1.454	1.597	1.743	1.857	2.018	2.030	
.6000	1.387	1.434	1.489	1.484	1.487	1.600	1.726	1.822	1.968	2.003	.6500	1.462	1.520	1.580	1.581	1.546	1.636	1.714	1.801	1.912	1.977	
.7034	2.030	2.040	2.031	1.949	1.949	1.764	1.725	1.809	1.803	1.842	.7534	2.259	2.246	2.228	2.236	2.188	2.303	2.280	2.218	2.271	2.195	
.8099	2.259	2.246	2.228	2.236	2.236	2.309	2.266	2.385	2.364	2.303	.8141	2.146	2.320	2.308	2.309	2.266	2.364	2.303	2.122	1.923	1.921	
.8182	2.404	2.362	2.366	2.368	2.316	2.439	2.399	2.392	2.364	2.160	.8224	2.560	2.542	2.505	2.506	2.492	2.484	2.270	2.036	1.991	1.991	
<b>Vane</b>																						
.0000	4.581	4.432	4.317	4.297	4.224	4.475	4.303	4.192	3.740	3.267	.0250	5.009	5.566	5.649	5.683	5.453	5.451	4.409	3.498	3.267	3.000	
.0500	5.009	5.566	5.649	5.683	5.453	5.795	5.795	5.451	4.409	3.498	.1000	5.744	5.742	5.732	5.732	5.732	5.732	5.732	5.732	5.732	5.732	
.1500	6.004	5.742	5.594	5.624	5.427	5.855	5.775	5.416	4.362	3.276	.2000	6.882	5.732	5.732	5.732	5.732	5.732	5.732	5.732	5.732	5.732	
.2500	5.940	5.686	5.458	5.468	5.161	5.775	5.731	5.419	4.367	3.295	.3000	5.422	4.954	4.929	4.929	4.929	4.929	4.929	4.929	4.929	4.929	
.3500	5.422	4.738	4.523	4.283	4.015	4.751	4.731	4.559	4.856	2.764	.4000	4.523	4.251	4.015	4.015	4.015	4.015	4.015	4.015	4.015	4.015	
.4500	4.219	2.332	2.303	2.287	2.743	2.804	2.452	2.478	2.349	2.055	.5000	3.593	3.379	3.160	3.027	3.718	3.757	3.644	3.216	2.480	2.000	
.5500	3.274	2.874	2.686	2.647	2.609	3.213	3.253	3.218	2.903	2.332	.6000	3.072	2.874	2.686	2.647	2.609	2.609	2.609	2.609	2.609	2.609	
.6500	2.575	2.185	2.123	2.101	2.048	2.409	2.452	2.478	2.349	2.055	.7000	2.142	1.105	1.05	1.13	1.13	1.13	1.13	1.13	1.13	1.13	
.7500	1.485	1.471	1.474	1.349	1.233	1.427	1.450	1.455	1.408	1.290	.8000	1.219	1.212	1.212	1.212	1.212	1.212	1.212	1.212	1.212	1.212	
<b>Flap</b>																						
.0000	4.437	4.317	4.224	4.297	4.224	4.475	4.303	4.192	3.740	3.267	.0125	.214	.203	.203	.156	.143	.188	.195	.201	.178	.148	
.0250	5.030	5.566	5.649	5.683	5.453	5.795	5.795	5.451	4.409	3.498	.0500	.766	.760	.620	.501	.430	.385	.347	.325	.309	.269	
.0750	.018	.006	.000	.000	.000	.000	.000	.000	.000	.000	.1000	.277	.308	.329	.205	.113	.058	.047	.032	.030	.033	
.1500	.042	.018	.010	.003	.000	.000	.000	.000	.000	.000	.2000	.142	.182	.197	.107	.027	.015	.012	.006	.000	.000	
.2000	.050	.028	.019	.009	.002	.000	.000	.000	.000	.000	.2500	.050	.028	.040	.018	.018	.018	.027	.029	.020	.021	
.3000	.039	.043	.040	.021	.003	.000	.000	.000	.000	.000	.3500	.054	.043	.040	.021	.036	.045	.061	.064	.036	.036	
.4000	.075	.049	.049	.046	.060	.088	.079	.079	.079	.079	.4500	.124	.124	.124	.124	.124	.124	.124	.124	.124	.124	
.5000	.717	.720	.683	.693	.666	.666	.666	.666	.666	.666	.5500	.258	.258	.258	.258	.258	.258	.258	.258	.258	.258	
<b>Spoiler</b>																						
.0287	.974	1.548	1.584	1.584	1.584	1.584	1.584	1.584	1.584	1.584	.5024	1.020	1.560	1.581	1.629	1.650	1.687	1.687	1.687	1.687	1.687	
.7531	1.099	1.579	1.587	1.638	1.638	1.638	1.638	1.638	1.638	1.638	.9940	1.124	1.612	1.627	1.667	1.637	1.743	1.761	1.755	1.646	1.646	

TABLE X.- PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(d) Double slotted flap configuration with spoiler;  $h_s = -0.010\bar{c}$ ;  $\frac{V}{b/2} = 0.43$

Upper surface												Lower surface																																																																																																																																																																																																																																																																																																																																																																			
x/c	$C_p$ for -											x/c	$C_p$ for -																																																																																																																																																																																																																																																																																																																																																																		
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$																																																																																																																																																																																																																																																																																																																																																							
<b>Wing</b>																																																																																																																																																																																																																																																																																																																																																																															
.0000	.862	1.856	2.163	2.640	2.907	2.886	2.709	2.623	2.479	2.178	.0125	1.123	1.821	2.150	2.619	2.883	2.823	2.699	2.488	2.202	.0250	1.139	1.771	2.200	2.660	2.889	2.937	2.877	2.717	2.482	2.190																																																																																																																																																																																																																																																																																																																																																
.0375	1.120	1.745	2.253	2.722	3.027	3.006	2.883	2.749	2.476	2.226	.0500	1.142	1.745	2.325	2.803	2.967	2.904	2.871	2.761	2.476	2.208	.0750	1.158	1.573	2.075	2.582	2.913	2.839	2.892	2.776	2.473	2.196																																																																																																																																																																																																																																																																																																																																															
.1000	1.162	1.745	2.325	2.803	2.967	2.904	2.839	2.882	2.776	2.473	.1250	1.162	1.745	2.419	2.929	3.052	3.010	2.976	2.770	2.473	2.208	.1500	1.162	1.745	2.419	2.929	3.052	3.010	2.976	2.770	2.473	2.208																																																																																																																																																																																																																																																																																																																																															
.1750	1.162	1.745	2.327	2.807	2.967	2.904	2.839	2.882	2.776	2.473	.2000	1.162	1.745	2.419	2.929	3.052	3.010	2.976	2.770	2.473	2.208	.2500	1.162	1.745	2.419	2.929	3.052	3.010	2.976	2.770	2.473	2.208																																																																																																																																																																																																																																																																																																																																															
.3000	1.162	1.745	2.327	2.807	2.967	2.904	2.839	2.882	2.776	2.473	.3500	1.162	1.745	2.419	2.929	3.052	3.010	2.976	2.770	2.473	2.208	.4000	1.162	1.745	2.419	2.929	3.052	3.010	2.976	2.770	2.473	2.208																																																																																																																																																																																																																																																																																																																																															
.4500	1.162	1.745	2.327	2.807	2.967	2.904	2.839	2.882	2.776	2.473	.5000	1.162	1.745	2.419	2.929	3.052	3.010	2.976	2.770	2.473	2.208	.5500	1.162	1.745	2.419	2.929	3.052	3.010	2.976	2.770	2.473	2.208																																																																																																																																																																																																																																																																																																																																															
.6000	1.162	1.745	2.327	2.807	2.967	2.904	2.839	2.882	2.776	2.473	.6500	1.162	1.745	2.419	2.929	3.052	3.010	2.976	2.770	2.473	2.208	.7000	1.162	1.745	2.419	2.929	3.052	3.010	2.976	2.770	2.473	2.208																																																																																																																																																																																																																																																																																																																																															
.7500	1.162	1.745	2.327	2.807	2.967	2.904	2.839	2.882	2.776	2.473	.8000	1.162	1.745	2.419	2.929	3.052	3.010	2.976	2.770	2.473	2.208	.8500	1.162	1.745	2.419	2.929	3.052	3.010	2.976	2.770	2.473	2.208																																																																																																																																																																																																																																																																																																																																															
.9000	1.162	1.745	2.327	2.807	2.967	2.904	2.839	2.882	2.776	2.473	.9500	1.162	1.745	2.419	2.929	3.052	3.010	2.976	2.770	2.473	2.208	.0000	1.571	1.635	1.653	1.596	1.586	1.872	2.016	2.071	2.048	2.021	.0500	1.802	1.906	1.941	1.883	1.760	1.842	1.928	1.991	1.973	1.979	.1000	2.237	2.330	2.372	2.291	2.102	2.069	2.075	2.050	1.985	1.985	.1500	2.365	2.683	2.719	2.701	2.414	2.292	2.237	2.156	2.021	1.997	.2000	2.647	2.774	2.813	2.763	2.501	2.379	2.285	2.177	2.036	1.988	.2500	2.798	2.893	2.941	2.624	2.631	2.466	2.348	2.215	2.039	2.000																																																																																																																																																																																																																																																																																								
<b>Vane</b>																																																																																																																																																																																																																																																																																																																																																																															
.0000	4.143	4.117	4.100	4.051	3.673	3.355	3.108	2.870	2.548	2.488	.0250	6.756	6.664	6.722	6.794	6.031	5.337	4.766	4.213	3.314	3.065	.0500	6.756	6.664	6.722	6.794	6.031	5.337	4.766	4.213	3.314	3.065	.0750	6.958	6.897	6.958	6.958	6.264	5.466	4.829	4.221	3.168	2.657	.1000	7.121	6.986	7.072	7.162	6.456	5.546	4.781	4.089	2.886	2.553	.1250	7.164	7.144	7.253	7.387	6.673	5.710	4.883	4.127	2.929	2.452	.1500	7.201	7.184	7.253	7.387	6.673	5.710	4.883	4.127	2.929	2.452	.2000	7.201	7.184	7.253	7.387	6.673	5.710	4.883	4.127	2.929	2.452	.2500	6.784	6.664	6.998	6.848	5.522	4.736	3.924	3.122	2.455	2.055	.3000	6.642	6.402	6.004	6.241	5.781	5.045	4.348	3.667	2.595	2.309	.3500	5.433	5.318	5.163	5.423	5.114	4.522	3.935	3.375	2.467	2.226	.4000	6.174	6.042	6.004	6.241	5.781	5.045	4.348	3.667	2.595	2.309	.4500	6.003	5.875	5.714	6.447	5.467	4.883	4.221	3.667	2.595	2.309	.5000	4.723	4.611	4.647	4.788	4.456	3.997	3.562	3.095	2.350	2.170	.5500	3.489	3.428	3.447	3.520	3.330	3.107	2.883	2.611	2.138	2.042	.6000	3.004	2.958	2.997	3.049	2.853	2.692	2.535	2.331	1.997	1.958																																																																																																																																																																																																
<b>Flap</b>																																																																																																																																																																																																																																																																																																																																																																															
.0000	2.284	2.365	2.538	2.537	2.300	2.134	2.012	1.797	1.578	1.553	.0250	2.284	2.365	2.538	2.537	2.300	2.134	2.012	1.797	1.578	1.553	.0500	2.284	2.365	2.538	2.537	2.300	2.134	2.012	1.797	1.578	1.553	.0750	2.284	2.365	2.538	2.537	2.300	2.134	2.012	1.797	1.578	1.553	.1000	2.284	2.365	2.538	2.537	2.300	2.134	2.012	1.797	1.578	1.553	.1250	2.284	2.365	2.538	2.537	2.300	2.134	2.012	1.797	1.578	1.553	.1500	2.284	2.365	2.538	2.537	2.300	2.134	2.012	1.797	1.578	1.553	.2000	2.284	2.365	2.538	2.537	2.300	2.134	2.012	1.797	1.578	1.553	.2500	2.284	2.365	2.538	2.537	2.300	2.134	2.012	1.797	1.578	1.553	.3000	2.284	2.365	2.538	2.537	2.300	2.134	2.012	1.797	1.578	1.553	.3500	2.284	2.365	2.538	2.537	2.300	2.134	2.012	1.797	1.578	1.553	.4000	2.284	2.365	2.538	2.537	2.300	2.134	2.012	1.797	1.578	1.553	.4500	2.284	2.365	2.538	2.537	2.300	2.134	2.012	1.797	1.578	1.553	.5000	2.284	2.365	2.538	2.537	2.300	2.134	2.012	1.797	1.578	1.553	.5500	2.284	2.365	2.538	2.537	2.300	2.134	2.012	1.797	1.578	1.553	.6000	2.284	2.365	2.538	2.537	2.300	2.134	2.012	1.797	1.578	1.553	.6500	2.284	2.365	2.538	2.537	2.300	2.134	2.012	1.797	1.578	1.553	.7000	2.284	2.365	2.538	2.537	2.300	2.134	2.012	1.797	1.578	1.553	.7500	2.284	2.365	2.538	2.537	2.300	2.134	2.012	1.797	1.578	1.553	.8000	2.284	2.365	2.538	2.537	2.300	2.134	2.012	1.797	1.578	1.553	.8500	2.284	2.365	2.538	2.537	2.300	2.134	2.012	1.797	1.578	1.553	.9000	2.284	2.365	2.538	2.537	2.300	2.134	2.012	1.797	1.578	1.553	.9500	2.284	2.365	2.538	2.537	2.300	2.134	2.012	1.797	1.578	1.553	.0000	1.571	1.635	1.653	1.596	1.586	1.872	2.016	2.071	2.048	2.021	.0500	1.802	1.906	1.941	1.883	1.760	1.842	1.928	1.991	1.973	1.979	.1000	1.855	1.959	2.001	2.040	1.912	1.994	2.080	2.147	2.129	2.119	.1500	1.898	1.992	2.034	2.073	1.944	2.025	2.112	2.179	2.161	2.159	.2000	1.941	2.043	2.085	2.124	1.995	2.076	2.163	2.230	2.212	2.200	.2500	1.984	2.086	2.128	2.167	2.038	2.119	2.196	2.263	2.244	2.232	.3000	2.026	2.128	2.170	2.209	1.980	2.059	2.137	2.204	2.185	2.173	.3500	2.068	2.170	2.212	2.251	2.024	2.093	2.171	2.238	2.219	2.207	.4000	2.110	2.212	2.254	2.293	2.075	2.144	2.223	2.290	2.271	2.259	.4500	2.152	2.254	2.296	2.335	2.124	2.193	2.272	2.339	2.318	2.306	.5000	2.194	2.296	2.338	2.37

TABLE X.- PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(e) Double slotted flap configuration with spoiler;  $h_s = -0.010\bar{c}$ ;  $\frac{y}{b/2} = 0.55$

Upper surface												Lower surface																																		
x/c	$C_p$ for -											x/c	$C_p$ for -																																	
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 15^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 25^\circ$	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 15^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 25^\circ$	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 15^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 25^\circ$																
Wing																																														
.0000	1.292	1.908	2.071	2.389	2.442	2.366	2.180	2.178	2.190	2.086	.0125	.922	.818	.923	1.193	1.355	1.446	1.490	1.551	1.601	1.650	.0150	1.340	1.917	2.117	2.481	2.406	2.253	2.230	2.210	2.080															
.0250	1.235	1.877	2.089	2.414	2.430	2.397	2.256	2.227	2.216	2.089	.0375	.940	.849	.754	.943	1.009	1.035	1.073	1.127	1.171	1.227	1.274	.0500	1.220	1.840	2.145	2.473	2.463	2.497	2.297	2.262	2.222	2.092													
.0750	1.208	1.877	2.215	2.539	2.495	2.469	2.297	2.259	2.216	2.077	.1000	.949	.843	.738	.699	.672	.667	.662	.673	.686	.707	.1500	1.229	1.794	2.274	2.592	2.534	2.445	2.317	2.265	2.210	2.077														
.1500	1.247	1.594	2.652	2.635	2.463	2.436	2.349	2.274	2.222	2.077	.2050	1.268	1.428	2.895	2.427	2.430	2.347	2.262	2.210	2.071	.2500	1.307	1.403	2.708	3.545	2.579	2.479	2.355	2.262	2.219	2.077															
.3000	1.323	1.409	2.215	3.588	2.872	2.506	2.341	2.245	2.204	2.074	.3500	1.326	1.406	2.907	2.518	2.293	2.227	2.193	2.068	.4000	1.340	1.406	2.874	2.518	2.293	2.227	2.193	2.068	.4500	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068	.5000	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068
.6000	1.343	1.409	2.215	3.588	2.872	2.506	2.341	2.245	2.204	2.074	.6500	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068	.7000	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068	.7500	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068									
.8000	1.347	1.409	2.215	3.588	2.872	2.506	2.341	2.245	2.204	2.074	.8500	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068	.9000	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068	.9500	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068									
1.0000	1.347	1.409	2.215	3.588	2.872	2.506	2.341	2.245	2.204	2.074	1.050	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068	1.1000	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068	1.1500	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068									
1.2000	1.347	1.409	2.215	3.588	2.872	2.506	2.341	2.245	2.204	2.074	1.2500	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068	1.3000	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068	1.3500	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068									
1.4000	1.347	1.409	2.215	3.588	2.872	2.506	2.341	2.245	2.204	2.074	1.4500	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068	1.5000	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068	1.5500	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068									
1.6000	1.347	1.409	2.215	3.588	2.872	2.506	2.341	2.245	2.204	2.074	1.6500	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068	1.7000	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068	1.7500	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068									
1.8000	1.347	1.409	2.215	3.588	2.872	2.506	2.341	2.245	2.204	2.074	1.8500	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068	1.9000	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068	1.9500	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068									
2.0000	1.347	1.409	2.215	3.588	2.872	2.506	2.341	2.245	2.204	2.074	2.0500	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068	2.1000	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068	2.1500	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068									
2.1000	1.347	1.409	2.215	3.588	2.872	2.506	2.341	2.245	2.204	2.074	2.1500	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068	2.2000	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068	2.2500	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068									
2.3000	1.347	1.409	2.215	3.588	2.872	2.506	2.341	2.245	2.204	2.074	2.3500	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068	2.4000	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068	2.4500	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068									
2.5000	1.347	1.409	2.215	3.588	2.872	2.506	2.341	2.245	2.204	2.074	2.5500	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068	2.6000	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068	2.6500	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068									
2.7000	1.347	1.409	2.215	3.588	2.872	2.506	2.341	2.245	2.204	2.074	2.7500	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068	2.8000	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068	2.8500	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068									
2.9000	1.347	1.409	2.215	3.588	2.872	2.506	2.341	2.245	2.204	2.074	2.9500	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068	3.0000	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068	3.0500	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068									
3.1000	1.347	1.409	2.215	3.588	2.872	2.506	2.341	2.245	2.204	2.074	3.1500	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068	3.2000	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068	3.2500	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068									
3.3000	1.347	1.409	2.215	3.588	2.872	2.506	2.341	2.245	2.204	2.074	3.3500	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068	3.4000	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068	3.4500	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068									
3.5000	1.347	1.409	2.215	3.588	2.872	2.506	2.341	2.245	2.204	2.074	3.5500	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068	3.6000	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068	3.6500	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068									
3.7000	1.347	1.409	2.215	3.588	2.872	2.506	2.341	2.245	2.204	2.074	3.7500	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068	3.8000	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068	3.8500	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068									
3.9000	1.347	1.409	2.215	3.588	2.872	2.506	2.341	2.245	2.204	2.074	3.9500	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068	4.0000	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068	4.0500	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068									
4.1000	1.347	1.409	2.215	3.588	2.872	2.506	2.341	2.245	2.204	2.074	4.1500	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068	4.2000	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068	4.2500	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068									
4.3000	1.347	1.409	2.215	3.588	2.872	2.506	2.341	2.245	2.204	2.074	4.3500	1.347	1.409	2.874	2.539	2.293	2.227	2.193	2.068																											

TABLE X.- PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Concluded

(f) Double slotted flap configuration with spoiler;  $h_s = -0.010\bar{c}$ ;  $\frac{V}{b/2} = 0.72$

x/c	Upper surface										Lower surface									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 23^\circ$	$\alpha = 24^\circ$	$\alpha = 25^\circ$	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 23^\circ$	$\alpha = 24^\circ$	$\alpha = 25^\circ$
<b>Wing</b>																				
.0000	1.811	2.237	2.434	2.383	2.166	2.042	1.929	1.934	1.889	1.893	.598	.874	1.058	1.165	1.224	1.281	1.310	1.385	1.389	1.427
.0125	1.607	2.128	2.313	2.274	2.108	2.076	1.905	1.937	1.883	1.899	.622	.788	.887	.947	.988	1.030	1.074	1.143	1.158	1.190
.0250	1.469	2.225	2.370	2.314	2.113	2.044	1.883	1.912	1.862	1.882	.628	.745	.777	.786	.807	.835	.850	.904	.918	.937
.0375	1.456	2.265	2.413	2.330	2.133	2.064	1.959	1.970	1.905	1.911	.640	.757	.755	.751	.747	.757	.767	.815	.822	.839
.0750	1.437	2.320	2.480	2.367	2.154	2.084	1.977	1.976	1.898	1.900	.651	.721	.777	.777	.776	.775	.775	.814	.821	.830
.1000	1.436	2.314	2.459	2.355	2.142	2.063	1.971	1.976	1.904	1.902	.650	.714	.766	.765	.765	.765	.765	.806	.813	.822
.1500	1.411	2.455	2.532	2.389	2.148	2.072	1.994	1.991	1.901	1.911	.655	.700	.754	.754	.754	.754	.754	.801	.807	.816
.2000	1.207	2.659	2.624	2.386	2.133	2.073	2.009	2.000	1.909	1.917	.653	.706	.732	.732	.732	.732	.732	.732	.732	.732
.2500	1.248	2.683	2.712	2.364	2.124	2.090	2.006	2.000	1.909	1.919	.654	.711	.737	.737	.737	.737	.737	.737	.737	.737
.3000	1.254	2.532	2.720	2.352	2.133	2.090	2.000	1.997	1.918	1.919	.655	.706	.731	.731	.731	.731	.731	.731	.731	.731
.3500	1.356	2.342	3.131	2.332	2.154	2.090	2.003	2.000	1.924	1.928	.655	.711	.736	.736	.736	.736	.736	.736	.736	.736
.4000	1.353	2.037	3.226	2.412	2.169	2.084	1.997	1.994	1.927	1.934	.655	.711	.736	.736	.736	.736	.736	.736	.736	.736
.4500	1.402	1.775	3.238	2.556	2.175	2.075	1.991	1.997	1.933	1.934	.655	.711	.736	.736	.736	.736	.736	.736	.736	.736
.5000	1.424	1.699	3.032	2.654	2.208	2.084	1.991	1.988	1.936	1.945	.655	.711	.736	.736	.736	.736	.736	.736	.736	.736
.5500	1.444	1.645	3.012	2.656	2.184	2.089	1.979	1.985	1.930	1.942	.655	.711	.736	.736	.736	.736	.736	.736	.736	.736
.6000	1.450	1.572	2.780	2.472	2.101	2.059	1.978	1.986	1.936	1.942	.655	.711	.736	.736	.736	.736	.736	.736	.736	.736
.6500	1.393	1.572	2.480	2.324	2.130	2.056	1.953	1.962	1.903	1.953	.655	.711	.736	.736	.736	.736	.736	.736	.736	.736
.7000	1.356	1.517	2.254	2.258	2.118	2.018	1.947	1.970	1.950	1.958	.655	.711	.736	.736	.736	.736	.736	.736	.736	.736
.7500	1.347	1.471	2.098	2.246	2.100	2.006	1.947	1.976	1.959	1.968	.655	.711	.736	.736	.736	.736	.736	.736	.736	.736
.8000	1.260	1.375	1.942	2.193	2.066	1.994	1.947	1.964	1.952	1.974	.655	.711	.736	.736	.736	.736	.736	.736	.736	.736
.8500	1.257	1.317	1.832	2.065	2.015	1.955	1.929	1.964	1.965	1.977	.655	.711	.736	.736	.736	.736	.736	.736	.736	.736
.9000	1.139	1.298	1.737	1.990	1.991	1.958	1.923	1.964	1.962	1.974	.655	.711	.736	.736	.736	.736	.736	.736	.736	.736

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TABLE XI - PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE

(a) Double slotted flap configuration with spoiler;  $h_s = -0.035\bar{c}$ ;  $\frac{V}{b/2} = 0$

x/c	Upper surface										Lower surface										
	$C_p$ for -										$C_p$ for -										
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 24^\circ$	$\alpha = 28^\circ$	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 24^\circ$	$\alpha = 28^\circ$			
Fuselage																					
.0000	.506	.018	.009	.024	.015	.030	.076	.052	.124	.149	.0500	.853	.810	.743	.695	.622	.548	.496	.469	.419	.394
.0500	.768	.846	.900	.946	.976	1.009	1.058	1.067	1.072	1.111	.1000	.922	.907	.849	.796	.720	.665	.586	.571	.523	.510
.1000	.843	.931	.967	1.000	1.030	1.058	1.102	1.116	1.124	1.172	.1500	.994	.964	.915	.868	.792	.735	.665	.641	.590	.577
.1500	.941	.991	1.030	1.054	1.071	1.087	1.116	1.140	1.160	1.180	.2000	1.018	.997	.952	.893	.839	.793	.732	.703	.655	.614
.2000	1.050	1.070	1.095	1.098	1.108	1.125	1.145	1.162	1.171	1.185	.2500	1.038	1.069	1.091	1.102	1.113	1.151	1.142	1.163	.735	.682
.3000	1.069	1.087	1.103	1.101	1.105	1.116	1.125	1.107	1.154	1.174	.3500	1.028	1.021	.997	.979	.926	.892	.834	.816	.763	.740
.3500	1.075	1.072	1.082	1.090	1.098	1.087	1.096	1.116	1.098	1.151	.4000	.969	.961	.935	.916	.872	.834	.770	.761	.717	.689
.4000	1.038	1.036	1.048	1.060	1.071	1.079	1.095	1.116	1.116	1.160	.4500	.931	.925	.897	.860	.815	.775	.711	.705	.664	.619
.4500	1.041	1.048	1.059	1.099	1.113	1.137	1.189	1.218	1.220	1.268	.5000	.897	.892	.846	.823	.759	.705	.639	.621	.575	.548
.5000	1.047	1.053	1.103	1.144	1.190	1.224	1.306	1.350	1.376	1.609	.5500	.856	.852	.798	.764	.693	.653	.577	.551	.520	.490
.5500	1.053	1.096	1.160	1.219	1.262	1.303	1.381	1.414	1.532	2.035	.6000	.809	.783	.725	.684	.625	.564	.525	.501	.465	.440
.6000	1.110	1.139	1.214	1.257	1.301	1.309	1.361	1.402	1.575	2.023	.7500	.903	.898	.864	.832	.792	.752	.714	.703	.673	.673
.6500	1.144	1.190	1.245	1.293	1.301	1.326	1.318	1.344	1.451	1.901	.9500	1.060	1.078	1.106	1.195	1.146	1.143	1.165	1.183	1.205	1.271
.7000	1.188	1.232	1.275	1.311	1.327	1.320	1.315	1.318	1.402	1.819	.9940	1.028	1.111	1.184	1.222	1.247	1.274	1.318	1.355	1.361	1.411
.7500	1.251	1.266	1.311	1.338	1.342	1.353	1.377	1.370	1.440	1.732	.8500	1.347	1.313	1.295	1.255	1.240	1.240	1.242	1.242	1.242	1.242
.8000	1.317	1.313	1.320	1.325	1.326	1.330	1.340	1.340	1.492	1.559	.8500	1.317	1.313	1.320	1.325	1.330	1.340	1.340	1.342	1.342	1.342
.8500	1.229	1.232	1.220	1.240	1.274	1.309	1.361	1.402	1.428	1.452	.9000	1.154	1.163	1.157	1.162	1.202	1.236	1.303	1.335	1.370	1.411
.9000	1.216	1.199	1.194	1.195	1.211	1.246	1.320	1.353	1.399	1.431	.9500	1.154	1.163	1.157	1.162	1.202	1.236	1.303	1.335	1.370	1.411

