

~~11054~~
~~43~~

NATIONAL ADVISORY COMMITTEE FOR AERONAUTICS.

Technical Memorandum No. 86.

TESTING AIRPLANES IN FLIGHT:

Determining Position of Resultant of Action of Air
and Longitudinal Stability of an Airplane
at Different Angles of Attack.

By A Senouque.

From "Premier Congrès International de la Navigation
Aérienne," Paris, November, 1921, Vol. I.

FILE COPY

To be returned to
the files of the Langley
Memorial Aeronautical
Laboratory

1.8

April, 1932.



3 1176 01440 5105

TESTING AIRPLANES IN FLIGHT.*

Determining Position of Resultant of Action of Air and Longitudinal Stability of an Airplane at Different Angles of Attack.

By A. Senouque.

Measurements made during flight with the triple recording device (formerly employed by Mr. Toussaint and by ourselves), which gives the horizontal and vertical speeds of an airplane and the angle it makes with the horizon, render it possible to calculate its lift C_L , its drag C_D , and the resultant R , of the action of the air both in magnitude and direction, but, with these data alone, it is impossible to determine the position of this resultant in the plane of symmetry of the airplane. We will also see how we may determine the position of R during flight and then calculate the variations in the stability of an airplane.

In order that an airplane shall not turn about an axis perpendicular to its plane of symmetry and passing through its center of gravity, the moment of the action of the air with reference to this point must be zero. If, for example, it is desired to fly at an angle of 6° , while completely abandoning the elevator, the resultant of the action of the air for this angle of attack must pass through the center of gravity of the airplane. When, for any reason, the airplane attacks the air at a different angle for the above, 9° , for example, the resultant R , will no

* From "Premier Congrès International de la Navigation Aérienne, Paris, November, 1921, Vol. I, pp. 33-36.

longer pass through the center of gravity and will tend to make the airplane turn in one direction or the other. If the airplane is stable, the moment of R will bring the airplane back toward the angle of 6° . If it is poorly constructed, this moment will carry it further and further away from the original angle of attack. But when we would hold the airplane at this angle of attack of 9° , we must annul the moment of R . For this purpose, we have two ways at our disposal. By moving a weight on an airplane, we can change the location of its center of gravity and bring it on the resultant R for 9° , or we can adjust the lifting surfaces of the airplane, so that the resultant for 9° shall pass through the original center of gravity, as was the case when flying with an incidence of 6° . In practice the second method is always employed, the shape of the lifting surfaces being modified by the different positions given the elevator while climbing or descending. Hence, for each angle of attack, there is a corresponding shape of the lifting surfaces which causes the stability of the airplane to vary with the angle of attack. If we fly without changing the position of the elevator, but only move the weight on board and if we note the position of this weight for each angle of attack, we can, in each instance, determine the position of the center of gravity of the airplane, i. e., the point through which the resultant of the action of the air will pass for the given angle of attack.

This is how these determinations can be made in practice. The above-mentioned triple recorder is taken on board, a record-

ing thermometer and another recording instrument, on the drum of which is recorded the variations in the position of the elevator. One climbs to about 1000 meters and then descends by gliding with the engine at its very lowest speed and changes the angle of attack every 30 seconds. On the records obtained, care is taken to make measurements only after the lapse of 15 seconds after changing the angle of attack, for only after this interval of time can the airplane be considered to have a constant speed. Several series of experiments are thus made, the position of the center of gravity of the airplane being changed in each series by means of heavier or lighter weights situated as far as possible from the center of gravity, so that a slight variation in the additional load causes a relatively large variation in the position of the center of gravity.

The triple recorder furnishes the necessary data for calculating the different values of C_L and C_D for the different angles of attack. The barograph and thermograph curves enable the correction of these data for the density of the air. With the corrected values of C_L and C_D we can calculate the magnitude and direction of R with reference to the path followed by the airplane. The position of the elevator for the different angles of attack is shown by the position of its control lever. Lastly, a very simple calculation or a direct measurement gives the location of the center of gravity of the airplane for each flight.

Let us indicate, for each series of flights without changing

the position of the center of gravity, the positions of the elevator as ordinates and the corresponding angles of attack as abscissas (Fig. 1). Let us further indicate (Fig. 2) on a certain scale and with reference to an axis parallel to the wing chord, the different positions of the center of gravity of the airplane corresponding to the position of the elevator. When the airplane is in normal flight, i. e. without additional weight, the positions of the elevator for the angle of attack are represented by the line g (Fig. 1) and the resultant R of the action of the air passes through C.G. (Fig. 2). For the angle of attack I , the ordinate of the position of the elevator is F and the resultant of the action of the air is an angle with the path MN of the airplane, which is inclined to the wing chord by an angle I equal to the angle of attack. After placing a certain additional load on the tail of the airplane, the center of gravity is at C.G.' and the resultant of the action of the air at the different angles of attack passes through this point, when the path of the airplane is straight and the corresponding positions of the elevator are represented by the curve g' . For the same position of the elevator as in the first case, the angle of attack is I' , since we have $F = F'$ and the resultant of the action of the air is represented in magnitude, position and direction by the vector R' . Since the elevator occupies the same position in both cases, the shape of the lifting surfaces has not changed and, for this shape, we have all the elements of the reaction of the air

for the angles I and I' and can calculate the value of the stabilizing couple for these angles of attack.

