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RESEARCH MEMORANDUM

LOW-SPEED PRESSURE-DISTRIBUTION INVESTIGATION OF A THIN-DELTA-WING—FUSELAGE MODEL WITH DOUBLE SLOTTED FLAP, EXTENDED DOUBLE SLOTTED FLAP, AND CANARD

By Delwin R. Croom and Jarrett K. Huffman

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NATIONAL ADVISORY COMMITTEE FOR AERONAUTICS

WASHINGTON

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LOW-SPEED PRESSURE-DISTRIBUTION INVESTIGATION OF A THIN-DELTA-WING—FUSELAGE MODEL WITH DOUBLE SLOTTED FLAP, EXTENDED DOUBLE SLOTTED FLAP, AND CANARD

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SUMMARY

An investigation was made in the Langley 300 MPH 7- by 10-foot tunnel to determine the chordwise and spanwise load distribution on a thin 60° delta-wing—fuselage model equipped with double slotted and extended double slotted flaps. The effects of a canard on the loads of the wing with the extended double slotted flap were also determined. The wing had an aspect ratio of 2.31, taper ratio of 0, sweep of 60° at the leading edge, and NACA 65A003 airfoil section parallel to the free-airstream direction.

Results of the investigation without discussion, are presented in the form of longitudinal aerodynamic characteristics of the extended-double-slotted-flap configuration and in the form of normal-force coefficients and pitching-moment coefficients on the wing, vane, and flap; sample pressure plots; sample span load distributions; tabulated pressure coefficients; and tabulated section normal-force coefficients and section pitching-moment coefficients.

INTRODUCTION

Interest is being shown in the use of delta wings for high-speed airplanes because this plan form has some desirable aerodynamic and structural characteristics. Results of previous investigations (for example, see refs. 1 and 2) indicate that, by employing double slotted flaps on a 60° delta wing, the angle of attack necessary to obtain a given lift coefficient is considerably reduced, thereby making the use of double slotted flaps desirable for the landing configuration.

Although restricted to section loading at one spanwise station, previous investigations have shown very large load over the vane and

flap for double slotted flaps on delta, two-dimensional, and swept wings. (See refs. 1, 3, and 4, respectively.) The present investigation has been made in the Langley 300 MPH 7- by 10-foot tunnel to determine the loads at several spanwise stations on a 60° delta wing model with double slotted and extended double slotted flaps. The investigation also included the effect of sideslip on the loading on the plain-wing configuration and also the effect of a canard on the loading on the extended-double-slotted-flap configuration.

The present paper presents the results without discussion of the investigation in the form of longitudinal aerodynamic characteristics of the complete model and extended-double-slotted-flap configuration and in the form of normal-force and pitching-moment coefficients of the wing, vane, and flap. Included are sample pressure plots, span load distributions, tabulated pressure coefficients, and tabulated section normal-force coefficients and section pitching-moment coefficients.

SYMBOLS

b	wing span (based on theoretical tip, fig. 1), ft
b_f	flap span (exposed span of one flap), ft
b_v	vane span (exposed span of one vane), ft
c	chord, ft
$\bar{c}$	wing mean aerodynamic chord (based on theoretical tip, fig. 1), $\frac{2}{S}\int_0^{b/2} c_w^2 dy$, ft
c_{av}	average wing chord, ft
c_v	vane chord, ft
c_f	flap chord, ft
c_w	plain-wing chord, ft
c_F	fuselage length, ft
S	wing area (based on theoretical tip, fig. 1), sq ft
S_v	vane area, sq ft

S_f flap area, sq ft

C_L lift coefficient, $\frac{\text{Lift of model}}{q_o S}$

$C_{L_{max}}$ maximum lift coefficient

C_D drag coefficient, $\frac{\text{Drag of model}}{q_o S}$

ΔC_D jet-boundary correction applied to drag coefficient

C_m pitching-moment coefficient of model referred to quarter mean aerodynamic chord, $\frac{\text{Pitching moment of model}}{q_o S \bar{c}}$