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TABLE XI - PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(b) Double slotted flap configuration with spoiler;  $h_S = -0.035\bar{c}$ ;  $\frac{V}{b/2} = 0.21$

x/c	Upper surface										Lower surface										
	$C_p$ for -										$C_p$ for -										
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$		$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
<b>Wing</b>																					
.0300	.807	1.902	1.778	2.259	2.255	2.575	2.658	2.092	2.889	2.542	.0125	1.016	.786	.665	.585	.579	.450	.703	.690	.661	.371
.0125	.945	1.220	1.643	2.281	2.382	10.444	11.841	9.065	3.497	2.313	.0250	.997	.810	.693	.549	.460	.389	.342	.316	.243	.204
.0250	.952	1.159	1.502	2.500	3.620	6.136	8.390	7.971	3.509	2.345	.0500	1.010	1.156	1.423	1.616	1.794	2.676	4.003	5.198	3.150	2.258
.0750	1.026	1.147	1.373	1.534	1.801	2.157	2.814	3.758	3.015	2.261	.1.026	1.147	1.373	1.534	1.801	2.157	2.814	3.758	3.015	2.261	
.1000	1.042	1.165	1.361	1.506	1.724	1.982	2.531	3.227	2.916	2.244	.1.000	1.042	1.165	1.361	1.506	1.724	1.982	2.531	3.227	2.916	2.244
.1500	1.067	1.165	1.348	1.464	1.614	1.786	2.174	2.552	2.719	2.189	.1.500	1.067	1.165	1.348	1.464	1.614	1.786	2.174	2.552	2.719	2.189
.2000	1.100	1.190	1.354	1.439	1.558	1.715	2.003	2.280	2.560	2.139	.2.000	1.100	1.190	1.354	1.439	1.558	1.715	2.003	2.280	2.560	2.139
.2500	1.125	1.192	1.342	1.421	1.504	1.654	1.894	2.077	2.341	2.105	.3.000	1.150	1.220	1.354	1.409	1.485	1.605	1.808	1.923	2.210	2.073
.3000	1.150	1.220	1.354	1.409	1.485	1.625	1.880	2.123	2.456	2.200	.4.000	1.164	1.229	1.354	1.405	1.484	1.606	1.808	1.921	2.210	2.073
.3500	1.164	1.229	1.354	1.405	1.484	1.584	1.698	2.126	2.456	2.200	.4.750	1.174	1.229	1.354	1.405	1.484	1.605	1.808	1.921	2.210	2.073
.4000	1.174	1.231	1.356	1.421	1.443	1.608	1.741	2.120	2.454	2.202	.5.000	1.185	1.231	1.356	1.421	1.443	1.608	1.741	2.120	2.454	2.202
.4500	1.185	1.231	1.356	1.421	1.443	1.608	1.741	2.120	2.454	2.202	.5.500	1.195	1.231	1.356	1.421	1.443	1.608	1.741	2.120	2.454	2.202
.5000	1.195	1.231	1.356	1.421	1.443	1.608	1.741	2.120	2.454	2.202	.6.000	1.203	1.231	1.356	1.421	1.443	1.608	1.741	2.120	2.454	2.202
.6000	1.203	1.231	1.356	1.421	1.443	1.608	1.741	2.120	2.454	2.202	.6.500	1.213	1.231	1.356	1.421	1.443	1.608	1.741	2.120	2.454	2.202
.7000	1.213	1.231	1.356	1.421	1.443	1.608	1.741	2.120	2.454	2.202	.7.000	1.207	1.231	1.356	1.421	1.443	1.608	1.741	2.120	2.454	2.202
.8125	1.207	1.231	1.356	1.421	1.443	1.608	1.741	2.120	2.454	2.202	.8.325	1.213	1.231	1.356	1.421	1.443	1.608	1.741	2.120	2.454	2.202
.9375	1.213	1.231	1.356	1.421	1.443	1.608	1.741	2.120	2.454	2.202	.9.375	1.213	1.231	1.356	1.421	1.443	1.608	1.741	2.120	2.454	2.202
.9750	1.213	1.231	1.356	1.421	1.443	1.608	1.741	2.120	2.454	2.202	.9.750	1.213	1.231	1.356	1.421	1.443	1.608	1.741	2.120	2.454	2.202
.99375	1.213	1.231	1.356	1.421	1.443	1.608	1.741	2.120	2.454	2.202	.9.9375	1.213	1.231	1.356	1.421	1.443	1.608	1.741	2.120	2.454	2.202
.99750	1.213	1.231	1.356	1.421	1.443	1.608	1.741	2.120	2.454	2.202	.9.99750	1.213	1.231	1.356	1.421	1.443	1.608	1.741	2.120	2.454	2.202
.999375	1.213	1.231	1.356	1.421	1.443	1.608	1.741	2.120	2.454	2.202	.9.999375	1.213	1.231	1.356	1.421	1.443	1.608	1.741	2.120	2.454	2.202
.999750	1.213	1.231	1.356	1.421	1.443	1.608	1.741	2.120	2.454	2.202	.9.999750	1.213	1.231	1.356	1.421	1.443	1.608	1.741	2.120	2.454	2.202
<b>Vane</b>																					
.0300	2.334	2.214	2.251	2.232	2.289	2.415	2.395	2.404	2.314	2.099	.0280	1.821	1.257	1.276	1.177	1.095	1.063	1.012	1.000	.964	.904
.0250	2.298	3.752	3.762	3.720	3.911	4.418	4.514	4.578	4.260	3.514	.0500	1.820	1.255	1.275	1.185	1.140	1.044	1.030	1.034	.935	.875
.0500	2.774	3.529	3.502	3.427	3.655	4.160	4.245	4.295	3.985	3.203	.1.000	1.511	1.462	1.379	1.241	1.128	0.89	0.655	0.74	0.600	0.572
.1000	3.665	3.434	3.411	3.290	3.486	4.038	4.183	4.266	3.889	2.928	.1.500	1.405	1.352	1.270	1.146	1.045	0.933	0.930	0.906	0.806	0.723
.1500	3.533	3.336	3.202	3.168	3.382	3.905	4.106	4.207	3.871	2.882	.2.000	1.334	1.300	1.232	1.104	1.027	0.912	0.909	0.906	0.806	0.714
.2000	3.379	3.220	3.157	3.079	3.305	3.813	4.053	4.189	3.901	2.922	.3.000	1.260	1.229	1.188	1.055	0.921	0.936	0.932	0.909	0.809	0.723
.3000	2.884	2.023	2.765	2.717	2.979	3.430	3.749	3.944	3.772	2.899	.4.000	2.138	2.08	1.41	0.952	0.933	0.942	0.938	0.947	0.938	0.52
.4000	2.566	2.480	2.439	2.691	3.023	3.375	3.602	3.521	2.766	.5.000	1.744	1.77	1.29	0.664	0.650	0.655	0.656	0.658	0.654	0.67	
.5000	2.373	2.238	2.226	2.562	2.813	2.724	3.059	3.263	3.293	2.618	.6.000	1.54	1.59	1.32	0.70	0.74	1.01	0.89	0.97	0.90	0.96
.6000	2.151	2.043	2.047	2.031	2.199	2.391	2.614	2.785	2.850	2.406	.7.000	1.177	1.187	1.154	1.128	1.145	1.166	1.159	1.168	1.162	1.154
.7000	2.115	1.988	1.981	1.979	2.133	2.320	2.525	2.681	2.671	2.299	.8.000	1.213	2.31	2.235	2.19	2.23	2.279	2.283	3.04	2.78	2.55
.8000	2.128	1.969	1.959	1.959	2.127	2.305	2.543	2.626	2.614	2.289	.9.000	1.213	1.969	1.959	2.127	2.305	2.543	2.626	2.614	2.289	2.105
.9000	2.070	1.912	1.913	1.908	2.253	2.503	2.655	2.572	2.572	2.105	.9.200	1.678	1.664	1.652	1.662	1.712	1.801	1.861	1.920	1.844	1.736
<b>Flap</b>																					
.0000	.881	1.287	1.448	1.339	1.231	1.303	1.404	1.505	1.491	1.328	.0125	.476	.303	.182	.091	.065	.101	.112	.124	.102	.084
.0125	2.212	2.094	1.951	2.038	2.219	2.451	2.575	2.416	1.995	.0250	.154	.064	.019	.021	.015	.030	.024	.015	.032	.015	
.0250	2.392	2.275	2.256	2.359	2.590	2.862	3.015	2.938	2.357	.0500	.084	.064	.028	.049	.050	.065	.053	.051	.072	.051	
.0500	2.447	2.358	2.351	2.336	2.480	2.724	2.988	3.151	3.018	2.632	.0750	.090	.086	.053	.067	.080	.083	.080	.077	.093	
.0750	2.305	2.193	2.195	2.165	2.291	2.516	2.738	2.873	2.766	2.470	.1.000	1.106	1.07	1.085	1.04	1.01	1.119	1.106	1.098	1.122	
.1000	2.109	2.006	1.994	1.976	2.080	2.249	2.453	2.556	2.503	2.287	.1.500	1.32	1.44	1.38	1.43	1.42	1.48	1.42	1.48	1.40	
.1500	1.800	1.709	1.702	1.674	1.749	1.890	2.050	2.154	2.144	1.992	.2.000	.367	.382	.364	.369	.341</td					

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TABLE XI. - PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(c) Double slotted flap configuration with spoiler;  $h_s = -0.035\bar{c}$ ;  $\frac{V}{b/2} = 0.30$

x/c	Upper surface									
	$C_p$ for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	.573	1.399	2.250	2.863	3.520	4.240	4.317	3.960	2.900	2.368
.0125	1.016	1.594	2.348	2.890	3.465	3.934	3.991	3.874	2.897	2.371
.0250	1.035	1.498	2.475	2.991	3.592	4.084	4.011	3.885	2.900	2.374
.0375	1.061	1.328	3.083	3.894	3.928	3.976	4.005	3.895	2.891	2.374
.0500	1.077	1.260	1.991	3.897	5.213	5.688	4.424	3.894	2.901	2.374
.0625	1.088	1.238	1.271	2.408	4.020	4.684	4.153	3.882	2.892	2.377
.0750	1.116	1.158	1.248	1.328	1.359	1.737	3.496	2.826	2.354	
.0875	1.120	1.150	1.330	1.348	1.610	2.455	3.104	3.121	2.746	2.325
.1000	1.153	1.260	1.249	1.372	1.468	1.986	2.621	2.634	2.290	
.1125	1.172	1.285	1.364	1.378	1.435	1.814	2.438	2.561	2.542	2.244
.1250	1.197	1.285	1.361	1.375	1.420	1.671	2.081	2.317	2.385	2.200
.1375	1.231	1.307	1.392	1.403	1.433	1.626	1.968	2.185	2.290	2.151
.1500	1.256	1.347	1.416	1.412	1.456	1.620	1.906	2.094	2.210	2.119
.1625	1.256	1.367	1.441	1.427	1.468	1.608	1.819	1.956	2.074	2.058
.1750	1.344	1.387	1.538	1.436	1.477	1.593	1.755	1.895	2.008	2.026
.1875	1.389	1.446	1.481	1.470	1.508	1.599	1.747	1.859	1.923	1.963
.2000	1.471	1.511	1.555	1.534	1.556	1.670	1.723	1.807	1.871	1.915
.2125	1.496	1.537	1.559	1.551	1.571	1.772	1.799	1.889	1.863	1.905
.2250	2.248	2.248	2.013	1.997	2.009	2.048	2.078	2.091	1.935	1.937
.2375	2.249	2.214	2.197	2.189	2.195	2.248	2.256	2.258	2.015	1.963
.2500	2.344	2.297	2.277	2.265	2.270	2.329	2.341	2.323	2.057	1.977
.2625	2.399	2.357	2.336	2.314	2.321	2.395	2.384	2.376	2.071	1.997
.2750	2.338	2.505	2.469	2.457	2.459	2.512	2.521	2.511	2.160	2.029
Vane										
.0000	4.641	4.390	4.296	4.241	4.237	4.383	4.205	4.210	3.628	3.351
.0125	5.958	5.616	5.567	5.525	5.490	5.739	5.734	5.438	4.190	3.461
.0250	6.010	5.861	5.738	5.510	5.402	5.689	5.606	5.391	4.137	3.453
.0375	6.030	5.958	5.787	5.658	5.559	5.687	5.553	5.398	4.100	3.473
.0500	6.056	5.956	5.836	5.711	5.658	5.772	5.639	5.432	3.935	3.050
.0625	5.956	5.836	5.707	5.585	5.219	5.590	5.632	5.312	3.935	3.050
.0750	5.870	5.489	5.330	5.305	5.153	5.545	5.629	5.367	4.015	3.105
.0875	5.341	5.019	4.827	4.793	4.656	5.135	5.230	5.062	3.900	3.038
.1000	4.659	4.301	4.175	4.125	4.003	4.533	4.632	4.514	3.554	2.841
.1125	4.000	3.009	2.988	2.945	2.662	3.623	3.602	3.024	2.554	
.1250	3.405	2.780	2.620	2.570	2.577	3.078	3.179	2.767	2.397	
.1375	2.967	2.368	2.296	2.229	2.268	2.638	2.711	2.751	2.486	2.226
.1500	2.255	2.130	2.089	2.046	2.018	2.317	2.384	2.449	2.267	2.098
Flap										
.0000	1.452	1.449	1.478	1.351	1.249	1.386	1.452	1.496	1.332	1.293
.0125	2.083	2.081	2.046	1.979	1.926	2.105	2.201	2.270	2.090	1.992
.0250	2.398	2.517	2.435	2.384	2.342	2.635	2.728	2.830	2.619	2.516
.0375	2.494	2.505	2.416	2.363	2.384	2.698	2.825	2.980	2.793	2.655
.0500	2.427	2.427	2.395	2.284	2.336	2.674	2.810	2.971	2.805	2.644
.0625	2.344	2.279	2.194	2.143	2.225	2.542	2.670	2.813	2.665	2.499
.0750	1.879	1.895	1.870	1.863	1.967	2.204	2.297	2.440	2.332	2.244
.0875	1.729	1.663	1.648	1.637	1.716	1.886	1.968	2.036	1.977	1.919
.1000	1.494	1.548	1.595	1.595	1.696	1.711	1.766	1.887	1.728	
.1125	1.663	1.681	1.673	1.683	1.700	1.754	1.732	1.783	1.767	1.815
.1250	1.704	1.721	1.713	1.720	1.742	1.763	1.737	1.798	1.778	1.823
Spoiler										
.2475	1.411	1.533	1.517	1.554	1.574	1.627	1.664	1.694	1.659	1.729
.4009	1.477	1.572	1.550	1.593	1.613	1.679	1.726	1.755	1.728	1.790
.7396	1.445	1.539	1.517	1.560	1.571	1.632	1.673	1.711	1.688	1.755
.9794	1.586	1.732	1.692	1.695	1.711	1.764	1.813	1.860	1.844	1.924

x/c	Lower surface									
	$C_p$ for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	1.000	.774	.741	.814	.970	1.168	1.286	1.308	1.178	1.070
.0250	1.022	.817	.719	.701	.733	.707	.824	.852	.802	.745
.0375	1.010	.858	.738	.757	.821	.849	.874	.847	.819	
.0500	1.013	.879	.762	.774	.807	.841	.871	.849	.823	
.0625	1.013	.870	.755	.764	.806	.846	.883	.857	.833	
.0750	1.013	.929	.846	.753	.763	.587	.496	.475	.450	.435
.0875	1.013	.920	.846	.762	.682	.596	.525	.499	.457	.452
.1000	1.026	.929	.846	.762	.682	.596	.525	.499	.457	.452
.1125	1.013	.929	.846	.777	.703	.623	.548	.516	.473	.470
.1250	1.020	.929	.846	.777	.703	.623	.551	.516	.473	.470
.1375	1.020	.929	.846	.777	.703	.623	.551	.516	.473	.470
.1500	1.020	.929	.846	.777	.703	.623	.551	.516	.473	.470
.1625	1.020	.929	.846	.777	.703	.623	.551	.516	.473	.470
.1750	1.020	.929	.846	.777	.703	.623	.551	.516	.473	.470
.1875	1.020	.929	.846	.777	.703	.623	.551	.516	.473	.470
.2000	1.020	.929	.846	.777	.703	.623	.551	.516	.473	.470
.2125	1.020	.929	.846	.777	.703	.623	.551	.516	.473	.470
.2250	1.020	.929	.846	.777	.703	.623	.551	.516	.473	.470
.2375	1.020	.929	.846	.777	.703	.623	.551	.516	.473	.470
.2500	1.020	.929	.846	.777	.703	.623	.551	.516	.473	.470
.2625	1.020	.929	.846	.777	.703	.623	.551	.516	.473	.470
.2750	1.020	.929	.846	.777	.703	.623	.551	.516	.473	.470
.2875	1.020	.929	.846	.777	.703	.623	.551	.516	.473	.470
.3000	1.020	.929	.846	.777	.703	.623	.551	.516	.473	.470
.3125	1.020	.929	.846	.777	.703	.623	.551	.516	.473	.470
.3250	1.020	.929	.846	.777	.703	.623	.551	.516	.473	.470
.3375	1.020	.929	.846	.777	.703	.623	.551	.516	.473	.470
.3500	1.020	.929	.846	.777	.703	.623	.551	.516	.473	.470
.3625	1.020	.929	.846	.777	.703	.623	.551	.516	.473	.470
.3750	1.020	.929	.846	.777	.703	.623	.551	.516	.473	.470
.3875	1.020	.929	.846	.777	.703	.623	.551	.516	.473	.470
.4000	1.020	.929	.846	.777	.703	.623	.551	.516	.473	.470
.4125	1.020	.929	.846	.777	.703	.623	.551			

TABLE XI.- PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(d) Double slotted flap configuration with spoiler;  $h_s = -0.035\bar{c}$ ;  $\frac{V}{b/2} = 0.43$

Upper surface											Lower surface										
x/c	$C_p$ for -										x/c	$C_p$ for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 23^\circ$	$\alpha = 24^\circ$	$\alpha = 25^\circ$		$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 23^\circ$	$\alpha = 24^\circ$	$\alpha = 25^\circ$
<b>Wing</b>																					
.0000	1.871	1.768	2.091	2.506	2.896	2.881	2.685	2.658	2.510	2.145	.0125	1.010	.774	.799	.936	1.130	1.258	1.316	1.378	1.386	1.249
.0125	1.106	1.706	2.057	2.482	2.857	2.970	2.805	2.744	2.542	2.177	.0250	1.022	.813	.762	.774	.834	.875	.906	.941	.961	.890
.0250	1.154	1.660	2.110	2.537	2.859	2.928	2.862	2.752	2.536	2.148	.0375	1.029	.844	.765	.777	.807	.852	.884	.953	.974	.878
.0500	1.087	1.560	2.129	2.616	2.982	3.009	2.856	2.805	2.548	2.195	.0750	1.122	1.544	2.405	2.689	2.946	2.881	2.811	2.536	2.148	2.148
.1000	1.164	1.446	2.684	2.899	2.836	2.867	2.623	2.536	2.163	2.933	.1250	1.035	.875	.704	.730	.750	.800	.834	.852	.874	.802
.1500	1.161	1.269	2.182	3.641	3.563	3.074	2.968	2.835	2.548	2.193	.2000	1.209	1.517	2.869	3.341	2.968	2.782	2.527	2.154	2.250	2.250
.2500	1.226	1.284	1.257	1.927	3.095	3.184	2.838	2.667	2.497	2.151	.3000	1.267	1.327	1.307	1.534	2.492	2.970	2.730	2.608	2.464	2.125
.4000	1.249	1.357	1.356	1.534	2.492	2.970	2.730	2.667	2.497	2.151	.5000	1.303	1.337	1.356	2.015	2.644	2.475	2.389	2.110	2.079	2.079
.6000	1.386	1.348	1.369	1.535	2.478	2.918	2.428	2.404	2.344	2.105	.7000	1.348	1.353	1.363	2.022	2.622	2.435	2.278	2.079	2.079	2.079
.8000	1.302	1.416	1.444	1.393	1.964	2.038	2.159	2.077	2.024	2.146	.8500	1.472	1.505	1.539	1.570	1.955	2.083	2.151	2.147	2.044	2.044
.9000	1.569	1.572	1.610	1.537	1.984	1.890	2.003	2.068	2.090	2.013	.9500	1.611	1.777	1.872	1.926	1.739	1.840	1.932	1.991	2.015	1.986
.7000	2.032	2.185	2.273	2.220	2.080	2.050	2.059	2.062	2.039	1.980	.7700	2.277	2.480	2.599	2.555	2.388	2.255	2.224	2.186	2.084	2.003
.7750	2.350	2.556	2.684	2.659	2.477	2.341	2.272	2.224	2.193	2.000	.7800	2.472	2.682	2.812	2.778	2.587	2.418	2.339	2.277	2.114	2.009
<b>Vane</b>																					
.0000	4.032	3.899	3.976	3.646	3.605	3.702	3.086	2.932	2.638	2.476	.0250	4.034	4.003	4.069	.970	.852	.736	.645	.605	.590	.542
.0500	6.411	6.186	6.392	6.351	5.943	5.210	4.770	4.301	3.470	3.018	.0750	4.469	4.74	5.64	.476	.401	.306	.245	.224	.220	.194
.1000	6.642	6.336	6.549	6.513	6.133	5.332	4.847	4.398	3.338	2.789	.1000	1.199	.223	.310	.268	.187	.113	.096	.082	.078	.055
.1500	6.051	6.480	6.706	6.702	6.320	5.400	4.809	4.278	3.066	2.484	.2000	5.090	.086	.179	.129	.086	.033	.047	.032	.054	.009
.2000	7.028	6.636	6.881	6.672	6.524	5.548	4.909	4.328	3.015	2.380	.2500	.042	.661	.103	.088	.039	.050	.027	.026	.009	.009
.2500	7.218	6.853	7.132	7.165	6.818	5.839	5.186	4.355	3.156	2.502	.3000	.026	.028	.056	.064	.000	.036	.035	.018	.042	.038
.3000	6.543	6.229	6.446	6.438	6.240	5.355	4.752	4.174	2.919	2.322	.4000	.058	.055	.053	.043	.027	.036	.047	.035	.066	.052
.4000	5.790	5.520	5.696	5.763	5.599	4.875	4.354	3.853	2.751	2.247	.5000	.064	.070	.041	.073	.077	.071	.068	.074	.081	.061
.6000	5.188	4.853	5.000	5.061	4.961	4.373	3.935	3.519	2.605	2.168	.7000	.167	.144	.138	.149	.159	.149	.148	.130	.129	.116
.8000	3.247	3.092	3.163	3.220	3.104	2.991	2.464	2.664	2.231	2.015	.9000	.315	.308	.266	.308	.317	.288	.283	.251	.249	.226
.9000	2.800	2.676	2.765	2.787	2.747	2.599	2.496	2.389	2.069	1.931	.9200	1.003	.945	.950	.976	.961	.902	.844	.805	.746	.710
<b>Flap</b>																					
.0000	1.910	2.095	2.317	2.360	2.228	2.059	1.959	1.856	1.644	1.542	.0125	.215	.229	.205	.311	.285	.258	.242	.215	.198	.165
.0125	2.347	2.553	2.718	2.755	2.673	2.504	2.404	2.286	2.069	1.927	.0250	.035	.021	.019	.040	.036	.035	.030	.027	.024	.055
.0250	2.855	3.104	3.270	3.323	3.261	3.118	3.006	2.862	2.614	2.476	.0375	.039	.012	.016	.021	.018	.009	.009	.000	.024	.038
.0500	2.852	3.064	3.201	3.278	3.258	3.175	3.098	2.971	2.692	2.508	.0750	.061	.052	.022	.019	.045	.021	.020	.015	.045	.038
.0750	2.694	2.936	3.057	3.144	3.139	3.098	3.062	2.953	2.656	2.455	.1000	.051	.061	.028	.018	.047	.045	.035	.015	.039	.046
.1000	2.723	2.819	2.890	2.970	2.964	2.943	2.921	2.832	2.554	2.351	.1500	.051	.083	.056	.076	.039	.059	.041	.058	.057	.046
.1500	2.070	2.113	2.151	2.201	2.255	2.380	2.431	2.404	2.231	2.073	.2000	.084	.095	.066	.076	.089	.095	.077	.089	.081	.067
.2000	1.910	1.801	1.845	1.884	1.920	1.955	2.000	1.997	1.910	1.829	.4000	.225	.217	.191	.204	.187	.181	.185	.136	.159	.157
.4000	1.151	1.211	1.251	1.293	1.335	1.427	1.505	1.578	1.710	1.751	.6000	.350	.358	.329	.329	.312	.285	.277	.268	.272	.258
.6000	1.630	1.703	1.721	1.710	1.697	1.739	1.738	1.758	1.802	1.815	.8000	.617	.566	.536	.531	.510	.481	.448	.445	.473	.461
.8000	1.492	1.685	1.746	1.726	1.709	1.739	1.752	1.770	1.817	1.815	.9000	.752	.755	.737	.732	.703	.682	.651	.628	.677	.664
<b>Spoiler</b>																					
.2443	1.125	1.295	1.296	1.353	1.423	1.481	1.557	1.612	1.694	1.834	.2507	1.479	1.678	1.652	1.647	1.655	1.676	1.708	1.720	1.751	1.816
.4905	1.210	1.352	1.369	1.428	1.491	1.548	1.629	1.667	1.708	1.759	.5039	1.613	1.705	1.668	1.677	1.679	1.697	1.717	1.749	1.757	1.831
.7370	1.292	1.407	1.393	1.458	1.512	1.548	1.612	1.644	1.682	1.778	.9854	1.618	1.789	1.749	1.760	1.747	1.804	1.778	1.747	1.755	1.836
.7508	1.462	1.602	1.741	1.701	1.695	1.696	1.691	1.729	1.759	1.813	.9940	1.834	1.873	1.804	1.778	1.747	1.758	1.813	1.839	1.858	1.898

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TABLE XI - PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(e) Double slotted flap configuration with spoiler;  $h_s = -0.035c$ ;  $\frac{V}{b/2} = 0.55$

Upper surface											Lower surface										
x/c	$C_p$ for -										x/c	$C_p$ for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$		$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
<b>Wing</b>																					
.0000	1.252	1.895	2.095	2.357	2.484	2.356	2.188	2.214	2.181	2.087	.0125	.920	.624	.944	1.159	1.345	1.455	1.510	1.546	1.592	1.580
.0250	1.328	1.904	2.165	2.412	2.524	2.386	2.259	2.267	2.196	2.093	.0250	.648	.624	.621	.649	.931	1.009	1.038	1.079	1.101	1.107
.0500	1.229	1.701	2.129	2.378	2.471	2.396	2.247	2.270	2.193	2.103	.0500	.981	.842	.772	.744	.763	.749	.770	.798	.808	.803
.0750	1.210	1.932	2.265	2.503	2.532	2.452	2.309	2.311	2.190	2.093	.0750	.956	.830	.759	.698	.670	.674	.659	.660	.681	.661
.1000	1.220	1.867	2.327	2.568	2.428	2.338	2.303	2.190	2.190	2.081	.1000	.952	.842	.747	.695	.649	.629	.612	.613	.621	.604
.1250	1.247	1.678	2.722	2.604	2.495	2.416	2.387	2.303	2.190	2.084	.1250	.943	.842	.759	.680	.634	.593	.560	.563	.556	.557
.1500	1.242	1.849	2.971	2.624	2.442	2.416	2.358	2.285	2.190	2.081	.1500	.982	.817	.747	.671	.625	.572	.528	.525	.515	.510
.1750	1.203	1.999	2.430	3.468	2.913	2.471	2.349	2.250	2.190	2.084	.1750	.781	.713	.640	.604	.563	.516	.499	.500	.481	
.2000	1.328	1.409	2.308	3.462	2.913	2.471	2.349	2.250	2.190	2.077	.2000	.731	.628	.571	.542	.501	.484	.455	.453	.442	
.2250	1.414	1.880	2.463	2.427	2.790	2.428	2.356	2.220	2.181	2.050	.2250	.720	.650	.614	.585	.547	.513	.499	.481		
.2500	1.452	1.517	1.379	1.964	2.593	2.359	2.210	2.185	2.119	2.047	.2500	.656	.638	.596	.549	.520	.484	.437	.440	.435	
.2750	1.342	1.444	2.710	2.429	2.284	2.151	2.159	2.089	2.029	.2750	.720	.650	.614	.585	.547	.513	.499	.481			
.3000	1.643	1.808	1.725	1.622	1.982	2.069	2.003	2.059	2.012	2.000	.3000	.781	.713	.640	.604	.563	.516	.499	.500	.481	
.3250	2.003	2.248	2.194	2.012	2.114	2.079	2.035	2.085	2.006	1.995	.3250	.731	.650	.614	.585	.547	.513	.499	.481		
.3500	2.389	2.537	2.518	2.478	2.420	2.138	2.067	2.094	2.015	2.009	.3500	.720	.650	.614	.585	.547	.513	.499	.481		
.3750	2.411	2.594	2.543	2.336	2.252	2.138	2.064	2.112	2.015	2.006	.3750	.720	.650	.614	.585	.547	.513	.499	.481		
.4000	2.458	2.740	2.728	2.518	2.351	2.210	2.094	2.135	2.030	2.009	.4000	.720	.650	.614	.585	.547	.513	.499	.481		
.4270	2.508	2.642	2.682	2.465	2.219	2.110	2.144	2.034	2.015	.4270	.720	.650	.614	.585	.547	.513	.499	.481			
<b>Vane</b>																					
.0000	3.322	3.282	3.154	2.793	2.471	2.284	2.198	2.182	2.065	2.035	.0250	1.319	1.180	1.182	1.018	.893	.799	.723	.701	.675	.652
.0250	6.185	6.211	6.237	5.604	4.544	3.883	3.594	3.512	3.083	2.960	.0500	.659	.593	.605	.503	.420	.365	.306	.296	.266	.261
.0500	6.338	6.303	6.400	5.753	4.601	3.841	3.554	3.420	2.929	2.772	.0750	.325	.235	.318	.268	.207	.150	.125	.111	.109	.099
.0750	6.406	6.474	6.870	6.226	4.781	3.781	3.399	3.250	2.693	2.549	.1000	.188	.111	.201	.159	.129	.087	.099	.076	.065	.057
.1000	6.474	6.874	7.020	6.446	5.071	4.446	3.845	3.424	2.914	2.546	.1250	.099	.077	.142	.116	.084	.069	.064	.076	.055	
.1250	6.707	6.700	6.968	5.379	4.733	3.889	3.426	3.026	2.612	2.412	.1500	.041	.084	.079	.059	.061	.076	.059	.064	.064	
.1500	6.744	6.362	6.623	6.189	4.592	3.542	3.154	2.986	2.735	2.508	.1750	.057	.087	.080	.050	.064	.084	.064	.076	.076	
.1750	5.679	5.613	5.870	5.567	4.147	3.213	2.886	2.760	2.558	2.330	.2000	.089	.144	.130	.128	.138	.123	.137	.141	.121	.125
.2000	5.128	5.059	5.333	5.104	3.850	2.754	2.625	2.199	2.134	.2250	.090	.144	.130	.128	.139	.135	.133	.140	.130	.135	
.2250	4.408	4.353	4.573	4.421	3.393	2.754	2.545	2.447	2.130	2.081	.2500	.166	.204	.207	.189	.180	.183	.149	.173	.157	.160
.2500	3.781	3.752	3.922	3.845	3.054	2.542	2.373	2.329	2.065	2.032	.2750	.104	.350	.358	.351	.315	.296	.268	.276	.263	.255
.2800	3.287	3.257	3.388	3.339	2.772	2.368	2.236	2.111	2.009	1.997	.3000	1.032	1.055	1.086	1.049	.916	.832	.802	.793	.781	.791
<b>Flap</b>																					
.0125	.255	.260	.346	.332	.291	.249	.251	.243	.234	.229	.0250	.048	.074	.079	.069	.069	.073	.079	.055	.058	
.0250	.064	.056	.065	.058	.075	.072	.055	.067	.061	.064	.0500	.084	.088	.079	.069	.063	.061	.076	.059	.064	
.0500	.264	.264	.264	.264	.264	.264	.264	.264	.264	.264	.0750	.099	.077	.142	.116	.084	.064	.076	.055	.058	
.0750	.076	.076	.076	.076	.076	.076	.076	.076	.076	.076	.1000	.076	.076	.076	.076	.076	.076	.076	.076	.076	
.1000	.073	.121	.073	.088	.093	.117	.108	.103	.103	.103	.1250	.073	.121	.088	.135	.135	.122	.114	.112	.110	
.1250	.102	.142	.142	.142	.142	.142	.142	.142	.142	.142	.1500	.102	.142	.142	.142	.142	.142	.142	.142	.142	
.1500	.264	.268	.250	.231	.231	.231	.231	.231	.231	.231	.1750	.264	.268	.250	.231	.231	.231	.231	.231	.231	
.1750	.430	.421	.407	.363	.351	.336	.324	.314	.314	.314	.2000	.430	.421	.407	.363	.351	.336	.324	.314	.308	
.2000	.618	.638	.602	.561	.544	.521	.498	.487	.487	.487	.2250	.618	.638	.602	.561	.544	.521	.498	.487	.481	
.2250	.806	.830	.796	.759	.745	.716	.688	.668	.668	.668	.2500	.806	.830	.796	.759	.745	.716	.688	.668	.673	
<b>Spoiler</b>																					
.2584	1.655	1.807	1.770	1.737	1.699	1.679	1.672	1.711	1.720	1.807	.5039	1.643	1.828	1.791	1.748	1.705	1.685	1.677	1.729	1.740	1.819
.2750	1.655	1.807	1.770	1.737	1.699	1.679	1.672	1.711	1.720	1.807	.5708	1.668	1.867	1.828	1.760	1.702	1.697	1.677	1.732	1.745	1.822
.3000	1.689	1.864	1.790	1.757	1.751	1.711	1.772	1.792	1.798	1.864	.5940	1.850	2.090	2.094	1.934	1.777	1.743	1.764	1.793	1.792	1.866

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TABLE XI - PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Concluded

(f) Double slotted flap configuration with spoiler;  $h_S = -0.035\bar{c}$ ;  $\frac{V}{b/2} = 0.72$