Let us suppose that the normal angle of attack is I and that the center of gravity is at C.G. If, in a gust, the wind attacks the lifting surfaces suddenly with the angle of attack I' , the resultant of the action of the air, which was R an instant before, is now represented by the vector R' . It no longer passes through the center of gravity of the airplane. It has, therefore, with reference to the center of gravity, a moment $M = R'L$. L is measured on Fig. 1. This is a stabilizing moment, since it tends to return the airplane to the position of normal flight with reference to the new direction of the wind.

For other loads placed on the tail of the airplane, we obtain other curves for the positions of the elevator with reference to the angles of attack and we can determine the stabilizing moment for other inclinations of the wind. When the airplane flies, without additional load, at the angle of incidence I , we can then trace the longitudinal stability curve of the airplane, when it flies in a straight line with its elevator in the position corresponding to the ordinate F .

We can make the same determinations for other positions of the elevator and find whether the moment of the resultant of the air with reference to the center of gravity of the airplane is always stabilizing for all these positions, i. e. for all angles of attack. If this moment does not give stability for all angles

of attack, it will be necessary to reduce the angle of attack of the tail plane. On the contrary, if this stabilizing moment is too high, it may be advantageous to reduce it by increasing the angle of attack of the tail plane, so as to render the airplane more maneuverable and thus lessen the work of the pilot. All these tests give the value of the stability of the airplane only during gliding flight. It is therefore desirable to try experiments for determining this stability in horizontal flight with the engine running at different speeds, since the slip stream against the different parts of an airplane changes the value of the reaction of the air. Moreover, if the axis of the propeller is not parallel to the path followed by the airplane, the lifting force of the latter may be either increased or diminished. Lastly, if the axis of the propeller does not pass through the center of gravity of the airplane, it produces a couple which varies with its speed of rotation. The stability also varies with the distribution of the loads on the airplane. Thus with only the pilot, the airplane may be either more or less stable than with several passengers. We must therefore repeat the above tests with a smaller supply of fuel and oil (to reduce the load) and also with as large a load as the airplane can carry.

Translated by the National Advisory Committee for Aeronautics.

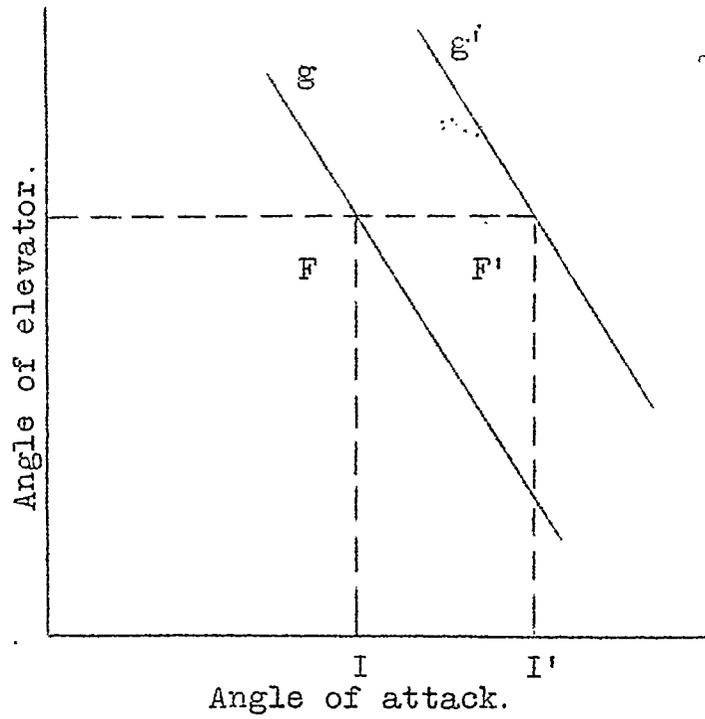


Fig. 1.

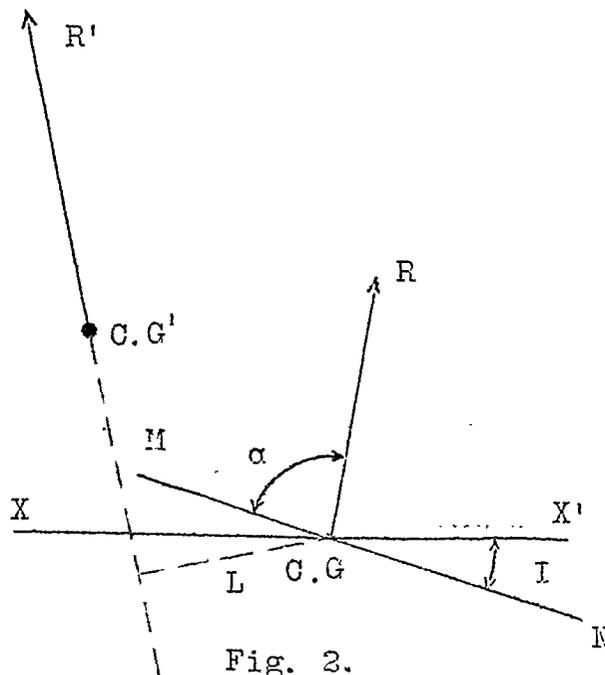


Fig. 2.

NASA Technical Library



3 1176 01440 5105

01 11 76
01 44 0
51 05