C_p pressure coefficient, $\frac{H_o - p}{q_o}$

l_v distance from wing quarter chord to vane nose, measured parallel to vane chord, ft

l_f distance from wing quarter chord to flap nose, measured parallel to flap chord, ft

x longitudinal distance, ft

x_v distance from vane nose to center of load on vane, ft

x_f distance from flap nose to center of load on flap, ft

y lateral distance, ft

z vertical distance, ft

H_o free-stream total pressure, lb/sq ft

p local static pressure, lb/sq ft

q_o free-stream dynamic pressure, $\frac{\rho V_o^2}{2}$, lb/sq ft

ρ mass density of air, slugs/cu ft

V_o free-stream velocity, ft/sec

δ_c canard deflection (positive direction, trailing edge down), deg

δ_f flap deflection (positive direction, trailing edge down), deg

δ_v vane deflection (angle between vane chord line and wing chord line; positive direction, trailing edge down; see fig. 2), deg

α angle of attack set in tunnel

$\Delta\alpha$ jet-boundary correction applied to angle of attack

α_c corrected angle of attack

β sideslip angle, deg

$c_{n,v}$ vane section normal-force coefficient, based on vane chord

$c_{n,f}$ flap section normal-force coefficient, based on flap chord

$c_{n,WF}$ section normal-force coefficient of wing forward of slot lip, based on plain-wing chord

$c_{n,w}$ wing section normal-force coefficent, based on plain wing chord,

$$c_{n,WF} + c_{n,v}\left(\frac{c_v}{c_w}\right)\cos\delta_v + c_{n,f}\left(\frac{c_f}{c_w}\right)\cos\delta_f$$

$c_{m,v}$ vane section pitching-moment coefficient, based on vane chord (moments taken about vane nose)

$c_{m,f}$ flap section pitching-moment coefficient, based on flap chord (moments taken about flap nose)

$c_{m,WF}$ section pitching-moment coefficient of wing forward of slot lip, based on plain-wing chord

$c_{m,w}$ wing section pitching-moment coefficient, based on plain-wing chord (moments taken about wing quarter chord),

$$c_{m,WF} - \frac{c_{n,v}\left(l_v + x_v\right)c_v}{c_w^2} - \frac{c_{n,f}\left(l_f + x_f\right)c_f}{c_w^2}$$

$C_{N,w}$ wing normal-force coefficient, $\dfrac{\text{Wing normal force}}{q_oS}$

$C_{N,v}$ vane normal-force coefficient, $\dfrac{\text{Vane normal force}}{q_oS_v}$

$C_{N,f}$ flap normal-force coefficient, $\dfrac{\text{Flap normal force}}{q_oS_f}$

$C_{m,w}$ pitching-moment coefficient of wing, referred to quarter mean aerodynamic chord, $\frac{\text{Pitching moment of wing}}{q_o S \bar{c}}$

$C_{m,v}$ pitching-moment coefficient of vane, referred to vane nose, $\frac{\text{Pitching moment of vane}}{q_o S_v c_v}$

$C_{m,f}$ pitching-moment coefficient of flap, referred to flap nose, $\frac{\text{Pitching moment of flap}}{q_o S_f c_f}$

MODEL AND APPARATUS

The model was tested on the single-support-strut system in the Langley 300 MPH 7- by 10-foot tunnel. The geometric and physical characteristics of the wing-fuselage configuration are given in figure 1 and table I.

The wing of the model had a 60° apex angle, an aspect ratio of 2.31 (based on the theoretical tip), a taper ratio of 0, and an NACA 65A003 airfoil section parallel to the free-stream direction.

The double-slotted-flap configuration and the extended-double-slotted-flap configuration used for this investigation are shown in figure 2. The general arrangement, that is, relation of flap to vane to wing, were obtained from preliminary explorative tests based on the information of the systematic investigations of references 2 and 5.