$x/c$	Upper surface										Lower surface										
	$C_p$ for -										$C_p$ for -										
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 25^\circ$		$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 25^\circ$
<b>Wing</b>																					
.0125	2.141	2.196	2.378	2.246	2.122	1.982	1.930	1.930	1.879	1.953	.0125	.749	.856	1.021	1.114	1.193	1.248	1.318	1.364	1.387	1.478
.0250	2.063	2.184	2.305	2.192	2.077	1.953	1.936	1.947	1.893	1.953	.0250	.719	.816	.925	.970	1.035	1.084	1.134	1.144	1.230	
.0500	2.057	2.199	2.323	2.204	2.083	1.968	1.947	1.959	1.899	1.945	.0500	.760	.850	.951	.981	1.050	1.104	1.154	1.164	1.249	.977
.0750	2.025	2.247	2.365	2.216	2.092	2.011	1.971	1.971	1.902	1.968	.0750	.784	.852	.940	.980	1.050	1.104	1.154	1.164	1.249	.977
.1000	1.979	2.349	2.414	2.272	2.092	2.000	1.973	1.976	1.922	1.956	.1000	.796	.859	.928	.970	.976	1.025	1.075	1.125	1.135	1.220
.1500	1.959	2.518	2.483	2.263	2.107	2.011	2.000	2.003	1.910	1.968	.1500	.818	.873	.946	.979	.976	1.025	1.075	1.125	1.135	1.220
.2000	1.939	2.663	2.565	2.248	2.101	2.043	2.017	2.006	1.913	1.971	.2000	.830	.877	.948	.981	.976	1.025	1.075	1.125	1.135	1.220
.2500	1.946	2.626	2.668	2.231	2.092	2.046	2.017	2.006	1.913	1.976	.2500	.850	.887	.953	.986	.976	1.025	1.075	1.125	1.135	1.220
.3000	1.946	2.424	2.855	2.734	2.107	2.038	2.011	1.994	1.922	1.979	.3000	.878	.840	.807	.772	.726	.708	.705	.694	.717	
.3500	1.911	2.154	3.051	2.216	2.125	2.035	2.006	1.994	1.910	1.982	.3500	.941	.925	.906	.890	.798	.778	.770	.764	.760	.775
.4000	1.891	1.988	3.145	2.427	2.125	2.032	1.974	1.988	1.928	1.991	.4000	1.063	1.051	1.051	1.051	.976	.914	.880	.866	.872	.872
.4500	1.899	1.687	3.147	2.427	2.125	2.032	1.974	1.988	1.928	1.991	.4500	1.248	1.247	1.245	1.135	1.042	.997	.991	.994	.959	.974
.5000	1.862	1.626	3.130	2.343	2.137	2.017	1.974	1.991	1.944	2.000	.5000	1.464	1.452	1.462	1.305	1.199	1.154	1.125	1.137	1.107	1.108
.5500	1.870	1.576	2.930	2.389	2.119	2.004	1.973	1.979	1.936	2.003	.5500	1.636	1.660	1.659	1.515	1.381	1.332	1.315	1.332	1.298	1.300
.6000	1.893	1.572	2.680	2.132	2.101	1.988	1.965	1.968	1.935	2.008	.6000	1.843	1.837	1.728	1.592	1.527	1.527	1.548	1.520	1.539	
.6500	1.865	1.536	2.405	2.207	2.074	1.971	1.962	1.973	1.954	2.029	.6500	1.975	1.987	1.987	1.820	1.720	1.688	1.720	1.764	1.728	1.729
.7000	1.862	1.467	2.184	2.150	2.053	1.962	1.947	1.968	1.954	2.032	.7000	2.020	2.016	1.965	1.780	1.781	1.624	1.857	1.809	1.784	
.7500	1.850	1.413	2.000	2.108	2.027	1.947	1.936	1.959	1.936	2.038	.7500	1.988	1.994	1.988	1.835	1.782	1.746	1.684	1.842	1.798	1.801
.8000	1.842	1.337	1.885	2.066	2.006	1.941	1.938	1.968	1.954	2.043	.8000	1.659	1.876	1.831	1.823	1.756	1.737	1.740	1.754	1.773	
.8500	1.863	1.292	1.758	1.964	1.961	1.912	1.918	1.956	1.954	2.041	.8500	1.724	1.762	1.767	1.790	1.726	1.702	1.711	1.734	1.720	1.731
.9000	1.839	1.277	1.683	1.916	1.925	1.901	1.921	1.953	1.962	2.038	.9000	1.608	1.648	1.701	1.760	1.708	1.695	1.699	1.723	1.711	1.775
.9500	1.473	1.521	1.625	1.731	1.717	1.702	1.708	1.740	1.734	1.781	.9500										

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TABLE XII. - PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE

(a) Double slotted flap configuration with spoiler;  $h_s = -0.050\bar{c}$ ;  $\frac{y}{b/2} = 0$

		Upper surface										Lower surface									
		$C_p$ for -										$C_p$ for -									
$x/c$	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$	$x/c$	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
<b>Fuselage</b>																					
.0000	.000	.006	.000	.000	.015	.018	.041	.076	.116	.152	.0500	.784	.810	.752	.708	.640	.587	.474	.455	.419	.393
.1000	.703	.448	.890	.948	.986	1.000	1.029	1.042	1.078	1.092	.1100	.944	.579	.654	.796	.739	.641	.579	.546	.523	.496
.2000	.780	.910	.956	1.004	1.039	1.050	1.076	1.115	1.130	1.144	.1150	.941	.752	.674	.885	.805	.729	.641	.622	.595	.564
.3000	.807	.879	1.021	1.064	1.084	1.092	1.099	1.126	1.139	1.155	.1200	.944	.767	.643	.911	.825	.722	.659	.659	.616	
.4000	.941	1.009	1.055	1.097	1.099	1.103	1.099	1.126	1.127	1.135	.1250	.997	1.015	.991	.941	.910	.826	.742	.754	.754	.716
.5000	.985	1.051	1.092	1.104	1.114	1.112	1.117	1.132	1.130	1.155	.1300	1.022	1.027	1.018	.991	.937	.870	.813	.774	.754	
.6000	1.040	1.072	1.089	1.113	1.114	1.097	1.099	1.112	1.118	1.132	.1350	1.043	1.075	1.086	1.097	1.102	1.082	.827	.801	.769	.734
.7000	1.043	1.075	1.086	1.097	1.102	1.092	1.088	1.100	1.107	1.137	.1400	1.015	1.018	1.000	.985	.949	.820	.760	.733	.723	.645
.8000	1.015	1.039	1.055	1.070	1.075	1.062	1.088	1.109	1.116	1.146	.1450	1.032	1.055	1.070	1.093	1.021	1.004	.876	.872	.767	.672
.9000	1.023	1.045	1.047	1.097	1.120	1.115	1.158	1.203	1.225	1.281	.1500	1.057	1.107	1.184	1.198	1.230	1.287	1.326	1.381	1.590	
.9500	1.021	1.057	1.107	1.184	1.198	1.230	1.287	1.326	1.381	1.590	.1550	1.058	1.095	1.171	1.231	1.270	1.298	1.357	1.399	1.514	1.960
.6900	1.040	1.072	1.089	1.113	1.114	1.097	1.099	1.112	1.118	1.132	.1600	1.142	1.214	1.274	1.333	1.381	1.420	1.470	1.520	1.580	
.7500	1.130	1.181	1.190	1.209	1.204	1.204	1.225	1.299	1.329	1.351	.1650	1.180	1.209	1.230	1.294	1.350	1.405	1.467	1.520	1.580	
.8000	1.180	1.229	1.230	1.231	1.232	1.232	1.236	1.301	1.305	1.367	.1700	1.227	1.315	1.342	1.348	1.339	1.343	1.410	1.462		
.7500	1.240	1.277	1.281	1.310	1.327	1.316	1.316	1.351	1.367	1.474	.1750	1.240	1.315	1.342	1.348	1.339	1.343	1.410	1.462		
.8000	1.310	1.328	1.346	1.356	1.366	1.384	1.398	1.420	1.459	1.573	.1800	1.310	1.328	1.346	1.356	1.366	1.384	1.410	1.462		
.8500	1.285	1.304	1.321	1.344	1.343	1.372	1.430	1.478	1.504	1.513	.1850	1.285	1.304	1.321	1.344	1.343	1.372	1.430	1.478		
.9000	1.201	1.227	1.232	1.239	1.273	1.295	1.342	1.399	1.454	1.493	.1900	1.201	1.227	1.232	1.239	1.273	1.295	1.342	1.399	1.454	
.9500	1.130	1.157	1.165	1.186	1.210	1.233	1.263	1.335	1.384	1.481	.1950	1.170	1.193	1.187	1.198	1.216	1.242	1.288	1.343	1.398	

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TABLE XII.- PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE  
ANGLE-OF-ATTACK RANGE - Continued

(b) Double slotted flap configuration with spoiler;  $h_s = -0.050\bar{c}$ ;  $\frac{V}{b/2} = 0.21$

x/c	Upper surface										Lower surface										
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 15^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$	
<b>Wing</b>																					
.0000	.942	.926	.903	.8412	.8273	.74553	.9395	.8449	.8911	.2483	.0125	1.021	.809	.638	.582	.574	.625	.702	.673	.459	.379
.0125	.954	1.241	1.762	3.452	7.4395	10.377	11.553	6.228	3.506	2.415	.0250	.977	.830	.859	.595	.440	.390	.333	.310	.249	.206
.0250	.984	1.188	1.855	2.453	3.637	6.411	8.347	7.422	3.515	2.450	.0375	.984	.857	.697	.579	.440	.338	.249	.221	.222	.209
.0500	1.005	1.170	1.458	1.619	1.962	2.658	4.041	5.177	3.089	2.344	.0750	.978	.873	.715	.604	.464	.349	.267	.260	.275	.279
.0750	1.018	1.163	1.398	1.548	1.774	2.139	2.804	3.629	2.967	2.344	.1000	.978	.867	.740	.631	.497	.411	.342	.327	.337	.335
.1000	1.042	1.170	1.525	1.705	1.952	2.928	3.277	3.638	2.820	2.320	.1500	.960	.886	.792	.652	.551	.469	.426	.413	.417	.412
.1500	1.048	1.174	1.544	1.754	1.950	2.115	2.818	3.278	2.878	2.345	.2000	.969	.886	.786	.659	.592	.538	.473	.472	.450	.450
.2000	.988	1.377	1.337	1.449	1.631	1.813	2.072	2.883	2.471	2.345	.2500	.973	.901	.808	.732	.643	.580	.519	.507	.497	.482
.2500	1.105	1.213	1.424	1.452	1.649	1.854	2.074	2.813	2.437	2.347	.3000	.972	.914	.817	.744	.658	.587	.544	.524	.506	.506
.3000	1.124	1.228	1.431	1.469	1.401	1.783	1.929	2.166	2.091	2.345	.3500	.981	.929	.811	.793	.666	.586	.546	.553	.535	.535
.3500	1.139	1.241	1.437	1.457	1.426	1.565	1.647	1.808	2.033	2.056	.4000	.981	.932	.841	.796	.670	.609	.546	.510	.571	.565
.4000	1.176	1.254	1.334	1.403	1.427	1.550	1.609	1.741	1.950	2.026	.4500	.957	.923	.888	.802	.722	.692	.617	.605	.596	.570
.4500	1.204	1.284	1.353	1.406	1.426	1.535	1.674	1.679	1.876	2.009	.5000	.910	.889	.824	.777	.714	.671	.609	.596	.580	.570
.5000	1.247	1.318	1.390	1.415	1.427	1.517	1.554	1.644	1.820	1.973	.5500	.936	.821	.748	.673	.637	.574	.535	.544	.559	.559
.5500	1.345	1.345	1.393	1.421	1.471	1.519	1.587	1.749	1.947	2.026	.6000	.772	.770	.740	.692	.643	.607	.546	.543	.536	.535
.6000	1.312	1.370	1.421	1.439	1.434	1.492	1.531	1.626	1.704	1.926	.6500	.719	.704	.659	.622	.571	.553	.502	.507	.479	.483
.6500	1.355	1.416	1.485	1.454	1.467	1.507	1.574	1.640	1.753	1.955	.7000	.652	.614	.573	.549	.503	.463	.452	.457	.431	.444
.7000	1.423	1.469	1.520	1.500	1.515	1.535	1.603	1.664	1.752	1.959	.7500	.682	.652	.621	1.089	.841	.773	.733	.725	.735	.735
.7500	1.487	1.543	1.622	1.604	1.652	1.678	1.737	1.781	1.844	1.954	.8000	.790	1.700	1.768	1.809	1.849	1.889	1.911	1.931	1.950	1.950
.8000	1.504	1.626	1.784	1.764	1.784	1.809	1.857	1.881	1.924	1.959	.8500	1.922	2.089	2.000	2.018	1.980	1.950	1.930	1.910	1.890	1.874
.8500	1.901	1.874	1.836	1.855	1.909	1.922	2.039	2.080	2.107	2.152	.9000	1.972	1.994	1.951	1.991	1.948	1.977	1.947	1.935	1.915	1.894
.9000	2.015	1.979	1.951	1.973	2.115	2.084	2.107	2.154	2.197	2.233	.9500	1.972	1.994	1.951	1.991	1.948	1.977	1.947	1.935	1.915	1.894
<b>Vane</b>																					
.0000	2.305	2.268	2.242	2.256	2.292	2.426	2.351	2.369	2.287	2.144	.0250	1.318	1.281	1.251	1.189	1.092	1.046	.989	.988	.951	.906
.0500	3.941	3.633	3.731	3.762	3.907	4.402	4.404	4.404	4.163	3.612	.0750	.815	.765	.703	.445	.461	.396	.351	.365	.340	.324
.1000	3.529	3.477	3.482	3.634	4.142	4.157	4.180	3.891	3.395	3.395	.1500	.518	.459	.348	.236	.113	.079	.067	.077	.058	.059
.1500	3.643	3.427	3.422	3.547	3.682	4.024	4.052	4.130	3.793	3.073	.2000	.349	.309	.217	.101	.012	.006	.006	.006	.006	.006
.2000	3.481	3.302	3.111	3.104	3.100	3.850	3.945	3.945	3.850	3.045	.3000	.235	.198	.129	.055	.024	.018	.024	.015	.021	.021
.4000	2.465	2.561	2.409	2.470	2.684	2.991	3.279	3.475	3.421	2.873	.5000	.207	.176	.127	.067	.045	.045	.035	.051	.044	.044
.5000	2.398	2.315	2.195	2.272	2.482	2.674	2.949	3.168	3.187	2.747	.6000	.158	.173	.124	.076	.071	.094	.104	.115	.105	.102
.6000	2.164	2.083	2.003	2.031	2.133	2.371	2.849	2.726	2.779	2.494	.7000	.201	.182	.149	.128	.111	.151	.159	.180	.187	.150
.7000	2.120	2.037	1.944	1.973	1.973	2.107	2.323	2.493	2.614	2.807	.8000	.248	.247	.217	.220	.235	.263	.281	.304	.275	.254
.8000	2.095	2.015	1.957	1.957	1.901	2.296	2.922	2.628	2.860	2.977	.9000	.677	.670	.641	.671	.705	.779	.847	.891	.894	.744
.9000	1.968	1.885	1.927	2.053	2.242	2.458	2.595	2.500	2.197	2.046	.9500	1.972	1.994	1.951	1.991	1.948	1.977	1.947	1.935	1.915	1.894
<b>Flap</b>																					
.0125	.444	.299	.180	.088	.088	.088	.076	.104	.115	.101	.0250	.444	.299	.180	.088	.088	.088	.088	.088	.088	.076
.0500	.448	.348	.248	.188	.188	.188	.188	.188	.188	.188	.0750	.111	.077	.059	.070	.062	.062	.062	.062	.062	.062
.1000	.445	.345	.245	.185	.185	.185	.185	.185	.185	.185	.1500	.127	.105	.093	.116	.092	.103	.099	.112	.115	.112
.1500	.131	.148	.124	.124	.124	.124	.124	.124	.124	.124	.2000	.131	.148	.124	.137	.125	.148	.130	.162	.145	.144
.2000	.134	.148	.124	.124	.124	.124	.124	.124	.124	.124	.3000	.384	.320	.188	.372	.248	.356	.316	.333	.328	.324
.4000	.546	.431	.311	.212	.176	.147	.114	.092	.076	.059	.6000	.546	.431	.311	.212	.176	.147	.114	.437	.426	.426
.8000	.635	.654	.635	.646	.583	.583	.583	.583	.583	.583	.9000	.782	.745	.743	.735	.699	.698	.638	.631	.654	.650
<b>Spoiler</b>																					
.2573	1.307	1.361	1.609	1.605	1.625	1.652	1.678	1.725	1.751	1.783	.4252	1.307	1.361	1.609	1.605	1.625	1.652	1.678	1.725	1.751	1.783
.4954	1.474	1.642	1.684	1.683	1.683	1.679	1.679	1.679	1.679	1.679	.5748	1.474	1.642	1.684	1.683	1.683	1.679	1.679	1.683	1.683	1.683
.7449	1.492	1.686	1.691	1.693	1.693	1.672	1.672	1.672	1.672	1.672	.6500	1.492	1.686	1.691	1.693	1.693	1.672	1.672	1.684	1.684	1.684
.9740	1.711	1.637	1.637	1.637	1.637	1.637	1.637	1.637	1.637	1.637	.7500	1.711	1.637	1.637	1.637	1.637	1.637	1.637	1.637	1.637	1.637

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TABLE XII. - PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(c) Double slotted flap configuration with spoiler;  $h_s = -0.050\bar{c}$ ;  $\frac{V}{b/2} = 0.30$

Upper surface													Lower surface													
x/c	$C_p$ for -												$C_p$ for -													
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$		
Wing																										
.0000	.572	1.325	2.145	2.812	3.379	4.231	4.406	3.883	2.909	2.294	.0125	1.013	.788	.723	.800	.946	1.154	1.293	1.286	1.183	1.052	.0250	1.032	1.528	2.248	
.0125	.985	1.528	2.245	2.832	3.349	4.154	4.056	3.793	2.897	2.277	.0250	1.013	.859	.732	.657	.607	.581	.574	.557	.555	.525	.0375	1.044	1.300	2.035	
.0250	1.031	1.406	2.372	2.943	3.483	4.066	4.084	3.813	2.909	2.277	.0750	1.013	.884	.754	.660	.601	.539	.494	.473	.449	.451	.1000	1.052	1.231	1.895	
.0375	1.040	1.263	1.938	2.482	3.052	3.671	3.979	4.045	3.854	2.858	.0750	1.013	.884	.754	.660	.601	.539	.494	.473	.449	.451	.1000	1.062	1.231	1.895	
.0500	1.044	1.300	1.925	2.483	3.052	3.671	3.979	4.045	3.854	2.858	.0750	1.013	.884	.754	.660	.601	.539	.494	.473	.449	.451	.1000	1.062	1.231	1.895	
.0750	1.060	1.263	1.823	2.482	3.052	3.671	3.979	4.045	3.854	2.858	.1000	1.013	.884	.754	.660	.601	.539	.494	.473	.449	.451	.1000	1.062	1.231	1.895	
.1000	1.062	1.231	1.252	2.332	3.053	3.673	4.046	4.424	3.787	2.857	.1000	1.013	.919	.803	.699	.628	.542	.479	.452	.445	.429	.1000	1.062	1.231	1.895	
.1250	1.097	1.244	1.277	1.401	2.093	3.347	3.790	3.460	2.805	2.268	.1000	1.013	.919	.803	.699	.628	.542	.479	.452	.445	.429	.1000	1.062	1.231	1.895	
.1500	1.116	1.259	1.326	1.344	1.974	2.422	3.104	3.102	2.714	2.239	.1000	1.013	.934	.811	.742	.637	.551	.485	.458	.445	.429	.1000	1.062	1.231	1.895	
.1750	1.141	1.263	1.338	1.342	1.974	1.982	2.654	2.784	2.671	2.236	.1000	1.013	.934	.811	.742	.637	.551	.485	.458	.445	.429	.1000	1.062	1.231	1.895	
.2000	1.166	1.288	1.354	1.371	1.917	1.795	2.539	2.684	2.672	2.212	.1000	1.013	.934	.811	.742	.637	.551	.485	.458	.445	.429	.1000	1.062	1.231	1.895	
.2250	1.189	1.301	1.381	1.389	1.908	1.623	1.985	2.151	2.021	1.887	.1000	1.013	.934	.811	.742	.637	.551	.485	.458	.445	.429	.1000	1.062	1.231	1.895	
.2500	1.214	1.315	1.381	1.389	1.908	1.623	1.985	2.151	2.021	1.887	.1000	1.013	.934	.811	.742	.637	.551	.485	.458	.445	.429	.1000	1.062	1.231	1.895	
.2750	1.240	1.310	1.394	1.404	1.929	1.605	1.917	2.075	2.201	2.137	.1000	1.013	.934	.811	.742	.637	.551	.485	.458	.445	.429	.1000	1.062	1.231	1.895	
.3000	1.249	1.318	1.423	1.423	1.944	1.581	1.820	1.944	2.080	2.090	.1000	1.013	.934	.811	.742	.637	.551	.485	.458	.445	.429	.1000	1.062	1.231	1.895	
.3250	1.256	1.400	1.428	1.429	1.947	1.572	1.775	1.877	2.015	2.055	.1000	1.013	.934	.811	.742	.637	.551	.485	.458	.445	.429	.1000	1.062	1.231	1.895	
.3500	1.373	1.456	1.480	1.471	1.977	1.592	1.744	1.891	2.031	2.042	.1000	1.013	.934	.811	.742	.637	.551	.485	.458	.445	.429	.1000	1.062	1.231	1.895	
.3750	1.452	1.525	1.548	1.529	1.952	1.617	1.716	1.801	1.900	1.991	.1000	1.013	.934	.811	.742	.637	.551	.485	.458	.445	.429	.1000	1.062	1.231	1.895	
.4000	1.452	1.738	1.739	1.736	1.697	1.743	1.835	1.848	1.876	1.906	.1000	1.013	.934	.811	.742	.637	.551	.485	.458	.445	.429	.1000	1.062	1.231	1.895	
.4250	1.457	2.021	2.028	2.195	2.163	2.235	2.285	2.319	2.424	2.405	.1000	1.013	.934	.811	.742	.637	.551	.485	.458	.445	.429	.1000	1.062	1.231	1.895	
.4500	1.457	2.031	2.031	2.195	2.163	2.235	2.285	2.319	2.424	2.405	.1000	1.013	.934	.811	.742	.637	.551	.485	.458	.445	.429	.1000	1.062	1.231	1.895	
.4750	1.459	2.031	2.031	2.195	2.163	2.235	2.285	2.319	2.424	2.405	.1000	1.013	.934	.811	.742	.637	.551	.485	.458	.445	.429	.1000	1.062	1.231	1.895	
.5000	1.459	2.031	2.031	2.195	2.163	2.235	2.285	2.319	2.424	2.405	.1000	1.013	.934	.811	.742	.637	.551	.485	.458	.445	.429	.1000	1.062	1.231	1.895	
.5250	1.459	2.031	2.031	2.195	2.163	2.235	2.285	2.319	2.424	2.405	.1000	1.013	.934	.811	.742	.637	.551	.485	.458	.445	.429	.1000	1.062	1.231	1.895	
.5500	1.459	2.031	2.031	2.195	2.163	2.235	2.285	2.319	2.424	2.405	.1000	1.013	.934	.811	.742	.637	.551	.485	.458	.445	.429	.1000	1.062	1.231	1.895	
.5750	1.459	2.031	2.031	2.195	2.163	2.235	2.285	2.319	2.424	2.405	.1000	1.013	.934	.811	.742	.637	.551	.485	.458	.445	.429	.1000	1.062	1.231	1.895	
.6000	1.459	2.031	2.031	2.195	2.163	2.235	2.285	2.319	2.424	2.405	.1000	1.013	.934	.811	.742	.637	.551	.485	.458	.445	.429	.1000	1.062	1.231	1.895	
.6250	1.459	2.031	2.031	2.195	2.163	2.235	2.285	2.319	2.424	2.405	.1000	1.013	.934	.811	.742	.637	.551	.485	.458	.445	.429	.1000	1.062	1.231	1.895	
.6500	1.459	2.031	2.031	2.195	2.163	2.235	2.285	2.319	2.424	2.405	.1000	1.013	.934	.811	.742	.637	.551	.485	.458	.445	.429	.1000	1.062	1.231	1.895	
.6750	1.459	2.031	2.031	2.195	2.163	2.235	2.285	2.319	2.424	2.405	.1000	1.013	.934	.811	.742	.637	.551	.485	.458	.445	.429	.1000	1.062	1.231	1.895	
.7000	1.459	2.031	2.031	2.195	2.163	2.235	2.285	2.319	2.424	2.405	.1000	1.013	.934	.811	.742	.637	.551	.485	.458	.445	.429	.1000	1.062	1.231	1.895	
.7250	1.459	2.031	2.031	2.195	2.163	2.235	2.285	2.319	2.424	2.405	.1000	1.013	.934	.811	.742	.637	.551	.485	.458	.445	.429	.1000	1.062	1.231	1.895	
.7500	1.459	2.031	2.031	2.195	2.163	2.235	2.285	2.319	2.424	2.405	.1000	1.013	.934	.811	.742	.637	.551	.485	.458	.445	.429	.1000	1.062	1.231	1.895	
.7750	1.459	2.031	2.031	2.195	2.163	2.235	2.285	2.319	2.424	2.405	.1000	1.013	.934	.811	.742	.637	.551	.485	.458	.445	.429	.1000	1.062	1.231	1.895	
.8000	1.459	2.031	2.031	2.195	2.163	2.235	2.285	2.319	2.424	2.405	.1000	1.013	.934	.811	.742	.637	.551	.485	.458	.445	.429	.1000	1.062	1.231	1.895	
Flap																										
.0000	1.441	1.456	1.474	1.593	1.877	1.850	1.477	1.458	1.360	1.271	.0125	.194	.191	.203	.167	.153	.153	.184	.192	.165	.154	.0250	.191	.186	.206	
.0125	2.094	2.116	2.074	1.971	1.801	2.094	2.149	2.204	2.097	1.973	.0250	.787	.769	.635	.722	.622	.622	.779	.735	.722	.706	.0375	.787	.769	.635	
.0250	2.040	2.049	2.043	2.033	2.032	2.043	2.048	2.052	2.052	2.052	.0375	.288	.309	.322	.219	.123	.066	.014	.014	.006	.006	.0500	.288	.309	.322	
.0375	2.040	2.049	2.043	2.033	2.032	2.043	2.048	2.052	2.052	2.05																

TABLE XII. - PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(d) Double slotted flap configuration with spoiler;  $h_s = -0.050\bar{c}$ ;  $\frac{V}{b/2} = 0.43$

x/c	Upper surface										Lower surface									
	$C_p$ for -											$C_p$ for -								
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 24^\circ$	$\alpha = 28^\circ$	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 24^\circ$	$\alpha = 28^\circ$		
Wing																				
0000	.805	1.771	2.130	2.340	2.848	2.632	2.593	2.459	2.182	.0125	.988	.784	.814	.951	1.104	1.260	1.310	1.354	1.340	1.248
.0125	1.074	1.703	2.111	2.322	2.821	2.694	2.742	2.685	2.486	.0250	.997	.815	.752	.781	.809	.882	.902	.926	.935	.905
.0250	1.089	1.664	2.164	2.353	2.821	2.934	2.803	2.690	2.480	.0375	.994	.849	.742	.774	.817	.879	.895	.947	.945	.912
.0500	1.096	1.564	2.223	2.657	2.943	3.003	2.789	2.735	2.494	.0750	.994	.887	.774	.717	.849	.857	.891	.902	.915	.934
.0750	1.102	1.546	2.492	2.711	2.908	2.900	2.792	2.761	2.450	.1000	.984	.876	.780	.713	.837	.867	.891	.923	.931	.944
.1000	1.111	1.494	2.602	2.908	2.893	2.852	2.824	2.749	2.474	.1250	.978	.859	.784	.724	.843	.889	.901	.925	.933	.936
.1500	1.121	1.459	2.602	2.478	2.711	2.700	2.670	2.649	2.404	.2000	.969	.897	.796	.729	.837	.864	.892	.919	.927	.937
.2000	1.170	1.387	1.666	1.994	1.774	3.041	2.780	2.637	2.435	.3000	.926	.854	.783	.729	.843	.864	.894	.910	.915	.949
.3000	1.213	1.330	1.291	1.664	2.512	2.970	2.676	2.552	2.400	.4000	.895	.842	.777	.723	.849	.8610	.8533	.8522	.8518	.8491
.3500	1.241	1.333	1.319	1.357	2.039	2.649	2.499	2.440	2.341	.4000	.855	.818	.759	.713	.837	.864	.894	.910	.912	.941
.4000	1.271	1.370	1.365	1.359	1.798	2.441	2.393	2.365	2.284	.4500	.802	.762	.718	.674	.716	.764	.809	.837	.852	.849
.4500	1.318	1.407	1.418	1.351	1.678	2.277	2.284	2.295	2.234	.5000	.741	.713	.672	.634	.734	.754	.794	.827	.845	.836
.5000	1.358	1.422	1.443	1.375	1.945	2.048	2.131	2.177	2.164	.5500	.716	.663	.682	.573	.682	.748	.796	.844	.871	.842
.5500	1.416	1.494	1.514	1.442	1.853	1.970	2.092	2.121	2.116	.6000	.597	.602	.545	.524	.647	.656	.713	.743	.764	.774
.6000	1.497	1.571	1.585	1.509	1.559	1.685	1.981	2.044	2.052	.7000	.420	.410	.396	.387	.433	.438	.525	.548	.548	.526
.7000	1.771	1.815	1.856	1.793	1.720	1.849	1.876	1.962	1.977	.7500	.358	.352	.353	.320	.388	.424	.525	.542	.540	.520
.7750	2.366	2.386	2.382	2.607	2.428	2.341	2.197	2.159	2.059	.8750	.249	.246	.237	.237	.288	.288	.327	.348	.340	.320
.8750	2.691	2.719	2.743	2.732	2.942	2.417	2.295	2.207	2.068	.9750	.370	.373	.375	.332	.393	.320	.324	.374	.357	.329
Vane																				
0000	3.916	3.885	3.858	3.802	3.541	3.308	3.009	2.856	2.601	.0250	.963	.994	1.053	.367	.533	.737	.841	.802	.568	.541
.0250	6.184	6.136	6.173	6.244	5.800	5.172	4.654	4.204	3.421	.0500	.451	.473	.551	.581	.599	.621	.730	.722	.725	.715
.0500	6.244	6.264	6.204	6.243	5.978	5.287	4.737	4.213	3.482	.0750	.424	.447	.527	.559	.576	.604	.705	.714	.715	.715
.1000	6.251	6.382	6.477	6.580	6.256	5.516	4.941	4.421	3.670	.1250	.389	.403	.494	.514	.537	.562	.671	.681	.682	.681
.1500	6.460	6.521	6.625	6.754	6.339	5.495	4.794	4.121	2.984	.2000	.321	.321	.421	.485	.542	.620	.710	.727	.730	.726
.2000	6.611	6.663	6.768	6.924	6.553	5.710	4.972	4.254	3.077	.3000	.268	.268	.368	.455	.524	.605	.705	.723	.735	.735
.3000	6.237	6.094	6.180	6.360	6.045	5.108	4.609	3.956	2.858	.4000	.037	.033	.037	.033	.039	.039	.053	.071	.088	.088
.4000	5.509	5.391	5.443	5.630	5.467	4.828	4.218	3.652	2.728	.5000	.042	.043	.046	.061	.077	.073	.072	.074	.083	.053
.5000	4.888	4.712	4.768	4.949	4.803	4.223	3.693	3.345	2.565	.6000	.045	.053	.068	.098	.104	.100	.109	.089	.095	.082
.6000	4.126	4.049	4.074	4.259	4.175	3.806	3.421	3.053	2.447	.7000	.134	.139	.118	.143	.149	.163	.133	.139	.136	.121
.7000	3.561	3.503	3.502	3.692	3.640	3.390	3.103	2.823	2.346	.8000	.275	.293	.276	.302	.295	.287	.287	.251	.231	.222
.8000	3.000	2.959	2.978	3.125	3.077	2.942	2.751	2.561	2.190	.9000	.913	.907	.910	.939	.923	.994	.828	.776	.728	.709
.9000	2.655	2.686	2.629	2.723	2.649	2.412	2.272	2.033	1.935											
Flap																				
0000	2.052	2.048	2.248	2.245	2.149	2.027	1.873	1.773	1.598	.0125	.191	.210	.291	.296	.265	.248	.220	.212	.186	.159
.0125	2.824	2.931	2.610	2.664	2.577	2.468	2.297	2.189	2.012	.0250	.000	.006	.050	.064	.021	.033	.026	.009	.027	.018
.0250	3.067	3.102	3.201	3.155	3.057	2.951	2.749	2.530	2.450	.0500	.000	.012	.025	.018	.030	.006	.029	.012	.038	.018
.0500	3.024	2.993	3.009	3.131	3.137	3.115	2.937	2.847	2.622	.0750	.000	.025	.040	.018	.033	.023	.018	.056	.047	
.0750	2.873	2.858	2.861	2.991	2.030	3.027	2.877	2.626	2.595	.1000	.036	.037	.031	.030	.042	.027	.032	.030	.056	.056
.1000	2.716	2.688	2.697	2.853	2.842	2.861	2.760	2.708	2.483	.1250	.049	.065	.061	.048	.033	.043	.024	.074	.075	.075
.1500	1.997	1.972	1.978	2.076	2.133	2.293	2.276	2.293	2.164	.2000	.083	.103	.107	.076	.077	.085	.064	.059	.086	.074
.2000	1.657	1.657	1.672	1.762	1.795	1.676	1.690	1.669	1.686	.3000	.235	.219	.192	.195	.188	.175	.153	.163	.144	
.4000	.997	1.049	1.084	1.132	1.167	1.287	1.316	1.437	1.574	.6000	.376	.383	.337	.338	.348	.348	.323	.271	.227	.250
.6000	1.787	1.765	1.737	1.732	1.690	1.758	1.696	1.743	1.778	.8000	.614	.602	.548	.546	.609	.511	.441	.440	.482	.447
.8000	1.836	1.818	1.785	1.771	1.726	1.770	1.710	1.752	1.781	.9000	.627	.609	.672	.674	.711	.713	.664	.635	.672	.665
Spoiler																				
2243	.811	1.048	1.113	1.153	1.240	1.293	1.348	1.490	1.592	.2557	1.331	1.657	1.712	1.687	1.709	1.699	1.716	1.772	1.754	1.785
.6906	.938	1.169	1.229	1.271	1.373	1.440	1.503	1.594	1.653	.5039	1.471	1.723	1.768	1.733	1.727	1.717	1.731	1.772	1.763	1.785
.7370	1.074	1.107	1.164	1.404	1.486	1.528	1.561	1.634	1.682	.7508	1.424	1.738	1.786	1.748	1.742	1.735	1.731	1.780	1.780	1.802
.8756	1.433	1.792	1.847	1.809	1.811	1.791	1.845	1.838	1.891	.9940	1.656	1.879	1.899	1.816	1.798	1.779	1.777	1.824	1.821	1.845