The flap which extended from the fuselage to 0.67b/2 had a constant chord of 6.86 inches and an exposed area equal to 12.78 percent of the total wing area. The flap leading edge was constructed to the ordinates given in table II. The vane had a constant chord of 1.768 inches and was constructed to the ordinates given in table III. The vane and flap were deflected as a unit about the pivot point shown in figure 2.

The wing, vane, and flap were constructed with flush surface pressure orifices located on the right semispan at the 21-, 30-, 43-, and 55-percent-semispan station. Orifices were also located on the wing at the 72-percent-semispan station and along the fuselage at the plane of symmetry.

TESTS

The tests were performed at a dynamic pressure of approximately 25 pounds per square foot which corresponds to a Mach number of approximately 0.13. Reynolds number based on the mean aerodynamic chord of the model was approximately 2.7×10^6. The tests were run through an angle-of-attack range of approximately -4° through the stall. Flap deflections for the double-slotted-flap configuration were 50°, 60°, and 65°; and for the extended-double-slotted-flap configuration the flap-deflection range was from 45° through the flap deflection for maximum-lift increment at zero angle of attack. The plain-wing configuration was tested at sideslip angles of 0°, -10°, and 10°.

A canard at 20° deflection was tested on the extended-double-slotted-flap configuration. The flap deflection for this test was 60°. (See fig. 2 for detail of canard arrangement.)

CORRECTIONS

The jet-boundary corrections applied to the data of this paper were obtained by the method outlined in reference 6. Jet-boundary corrections applied are as follows:

$$\Delta\alpha = 1.028C_L$$

$$\Delta C_D = 0.0179C_L^2$$

The blockage correction as applied to the dynamic pressure was obtained by the method outlined in reference 7. The buoyancy correction due to the longitudinal-static-pressure gradient in the tunnel as applied to the data increased the drag coefficient by 0.001.

RESULTS

The results of this investigation are presented without discussion. Sample data figures are presented in order to give the reader a general idea of the chordwise and spanwise loadings to expect over a delta wing that is equipped with double slotted or extended double slotted flaps or when a canard is used to trim out the large diving moments caused by using an extended double slotted flap.

The model aerodynamic characteristics in pitch for the double-slotted-flap configuration has been presented in reference 1. The data in the form of figures and tables are as listed below.

Figures

Extended double-slotted-flap configuration longitudinal aerodynamic data 3

Comparison of lift data for double-slotted and extended-double-slotted-flap configurations 4

Sample chordwise pressure distribution . . . . . 5 and 6

Sample span load distributions on wing, vane, and flaps at several angles of attack . . . . . 7, 8, and 9

Sample span load distribution on wing at several flap deflections 10

Normal-force and pitching-moment coefficients of wing, vane, and flaps . . . . . 11 to 20

The pressure coefficients are presented in tables IV to XV.

The section data are presented in tables XVI to XVIII.

Langley Aeronautical Laboratory,
National Advisory Committee for Aeronautics,
Langley Field, Va., August 14, 1956.

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2. Riebe, John M., and MacLeod, Richard G.: Low-Speed Wind-Tunnel Investigation of a Thin 60° Delta Wing With Double Slotted, Single Slotted, Plain, and Split Flaps. NACA RM L52J29, 1953.

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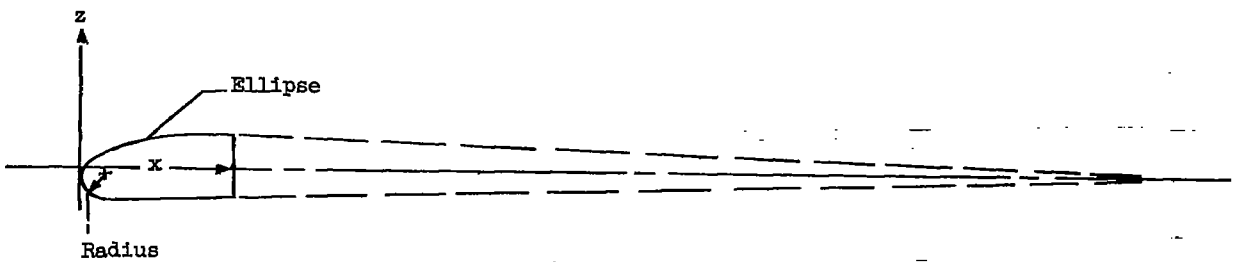
TABLE I.- PHYSICAL CHARACTERISTICS OF THE TEST MODEL