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TABLE XII - PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(e) Double slotted flap configuration with spoiler;  $h_s = -0.050\bar{c}$ ;  $\frac{V}{b/2} = 0.55$

x/c	Upper surface										Lower surface										
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$	
<b>Wing</b>																					
.0000	1.198	1.916	2.055	2.332	2.455	2.347	2.237	2.183	2.198	2.096	.0250	.947	.804	.929	1.140	1.323	1.446	1.515	1.584	1.617	
.0125	1.320	1.919	2.120	2.356	2.474	2.383	2.302	2.242	2.213	2.102	.0350	.984	.797	.803	.840	.843	1.006	1.042	1.076	1.112	1.152
.0250	1.226	1.928	2.089	2.341	2.432	2.371	2.298	2.230	2.221	2.099	.0450	.969	.831	.772	.751	.757	.764	.784	.793	.811	.845
.0500	1.213	1.878	2.142	2.405	2.453	2.455	2.344	2.252	2.218	2.096	.0750	.969	.831	.754	.696	.673	.665	.675	.676	.673	.682
.0750	1.204	1.819	2.200	2.475	2.492	2.446	2.349	2.260	2.204	2.096	.1000	.964	.838	.751	.690	.664	.629	.615	.624	.622	.624
.1000	1.216	1.728	2.248	2.529	2.529	2.425	2.367	2.277	2.198	2.096	.1500	.947	.834	.735	.678	.637	.574	.571	.553	.563	.563
.1500	1.245	1.559	2.631	2.567	2.488	2.419	2.397	2.282	2.198	2.093	.2000	.928	.834	.735	.681	.687	.559	.548	.522	.522	.522
.2000	1.280	1.431	2.843	2.870	2.432	2.425	2.397	2.277	2.201	2.096	.2500	.903	.822	.748	.678	.651	.575	.544	.531	.513	.516
.2500	1.258	1.411	2.695	3.454	2.601	2.455	2.415	2.277	2.204	2.093	.3000	.872	.757	.723	.660	.643	.572	.533	.516	.507	.507
.3000	1.200	1.419	2.620	3.436	2.877	2.485	2.388	2.265	2.195	2.087	.3500	.857	.759	.729	.645	.604	.555	.519	.507	.493	.496
.3500	1.318	1.410	2.602	2.954	2.697	2.497	2.352	2.233	2.171	2.081	.4000	.857	.731	.677	.623	.569	.518	.506	.470	.472	.472
.4000	1.304	1.409	2.634	2.626	2.455	2.411	2.357	2.269	2.201	2.175	.4500	.780	.651	.578	.545	.515	.482	.475	.454	.455	.455
.4500	1.498	1.531	1.386	1.724	2.454	2.359	2.327	2.160	2.142	2.058	.5000	.648	.631	.552	.541	.517	.476	.454	.443	.431	.426
.5000	1.927	1.409	1.431	1.681	2.364	2.272	2.194	2.137	2.104	2.055	.6201	1.662	1.803	1.723	1.656	1.627	1.574	1.539	1.509	1.493	1.493
.6201	1.662	1.803	1.723	1.593	1.913	2.033	2.059	2.014	2.027	2.003	.6534	2.073	2.221	2.175	2.063	2.057	2.057	2.057	2.057	2.057	2.057
.6534	2.073	2.221	2.175	1.952	2.064	2.065	2.026	2.021	1.997	.7087	2.370	2.559	2.453	2.343	2.296	2.259	2.259	2.259	2.259	2.259	
.7087	2.370	2.919	2.480	2.253	2.174	2.099	2.101	2.049	2.036	2.006	.7151	2.359	2.556	2.511	2.493	2.499	2.032	2.038	2.000	2.000	2.000
.7214	2.448	2.694	2.686	2.473	2.300	2.154	2.133	2.078	2.044	2.008	.7278	2.564	2.797	2.672	2.563	2.342	2.162	2.145	2.078	2.050	2.011
<b>Vane</b>																					
.0000	3.284	3.250	3.105	2.757	2.432	2.272	2.294	2.151	2.083	2.041	.0250	1.323	1.156	1.157	1.003	.892	.814	.734	.703	.679	.670
.0125	6.047	6.103	6.102	5.490	4.459	3.820	3.672	3.443	3.127	2.987	.0350	.677	.488	.597	.508	.438	.365	.325	.294	.289	.274
.0500	6.176	6.188	6.231	5.627	4.517	3.817	3.613	3.323	2.971	2.854	.1000	.348	.194	.314	.261	.231	.174	.124	.120	.106	.095
.1000	6.548	6.538	6.674	6.083	4.682	3.748	3.474	3.180	2.746	2.571	.1500	.210	.091	.215	.170	.135	.102	.074	.096	.088	.067
.1500	6.596	6.558	6.676	6.156	4.625	3.677	3.397	3.087	2.617	2.465	.2000	.049	.044	.065	.045	.045	.041	.036	.036	.038	.038
.2000	6.356	6.516	6.717	6.211	4.631	3.674	3.358	3.087	2.572	2.408	.2500	.054	.073	.086	.082	.099	.094	.082	.082	.082	.082
.3000	6.210	6.153	6.613	6.016	4.686	3.503	3.225	2.926	2.645	2.255	.4000	.050	.073	.095	.076	.120	.132	.101	.114	.100	.105
.4000	5.986	5.951	6.620	6.048	4.626	3.500	2.787	2.589	2.326	2.131	.4500	.058	.125	.108	.128	.132	.127	.143	.124	.125	.125
.4500	6.188	6.172	6.235	4.263	3.307	2.734	2.592	2.422	2.148	2.070	.7000	.135	.154	.164	.164	.174	.175	.120	.126	.172	.172
.7000	3.593	3.561	3.708	3.685	2.779	2.527	2.409	2.280	2.280	2.052	.8000	.288	.331	.326	.319	.324	.287	.284	.292	.293	.296
.8000	3.113	3.097	3.209	3.189	2.701	2.352	2.284	2.165	2.033	2.000	.9200	.959	1.003	1.012	1.009	.889	.841	.808	.799	.768	.802
.9000	2.690	2.666	2.745	2.733	2.400	2.165	2.107	2.023	1.950	1.950											
<b>Flap</b>																					
.0000	1.671	1.688	1.825	1.651	1.473	1.446	1.406	1.355	1.333	1.333	.0125	.219	.225	.206	.201	.182	.140	.140	.251	.230	.245
.0125	2.314	2.613	2.705	2.677	2.460	2.179	2.136	2.087	2.041	2.082	.0250	.000	.028	.043	.072	.078	.074	.067	.068	.070	.070
.0250	2.903	3.075	3.184	2.922	2.593	2.536	2.452	2.426	2.427	2.427	.0500	.019	.028	.046	.046	.078	.057	.077	.047	.058	.058
.0500	2.869	2.949	3.068	2.849	2.621	2.571	2.482	2.451	2.448	2.448	.0750	.040	.046	.062	.064	.087	.066	.096	.045	.070	.070
.0750	2.803	2.884	2.982	2.612	3.018	2.722	2.639	2.536	2.449	2.373	.1000	.059	.078	.080	.084	.084	.077	.082	.080	.090	.090
.1000	2.684	2.816	2.859	2.874	2.671	2.624	2.540	2.457	2.346	2.285	.1500	.082	.094	.098	.100	.109	.102	.095	.108	.098	.093
.1500	1.953	1.996	2.034	2.201	2.312	2.173	2.113	2.042	1.994	1.994	.2000	.113	.134	.126	.138	.120	.124	.128	.118	.129	.129
.2000	1.742	1.819	1.821	1.914	1.982	1.972	1.912	1.871	1.880	1.873	.4000	.273	.268	.243	.249	.234	.228	.224	.215	.210	.210
.4000	1.097	1.138	1.129	1.213	1.346	1.447	1.533	1.539	1.632	1.647	.6000	.420	.431	.391	.383	.364	.341	.331	.335	.322	.306
.6000	1.850	1.859	1.840	1.770	1.723	1.707	1.716	1.714	1.744	1.778	.9000	.833	.850	.809	.772	.745	.713	.698	.691	.687	.688
<b>Spoiler</b>																					
.2443	.824	1.039	1.079	1.192	1.408	1.497	1.547	1.613	1.650	1.705	.2554	1.497	1.759	1.804	1.751	1.733	1.705	1.711	1.757	1.748	1.771
.4891	.974	1.223	1.251	1.392	1.848	1.847	1.849	1.695	1.717	1.782	.4939	1.337	1.753	1.824	1.763	1.733	1.729	1.739	1.769	1.784	1.782
.7448	1.087	1.381	1.413	1.411	1.622	1.640	1.681	1.698	1.705	1.875	.7508	1.421	1.771	1.830	1.772	1.745	1.729	1.728	1.750	1.766	1.793
.9554	1.448	1.622	1.908	1.693	1.795	1.770	1.778	1.830	1.809	1.881	.7740	1.474	1.883	2.000	1.867	1.757	1.778	1.807	1.803	1.822	

TABLE XII.- PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Concluded

(f) Double slotted flap configuration with spoiler;  $h_s = -0.050\bar{c}$ ;  $\frac{V}{b/2} = 0.72$

x/c	Upper surface										Lower surface												
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$	$C_p$ for -	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$		
<b>WING</b>																							
.0000	2.050	2.184	2.373	2.250	2.129	1.988	1.944	1.912	1.881	1.922		.0125	.721	.822	1.012	1.113	1.204	1.242	1.293	1.341	1.387	1.441	
.0125	1.923	2.169	2.300	2.180	2.073	1.947	1.924	1.924	1.856	1.928		.0250	.734	.756	.847	.921	.982	1.015	1.070	1.129	1.156	1.200	
.0375	1.872	2.190	2.336	2.195	2.096	1.974	1.947	1.936	1.902	1.923		.0500	.737	.723	.755	.775	.799	.826	.866	.895	.928	.957	
.0750	1.877	2.253	2.376	2.210	2.102	2.012	1.974	1.945	1.907	1.925		.1000	.712	.738	.731	.714	.724	.708	.725	.748	.769	.788	
.1000	1.803	2.181	2.282	2.228	2.120	2.030	1.985	1.956	1.905	1.928		.1500	.724	.744	.737	.731	.746	.724	.725	.748	.769	.788	
.1500	1.807	2.161	2.248	2.244	2.107	2.032	1.982	1.962	1.907	1.928		.2000	.735	.759	.746	.714	.691	.667	.644	.680	.682	.690	
.2000	1.872	2.266	2.378	2.234	2.105	2.041	2.018	1.971	1.940	1.940		.2500	.792	.780	.771	.730	.705	.684	.673	.683	.679	.690	
.2500	1.890	2.497	2.685	2.216	2.105	2.036	2.018	1.965	1.972	1.960		.3000	.830	.810	.807	.766	.735	.702	.690	.713	.705	.716	
.3000	1.809	2.274	2.856	2.207	2.117	2.041	2.006	1.974	1.923	1.948		.3500	.901	.889	.902	.842	.814	.779	.766	.777	.775	.771	
.3500	1.848	2.051	3.049	2.192	2.123	2.038	2.009	1.948	1.922	1.964		.4000	.878	.878	.878	.878	.878	.863	.871	.864	.856	.850	
.4000	1.886	1.789	3.107	2.253	2.126	2.030	2.000	1.948	1.936	1.951		.4500	1.617	3.107	2.323	2.111	2.030	1.988	1.965	1.936	1.957		
.4500	1.853	1.617	3.107	2.323	2.111	2.030	1.988	1.965	1.936	1.957		.5000	1.570	1.575	3.061	2.408	2.141	2.024	1.991	1.971	1.945	1.963	
.5000	1.854	1.567	3.061	2.408	2.141	2.024	1.991	1.971	1.945	1.963		.5500	1.604	1.548	2.878	2.362	2.123	2.006	1.980	1.965	1.942	1.945	
.5500	1.861	1.566	2.618	2.292	2.111	1.988	1.965	1.959	1.942	1.974		.6000	1.567	1.567	2.618	2.292	2.111	1.988	1.965	1.959	1.942	1.974	
.6000	1.854	1.533	2.152	2.171	2.072	1.974	1.959	1.952	1.954	1.985		.6500	1.533	1.533	2.152	2.171	2.072	1.974	1.959	1.952	1.954		
.6500	1.859	1.533	2.152	2.171	2.072	1.974	1.959	1.952	1.954	1.985		.7000	1.836	1.925	1.939	1.839	1.805	1.793	1.822	1.854	1.815	1.768	
.7000	1.853	1.533	2.152	2.171	2.072	1.974	1.959	1.952	1.954	1.985		.7500	1.833	1.901	1.911	1.845	1.817	1.791	1.801	1.830	1.800	1.773	
.7500	1.859	1.401	1.948	2.079	2.04	1.941	1.939	1.942	1.948	1.991		.8000	1.709	1.786	1.829	1.812	1.781	1.733	1.734	1.763	1.746	1.745	
.8000	1.875	1.328	1.848	1.904	1.909	1.929	1.933	1.935	1.933	1.965		.8500	1.573	1.681	1.732	1.772	1.797	1.699	1.699	1.736	1.722	1.745	
.8500	1.831	1.274	1.749	1.951	1.961	1.912	1.920	1.920	1.971	1.994		.9000	1.437	1.572	1.676	1.742	1.736	1.696	1.702	1.719	1.720	1.733	
.9000	1.266	1.262	1.657	1.906	1.925	1.897	1.924	1.948	1.971	1.997		.9500	1.334	1.443	1.605	1.730	1.739	1.711	1.713	1.736	1.731	1.745	

TABLE XIII.- PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE

(a) Double slotted flap configuration with spoiler;  $h_S = -0.100\bar{c}$ ;  $\frac{y}{b/2} = 0$

Upper surface												Lower surface																					
$x/c$	$C_p$ for -											$x/c$	$C_p$ for -																				
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$			
Fuselage																																	
.0000	.000	.006	.000	.000	.022	.039	.059	.077	.111	.142	.0500	.055	.015	.765	.705	.634	.546	.485	.440	.404	.385	.350	.328	.293	.263	.234	.204	.174	.144	.114	.084		
.0500	.748	.840	.884	.947	.997	.015	.035	.050	.060	.080	.100	.1500	.192	.134	.104	.093	.083	.073	.062	.052	.042	.032	.022	.012	.002	.002	.002	.002	.002	.002	.002	.002	.002
.1000	.832	.929	.947	.997	.009	.045	.063	.083	.109	.127	.136	.1500	.937	.991	.026	.072	.089	.113	.115	.132	.142	.152	.162	.172	.182	.192	.202	.212	.222	.232	.242	.252	.262
.2000	.984	.040	.058	.106	.108	.101	.110	.115	.127	.138	.148	.2500	.1.032	.1.080	.1.072	.1.131	.1.123	.1.122	.1.130	.1.124	.1.126	.1.136	.1.130	.1.135	.1.138	.1.142	.1.145	.1.148	.1.152	.1.155	.1.158	.1.162	.1.165
.3000	.1.057	.1.084	.1.107	.1.131	.1.117	.1.086	.1.092	.1.103	.1.111	.1.118	.1.128	.3500	.1.063	.1.083	.1.105	.1.090	.1.086	.1.092	.1.102	.1.105	.1.110	.1.115	.1.120	.1.125	.1.130	.1.135	.1.140	.1.145	.1.150	.1.155	.1.160	.1.165	
.4000	.1.032	.1.043	.1.058	.1.072	.1.065	.1.078	.1.083	.1.094	.1.102	.1.105	.1.113	.4500	.1.038	.1.046	.1.070	.1.115	.1.129	.1.125	.1.169	.1.192	.1.222	.1.260	.1.280	.1.290	.1.300	.1.310	.1.320	.1.330	.1.340	.1.350	.1.360	.1.370	
.5000	.1.044	.1.065	.1.107	.1.165	.1.229	.1.233	.1.267	.1.313	.1.398	.1.334	.5500	.1.117	.1.182	.1.233	.1.268	.1.298	.1.349	.1.387	.1.576	.1.676	.1.776	.1.876	.1.976	.2.076	.2.176	.2.276	.2.376	.2.476	.2.576	.2.676	.2.776		
.6000	.1.092	.1.152	.1.241	.1.283	.1.305	.1.307	.1.337	.1.363	.1.392	.1.422	.1.452	.6500	.1.164	.1.218	.1.271	.1.321	.1.371	.1.427	.1.466	.1.506	.1.546	.1.586	.1.626	.1.666	.1.706	.1.746	.1.786	.1.826	.1.866	.1.906	.1.946		
.7000	.1.164	.1.218	.1.271	.1.321	.1.371	.1.317	.1.357	.1.407	.1.457	.1.507	.1.557	.7500	.1.221	.1.242	.1.304	.1.329	.1.324	.1.331	.1.317	.1.336	.1.348	.1.369	.1.389	.1.409	.1.429	.1.449	.1.469	.1.489	.1.509	.1.529	.1.549	.1.569	
.8000	.1.268	.1.305	.1.329	.1.339	.1.357	.1.352	.1.362	.1.427	.1.457	.1.476	.1.516	.8500	.1.294	.1.305	.1.314	.1.336	.1.354	.1.367	.1.414	.1.454	.1.491	.1.531	.1.571	.1.611	.1.651	.1.691	.1.731	.1.771	.1.811	.1.851	.1.891	.1.931	
.9000	.1.224	.1.231	.1.238	.1.258	.1.277	.1.298	.1.343	.1.389	.1.428	.1.468	.1.508	.9500	.1.224	.1.194	.1.180	.1.193	.1.218	.1.224	.1.251	.1.293	.1.365	.1.387	.1.428	.1.468	.1.508	.1.548	.1.588	.1.628	.1.668	.1.708	.1.748	.1.788	
.9940	.1.221	.1.212	.1.192	.1.199	.1.215	.1.245	.1.296	.1.321	.1.377	.1.417	.1.467	.9940	.1.088	.1.078	.1.068	.1.083	.1.093	.1.103	.1.113	.1.123	.1.133	.1.143	.1.153	.1.163	.1.173	.1.183	.1.193	.1.203	.1.213	.1.223	.1.233	.1.243	

~~CONFIDENTIAL~~  
TABLE XIII. - PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(b) Double slotted flap configuration with spoiler;  $h_s = -0.100\bar{c}$ ;  $\frac{V}{b/2} = 0.21$

x/c	Upper surface										Lower surface										
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$	
<b>Wing</b>																					
.0000	.876	.908	1.842	3.168	5.397	7.416	9.440	7.392	3.347	2.804	.0125	1.039	.800	.635	.592	.572	.627	.664	.633	.431	.361
.0125	.879	1.252	1.648	3.046	7.582	10.266	11.121	7.078	3.033	2.291	.0250	1.069	.825	.654	.564	.449	.398	.310	.311	.243	.210
.0250	.918	1.200	1.502	2.412	3.659	5.871	8.392	6.507	3.050	2.312	.0375	.988	.849	.705	.601	.443	.339	.229	.237	.232	.222
.0500	.957	1.161	1.404	1.637	2.006	2.592	4.220	4.835	2.780	2.196	.0750	.976	.855	.724	.625	.480	.373	.271	.284	.279	.266
.0750	.970	1.145	1.341	1.487	1.812	2.104	2.940	3.752	2.687	2.192	.1000	.985	.874	.742	.649	.511	.416	.321	.340	.340	.350
.1000	.988	1.110	1.319	1.509	1.718	1.951	2.607	3.279	2.619	2.175	.1250	.970	.874	.874	.763	.486	.421	.422	.416	.417	.422
.1500	1.029	1.157	1.304	1.440	1.802	1.954	2.684	3.495	2.877	2.375	.2000	.970	.889	.795	.697	.595	.483	.388	.388	.388	.388
.2000	1.042	1.175	1.301	1.436	1.860	1.964	2.706	2.913	2.935	2.322	.2500	.971	.868	.803	.738	.652	.593	.432	.424	.419	.437
.2500	1.057	1.185	1.295	1.415	1.498	1.627	1.884	2.048	2.258	2.105	.3000	.970	.911	.810	.759	.671	.624	.545	.553	.531	.522
.3000	1.088	1.197	1.301	1.396	1.668	1.581	1.803	1.938	2.147	2.081	.3250	.985	.829	.790	.714	.651	.580	.559	.557	.545	.545
.3500	1.094	1.209	1.292	1.378	1.443	1.541	1.678	1.799	2.034	2.047	.4000	.976	.932	.845	.808	.735	.685	.601	.610	.578	.571
.4000	1.118	1.218	1.292	1.384	1.423	1.521	1.625	1.729	1.959	2.049	.4500	.951	.923	.845	.814	.745	.700	.619	.624	.592	.593
.4500	1.139	1.237	1.304	1.387	1.424	1.503	1.565	1.675	1.892	2.043	.5000	.988	.880	.815	.787	.726	.683	.601	.610	.592	.583
.5000	1.173	1.268	1.332	1.406	1.434	1.480	1.839	1.648	1.854	2.026	.5500	.833	.862	.751	.771	.677	.648	.589	.615	.593	.560
.5500	1.188	1.277	1.332	1.387	1.431	1.444	1.500	1.592	1.777	1.997	.6000	.761	.749	.720	.707	.652	.627	.554	.574	.546	.536
.6000	1.212	1.299	1.341	1.406	1.434	1.454	1.500	1.577	1.728	1.965	.6500	.706	.680	.635	.631	.586	.500	.518	.499	.490	.490
.6500	1.239	1.311	1.356	1.403	1.443	1.455	1.509	1.604	1.684	1.947	.7000	.674	.593	.553	.549	.514	.495	.440	.473	.454	.440
.7000	1.268	1.338	1.368	1.408	1.448	1.480	1.505	1.653	1.646	1.904	.7500	.656	.592	.542	.542	.504	.474	.437	.432	.427	.427
.7500	1.285	1.355	1.386	1.426	1.454	1.472	1.522	1.622	1.678	1.880	.8000	.632	.592	.542	.542	.504	.467	.437	.432	.427	.427
.8000	1.312	1.382	1.391	1.418	1.438	1.464	1.509	1.659	1.675	1.828	.8250	.612	.589	.542	.542	.504	.467	.437	.432	.427	.427
.8500	1.329	1.387	1.392	1.423	1.442	1.473	1.522	1.622	1.678	1.828	.9000	.592	.577	.537	.537	.504	.467	.437	.432	.427	.427
.9000	1.334	1.392	1.397	1.427	1.447	1.478	1.523	1.623	1.678	1.828	.9250	.572	.557	.522	.522	.504	.467	.437	.432	.427	.427
.9500	1.339	1.402	1.397	1.432	1.447	1.482	1.523	1.623	1.678	1.828	.9750	.552	.537	.507	.507	.504	.467	.437	.432	.427	.427
.1000	1.344	1.407	1.402	1.437	1.452	1.487	1.523	1.623	1.678	1.828	.1000	.532	.517	.507	.507	.504	.467	.437	.432	.427	.427
.1000	1.344	1.407	1.402	1.437	1.452	1.487	1.523	1.623	1.678	1.828	.1250	.512	.497	.487	.487	.484	.467	.437	.432	.427	.427
.1250	1.349	1.412	1.407	1.442	1.457	1.492	1.523	1.623	1.678	1.828	.1500	.502	.487	.477	.477	.474	.467	.437	.432	.427	.427
.1500	1.354	1.417	1.412	1.447	1.462	1.497	1.523	1.623	1.678	1.828	.1750	.502	.487	.477	.477	.474	.467	.437	.432	.427	.427
.2000	1.359	1.422	1.417	1.452	1.467	1.497	1.523	1.623	1.678	1.828	.2500	.482	.467	.457	.457	.454	.467	.437	.432	.427	.427
.2500	1.364	1.427	1.422	1.457	1.467	1.497	1.523	1.623	1.678	1.828	.3000	.462	.447	.437	.437	.434	.467	.437	.432	.427	.427
.3000	1.369	1.432	1.427	1.462	1.472	1.497	1.523	1.623	1.678	1.828	.3500	.442	.427	.417	.417	.414	.467	.437	.432	.427	.427
.4000	1.374	1.437	1.432	1.467	1.477	1.497	1.523	1.623	1.678	1.828	.4500	.422	.407	.397	.397	.394	.467	.437	.432	.427	.427
.5000	1.379	1.442	1.437	1.472	1.482	1.497	1.523	1.623	1.678	1.828	.5500	.402	.387	.377	.377	.374	.467	.437	.432	.427	.427
.6000	1.384	1.447	1.442	1.477	1.487	1.497	1.523	1.623	1.678	1.828	.6500	.382	.367	.357	.357	.354	.467	.437	.432	.427	.427
.7000	1.389	1.452	1.447	1.482	1.492	1.497	1.523	1.623	1.678	1.828	.7500	.362	.347	.337	.337	.334	.467	.437	.432	.427	.427
.8000	1.394	1.457	1.452	1.487	1.497	1.497	1.523	1.623	1.678	1.828	.8500	.342	.327	.317	.317	.314	.467	.437	.432	.427	.427
.9000	1.399	1.462	1.457	1.492	1.507	1.497	1.523	1.623	1.678	1.828	.9500	.322	.307	.297	.297	.294	.467	.437	.432	.427	.427
.9500	1.404	1.467	1.462	1.497	1.507	1.497	1.523	1.623	1.678	1.828	.1000	.302	.287	.277	.277	.274	.467	.437	.432	.427	.427
.1000	1.409	1.472	1.467	1.502	1.507	1.497	1.523	1.623	1.678	1.828	.1250	.282	.267	.257	.257	.254	.467	.437	.432	.427	.427
.1250	1.414	1.477	1.472	1.507	1.507	1.497	1.523	1.623	1.678	1.828	.1500	.262	.247	.237	.237	.234	.467	.437	.432	.427	.427
.1500	1.419	1.482	1.477	1.507	1.507	1.497	1.523	1.623	1.678	1.828	.2000	.242	.227	.217	.217	.214	.467	.437	.432	.427	.427
.2000	1.424	1.487	1.482	1.507	1.507	1.497	1.523	1.623	1.678	1.828	.2500	.222	.207	.197	.197	.194	.467	.437	.432	.427	.427
.3000	1.429	1.492	1.487	1.507	1.507	1.497	1.523	1.623	1.678	1.828	.3500	.202	.187	.177	.177	.174	.467	.437	.432	.427	.427
.4000	1.434	1.497	1.492	1.507	1.507	1.497	1.523	1.623	1.678	1.828	.4500	.182	.167	.157	.157	.154	.467	.437	.432	.427	.427
.5000	1.439	1.502	1.497	1.507	1.507	1.497	1.523	1.623	1.678	1.828	.5500	.162	.147	.137	.137	.134	.467	.437	.432	.427	.427
.6000	1.444	1.507	1.502	1.507	1.507	1.497	1.523	1.623	1.678	1.828	.6500	.142	.127	.117	.117	.114	.467	.437	.432	.427	.427
.7000	1.449	1.512	1.507	1.507	1.507	1.497	1.523	1.623	1.678	1.828	.7500	.122	.107	.097	.097	.094	.467	.437	.432	.427	.427
.8000	1.454	1.517	1.512	1.507	1.507	1.497	1.523	1.623	1.678	1.828	.8500	.102	.087	.077	.077	.074	.467	.437	.432	.427	.427
.9000	1.459	1.522	1.517	1.507	1.507	1.497	1.523	1.623	1.678	1.828	.9500	.082	.067	.057	.057	.054	.467	.437	.432	.427	.427
.9500	1.464	1.527	1.522																		

TABLE XIII. - PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(c) Double slotted flap configuration with spoiler;  $h_S = -0.100\bar{c}$ ;  $\frac{V}{b/2} = 0.30$

Upper surface													Lower surface												
Wing	$C_p$ for -												Wing	$C_p$ for -											
	$x/c$	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$	$x/c$	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$			
Wing																									
0000	.920	1.179	2.015	2.836	3.468	4.141	4.235	3.740	2.699	2.129															
.0125	.963	1.407	2.168	2.859	3.414	4.071	3.939	3.686	2.687	2.123															
.0250	1.000	1.408	2.169	2.860	3.425	4.082	3.943	3.701	2.693	2.129															
.0375	1.037	1.420	2.170	2.817	3.443	4.132	3.982	3.751	2.683	2.120															
.0500	1.031	1.420	2.169	2.817	3.443	4.132	3.982	3.751	2.683	2.120															
.0750	1.031	1.420	2.169	2.817	3.443	4.132	3.982	3.751	2.683	2.120															
1.000	1.053	1.497	2.210	2.912	3.958	4.758	4.263	3.630	2.681	2.132															
1.500	1.071	1.213	1.262	1.378	2.021	3.353	3.643	3.322	2.598	2.135															
2.000	1.102	1.219	1.299	1.314	1.345	2.025	3.045	3.012	2.527	2.135															
2.500	1.118	1.222	1.308	1.342	1.441	1.975	2.598	2.743	2.480	2.123															
3.000	1.146	1.234	1.320	1.332	1.394	1.804	2.371	2.522	2.397	2.129															
3.500	1.144	1.241	1.308	1.323	1.360	1.402	1.615	1.970	2.176	2.225	2.088														
4.000	1.167	1.259	1.323	1.360	1.402	1.615	1.970	2.176	2.225	2.088															
4.500	1.207	1.264	1.345	1.369	1.408	1.610	1.900	2.075	2.145	2.088															
5.000	1.229	1.305	1.363	1.378	1.417	1.978	1.813	1.955	2.033	2.065															
5.500	1.237	1.315	1.364	1.378	1.417	1.978	1.743	1.890	1.988	2.018															
6.000	1.259	1.340	1.380	1.435	1.435	1.981	1.822	1.822	1.932	1.944															
6.500	1.272	1.340	1.430	1.439	1.439	1.981	1.820	1.797	1.859	1.945															
7.000	1.421	1.477	1.518	1.554	1.519	1.676	1.751	1.824	1.841	1.865															
7.500	1.628	1.703	1.710	1.762	1.825	1.917	1.952	1.979	1.892	1.887															
8.000	1.740	1.805	1.817	1.872	1.864	2.082	2.100	2.087	1.935	1.900															
8.141	1.789	1.854	1.869	1.933	2.036	2.156	2.165	2.146	1.959	1.915															
8.182	1.814	1.858	1.903	1.964	2.072	2.193	2.196	2.188	1.985	1.912															
8.224	1.904	1.990	2.020	2.059	2.193	2.294	2.317	2.295	2.048	1.920															
Vane																									
0000	3.480	3.555	3.447	3.586	3.810	4.067	4.024	3.916	3.432	3.205															
.0250	4.127	4.243	4.159	4.209	4.323	5.177	5.241	4.928	3.823	3.235															
.0500	4.056	4.203	4.147	4.421	4.764	5.140	5.187	4.857	3.734	3.150															
1.000	3.941	4.092	4.031	4.327	4.661	5.004	5.127	4.755	3.512	2.676															
1.500	3.817	3.951	3.915	4.208	4.577	4.929	5.056	4.695	3.448	2.653															
2.000	3.647	3.833	3.762	4.073	4.477	4.856	5.015	4.692	3.521	2.688															
3.000	3.059	3.287	3.223	3.506	3.982	4.428	4.640	4.391	3.409	2.629															
4.000	2.396	2.657	2.598	2.860	3.165	3.847	4.048	3.684	3.101	2.467															
5.000	1.910	2.179	2.104	2.320	2.685	3.413	3.565	3.424	2.884	2.373															
6.000	1.672	1.852	1.811	1.945	2.048	2.899	3.045	2.979	2.604	2.259															
7.000	1.610	1.722	1.720	1.799	2.072	2.465	2.607	2.606	2.382	2.135															
8.000	1.598	1.676	1.686	1.844	2.104	2.229	2.251	2.122	2.023	1.803															
9.000	1.610	1.640	1.680	1.720	1.896	1.985	2.006	1.941	1.809	1.709															
Flap																									
0000	1.046	1.120	1.305	1.214	1.088	1.072	1.145	1.170	1.145	1.141															
.0125	1.536	1.648	1.717	1.741	1.701	1.728	1.794	1.848	1.805	1.794															
.0250	1.855	1.981	2.040	2.082	2.073	2.125	2.226	2.301	2.264	2.244															
.0500	1.749	1.870	1.924	1.970	2.012	2.079	2.184	2.322	2.355	2.329															
.0750	1.542	1.703	1.787	1.876	1.876	1.929	2.060	2.218	2.290	2.276															
1.000	1.433	1.537	1.651	1.619	1.746	1.826	1.933	2.003	2.142	2.153															
1.500	1.279	1.308	1.325	1.353	1.384	1.324	1.390	1.484	1.753	1.809															
2.000	1.252	1.268	1.285	1.293	1.313	1.205	1.328	1.491	1.650	1.745															
3.000	1.253	1.270	1.285	1.293	1.313	1.213	1.328	1.491	1.650	1.745															
4.000	1.209	1.279	1.299	1.316	1.335	1.234	1.354	1.470	1.614	1.700															
5.000	1.239	1.305	1.313	1.324	1.335	1.234	1.354	1.470	1.614	1.700															
6.000	1.239	1.305	1.313	1.324	1.335	1.234	1.354	1.470	1.614	1.700															
8.000	1.257	1.329	1.335	1.346	1.356	1.251	1.371	1.487	1.622	1.708															
9.000	1.221	1.351	1.363	1.374	1.384	1.251	1.371	1.487	1.622	1.708															
Spoiler																									
0275	.984	.972	.975	.980	.958	.648	.589	.555	.528	.493															
.4909	1.000	.972	1.015	.960	.846	.612	.654	.714	.991	1.287															
.7396	.909	.935	.957	.919	.892	.678	.755	.844	1.120	1.325															
.9796	1.215	1.351	1.363	1.424	1.529	1																			

TABLE XIII. - PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(d) Double slotted flap configuration with spoiler;  $h_s = -0.100\bar{c}$ ;  $\frac{V}{b/2} = 0.43$

x/c	Upper surface										Lower surface										
	$C_p$ for -										$C_p$ for -										
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$		$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
<b>Wing</b>																					
.0000	.764	1.782	2.009	2.439	2.874	2.813	2.580	2.512	2.352	2.108	.0125	1.067	.803	.787	.909	1.108	1.214	1.280	1.311	1.288	1.242
.0125	.957	1.742	1.982	2.418	2.846	2.823	2.690	2.577	2.352	2.124	.0250	1.042	.840	.742	.759	.918	.953	.981	.908	.895	.898
.0375	1.006	1.655	2.040	2.470	2.837	2.853	2.729	2.586	2.364	2.110	.0500	1.030	.877	.734	.720	.698	.673	.667	.678	.675	.633
.0500	1.009	1.600	2.070	2.540	2.966	2.945	2.734	2.601	2.379	2.151	.0750	1.027	.895	.765	.707	.658	.621	.595	.592	.590	.600
.0750	1.034	1.514	2.307	2.619	2.911	2.841	2.754	2.613	2.370	2.137	.1000	1.024	.898	.784	.707	.649	.632	.559	.553	.551	.583
.1000	1.061	1.382	2.617	2.890	2.837	2.795	2.765	2.613	2.364	2.131	.1500	1.006	.911	.793	.729	.649	.590	.530	.524	.519	.516
.1500	1.073	1.268	2.259	3.052	3.440	2.997	2.815	2.619	2.376	2.134	.2000	.988	.914	.793	.732	.655	.590	.535	.536	.502	.510
.2000	1.112	1.268	1.911	2.750	3.757	3.220	2.771	2.574	2.352	2.122	.2500	.973	.904	.790	.735	.658	.593	.539	.536	.499	.504
.2500	1.115	1.274	1.932	1.851	3.103	3.067	2.649	2.500	2.317	2.116	.3000	.959	.879	.778	.726	.652	.593	.539	.536	.499	.504
.3000	1.157	1.311	1.214	1.470	2.563	2.671	2.565	2.450	2.291	2.105	.3500	.942	.865	.772	.716	.665	.605	.545	.541	.510	.516
.3500	1.167	1.308	1.249	1.311	2.092	2.364	2.417	2.349	2.239	2.077	.4000	.970	.831	.748	.707	.649	.593	.539	.527	.504	.507
.4000	1.194	1.323	1.295	1.620	2.270	2.375	2.325	2.245	2.194	2.057	.4500	.918	.791	.708	.671	.622	.584	.524	.512	.484	.493
.4500	1.224	1.335	1.305	1.705	2.321	2.329	2.229	2.110	2.050	1.964	.5000	.741	.732	.657	.619	.585	.533	.494	.463	.475	
.5000	1.235	1.345	1.311	1.548	2.006	2.083	2.110	2.058	2.035	1.956	.5500	.703	.654	.578	.607	.551	.492	.443	.456	.440	.446
.5500	1.257	1.400	1.383	1.343	1.551	1.622	2.024	2.042	2.038	2.020	.6000	.613	.591	.525	.509	.489	.450	.417	.423	.394	.399
.6000	1.530	1.426	1.420	1.409	1.545	1.538	1.952	1.997	1.989	1.984	.6500	1.412	.416	.380	.360	.351	.315	.292	.293	.278	.292
.7000	1.342	1.559	1.514	1.558	1.612	1.764	1.863	1.926	1.912	1.926	.7500	.361	.354	.340	.311	.295	.260	.244	.249	.243	.239
.7700	1.645	1.659	1.818	1.912	1.994	2.061	2.042	2.039	1.956	1.944	.7750	1.673	1.680	1.687	2.071	2.057	1.959	1.956	1.956	1.956	
.7750	1.673	1.680	1.857	1.951	2.046	2.110	2.071	2.057	1.959	1.956	.8000	1.727	1.935	1.915	2.012	2.114	2.174	2.080	1.977	1.956	1.956
<b>Panel</b>																					
.0000	2.630	2.092	2.788	2.903	3.019	3.076	2.914	2.790	2.552	2.510	.0250	.976	.957	.970	.924	.812	.743	.649	.624	.569	.574
.0250	3.751	3.074	3.731	4.119	4.442	4.498	4.181	3.870	3.226	3.014	.0500	.958	.929	.950	.918	.812	.725	.648	.620	.564	.574
.0500	3.778	3.043	3.863	4.159	4.523	4.547	4.205	3.829	3.068	2.787	.1000	.921	.905	.938	.923	.815	.735	.631	.600	.576	.576
.1000	3.763	3.615	3.760	4.141	4.557	4.529	4.101	3.663	2.801	2.463	.1500	.982	.143	.143	.143	.098	.003	.024	.044	.044	.026
.1500	3.672	3.745	3.669	4.080	4.539	4.581	4.128	3.651	2.731	2.364	.2000	.127	.102	.204	.137	.052	.006	.035	.026	.000	.000
.2000	3.621	3.708	4.612	4.040	4.575	4.715	4.262	3.773	2.839	2.463	.3000	.058	.055	.134	.076	.082	.015	.030	.047	.053	.041
.3000	2.927	3.062	2.900	3.293	3.908	4.213	3.845	3.435	2.622	2.300	.4000	.092	.040	.062	.049	.021	.039	.051	.050	.050	.050
.4000	2.300	2.462	2.277	2.604	3.145	3.679	3.461	3.128	2.475	2.221	.5000	.048	.055	.098	.049	.029	.032	.043	.051	.050	.050
.6000	1.818	1.966	1.627	2.058	2.474	3.150	3.062	2.844	2.341	2.166	.6000	.059	.077	.078	.058	.039	.025	.077	.066	.079	.080
.7000	1.597	1.680	1.629	1.781	2.025	2.650	2.693	2.568	2.217	2.105	.7000	.094	.117	.122	.082	.098	.101	.113	.113	.117	.128
.8000	1.533	1.571	1.575	1.704	1.871	2.269	2.383	2.341	2.112	2.055	.8000	.179	.182	.182	.171	.218	.205	.214	.210	.208	.236
.9000	1.536	1.575	1.566	1.698	1.840	1.994	2.098	2.107	1.992	1.985	.9200	.970	.612	.554	.598	.680	.657	.643	.657	.714	.648
<b>Spoiler</b>																					
.2443	.928	.954	1.015	1.044	.969	.690	.651	.635	.404	1.487	.4000	.127	.145	.204	.189	.163	.128	.155	.157	.155	.178
.4906	.921	.951	1.040	1.075	.964	.818	.861	.926	1.281	1.475	.4500	.085	.025	.012	.037	.018	.000	.009	.033	.041	.035
.7370	.871	.951	1.015	1.062	.957	.887	.794	1.100	1.410	1.564	.4500	.036	.080	.058	.030	.043	.021	.012	.082	.038	.030
.9856	1.129	1.329	1.366	1.435	1.375	1.310	1.613	1.684	1.744	1.784	.4500	.073	.092	.084	.043	.049	.034	.026	.056	.044	.044
											.5000	.124	.124	.147	.124	.107	.079	.071	.080	.045	.087
											.5500	.256	.244	.274	.252	.207	.174	.171	.169	.141	.260
											.6000	.379	.400	.340	.323	.320	.250	.258	.254	.254	.271
											.6500	.612	.603	.556	.537	.502	.491	.482	.453	.469	.469
											.7000	.753	.775	.744	.729	.709	.684	.689	.680	.687	.687

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TABLE XIII. - PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(e) Double slotted flap configuration with spoiler;  $h_s = -0.100\bar{c}$ ;  $\frac{V}{b/2} = 0.55$

Upper surface													Lower surface																				
$x/c$	$C_p$ for -												$x/c$	$C_p$ for -																			
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$			
Wing																																	
0.0000	1.008	1.082	1.088	2.065	2.090	2.100	2.144	2.132	2.137	2.072	0.012	1.008	1.082	1.088	2.065	2.090	2.100	2.144	2.132	2.137	2.072	0.012	1.008	1.082	1.088	2.065	2.090	2.100	2.144	2.132	2.137		
0.1250	1.198	1.087	2.010	2.026	2.047	2.059	2.054	2.054	2.059	2.058	0.025	1.198	1.087	2.010	2.026	2.047	2.059	2.054	2.054	2.059	2.058	2.023	1.025	1.198	1.087	2.010	2.026	2.047	2.059	2.054	2.054	2.059	2.058
0.2500	1.136	1.087	1.982	2.099	2.193	2.333	2.233	2.233	2.233	2.023	0.050	1.136	1.087	1.982	2.099	2.193	2.333	2.233	2.233	2.233	2.023	0.050	1.070	1.072	2.029	2.029	2.029	2.029	2.029	2.029	2.029	2.029	2.029
0.3750	1.130	1.076	2.018	2.357	2.414	2.410	2.264	2.224	2.178	2.026	0.075	1.130	1.076	2.018	2.357	2.414	2.410	2.264	2.224	2.178	2.026	0.075	1.070	1.072	2.029	2.029	2.029	2.029	2.029	2.029	2.029	2.029	2.029
1.0000	1.142	1.050	2.156	2.454	2.459	2.382	2.293	2.221	2.145	2.026	1.000	1.142	1.050	2.156	2.454	2.459	2.382	2.293	2.221	2.145	2.026	1.000	1.096	1.142	2.029	2.029	2.029	2.029	2.029	2.029	2.029	2.029	2.029
1.1250	1.164	1.086	2.497	2.497	2.381	2.379	2.311	2.221	2.145	2.026	1.1250	1.164	1.086	2.497	2.497	2.381	2.379	2.311	2.221	2.145	2.026	1.1250	1.164	1.086	2.497	2.497	2.381	2.379	2.311	2.221	2.145	2.026	
2.0000	1.192	1.343	2.650	2.741	2.350	2.382	2.290	2.212	2.142	2.026	2.0000	1.192	1.343	2.650	2.741	2.350	2.382	2.290	2.212	2.142	2.026	2.0000	1.192	1.343	2.650	2.741	2.350	2.382	2.290	2.212	2.142	2.026	
2.1250	1.214	1.352	2.470	3.008	2.483	2.401	2.302	2.215	2.145	2.026	2.1250	1.214	1.352	2.470	3.008	2.483	2.401	2.302	2.215	2.145	2.026	2.1250	1.214	1.352	2.470	3.008	2.483	2.401	2.302	2.215	2.145	2.026	
3.0000	1.224	1.364	2.021	3.339	2.719	2.425	2.284	2.206	2.145	2.026	3.0000	1.224	1.364	2.021	3.339	2.719	2.425	2.284	2.206	2.145	2.026	3.0000	1.224	1.364	2.021	3.339	2.719	2.425	2.284	2.206	2.145	2.026	
3.1250	1.237	1.343	1.571	2.036	2.743	2.404	2.248	2.179	2.119	2.026	3.1250	1.237	1.343	1.571	2.036	2.743	2.404	2.248	2.179	2.119	2.026	3.1250	1.237	1.343	1.571	2.036	2.743	2.404	2.248	2.179	2.119	2.026	
4.0000	1.242	1.364	1.625	2.365	2.610	2.190	2.096	2.137	2.078	2.026	4.0000	1.242	1.364	1.625	2.365	2.610	2.190	2.096	2.137	2.078	2.026	4.0000	1.242	1.364	1.625	2.365	2.610	2.190	2.096	2.137	2.078	2.026	
4.1250	1.246	1.359	1.545	2.251	2.431	2.187	2.074	2.025	1.965	2.026	4.1250	1.246	1.359	1.545	2.251	2.431	2.187	2.074	2.025	1.965	2.026	4.1250	1.246	1.359	1.545	2.251	2.431	2.187	2.074	2.025	1.965	2.026	
5.0000	1.253	1.353	1.405	2.041	2.284	2.130	2.025	1.965	1.906	2.026	5.0000	1.253	1.353	1.405	2.041	2.284	2.130	2.025	1.965	1.906	2.026	5.0000	1.253	1.353	1.405	2.041	2.284	2.130	2.025	1.965	1.906	2.026	
5.1250	1.257	1.354	1.406	2.042	2.285	2.131	2.025	1.965	1.907	2.026	5.1250	1.257	1.354	1.406	2.042	2.285	2.131	2.025	1.965	1.907	2.026	5.1250	1.257	1.354	1.406	2.042	2.285	2.131	2.025	1.965	1.907	2.026	
6.0000	1.260	1.354	1.406	2.042	2.285	2.131	2.025	1.965	1.907	2.026	6.0000	1.260	1.354	1.406	2.042	2.285	2.131	2.025	1.965	1.907	2.026	6.0000	1.260	1.354	1.406	2.042	2.285	2.131	2.025	1.965	1.907	2.026	
6.1250	1.264	1.354	1.406	2.042	2.285	2.131	2.025	1.965	1.907	2.026	6.1250	1.264	1.354	1.406	2.042	2.285	2.131	2.025	1.965	1.907	2.026	6.1250	1.264	1.354	1.406	2.042	2.285	2.131	2.025	1.965	1.907	2.026	
7.0000	1.268	1.354	1.406	2.042	2.285	2.131	2.025	1.965	1.907	2.026	7.0000	1.268	1.354	1.406	2.042	2.285	2.131	2.025	1.965	1.907	2.026	7.0000	1.268	1.354	1.406	2.042	2.285	2.131	2.025	1.965	1.907	2.026	
7.1250	1.272	1.354	1.406	2.042	2.285	2.131	2.025	1.965	1.907	2.026	7.1250	1.272	1.354	1.406	2.042	2.285	2.131	2.025	1.965	1.907	2.026	7.1250	1.272	1.354	1.406	2.042	2.285	2.131	2.025	1.965	1.907	2.026	
8.0000	1.275	1.354	1.406	2.042	2.285	2.131	2.025	1.965	1.907	2.026	8.0000	1.275	1.354	1.406	2.042	2.285	2.131	2.025	1.965	1.907	2.026	8.0000	1.275	1.354	1.406	2.042	2.285	2.131	2.025	1.965	1.907	2.026	
8.1250	1.279	1.354	1.406	2.042	2.285	2.131	2.025	1.965	1.907	2.026	8.1250	1.279	1.354	1.406	2.042	2.285	2.131	2.025	1.965	1.907	2.026	8.1250	1.279	1.354	1.406	2.042	2.285	2.131	2.025	1.965	1.907	2.026	
9.0000	1.282	1.354	1.406	2.042	2.285	2.131	2.025	1.965	1.907	2.026	9.0000	1.282	1.354	1.406	2.042	2.285	2.131	2.025	1.965	1.907	2.026	9.0000	1.282	1.354	1.406	2.042	2.285	2.131	2.025	1.965	1.907	2.026	
9.1250	1.286	1.354	1.406	2.042	2.285	2.131	2.025	1.965	1.907	2.026	9.1250	1.286	1.354	1.406	2.042	2.285	2.131	2.025	1.965	1.907	2.026	9.1250	1.286	1.354	1.406	2.042	2.285	2.131	2.025	1.965	1.907	2.026	
10.0000	1.290	1.354	1.406	2.042	2.285	2.131	2.025	1.965	1.907	2.026	10.0000	1.290	1.354	1.406	2.042	2.285	2.131	2.025	1.965	1.907	2.026	10.0000	1.290	1.354	1.406	2.042	2.285	2.131	2.025	1.965	1.907	2.026	
10.1250	1.294	1.354	1.406	2.042	2.285	2.131	2.025	1.965	1.907	2.026	10.1250	1.294	1.354	1.406	2.042	2.285	2.131	2.025	1.965	1.907	2.026	10.1250	1.294	1.354	1.406	2.042	2.285	2.131	2.025	1.965	1.907	2.026	
11.0000	1.303	1.354	1.406	2.042	2.285	2.131	2.025	1.965	1.907	2.026	11.0000	1.303	1.354	1.406	2.042	2.285	2.131	2.025	1.965	1.907	2.026	11.0000	1.303	1.354	1.406	2.042	2.285	2.131	2.025	1.965	1.907	2.026	
11.1250	1.307	1.354	1.406	2.042	2.285	2.131	2.025	1.965	1.907	2.026	11.1250	1.307	1.354	1.406	2.042	2.285	2.131	2.025	1.965	1.907	2.026	11.1250	1.307	1.354	1.406	2.042	2.285	2.131	2.025	1.965	1.907	2.026	
12.0000	1.310	1.354	1.406	2.042	2.285	2.131	2.025	1.965	1.907	2.026	12.0000	1.310	1.354	1.406	2.042	2.285	2.131	2.025	1.965	1.907	2.026	12.0000	1.310	1.354	1.406	2.042	2.285	2.131	2.025	1.965	1.907	2.026	
12.1250	1.314	1.354	1.																														

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TABLE XIII. - PRESSURE COEFFICIENT  $C_p$  AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Concluded

(f) Double slotted flap configuration with spoiler;  $h_S = -0.100\bar{c}$ ;  $\frac{V}{b/2} = 0.72$

z/c	Upper surface									
	$C_p$ for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 28^\circ$
<b>Wing</b>										
.0000	2.079	2.105	2.223	2.192	2.042	1.928	1.846	1.820	1.862	1.897
.0125	1.877	2.077	2.150	2.111	2.003	1.872	1.838	1.852	1.890	
.0250	1.978	2.098	2.174	2.131	2.019	1.907	1.855	1.841	1.858	1.857
.0500	1.432	2.145	2.207	2.172	2.028	1.937	1.870	1.853	1.874	1.897
.0750	1.382	2.194	2.239	2.192	2.046	1.955	1.883	1.864	1.874	1.891
.1000	1.303	2.200	2.244	2.192	2.040	1.931	1.876	1.854	1.874	1.891
.1300	1.306	2.354	2.317	2.212	2.040	1.944	1.897	1.876	1.874	1.900
.2000	1.333	2.454	2.378	2.212	2.037	1.967	1.914	1.876	1.880	1.903
.2500	1.353	2.422	2.473	2.172	2.046	1.967	1.906	1.873	1.880	1.903
.3000	1.357	2.105	2.643	2.172	2.049	1.961	1.897	1.870	1.889	1.918
.3500	1.388	1.779	2.617	2.151	2.046	1.944	1.888	1.873	1.889	1.920
.4000	1.404	1.507	2.681	2.232	2.040	1.943	1.885	1.864	1.889	1.920
.4500	1.442	1.678	2.900	2.311	2.040	1.943	1.885	1.864	1.889	1.920
.5000	1.454	1.371	2.450	2.414	2.052	1.940	1.873	1.861	1.898	1.923
.5500	1.467	1.378	2.655	2.383	2.040	1.928	1.861	1.864	1.898	1.912
.6000	1.454	1.409	2.397	2.293	2.028	1.899	1.858	1.856	1.892	1.920
.6500	1.436	1.391	2.131	2.151	1.982	1.892	1.855	1.859	1.898	1.935
.7500	1.376	1.357	1.802	2.030	1.948	1.863	1.846	1.853	1.907	1.936
.8000	1.331	1.305	1.671	1.990	1.914	1.857	1.841	1.856	1.910	1.977
.8500	1.312	1.292	1.546	1.949	1.877	1.830	1.829	1.864	1.943	1.994
.9000	1.297	1.305	1.436	1.848	1.859	1.836	1.840	1.887	1.949	2.009

z/c	Lower surface									
	$C_p$ for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 28^\circ$
<b>Wing</b>										
.0125	.760	.812	.954	1.041	1.149	1.209	1.263	1.292	1.365	1.419
.0250	.782	.751	.814	.879	.957	.991	1.047	1.074	1.129	1.177
.0500	.792	.723	.726	.778	.803	.837	.858	.894	.947	
.0750	.776	.729	.698	.717	.726	.731	.755	.773	.799	.835
.1000	.773	.726	.695	.697	.695	.693	.707	.720	.746	.770
.1500	.773	.729	.692	.677	.645	.651	.654	.664	.683	.699
.2000	.782	.732	.704	.677	.668	.648	.654	.659	.662	.690
.2500	.788	.732	.698	.697	.671	.657	.660	.661	.674	
.3000	.792	.751	.713	.717	.695	.678	.681	.690	.698	.711
.3500	.823	.797	.774	.758	.740	.734	.737	.743	.763	.787
.4000	.909	.868	.851	.859	.840	.821	.823	.829	.844	.870
.4500	1.003	.966	.960	.960	.955	.945	.943	.945	.955	.975
.5000	1.117	1.172	1.180	1.191	1.194	1.042	1.056	1.062	1.084	1.100
.5500	1.193	1.172	1.180	1.222	1.191	1.179	1.213	1.238	1.260	1.283
.6000	1.253	1.265	1.271	1.323	1.326	1.340	1.385	1.410	1.470	1.505
.6500	1.328	1.357	1.384	1.444	1.468	1.513	1.571	1.623	1.683	1.723
.7000	1.351	1.397	1.451	1.525	1.572	1.609	1.646	1.705	1.760	1.800
.7500	1.364	1.415	1.494	1.586	1.585	1.609	1.648	1.687	1.748	1.791
.8000	1.335	1.397	1.476	1.505	1.517	1.534	1.577	1.608	1.680	1.732
.8500	1.309	1.382	1.494	1.543	1.477	1.495	1.530	1.561	1.632	1.684
.9000	1.316	1.397	1.513	1.545	1.477	1.475	1.506	1.534	1.608	1.658
.9500	1.335	1.403	1.506	1.586	1.529	1.519	1.548	1.569	1.644	1.684

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TABLE XIV.- INTEGRATED SECTION DATA FOR THE PLAIN WING CONFIGURATION  
WITH SPOILERS

(a)  $h_s = -0.005\bar{c}$ (b)  $h_s = -0.010\bar{c}$ 

$\frac{y}{b/2}$	$\alpha, \text{deg}$	$c_{n,w}$	$c_{m,w}$	$c_{n,s}$	$c_{m,s}$	$\frac{y}{b/2}$	$\alpha, \text{deg}$	$c_{n,w}$	$c_{m,w}$	$c_{n,s}$	$c_{m,s}$
0.21	-4	-0.1504	.0206	.0334	.0148	0.21	-4	-0.1940	.0435	-0.2783	.1414
0.21	0	-0.0285	.0128	.0046	-.0044	0.21	0	-0.0581	.0319	-0.2944	.1368
0.21	4	-0.1132	.0024	.0997	-.0116	0.21	4	.0577	.0211	-0.2758	.1199
0.21	8	-0.2968	-.0089	.0913	-.0092	0.21	8	.0752	.0144	-0.2358	.1039
0.21	12	.5405	.0003	.0715	-.0040	0.21	12	.0525	.0172	-0.2772	.0872
0.21	16	.8248	.0058	.1221	-.0164	0.21	16	.0311	.0280	-0.2525	.1164
0.21	20	1.1683	.0130	.0922	.0006	0.21	20	1.1321	.0356	-0.3025	.1405
0.21	22	1.3329	.0127	.0819	.0008	0.21	22	1.2909	.0374	-0.3350	.1542
0.21	24	1.4502	-.0174	.0869	.0009	0.21	24	1.3838	.0011	-0.3306	.1541
0.21	26	1.3531	-.0965	.1008	-.0114	0.21	26	1.3296	-.0750	-0.2532	.1206
0.30	-4	-.2057	.0329	.0564	.0100	0.30	-4	-.2384	.0543	-0.2129	.0967
0.30	0	-.0643	.0306	.0835	.0013	0.30	0	-0.1076	.0527	-0.2694	.1117
0.30	4	.0959	.0251	.0768	.0002	0.30	4	.0520	.0051	-0.2536	.0938
0.30	8	.3206	.0356	.0801	.0043	0.30	8	.2916	.0689	-0.2135	.0974
0.30	12	.6042	.0358	.0700	.0044	0.30	12	.5611	.0593	-0.1866	.0758
0.30	16	.9516	.0232	.0887	.0000	0.30	16	.9251	.0513	-0.2285	.0949
0.30	20	1.3263	-.0328	.1425	-.0234	0.30	20	1.2753	-.0053	-0.1760	.0729
0.30	22	1.4708	-.0675	.1419	-.0258	0.30	22	1.4127	-.0434	-0.1365	.0573
0.30	24	1.5753	-.1102	.1567	-.0239	0.30	24	1.5022	-.0891	-0.1344	.0604
0.30	26	1.4951	-.1684	.1238	-.0184	0.30	26	1.4756	-.1446	-.0677	.0324
0.43	-4	-.2332	.0132	.1006	-.0050	0.43	-4	-.2552	.0280	-0.1834	.0864
0.43	0	-.0471	.0078	.1180	-.0070	0.43	0	-.0895	.0308	-0.2810	.1242
0.43	4	.1324	.0100	.1116	-.0106	0.43	4	.1103	.0284	-0.2741	.1190
0.43	8	.4037	.0109	.1152	-.0112	0.43	8	.3798	.0369	-0.2705	.1163
0.43	12	.7702	-.0122	.1232	-.0154	0.43	12	.7227	.0226	-0.2782	.1198
0.43	16	1.1398	-.0836	.0945	-.0100	0.43	16	1.1200	-.0615	-0.2202	.0947
0.43	20	1.3795	-.1673	.1259	-.0248	0.43	20	1.3316	-.1451	-0.1363	.0645
0.43	22	1.4484	-.1961	.1516	-.0422	0.43	22	1.4089	-.1761	-.0659	.0409
0.43	24	1.5186	-.2242	.1370	-.0398	0.43	24	1.4467	-.2117	-.0017	.0126
0.43	26	1.4793	-.2426	.1340	-.0418	0.43	26	1.4934	-.2469	.0420	-.0019
0.55	-4	-.2979	.0306	.0743	-.0037	0.55	-4	-.3460	.0489	-0.1702	.0635
0.55	0	-.0758	.0243	.0683	.0099	0.55	0	-.1317	.0491	-0.2466	.0982
0.55	4	.1289	.0189	.0700	-.0069	0.55	4	.0797	.0431	-0.2347	.0979
0.55	8	.5023	.0028	.0626	.0169	0.55	8	.4620	.0323	-0.2865	.1102
0.55	12	.9042	-.0973	.0183	-.0369	0.55	12	.8435	-.0650	-0.3364	.1532
0.55	16	1.0851	-.1641	.0589	-.0030	0.55	16	1.0572	-.1413	-0.1847	.0894
0.55	20	1.1754	-.1772	.0887	-.0202	0.55	20	1.1406	-.1762	-.0406	.0276
0.55	22	1.2431	-.2029	.1058	-.0202	0.55	22	1.1927	-.1896	-.0206	.0217
0.55	24	1.2634	-.2180	.1184	-.0378	0.55	24	1.2193	-.2053	-.0191	.0261
0.55	26	1.2222	-.2198	.0965	-.0329	0.55	26	1.2256	-.2222	-.0414	.0409
0.72	-4	-.2353	-.0083			0.72	-4	-.2412	-.0047		
0.72	0	-.0244	-.0034			0.72	0	-.0903	.0013		
0.72	4	.2458	-.0067			0.72	4	.2060	-.0021		
0.72	8	.6912	-.1405			0.72	8	.6480	-.1298		
0.72	12	.7332	-.1620			0.72	12	.6927	-.1308		
0.72	16	.8281	-.1637			0.72	16	.7992	-.1441		
0.72	20	.8879	-.1667			0.72	20	.8740	-.1622		
0.72	22	.9031	-.1704			0.72	22	.9057	-.1676		
0.72	24	.9449	-.1798			0.72	24	.9319	-.1734		
0.72	28	.9504	-.1840			0.72	28	.9096	-.1786		