Wing:	
Section parallel to free-stream direction	NACA 65A003
Span, ft	5.00
Aspect ratio (based on theoretical tip)	2.31
Leading-edge sweep, deg	60.00
Trailing-edge sweep, deg	0
Area (based on theoretical tip), sq ft	10.83
Mean aerodynamic chord, ft	2.89
Root chord, ft	4.33
Vane:	
Span, ft	3.33
Chord, ft	0.15
Chord, percent wing root chord	3.40
Chord, percent flap chord	25.77
Flap:	
Span, ft	3.33
Chord, ft	0.57
Chord, percent wing root chord	13.20
Exposed area, sq ft	1.38
Exposed area, percent wing area	12.78
Canard:	
Span, ft	2.24
Area, sq ft	2.17
Area, percent wing area	20.0
Leading-edge sweep, deg	60.0
Trailing-edge sweep, deg	0

TABLE II.- ORDINATES OF THE LEADING EDGE OF THE TRAILING-EDGE FLAP

[All dimensions in inches]



c_f = 6.860 in.

Station, in. x	Lower z	Upper z
0	-0.107	-0.107
.010	Radius	-.059
.030		-.030
.050		-.005
.075		.022
.100		.041
.143	-.245	------
.200	Straight taper	.096
.300		.133
.400		.161
.500		.182
.600		.198
.700		.208
.800		.215
.900		.217
1.000	-.216	.216

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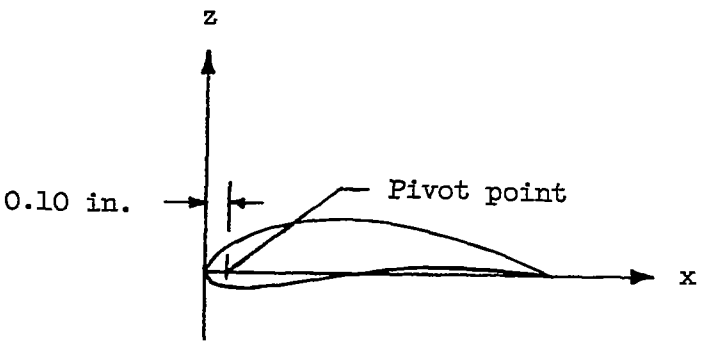
TABLE III.- ORDINATES OF THE VANE

[All dimensions in inches]



c_v = 1.768 in.

Station, in. x	Lower z	Upper z
0	0	0
.022	-.047	.067
.044	-.060	.092
.088	-.072	.131
.133	-.079	.160
.177	-.077	.185
.265	-.072	.224
.354	-.053	.255
.530	-.025	.288
.707	.002	.294
.884	.032	.283
1.061	.053	.255
1.238	.057	.207
1.414	.053	.147
1.591	.032	.080
1.680	.019	.046
1.768	0	0