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TABLE XIV.- INTEGRATED SECTION DATA FOR THE PLAIN WING CONFIGURATION  
WITH SPOILERS - Continued

(c)  $h_s = -0.035\bar{c}$ (d)  $h_s = -0.050\bar{c}$ 

$\frac{y}{b/2}$	$a, \text{deg}$	$c_{n,w}$	$c_{m,w}$	$c_{n,s}$	$c_{m,s}$	$\frac{y}{b/2}$	$a, \text{deg}$	$c_{n,w}$	$c_{m,w}$	$c_{n,s}$	$c_{m,s}$
0.21	-4	-0.2301	.0521	-0.8279	.2771	0.21	-4	-0.2888	.0987	-0.7428	.3243
0.21	0	-0.1162	.0547	-0.6355	.2481	0.21	0	-0.1788	.0918	-0.7679	.3230
0.21	4	.0264	.0577	-0.6384	.2665	0.21	4	-0.0306	.0817	-0.7660	.3160
0.21	8	.1956	.0556	-0.6731	.2819	0.21	8	.1437	.0752	-0.8149	.3384
0.21	12	.2229	.0510	-0.6812	.2867	0.21	12	.3817	.0745	-0.8224	.3478
0.21	16	.6668	.0723	-0.7350	.3151	0.21	16	.6461	.0876	-0.9058	.3829
0.21	20	1.0372	.0747	-0.6485	.3711	0.21	20	1.0020	.0907	-1.0400	.4541
0.21	22	1.1876	.0779	-0.7209	.4095	0.21	22	1.1539	.0865	-1.0723	.4779
0.21	24	1.2801	.0774	-0.6876	.3873	0.21	24	1.2275	.0322	-1.0749	.4811
0.21	26	1.2859	-.0060	-0.6904	.3203	0.21	26	1.1906	-.0721	-0.9112	.4214
0.30	-4	-.3569	.1099	-0.5872	.2379	0.30	-4	-.3697	.1200	-0.7245	.3014
0.30	0	-0.2127	.1057	-0.6316	.2503	0.30	0	-0.2660	.1310	-0.7980	.3213
0.30	4	-.0458	.1044	-0.6242	.2455	0.30	4	-.0916	.1203	-0.7853	.3160
0.30	8	.1918	.1113	-0.6549	.2614	0.30	8	.1203	.1295	-0.8564	.3472
0.30	12	.4473	.1069	-0.6797	.2763	0.30	12	.4084	.1259	-0.8335	.3425
0.30	16	.7812	.0997	-0.7100	.2910	0.30	16	.7371	.1182	-0.8884	.3646
0.30	20	1.1372	.0933	-0.6731	.2752	0.30	20	1.1007	.0566	-0.8279	.3331
0.30	22	1.2099	-.0059	-0.6731	.2585	0.30	22	1.2310	.0511	-0.8152	.3119
0.30	24	1.3997	-.0638	-0.6358	.2358	0.30	24	1.3369	-.0439	-0.7402	.3146
0.30	26	1.3925	-.1265	-0.4555	.1749	0.30	26	1.2360	-.1141	-0.5820	.2947
0.43	-4	-.3951	.0845	-0.5361	.2180	0.43	-4	-.4346	.1048	-0.6828	.2834
0.43	0	-0.2138	.0845	-0.6068	.2419	0.43	0	-0.2656	.1107	-0.7813	.3144
0.43	4	-.0082	.0834	-0.6024	.2383	0.43	4	-.0690	.1065	-0.7890	.3156
0.43	8	.2615	.0918	-0.6573	.2598	0.43	8	.1960	.1123	-0.8483	.3389
0.43	12	.5931	.0711	-0.7588	.3063	0.43	12	.5263	.0958	-0.9088	.3667
0.43	16	.911	-.0104	-0.7286	.3025	0.43	16	.6986	.0083	-0.8938	.3736
0.43	20	1.1268	-.0944	-0.6109	.2520	0.43	20	1.1313	-.0768	-0.7223	.2977
0.43	22	1.2879	-.1411	-0.3876	.1592	0.43	22	1.2314	-.1221	-0.5268	.2161
0.43	24	1.3621	-.2078	-0.2494	.1131	0.43	24	1.3081	-.1874	-0.3939	.1822
0.43	26	1.4132	-.2267	-0.1348	.0634	0.43	26	1.2690	-.2016	-0.2148	.0940
0.55	-4	-.5228	.1175	-0.5353	.2071	0.55	-4	-.5662	.1358	-0.6785	.2480
0.55	0	-0.2797	.1132	-0.6100	.2434	0.55	0	-.3544	.1441	-0.8107	.3238
0.55	4	-.0616	.1073	-0.6141	.2484	0.55	4	-.1276	.1272	-0.7839	.3199
0.55	8	.3069	.0978	-0.7095	.2888	0.55	8	.2226	.1221	-0.9040	.3496
0.55	12	.6404	.0066	-0.7500	.3129	0.55	12	.5869	.0227	-0.8087	.3385
0.55	16	.8574	-.0786	-0.5012	.2142	0.55	16	.6212	-.0641	-0.6101	.2602
0.55	20	.9788	-.1228	-0.3051	.1245	0.55	20	.9410	-.1044	-0.3988	.1658
0.55	22	1.1429	-.1459	-0.1883	.0744	0.55	22	1.0095	-.1291	-0.2985	.1228
0.55	24	1.1121	-.1758	-0.1155	.0470	0.55	24	1.0692	-.1579	-0.2347	.0987
0.55	26	1.1704	-.2009	-0.0284	.0138	0.55	26	1.0559	-.1782	-0.1040	.0448
0.72	-4	-.3671	.0061			0.72	-4	-.4121	.0037		
0.72	0	-.1319	.0154			0.72	0	-.1855	.0188		
0.72	4	.1486	.0098			0.72	4	.0898	.0138		
0.72	8	.4954	-.0967			0.72	8	.4506	-.0950		
0.72	12	.5808	-.1075			0.72	12	.5287	-.1004		
0.72	16	.6531	-.1229			0.72	16	.6401	-.1185		
0.72	20	.7826	-.1425			0.72	20	.7498	-.1378		
0.72	22	.8253	-.1539			0.72	22	.7627	-.1371		
0.72	24	.8465	-.1591			0.72	24	.8122	-.1500		
0.72	26	.8603	-.1490			0.72	26	.8442	-.1886		

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TABLE XIV.- INTEGRATED SECTION DATA FOR THE PLAIN  
WING CONFIGURATION WITH SPOILERS - Concluded

(e)  $h_s = -0.100c$ 

$\frac{y}{c}$	$\alpha, \deg$	$c_{D,w}$	$c_{m,w}$	$c_{D,s}$	$c_{m,s}$
0.21	-4	-0.3366	+0.1057	-1.0885	.5536
0.21	0	-0.2417	+0.0688	-1.0201	.5177
0.21	4	-0.1004	+0.0991	-0.8430	.4163
0.21	8	.0799	+0.0842	-0.7946	.3860
0.21	12	.0374	+0.0787	-0.8843	.4252
0.21	16	.0031	+0.0783	-1.0149	.4943
0.21	20	.0026	+0.0726	-1.0439	.5932
0.21	24	1.1394	+0.0573	-1.2751	.6030
0.21	28	1.1593	-0.0284	-1.2244	.6245
0.21	36	1.2192	-0.1362	-1.1827	.6029
0.30	-4	-0.4522	+0.1350	-1.2492	.6183
0.30	0	-0.3417	+0.1475	-1.2057	.6012
0.30	4	-0.1741	+0.1398	-1.0644	.5327
0.30	8	.0497	+0.1389	-1.1048	.5592
0.30	12	.0359	+0.1235	-1.2999	.6387
0.30	16	.0092	+0.1248	-1.2872	.6311
0.30	20	.0082	+0.1247	-1.2857	.6311
0.30	24	1.1348	+0.0133	-1.0849	.5171
0.30	28	1.2268	-0.0670	-0.8685	.4239
0.30	36	1.2097	-0.1482	-0.7070	.3487
0.43	-4	-0.5344	+0.1130	-1.1704	.5793
0.43	0	-0.3507	+0.1202	-1.1515	.5726
0.43	4	-0.1639	+0.1202	-1.1395	.5852
0.43	8	.0848	+0.1198	-1.1415	.5979
0.43	12	.0454	+0.1045	-1.1444	.6077
0.43	16	.0006	+0.0945	-1.1310	.5573
0.43	20	.0001	+0.0937	-0.9025	.4214
0.43	24	1.0829	-0.0986	-0.7070	.3352
0.43	28	1.1769	-0.1569	-0.4150	.2094
0.43	36	1.2262	-0.2057	-0.2510	.1335
0.55	-4	-0.6972	+0.1393	-1.2871	.6994
0.55	0	-0.4231	+0.1231	-1.1883	.6455
0.55	4	-0.2097	+0.1217	-1.2301	.6133
0.55	8	.0025	+0.1071	-1.2081	.6494
0.55	12	.0003	+0.1034	-0.8996	.4460
0.55	16	.0010	+0.0427	-0.6083	.2960
0.55	20	.0001	+0.0495	-0.5086	.2377
0.55	24	.0126	-0.0836	-0.4217	.2054
0.55	28	.0156	-0.1265	-0.2095	.1146
0.55	36	1.0040	-0.1627	-0.0862	.0550
0.72	-4	-0.5809	-0.0336		
0.72	0	-0.3485	-0.0114		
0.72	4	-0.1600	-0.0030		
0.72	8	.1192	-0.0499		
0.72	12	.2673	-0.0589		
0.72	16	.3607	-0.0716		
0.72	20	.5173	-0.0894		
0.72	24	.5838	-0.1005		
0.72	28	.6435	-0.1184		
0.72	36	.7452	-0.1360		

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TABLE XV.- INTEGRATED SECTION DATA FOR THE DOUBLE SLOTTED FLAP  
CONFIGURATION WITH SPOILERS

(a)  $h_S = -0.005\bar{c}$ 

$\frac{y}{b/2}$	$\alpha$ , deg	$c_{n,w}$	$c_{m,w}$	$c_{n,v}$	$c_{m,v}$	$c_{n,f}$	$c_{m,f}$	$c_{n,s}$	$c_{m,s}$
0.21	-4	.5357	-.2699	2.2330	-.9632	1.2170	-.4799	-.0547	.0111
0.21	0	.6811	-.2816	2.2912	-.9786	1.2241	-.4742	-.0423	.0063
0.21	4	.8312	-.2830	2.2851	-.9778	1.2048	-.4658	-.0200	-.0020
0.21	8	.9772	-.2724	2.3015	-.9814	1.1998	-.4678	-.0111	-.0031
0.21	12	1.2117	-.2632	2.6108	-.1.090	1.2818	-.4999	-.0113	-.0135
0.21	16	1.4687	-.2583	2.8901	-.1.2035	1.3762	-.5461	-.0248	-.0206
0.21	20	1.6885	-.2582	3.2000	-.1.2355	1.5546	-.6074	-.0306	-.0254
0.21	22	1.8154	-.2474	3.4732	-.1.4513	1.6856	-.6392	-.0499	-.0371
0.21	24	1.6498	-.3412	2.9937	-.1.3063	1.4703	-.6114	-.0228	-.0457
0.21	26	1.6744	-.3876	2.6177	-.1.1574	1.4568	-.6288	-.0226	-.0400
0.30	-4	.7212	-.3841	4.0150	-.1.5954	1.3751	-.5461	-.0475	-.0028
0.30	0	.8461	-.3632	3.6283	-.1.5225	1.3036	-.5567	-.0336	-.0064
0.30	4	1.0415	-.3488	3.4699	-.1.4540	1.3768	-.5601	-.0256	-.0045
0.30	8	1.2397	-.3239	3.6559	-.1.4312	1.3310	-.5302	-.0013	-.0110
0.30	12	1.4449	-.3073	3.6106	-.1.4187	1.3502	-.5320	-.0134	-.0155
0.30	16	1.7442	-.3263	3.9773	-.1.6010	1.4624	-.5651	-.0320	-.0405
0.30	20	1.9404	-.3610	4.0247	-.1.6382	1.5225	-.5834	-.0334	-.0348
0.30	22	2.0392	-.3886	3.9243	-.1.6135	1.5863	-.5956	-.0061	-.0325
0.30	24	1.8173	-.3926	3.1353	-.1.3403	1.4963	-.5675	-.0345	-.0217
0.30	26	1.6296	-.3915	2.4111	-.1.0739	1.4135	-.5410	-.0369	-.0073
0.43	-4	.8998	-.4138	4.9685	-.2.0027	1.3670	-.4569	-.2012	.0527
0.43	0	1.1475	-.4437	5.0297	-.2.0263	1.5191	-.5330	-.1983	.0533
0.43	4	1.3359	-.4187	4.9931	-.2.0165	1.5084	-.5344	-.1638	.0399
0.43	8	1.5538	-.3981	4.9858	-.2.0157	1.5166	-.5393	-.0805	.0147
0.43	12	1.8738	-.4123	4.8640	-.1.9796	1.5920	-.5855	-.0254	-.0042
0.43	16	1.9661	-.4347	4.1078	-.1.7035	1.5870	-.5953	-.0121	-.0229
0.43	20	1.9805	-.4463	3.6856	-.1.5472	1.6559	-.6436	-.0163	-.0256
0.43	22	1.8147	-.4357	4.1267	-.1.4240	1.5867	-.6070	-.0144	-.0147
0.43	24	1.6640	-.4051	2.2346	-.1.9076	1.6233	-.6342	-.0176	-.0071
0.43	26	1.6556	-.4264	2.1596	-.9887	1.6210	-.6877	-.0097	-.0102
0.55	-4	1.1603	-.5517	5.0218	-.2.0514	1.8061	-.7200	-.1040	.0155
0.55	0	1.3413	-.5471	4.9493	-.2.0198	1.6290	-.7338	-.1109	.0181
0.55	4	1.6827	-.5349	5.0699	-.2.0828	1.8570	-.7469	-.1045	.0160
0.55	8	1.7694	-.5393	4.7364	-.1.9722	1.8548	-.7325	-.0512	-.0345
0.55	12	1.9315	-.5238	3.5293	-.1.4830	1.7485	-.7078	-.1498	-.0688
0.55	16	1.7765	-.4769	2.8920	-.1.2046	1.6262	-.6680	-.1424	-.0610
0.55	20	1.6225	-.4176	2.5168	-.1.0564	1.3700	-.5084	-.1337	-.0594
0.55	22	1.6386	-.4465	2.4144	-.1.0434	1.5568	-.6405	-.1298	-.0592
0.55	24	1.3573	-.4200	2.0758	-.9131	1.4490	-.6251	-.0958	-.0469
0.55	26	1.4952	-.4232	1.9677	-.8807	1.4923	-.6336	-.0886	-.0422
0.72	-4	.1044	.1359						
0.72	0	.3511	.1698						
0.72	4	.8965	.0176						
0.72	8	.8967	.0497						
0.72	12	.8697	.0789						
0.72	16	.7708	.0657						
0.72	20	.7405	.0633						
0.72	22	.7084	.0587						
0.72	24	.6577	.0481						
0.72	26	.6726	.0754						

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TABLE XV.- INTEGRATED SECTION DATA FOR THE DOUBLE SLOTTED FLAP  
CONFIGURATION WITH SPOILERS - Continued

(b)  $h_s = -0.010\bar{c}$ 

$\frac{Y}{b/2}$	$a, \text{deg}$	$c_{n,w}$	$c_{m,w}$	$c_{n,v}$	$c_{m,v}$	$c_{n,f}$	$c_{m,f}$	$c_{n,s}$	$c_{m,s}$
0.21	-4	.5396	-.2726	2.3037	-.9915	1.2367	-.4880	-.0726	.0617
0.21	0	.6920	-.2869	2.2519	-.9662	1.2933	-.5272	.0105	.0094
0.21	4	.8184	-.2840	2.2464	-.9609	1.2700	-.5237	.0333	-.0065
0.21	8	1.0255	-.2599	2.3595	-.1.0091	1.3379	-.5619	.0261	-.0031
0.21	12	1.2050	-.2656	2.5069	-.1.0650	1.3510	-.5673	.0359	-.0064
0.21	16	1.4508	-.2645	2.5050	-.1.0622	1.3529	-.5642	.0531	-.0094
0.21	20	1.7494	-.2623	3.0797	-.1.3044	1.6271	-.6700	.1676	-.0011
0.21	22	1.8587	-.2854	3.2142	-.1.3741	1.7211	-.7264	.2222	-.1.168
0.21	24	1.6801	-.3471	2.9916	-.1.3037	1.6936	-.7206	.2740	-.1071
0.21	26	1.6119	-.3778	2.4780	-.1.0979	1.6456	-.7279	.2118	-.0786
0.30	-4	.7122	-.3609	3.9093	-.1.5505	1.4087	-.5813	.0184	.0044
0.30	0	.8424	-.3631	3.7110	-.1.4739	1.4494	-.6107	.0418	-.0162
0.30	4	1.0127	-.3468	3.5368	-.1.4023	1.4506	-.6203	.0330	-.0134
0.30	8	1.2114	-.3291	3.5523	-.1.3928	1.4486	-.6342	.0376	-.0154
0.30	12	1.4301	-.3108	3.4217	-.1.3458	1.4975	-.6498	.0438	-.0174
0.30	16	1.7856	-.3045	3.9782	-.1.5982	1.6894	-.7189	.0504	-.0175
0.30	20	1.9621	-.3757	3.8171	-.1.6070	1.7565	-.7477	.0577	-.0165
0.30	22	2.0001	-.4005	3.8362	-.1.5744	1.7611	-.7460	.0524	-.0137
0.30	24	1.8795	-.4189	3.2543	-.1.3842	1.7451	-.7539	.1234	-.0454
0.30	26	1.6377	-.4108	2.4156	-.1.0689	1.6880	-.7384	.1082	-.0434
0.43	-4	.9466	-.4378	5.0566	-.2.0360	1.4775	-.5101	.0476	.0008
0.43	0	1.1436	-.4403	4.9628	-.2.0018	1.5206	-.5428	.0253	.0095
0.43	4	1.4224	-.4217	4.9701	-.2.0669	1.5237	-.5464	.0222	.0077
0.43	8	1.6397	-.4036	5.0664	-.2.0672	1.6422	-.6115	.0121	.0121
0.43	12	1.8345	-.4034	5.0690	-.2.0670	1.6484	-.5918	.0109	.0118
0.43	16	1.9612	-.4358	4.1501	-.1.7166	1.6362	-.6206	.0497	-.0149
0.43	20	1.9340	-.4449	3.6485	-.1.5307	1.6731	-.6629	.1593	-.0580
0.43	22	1.8617	-.4406	3.1372	-.1.3331	1.6726	-.6870	.1942	-.0638
0.43	24	1.7227	-.4200	2.3210	-.1.0217	1.6264	-.6986	.1488	-.0498
0.43	26	1.5878	-.4141	2.0769	-.9292	1.6268	-.7128	.1244	-.0420
0.55	-4	1.1280	-.5247	4.9090	-.2.0045	1.6443	-.6100	.1541	-.0557
0.55	0	1.3269	-.5291	4.9129	-.1.9884	1.7219	-.6463	.1982	-.0469
0.55	4	1.6262	-.5200	4.9398	-.2.0476	1.7240	-.6463	.2254	-.0493
0.55	8	1.9207	-.5215	4.6030	-.1.9169	1.7444	-.6538	.2303	-.0745
0.55	12	1.9679	-.5023	3.3770	-.1.3228	1.6093	-.6440	.2244	-.0780
0.55	16	1.7551	-.4708	2.7523	-.1.1735	1.6076	-.6566	.2080	-.0770
0.55	20	1.6542	-.4492	2.5143	-.1.0803	1.5488	-.6501	.1935	-.0690
0.55	22	1.6014	-.4423	2.3515	-.1.0187	1.5436	-.6543	.1909	-.0698
0.55	24	1.5821	-.4413	2.1393	-.9419	1.5645	-.6747	.1327	-.0512
0.55	26	1.4912	-.4353	1.9586	-.8789	1.5771	-.6978	.1163	-.0420
0.72	-4	.1266	.0899						
0.72	0	.3916	.1711						
0.72	4	1.0395	.4222						
0.72	8	.9284	-.0649						
0.72	12	.6369	-.0763						
0.72	16	.7864	-.0668						
0.72	20	.7398	-.0645						
0.72	22	.7145	-.0604						
0.72	24	.6947	-.0691						
0.72	26	.7072	-.0747						

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TABLE XV.- INTEGRATED SECTION DATA FOR THE DOUBLE SLOTTED FLAP  
CONFIGURATION WITH SPOILERS - Continued

(c)  $h_S = -0.035\bar{c}$ 

$\frac{Y}{b/2}$	$a, \text{deg}$	$c_{n,w}$	$c_{m,w}$	$c_{n,y}$	$c_{m,y}$	$c_{n,f}$	$c_{m,f}$	$c_{n,s}$	$c_{m,s}$
0.21	-4	.5505	-.2808	2.2489	-.9846	1.4254	-.5630	-.1153	.0853
0.21	0	.6524	-.2744	2.1449	-.9219	1.3884	-.5571	-.0310	.0249
0.21	4	.8019	-.2826	2.1572	-.9240	1.4010	-.5655	-.0385	.0267
0.21	8	.9493	-.2726	2.1964	-.9393	1.3954	-.5632	-.0476	.0325
0.21	12	1.1800	-.2620	2.4254	-.1039	1.4412	-.5831	-.0340	.0281
0.21	16	1.4413	-.2560	2.7165	-.1302	1.5661	-.6389	-.0041	.0106
0.21	20	1.7350	-.2571	2.9624	-.1250	1.7026	-.6921	+.0768	-.0238
0.21	22	1.8602	-.2741	3.1024	-.1320	1.7936	-.7265	+.0958	-.0283
0.21	24	1.7059	-.3409	3.0118	-.1307	1.7919	-.7323	+.0649	-.0134
0.21	26	1.5724	-.3697	2.3728	-.1049	1.6497	-.6992	-.0063	.0149
0.30	-4	.6875	-.3531	3.8869	-.1538	1.3621	-.5444	-.0898	.0510
0.30	0	.8283	-.3509	3.5851	-.1422	1.3889	-.5859	-.0626	.0285
0.30	4	1.0088	-.3388	3.4961	-.1369	1.3944	-.5961	-.0518	.0261
0.30	8	1.1946	-.3181	3.4408	-.1371	1.4127	-.6122	-.0411	.0224
0.30	12	1.4249	-.3063	3.3857	-.1324	1.4623	-.6336	-.0350	.0196
0.30	16	1.7507	-.3285	3.8170	-.1508	1.5746	-.6708	-.0361	.0147
0.30	20	1.9409	-.3631	3.8859	-.1561	1.6201	-.6896	-.0314	.0101
0.30	22	2.0102	-.3938	3.7934	-.1553	1.7011	-.7193	-.0269	.0084
0.30	24	1.8357	-.4034	3.0472	-.1305	1.6428	-.6597	-.0370	.0188
0.30	26	1.6699	-.4043	2.4936	-.1014	1.6532	-.7234	-.0516	.0175
0.43	-4	.8342	-.3707	4.7447	-.1896	1.1797	-.3974	-.3399	.1560
0.43	0	1.0131	-.3893	4.5118	-.1871	1.2523	-.4712	-.3197	.1387
0.43	4	1.2463	-.3925	4.6551	-.1869	1.3847	-.5212	-.2857	.1217
0.43	8	1.4954	-.3756	4.6940	-.1882	1.4244	-.5394	-.2289	.0945
0.43	12	1.7777	-.3825	4.5501	-.1843	1.4675	-.5693	-.1750	.0708
0.43	16	1.9291	-.4212	4.0118	-.1654	1.5335	-.6110	-.1424	.0561
0.43	20	1.9152	-.4329	3.6388	-.1520	1.5839	-.6426	-.1066	.0428
0.43	22	1.8794	-.4366	3.2789	-.1385	1.6118	-.6664	-.0879	.0382
0.43	24	1.7470	-.4212	2.4295	-.1061	1.6089	-.6876	-.0529	.0248
0.43	26	1.5737	-.4119	2.0441	-.9159	1.6070	-.7064	-.0166	.0144
0.55	-4	1.0285	-.4621	4.6290	-.1682	1.3346	-.4985	-.3674	.1460
0.55	0	1.2367	-.4591	4.5696	-.1843	1.3791	-.5127	-.3982	.1451
0.55	4	1.5551	-.4466	4.7580	-.1925	1.3963	-.5112	-.3996	.1728
0.55	8	1.8781	-.4701	4.4785	-.1857	1.4897	-.5606	-.2478	.1090
0.55	12	1.8577	-.4675	3.4318	-.1441	1.5560	-.6215	-.1036	.0361
0.55	16	1.7245	-.4579	2.7439	-.1677	1.5380	-.6396	-.0631	.0239
0.55	20	1.6415	-.4394	2.5019	-.1073	1.4927	-.6328	-.0476	.0195
0.55	22	1.6327	-.4484	2.4065	-.1040	1.5483	-.6652	-.0312	.0145
0.55	24	1.5491	-.4299	2.0325	-.0977	1.5289	-.6690	-.0308	.0129
0.55	26	1.5005	-.4313	1.9724	-.0879	1.5504	-.6847	-.0222	.0104
0.72	-4	.2316	.0775						
0.72	0	.3778	.1222						
0.72	4	1.1128	-.0750						
0.72	8	.6898	-.0732						
0.72	12	.8208	-.0730						
0.72	16	.7578	-.0644						
0.72	20	.7352	-.0632						
0.72	22	.7246	-.0630						
0.72	24	.6957	-.0672						
0.72	26	.7401	-.0820						

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TABLE XV.- INTEGRATED SECTION DATA FOR THE DOUBLE SLOTTED FLAP  
CONFIGURATION WITH SPOILERS - Continued

(d)  $h_s = -0.050\bar{c}$ 

$\frac{Y}{\bar{c}/2}$	$\alpha, \text{ deg}$	$c_{n,w}$	$c_{m,w}$	$c_{n,v}$	$c_{m,v}$	$c_{n,f}$	$c_{m,f}$	$c_{n,s}$	$c_{m,s}$
0.21	-4	.5178	-.2646	2.2831	-.9814	1.2484	-.5078	-.1788	.1249
0.21	0	.6446	-.2733	2.2018	-.9456	1.2691	-.5270	-.1005	.0707
0.21	4	.8012	-.2732	2.1308	-.9134	1.2677	-.5351	-.0719	.0537
0.21	8	.9564	-.2671	2.2222	-.9459	1.2817	-.5474	-.0605	.0483
0.21	12	1.1686	-.2581	2.4243	-.10301	1.3374	-.5713	-.0237	.0307
0.21	16	1.4396	-.2570	2.7090	-.12186	1.4664	-.6312	.0204	.0096
0.21	20	1.7139	-.2512	2.8935	-.12254	1.5817	-.6772	.0724	-.0151
0.21	22	1.8270	-.2715	3.0104	-.12853	1.6405	-.7094	.0623	-.0159
0.21	24	1.6719	-.3330	2.9226	-.12483	1.6326	-.7923	.0295	.0088
0.21	26	1.5884	-.3644	2.4851	-.1002	1.5747	-.6909	-.0293	.0346
0.30	-4	.6486	-.3445	3.9265	-.15517	1.2992	-.5382	-.1872	.0911
0.30	0	.8209	-.3482	3.7057	-.14710	1.3293	-.5472	-.1894	.0742
0.30	4	.9934	-.3347	3.5219	-.13940	1.3393	-.5579	-.1392	.0623
0.30	8	1.1885	-.3167	3.5129	-.13761	1.3725	-.5809	-.1055	.0515
0.30	12	1.3913	-.2987	3.4022	-.13335	1.4069	-.5981	-.0741	.0360
0.30	16	1.7388	-.3205	3.8301	-.15369	1.5010	-.6297	.0990	.0375
0.30	20	1.9793	-.3629	3.9977	-.16069	1.5899	-.6616	-.1049	.0358
0.30	22	1.9763	-.3783	3.7441	-.15340	1.5776	-.6541	-.1305	.0448
0.30	24	1.8183	-.3967	3.0482	-.16037	1.5938	-.6723	-.1441	.0490
0.30	26	1.6369	-.4010	2.4154	-.10718	1.5943	-.6922	-.1087	.0379
0.43	-4	.8389	-.3830	4.5013	-.17916	1.2235	-.4478	-.4340	.1894
0.43	0	.9861	-.3722	4.4007	-.17544	1.2035	-.4361	-.4762	.1962
0.43	4	1.2328	-.3558	4.4320	-.17730	1.2349	-.4534	-.4622	.1880
0.43	8	1.4691	-.3486	4.5868	-.18408	1.3038	-.4840	-.3941	.1546
0.43	12	1.7400	-.3628	4.4166	-.17859	1.3856	-.5112	-.3193	.1182
0.43	16	1.9030	-.4046	3.9608	-.16521	1.4388	-.5639	-.2642	.0961
0.43	20	1.8879	-.4079	3.5557	-.16479	1.4423	-.5749	-.2218	.0783
0.43	22	1.8147	-.4148	3.1110	-.16167	1.5067	-.6180	-.1796	.0637
0.43	24	1.7016	-.4064	2.3363	-.16488	1.5210	-.6461	-.1078	.0394
0.43	26	1.5783	-.4051	2.0076	-.9313	1.5645	-.6821	-.0802	.0264
0.55	-4	.9807	-.4378	4.4422	-.17999	1.2333	-.4669	-.4083	.1513
0.55	0	1.1854	-.4311	4.4263	-.17742	1.2367	-.4532	-.5077	.1964
0.55	4	1.4767	-.4150	4.5459	-.18386	1.2907	-.4549	-.5248	.2072
0.55	8	1.7774	-.4378	4.3381	-.17884	1.3484	-.5055	-.3654	.1342
0.55	12	1.7676	-.4566	3.3426	-.16404	1.4240	-.5620	-.1884	.0627
0.55	16	1.6832	-.4588	2.7134	-.15653	1.4457	-.5974	-.1195	.0401
0.55	20	1.5550	-.4799	2.6864	-.17955	1.4617	-.6170	-.0474	.0314
0.55	22	1.5855	-.4274	2.3391	-.16021	1.5112	-.6210	-.0878	.0105
0.55	24	1.5550	-.4287	2.0446	-.9097	1.5083	-.6569	-.0591	.0213
0.55	26	1.4939	-.4258	1.9636	-.8749	1.5188	-.6693	-.0443	.0179
0.72	-4	.2411	.0540						
0.72	0	.5075	.1022						
0.72	4	1.1134	-.0739						
0.72	8	.8843	-.0733						
0.72	12	.8070	-.0666						
0.72	16	.7559	-.0613						
0.72	20	.7165	-.0644						
0.72	22	.7070	-.0598						
0.72	24	.5971	-.0686						
0.72	26	.7209	-.0776						

TABLE XV.- INTEGRATED SECTION DATA FOR THE DOUBLE SLOTTED FLAP  
CONFIGURATION WITH SPOILERS - Concluded

(e)  $h_s = -0.100\bar{c}$ 

$\frac{y}{h/2}$	$\alpha, \text{ deg}$	$c_{n,w}$	$c_{m,w}$	$c_{n,v}$	$c_{m,v}$	$c_{n,f}$	$c_{m,f}$	$c_{n,s}$	$c_{m,s}$
0.21	-4	+3757	-2748	1.0304	-7898	.0835	-3737	-0.4599	.2345
0.21	0	+3578	-2635	1.0407	-8141	1.0364	-4005	-0.4149	.2082
0.21	4	+6947	-1.118	1.0972	-8235	1.1183	-4349	-0.4309	.2143
0.21	8	+6631	-3275	2.0573	-8875	1.2052	-4784	-0.4624	.2145
0.21	12	1.1104	-3214	2.2091	-9445	1.2661	-5179	-0.4632	.1964
0.21	16	1.3335	-3161	2.4757	-1.0294	1.3443	-5570	-0.4963	.2102
0.21	20	1.6361	-3031	2.5078	-1.0439	1.3698	-5689	-0.5208	.2161
0.21	24	1.5911	-3964	2.4127	-1.0357	1.3494	-5654	-0.4798	.1896
0.21	26	1.5173	-4526	2.1552	-0.9523	1.4148	-6022	-0.4097	.1539
0.30	-4	+4767	-2368	2.1901	-8937	1.0470	-4100	-0.4704	.2426
0.30	0	+5010	-2403	2.0771	-8556	1.0880	-4366	-0.3940	.2401
0.30	4	+5779	-2180	2.0271	-9058	1.1246	-4385	-0.5014	.2874
0.30	8	1.0212	-2195	2.0335	-9732	1.1793	-4578	-0.7399	.3562
0.30	12	1.2638	-2133	2.8687	-1.1093	1.1826	-4531	-0.8631	.1997
0.30	16	1.5567	-2175	3.2343	-1.2667	1.1932	-4478	-1.0990	.5034
0.30	20	1.7746	-2580	3.3859	-1.9349	1.2564	-4764	-1.1103	.4943
0.30	22	1.6058	-2950	3.2504	-1.3062	1.3327	-5091	-1.0717	.4635
0.30	24	1.6341	-3292	2.6471	-1.1239	1.3629	-5404	-0.8387	.3457
0.30	26	1.5082	-3523	2.1909	-0.9718	1.4228	-5855	-0.6176	.2449
0.43	-4	+5089	-2346	2.1052	-8113	.9036	-3290	-0.4159	.2115
0.43	0	+7268	-2464	2.0426	-8308	1.0409	-4193	-0.4123	.2376
0.43	4	+979	-238	2.0980	-8214	1.0402	-4107	-0.4448	.2168
0.43	8	1.1950	-2306	2.3723	-9169	1.1138	-4972	-0.4983	.2316
0.43	12	1.4570	-2612	2.7214	-1.0356	1.1628	-4513	-0.6220	.2934
0.43	16	1.6970	-3024	3.0305	-1.1857	1.1127	-4360	-0.8030	.3467
0.43	20	1.6613	-3200	2.8959	-1.1724	1.1602	-4598	-0.8046	.3322
0.43	22	1.6159	-3311	2.6678	-1.1077	1.2344	-4908	-0.7653	.3046
0.43	24	1.5548	-3486	2.1917	-0.9553	1.3407	-5550	-0.5289	.1917
0.43	26	1.4788	-3650	2.0144	-0.9008	1.4324	-6035	-0.3686	.1232
0.55	-4	.6270	-2659	2.0947	-8315	.9611	-3515	-0.5706	.2892
0.55	0	.6220	-2666	2.1001	-8392	1.0052	-3701	-0.6157	.2958
0.55	4	1.1127	-2509	2.2149	-8636	1.0320	-3814	-0.6298	.3097
0.55	8	1.5114	-3011	2.5022	-1.0094	1.0961	-4108	-0.6153	.3058
0.55	12	1.5999	-3621	2.6551	-1.0466	1.1299	-4465	-0.5090	.2222
0.55	16	1.5274	-3498	2.4865	-1.0392	1.1198	-4494	-0.4566	.2073
0.55	20	1.4809	-3527	2.2984	-0.9747	1.1820	-4849	-0.5386	.1899
0.55	22	1.4568	-3633	2.1926	-0.9443	1.2591	-5199	-0.5007	.1731
0.55	24	1.4430	-3781	1.9661	-0.8703	1.3513	-5727	-0.3706	.1142
0.55	26	1.3867	-3798	1.8820	-0.8402	1.3639	-5843	-0.2752	.0718
0.72	-4	.3176	-0.0038						
0.72	0	.6097	.0338						
0.72	4	1.1351	-1028						
0.72	8	1.0125	-1287						
0.72	12	.9733	-1395						
0.72	16	.7920	-0.929						
0.72	20	.7270	-0.797						
0.72	22	.7043	-0.769						
0.72	24	.6998	-0.773						
0.72	26	.7033	-0.796						

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*Pressure orifice  
location*

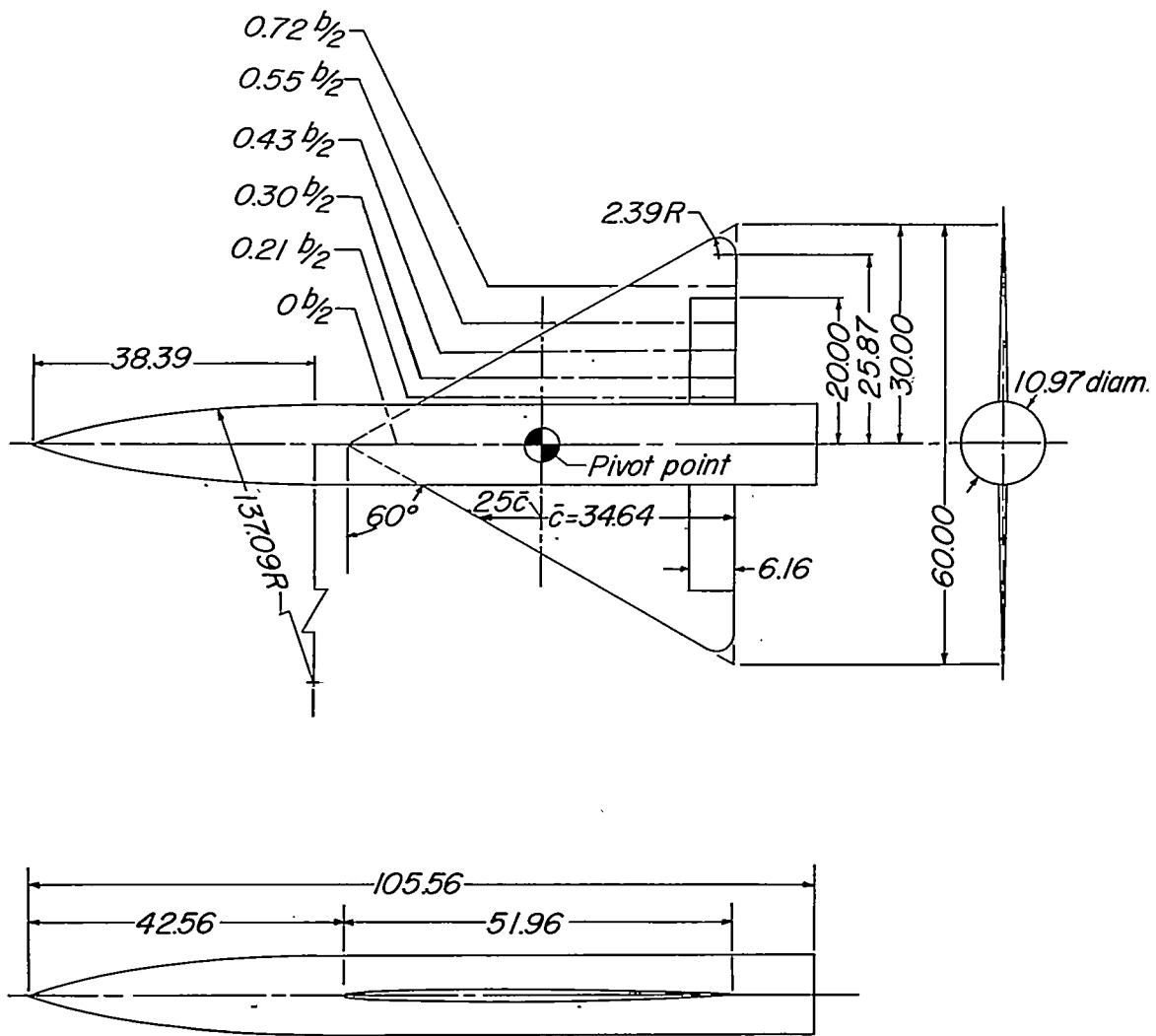


Figure 1.- General arrangement of  $60^\circ$  delta-wing model (all dimensions are in inches unless otherwise noted).

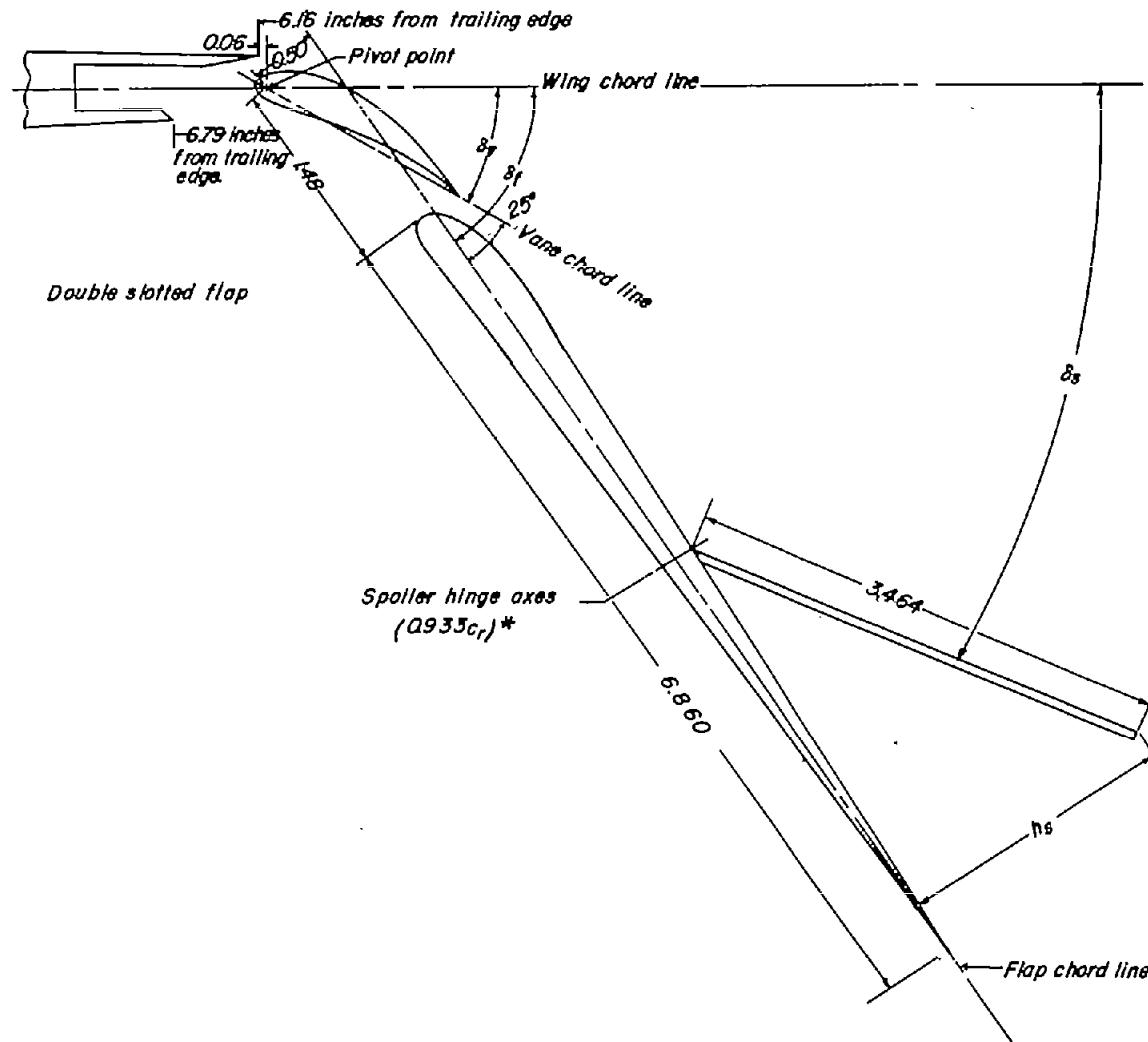


Figure 2.- Details of double slotted flap with spoiler. All dimensions are in inches unless otherwise noted. (\* indicates spoiler axes for pressure-distribution test. For force tests spoiler hinge axis at 0.700, 0.894, and  $0.933c_r$  on plain-wing configuration and at 0.894 and  $0.933c_r$  on double-slotted-flap configuration.)

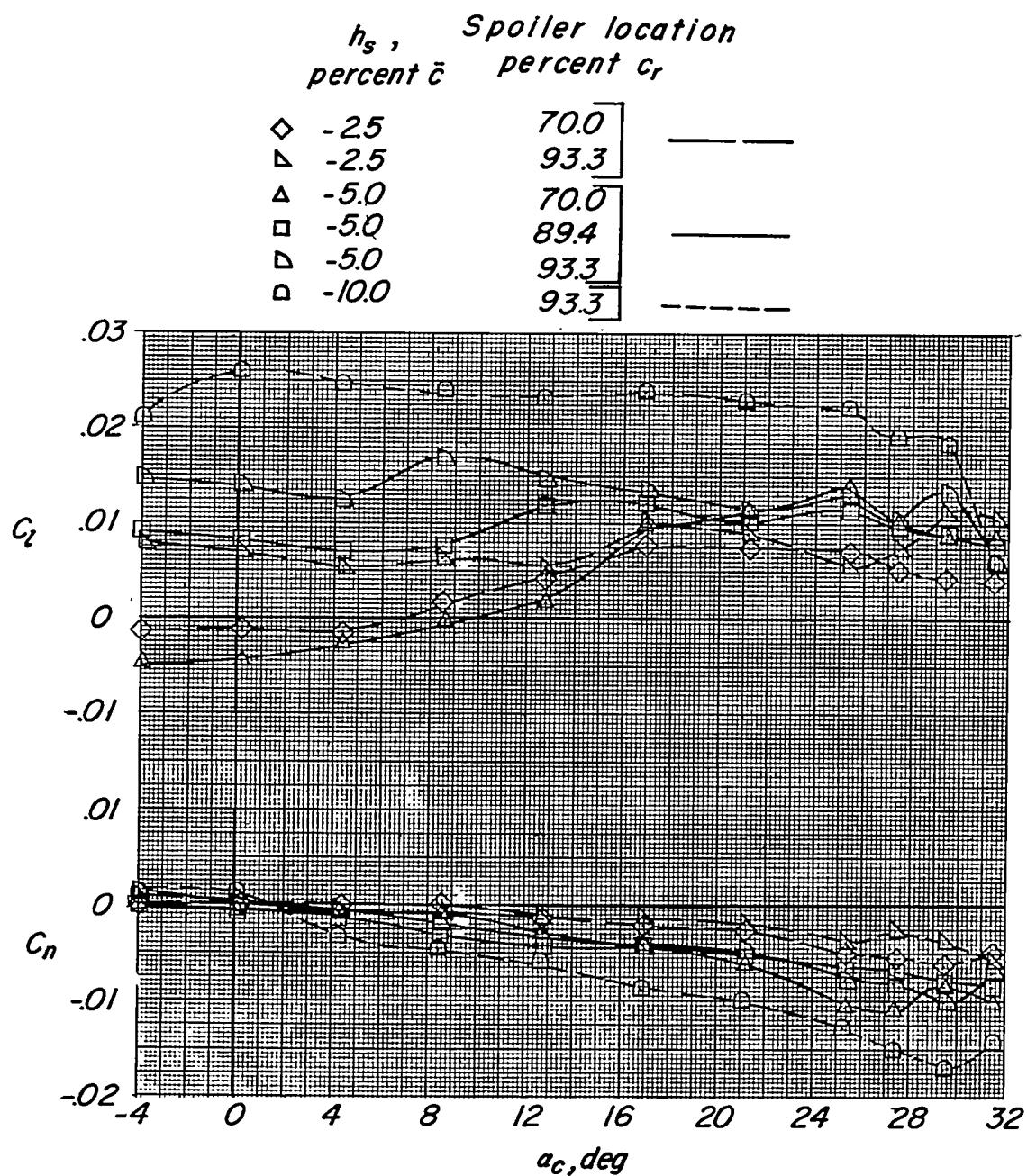


Figure 3.- Variation of rolling-moment coefficient and yawing-moment coefficient with angle of attack for plain-wing configuration having spoilers at various chordwise positions and at several projections.

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Spoiler location,  
percent  $c_r$

□ 89.4

◇ 93.3

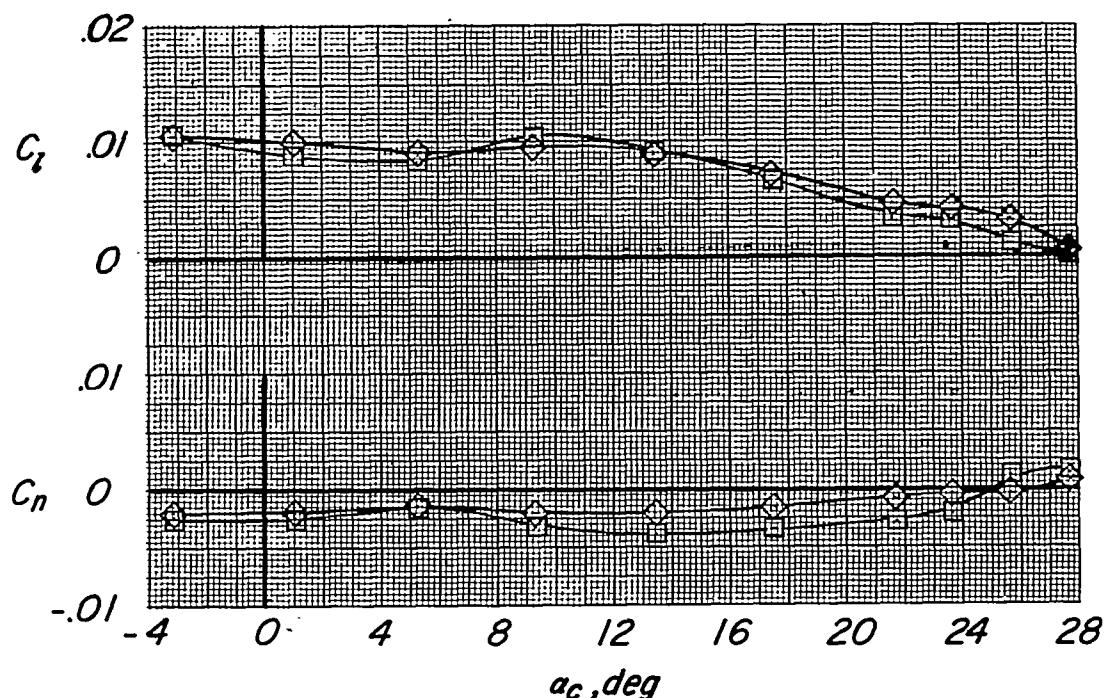
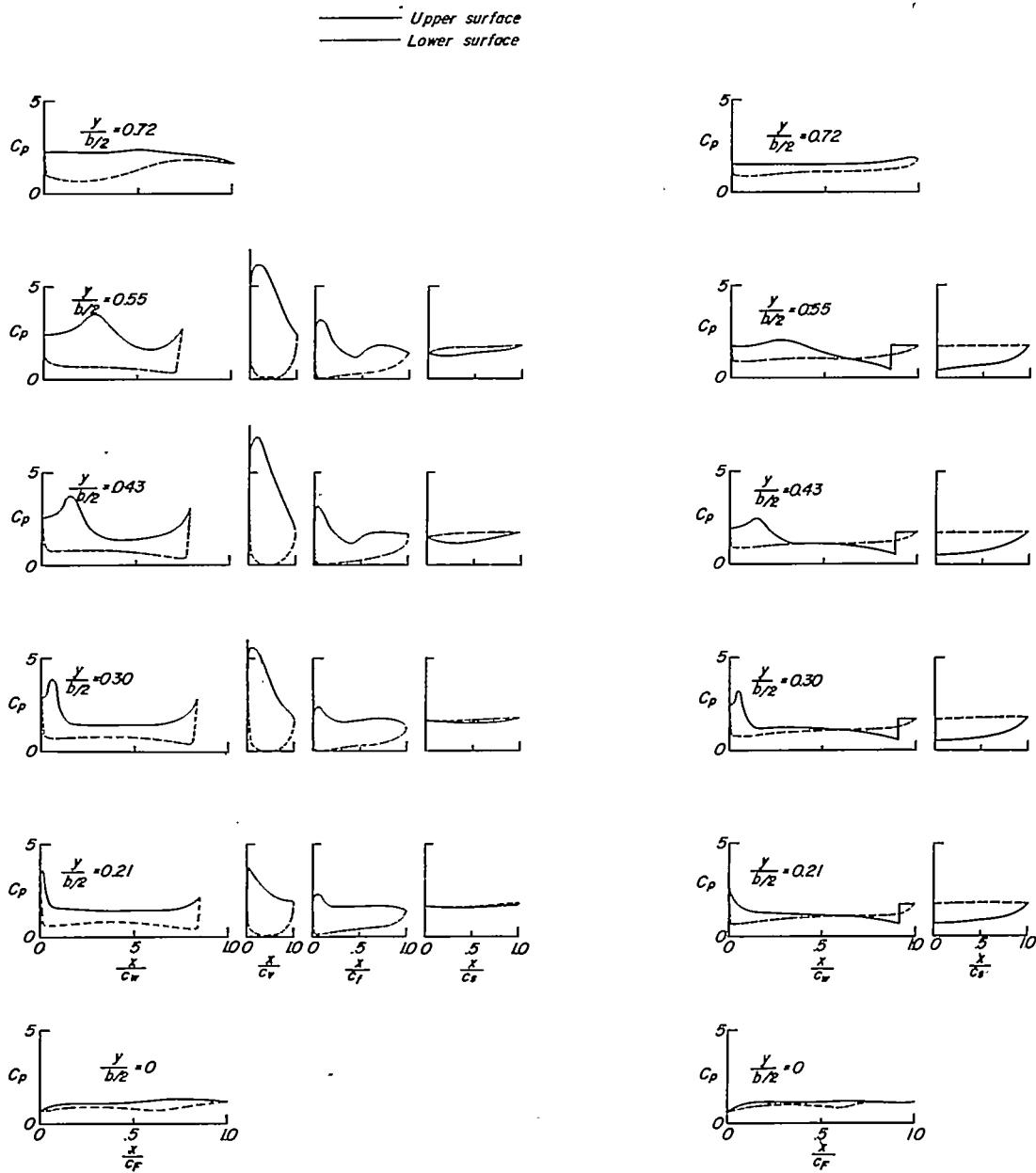


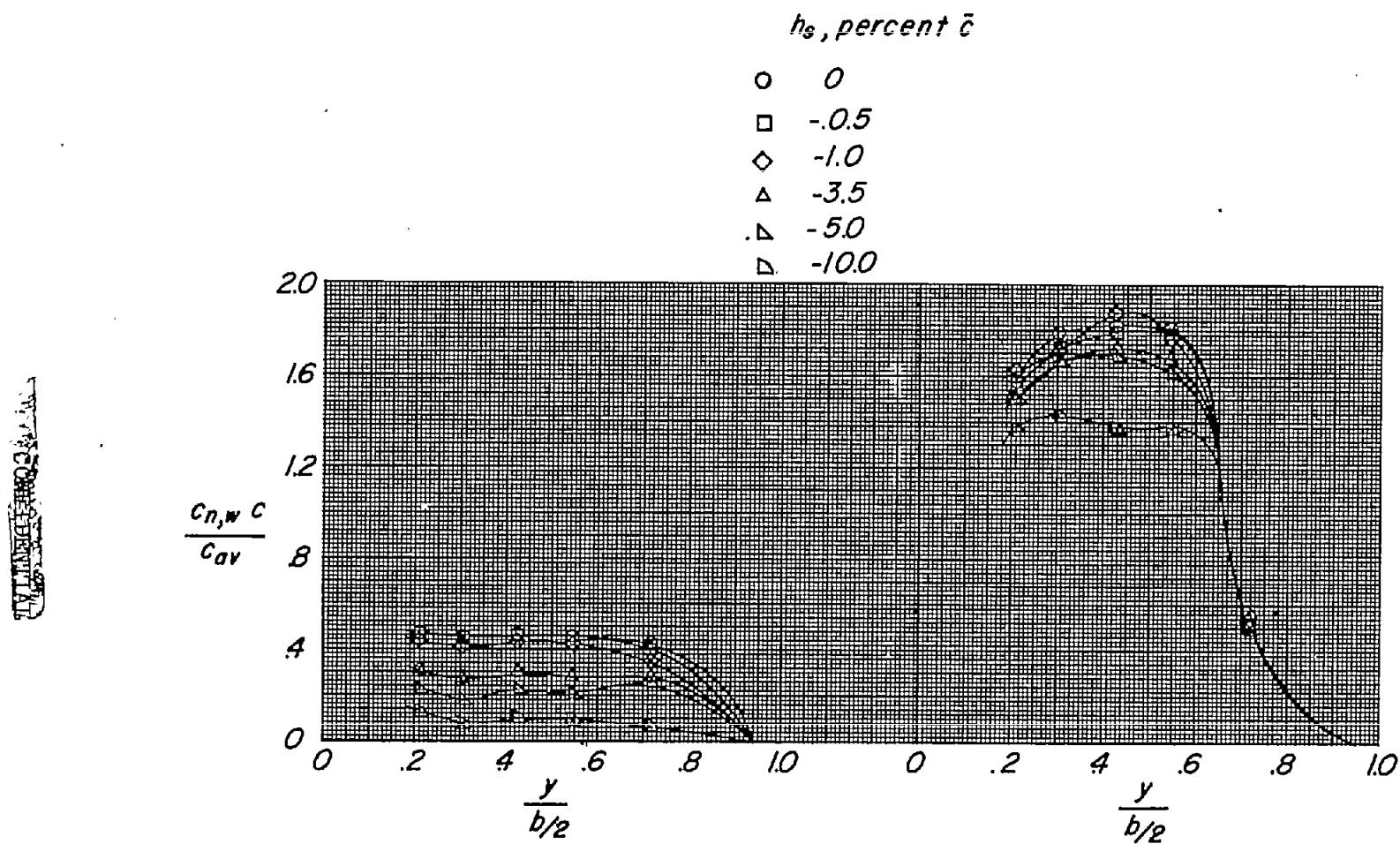
Figure 4.- Variation of rolling-moment coefficient and yawing-moment coefficient with angle of attack for the double-slotted-flap configuration having spoilers at two chordwise positions;  $\delta_f = 60^\circ$ ,  $h_s = -0.05\bar{c}$ .



(a) Double-slotted-flap configuration.

(b) Plain-wing configuration.

Figure 5.- Chordwise load distribution at several spanwise positions for plain-wing configuration and double-slotted-flap configuration at  $\alpha = 8^\circ$  and  $h_S = -0.05\bar{c}$ . (Note:  $x/c_F$ ,  $x/c_W$ ,  $x/c_V$ ,  $x/c_F$ , and  $x/c_S$  are not the same scale.)



(a) Plain-wing configuration.

(b) Double-slotted-flap configuration;  $\delta_f = 60^\circ$ .

Figure 6.- Span-load distribution over plain-wing and double-slotted-flap configuration at several spoiler projections;  $\alpha = 8^\circ$ .

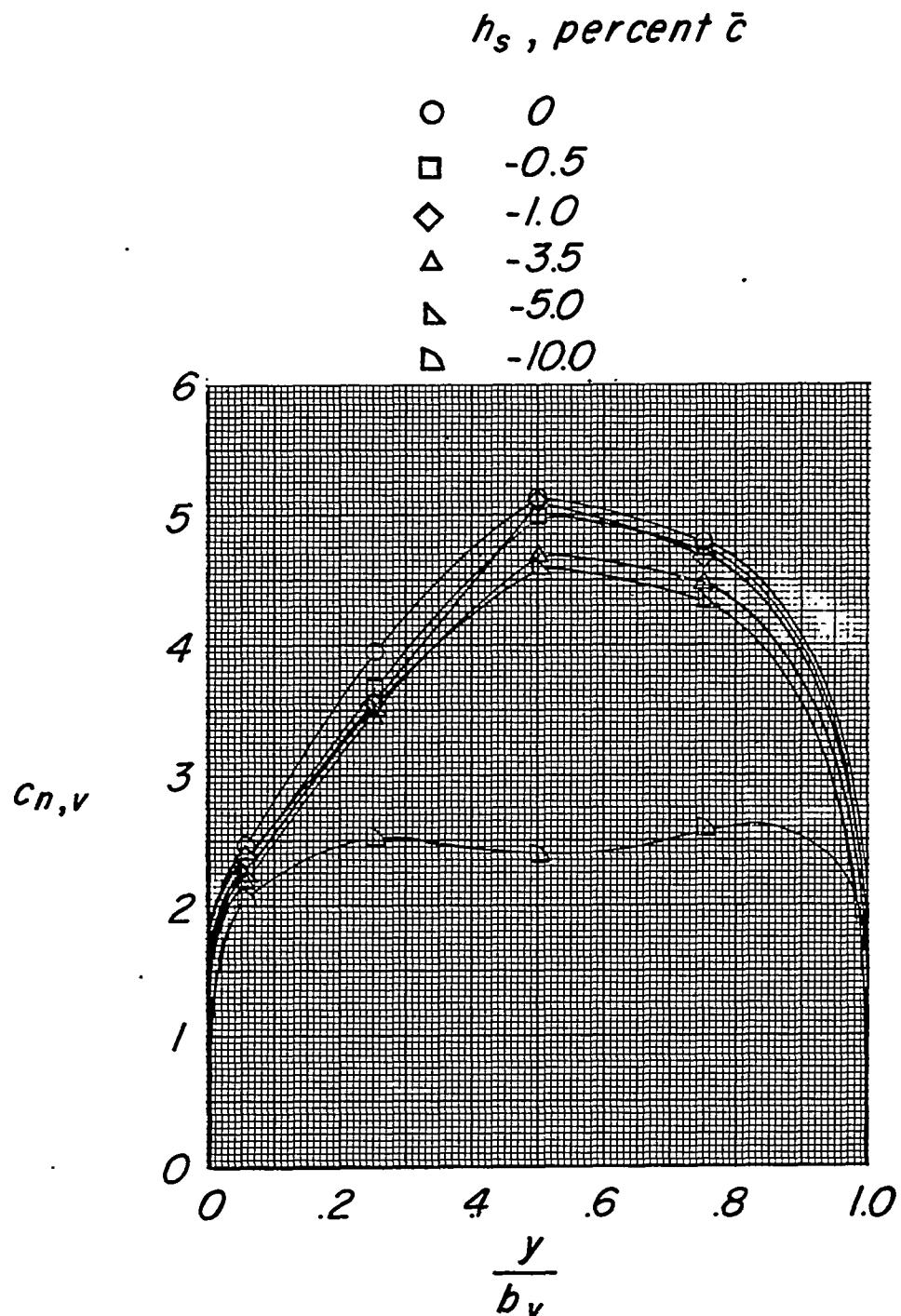


Figure 7.- Span-load distribution over vane of double-slotted-flap configuration at several spoiler projections;  $\delta_f = 60^\circ$ ;  $\alpha = 8^\circ$ .

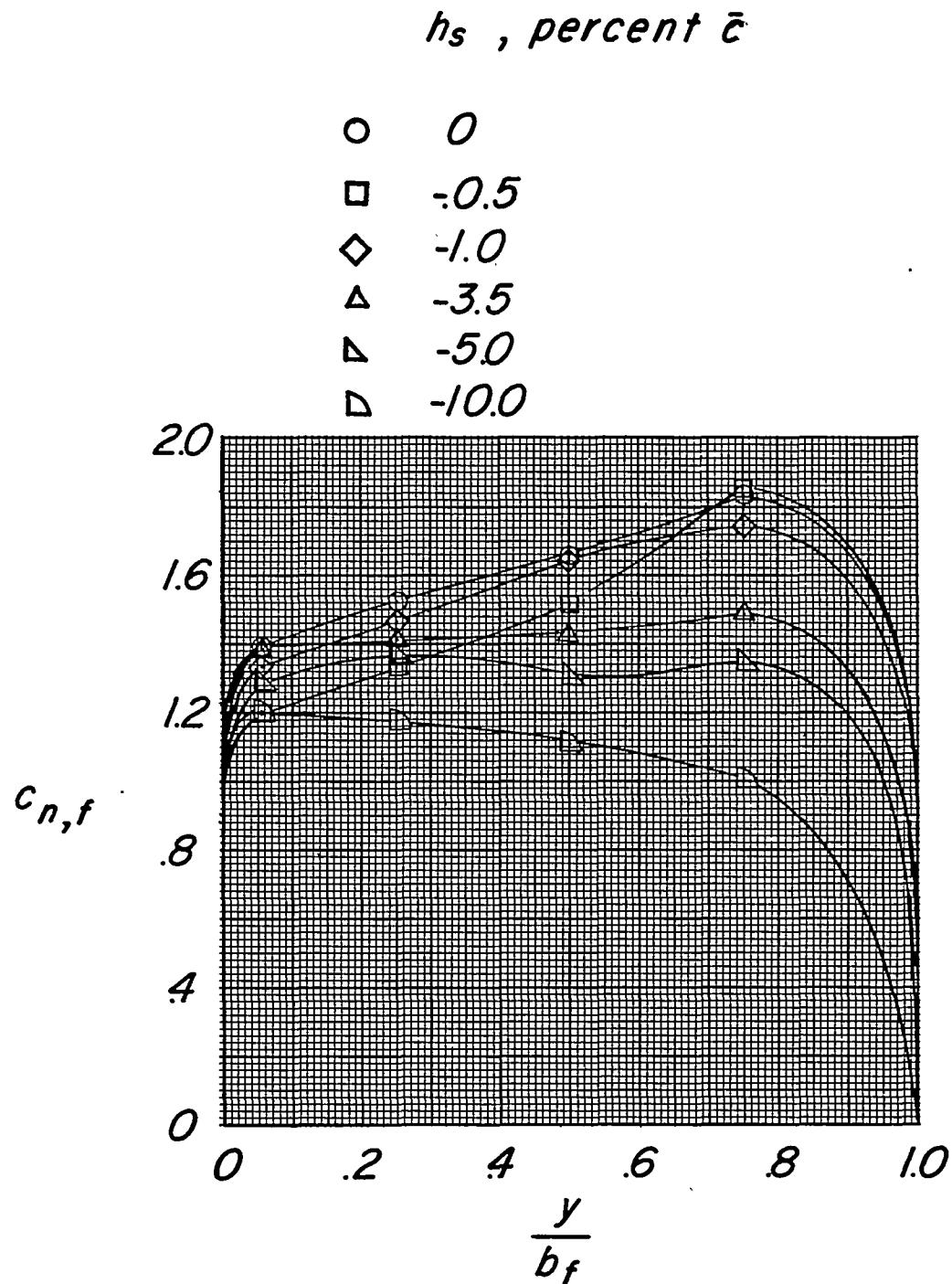


Figure 8.- Span-load distribution over flap of the double-slotted-flap configuration at several spoiler projections;  $\alpha = 8^\circ$ .

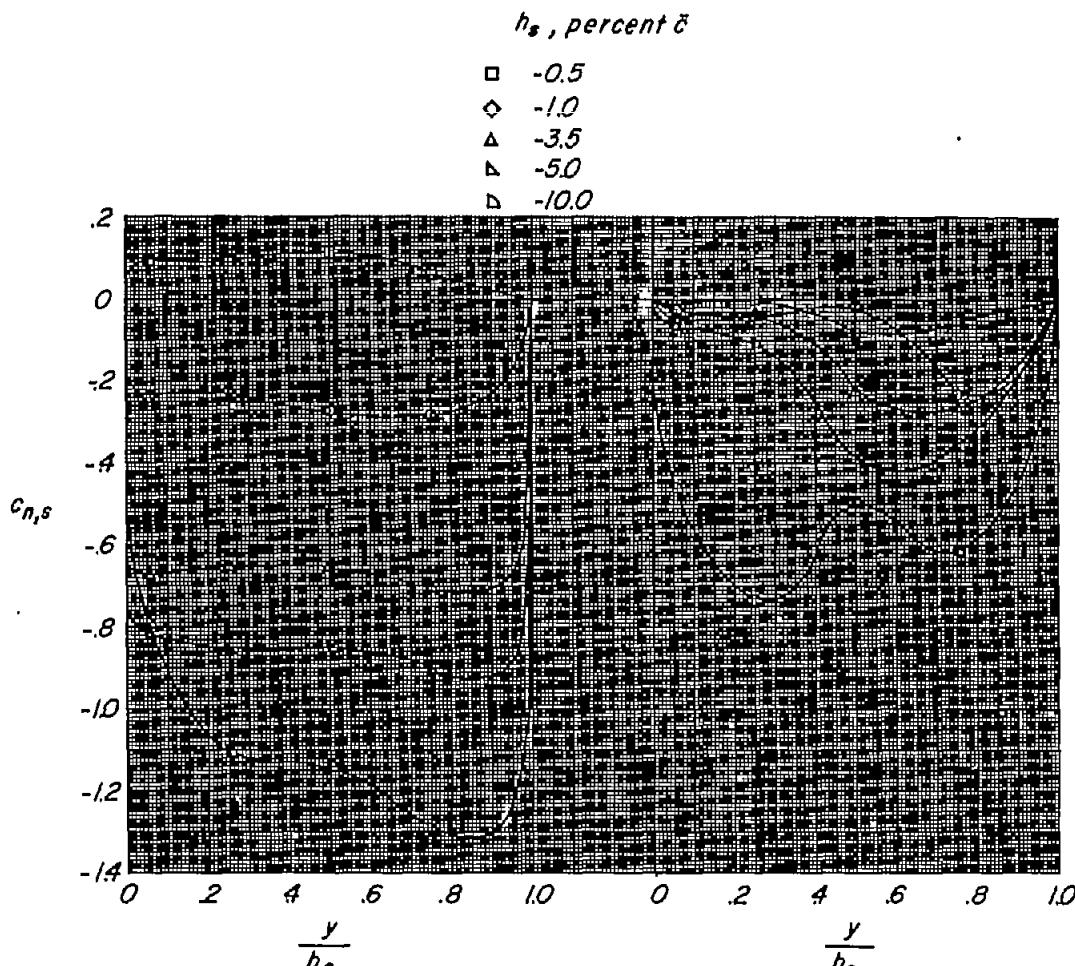


Figure 9.- Span-load distribution over spoiler for plain-wing and double-slotted-flap configurations at several spoiler projections;  $\alpha = 8^\circ$ .

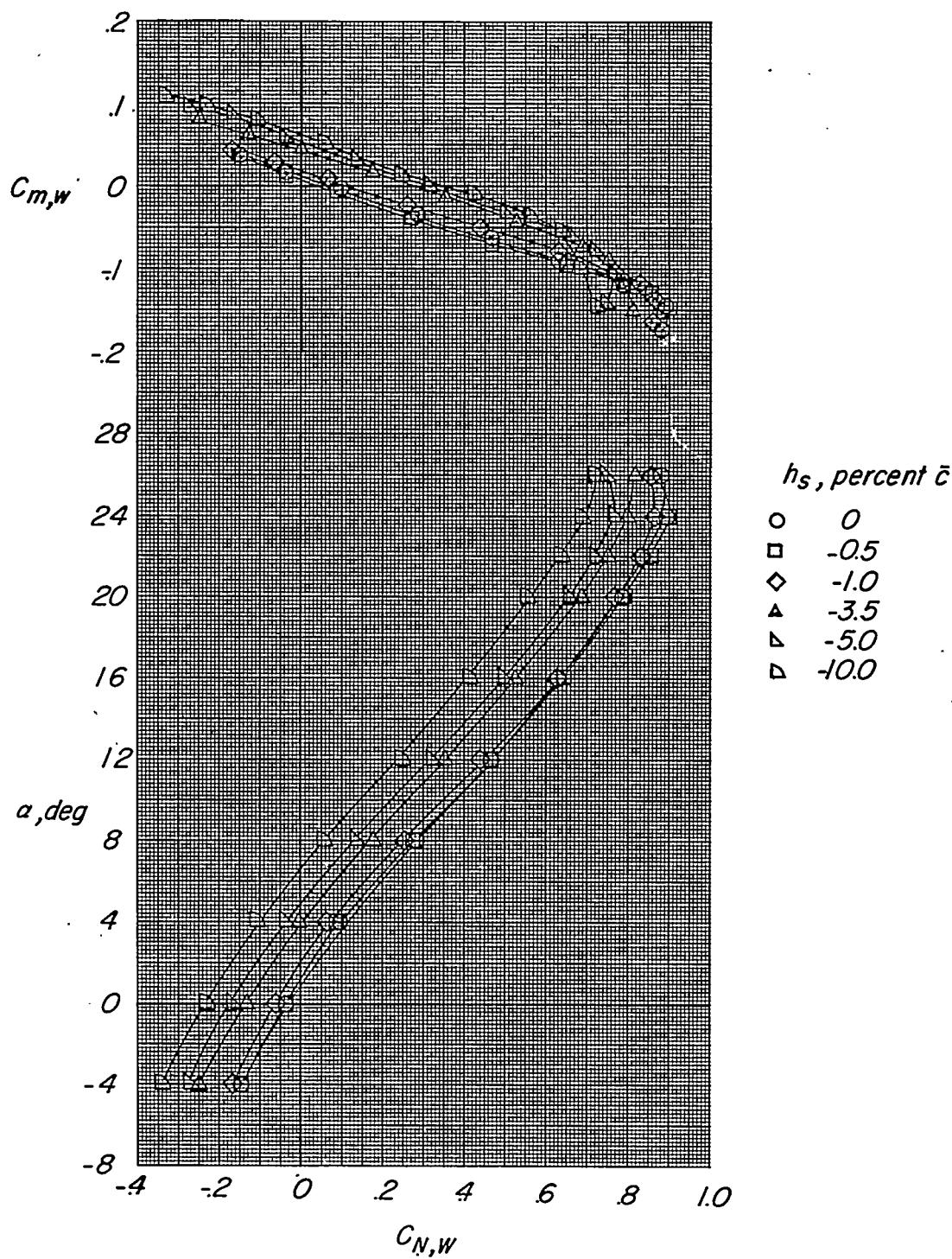


Figure 10.- Effect of spoiler projection on wing normal-force coefficients and pitching-moment coefficients of plain-wing configuration.

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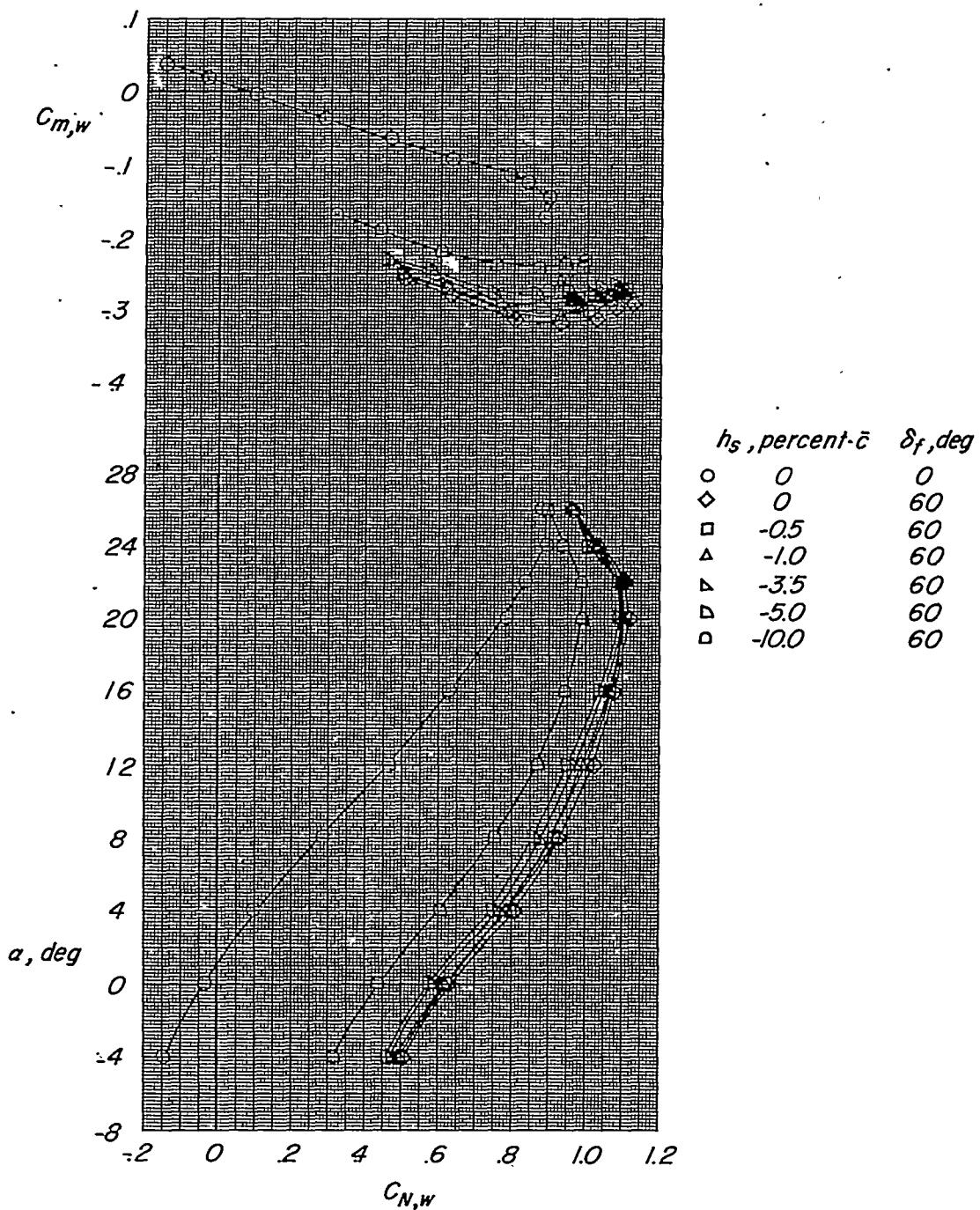


Figure 11.- Effect of spoiler projection on wing normal-force coefficients and pitching-moment coefficients of double-slotted-flap configuration;  $\delta_f = 60^\circ$ .

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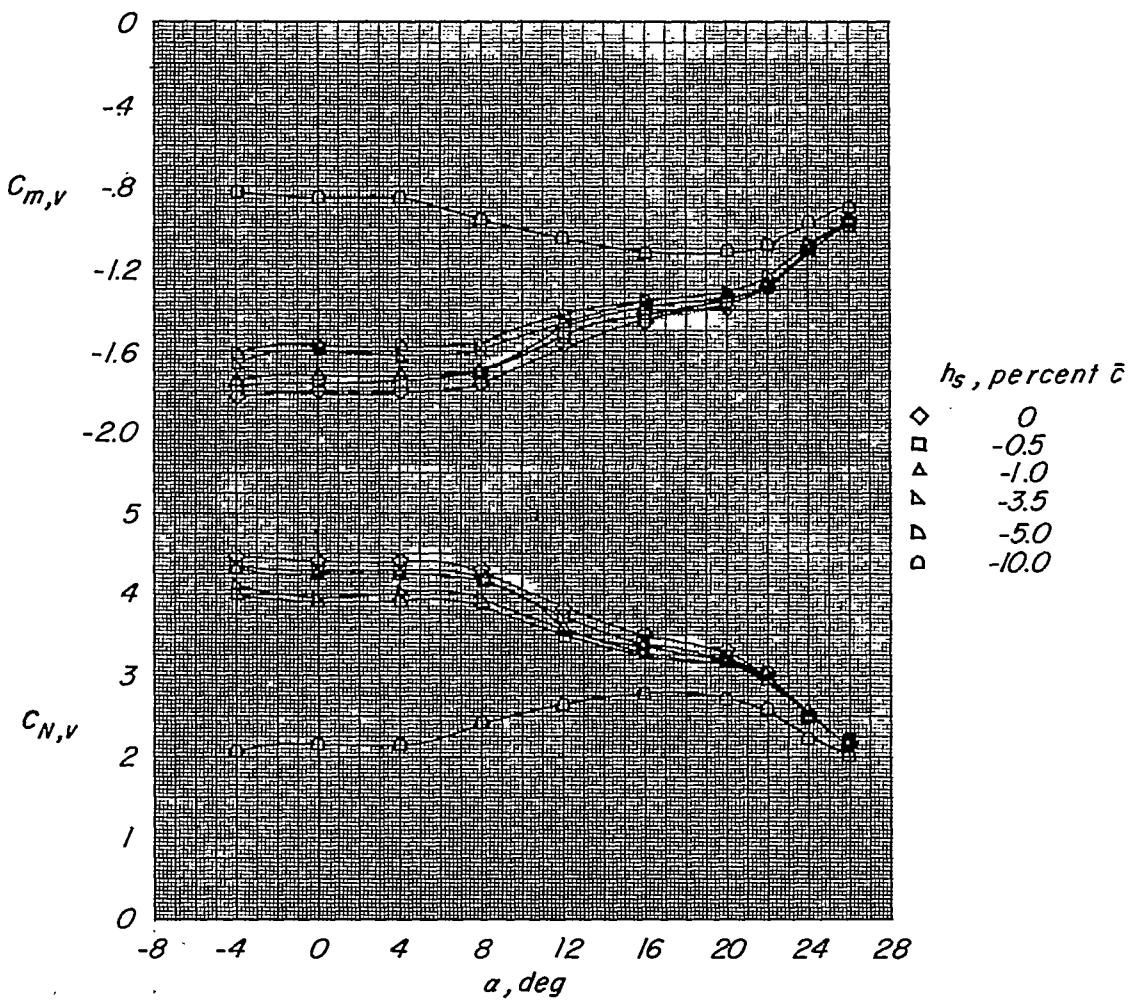


Figure 12.- Effect of spoiler projection on vane normal-force coefficients and pitching-moment coefficients of double-slotted-flap configuration;  $\delta_f = 60^\circ$ .

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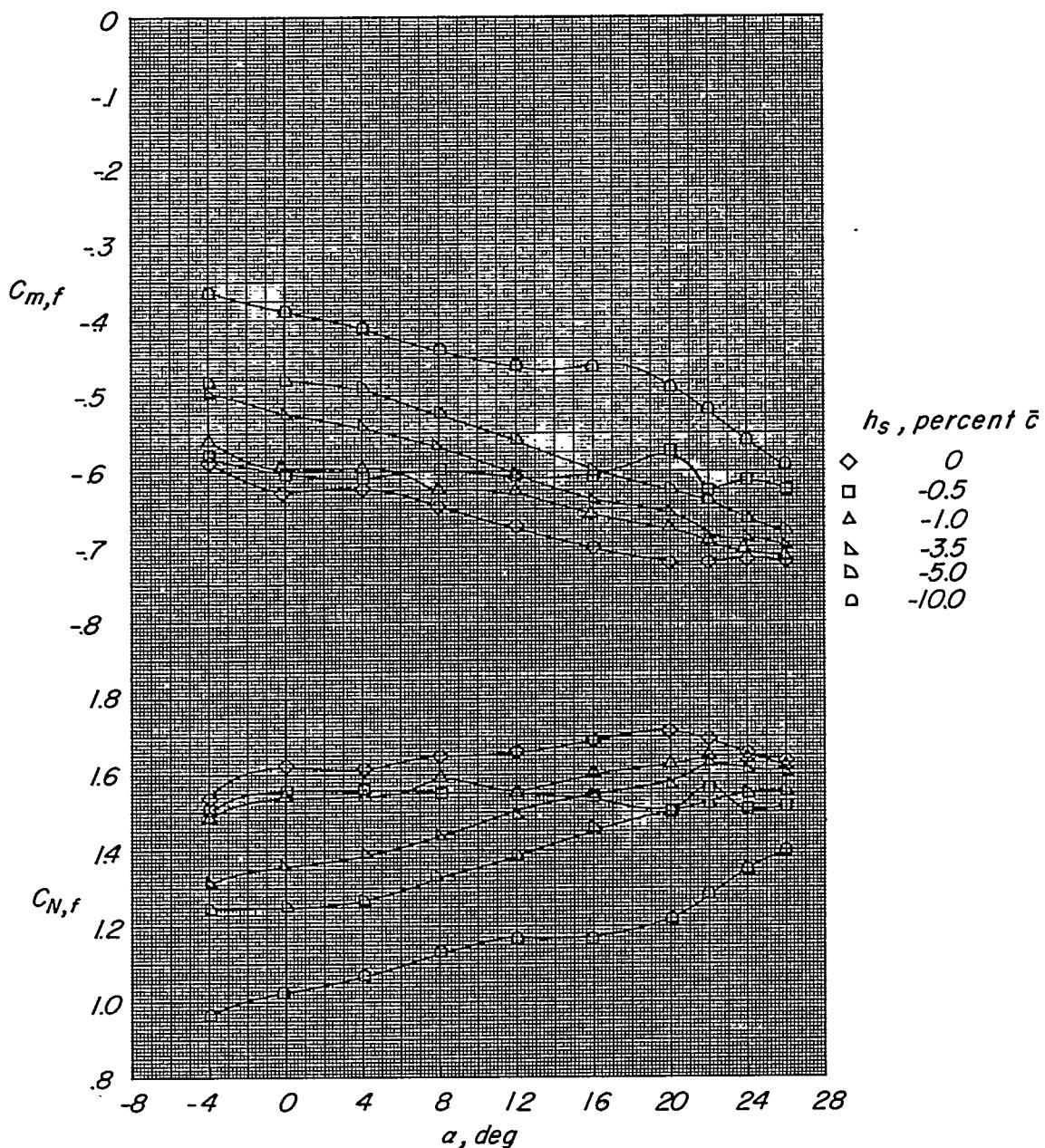


Figure 13.- Effect of spoiler projection on flap normal-force coefficients and pitching-moment coefficients of double-slotted-flap configuration;  $\delta_f = 60^\circ$ .

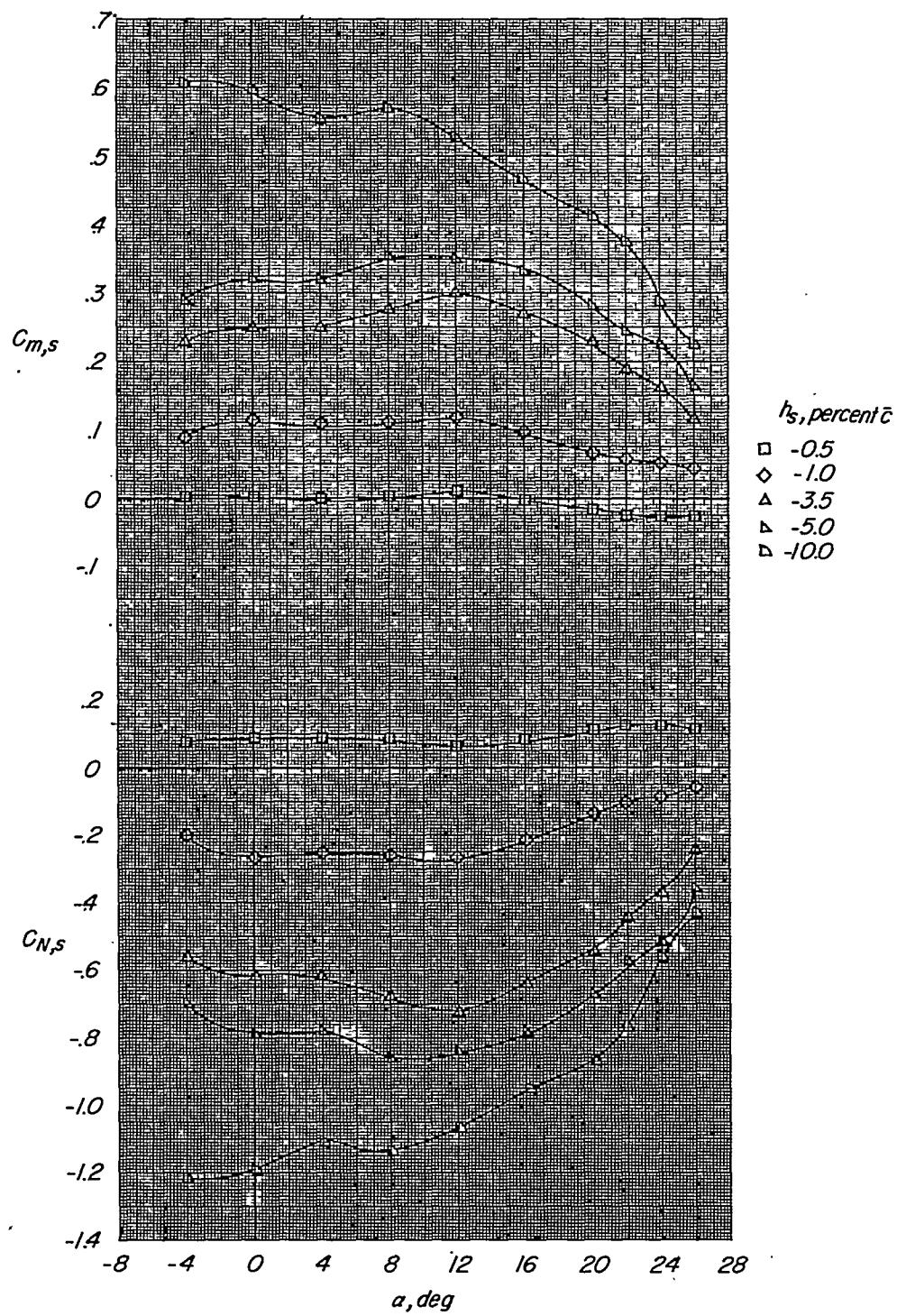


Figure 14.- Effect of spoiler projection on spoiler normal-force coefficients and pitching-moment coefficients of plain-wing configuration.

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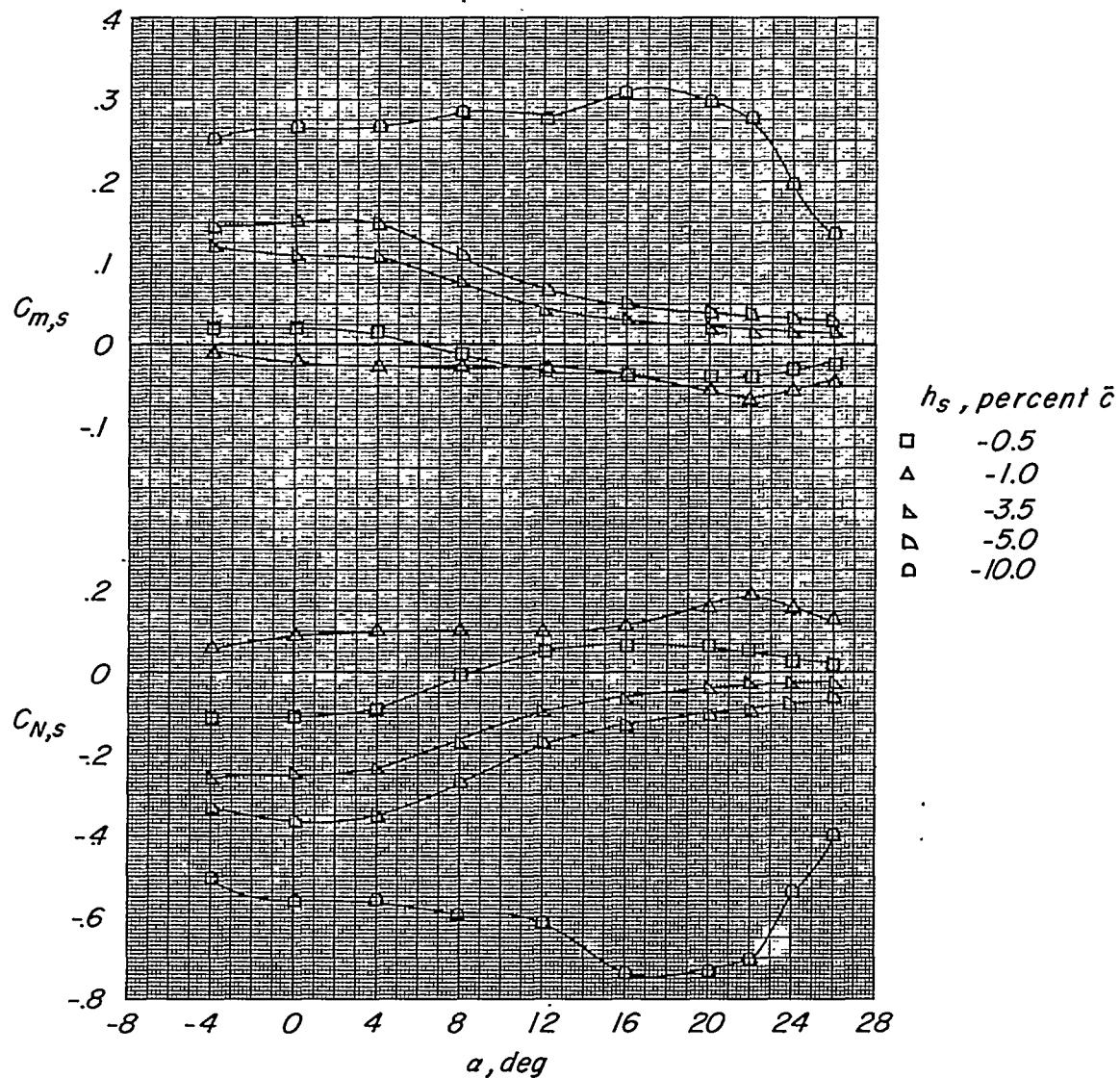


Figure 15.- Effect of spoiler projection on spoiler normal-force coefficients and pitching-moment coefficients of double-slotted-flap configuration;  $\delta_f = 60^\circ$ .

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