TABLE IV.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE

(a) Plain wing configuration; $\beta = 0^\circ$; $\frac{y}{b/2} = 0$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Fuselage										
.0000	.000	.000	.000	.000	.006	.000	.052	.103	.123	.142
.0500	.738	.825	.917	.983	.997	1.044	1.077	1.076	1.086	1.109
.1000	.855	.916	1.000	1.009	1.059	1.097	1.120	1.125	1.144	1.169
.1500	.947	.984	1.064	1.064	1.090	1.138	1.154	1.149	1.159	1.172
.2000	.891	1.038	1.099	1.100	1.115	1.144	1.148	1.138	1.147	1.163
.2500	1.035	1.078	1.131	1.116	1.130	1.163	1.151	1.152	1.141	1.178
.3000	1.069	1.094	1.131	1.110	1.118	1.144	1.126	1.122	1.113	1.180
.3500	1.066	1.078	1.115	1.094	1.102	1.122	1.129	1.107	1.123	1.154
.4000	1.028	1.041	1.077	1.055	1.077	1.100	1.117	1.110	1.132	1.178
.4500	1.025	1.038	1.070	1.070	1.105	1.150	1.188	1.204	1.239	1.284
.5000	1.013	1.028	1.086	1.104	1.161	1.231	1.293	1.319	1.368	1.459
.5500	1.018	1.041	1.122	1.155	1.211	1.284	1.349	1.368	1.426	1.604
.6000	1.013	1.050	1.131	1.170	1.217	1.278	1.330	1.316	1.374	1.643
.6500	1.025	1.069	1.147	1.161	1.207	1.250	1.262	1.256	1.303	1.498
.7000	1.028	1.072	1.122	1.149	1.180	1.231	1.225	1.210	1.264	1.402
.7500	1.035	1.078	1.112	1.131	1.155	1.203	1.216	1.219	1.264	1.350
.8000	1.058	1.081	1.112	1.113	1.139	1.184	1.210	1.222	1.276	1.320
.8500	1.054	1.072	1.106	1.104	1.108	1.166	1.194	1.216	1.251	1.272
.9000	1.057	1.053	1.083	1.076	1.074	1.119	1.157	1.161	1.178	1.208
.9500	1.060	1.069	1.093	1.064	1.065	1.100	1.126	1.131	1.147	1.134
.9940	1.130	1.113	1.141	1.097	1.096	1.119	1.123	1.119	1.116	1.145

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Fuselage										
.0500	.865	.856	.808	.702	.641	.591	.503	.456	.423	.408
.1000	.947	.930	.901	.812	.749	.706	.611	.565	.531	.517
.1500	1.019	.993	.952	.897	.833	.778	.694	.638	.607	.595
.2000	1.047	1.034	1.032	.936	.892	.847	.759	.711	.675	.665
.2500	1.066	1.063	1.067	.976	.941	.897	.821	.766	.724	.710
.3000	1.085	1.084	1.080	1.006	.978	.925	.855	.818	.779	.758
.3500	1.064	1.056	1.080	1.005	.969	.946	.883	.836	.797	.767
.4000	1.003	1.009	1.019	.961	.929	.891	.836	.784	.748	.740
.4500	.978	.978	.994	.915	.882	.850	.778	.733	.696	.677
.5000	.956	.972	.965	.882	.836	.781	.704	.634	.616	.604
.5500	.950	.953	.949	.848	.789	.731	.654	.602	.561	.556
.6000	.947	.931	.897	.800	.743	.678	.608	.553	.503	.523
.7500	1.205	1.184	1.179	1.091	1.059	1.018	.950	.906	.848	.847
.9500	1.019	1.013	1.026	.982	.972	.959	.944	.934	.911	.912
.9940	1.000	1.056	1.115	1.085	1.102	1.106	1.111	1.088	1.092	1.103

TABLE IV.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(b) Plain wing configuration; $\beta = 0^\circ$; $\frac{y}{b/2} = 0.21$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	1.302	.834	1.316	2.697	4.529	6.726	9.381	10.220	8.680	4.077
.0125	.829	1.056	1.428	2.299	6.954	9.464	12.300	11.720	8.420	3.722
.0250	.875	1.069	1.328	1.852	3.087	5.067	7.969	9.045	7.602	3.691
.0500	.903	1.063	1.261	1.562	1.881	2.375	3.949	4.551	5.348	5.456
.0750	.919	1.056	1.224	1.454	1.630	1.973	2.367	3.127	4.096	3.299
.1000	.941	1.060	1.206	1.404	1.603	1.929	2.341	2.742	3.551	3.197
.1500	.959	1.066	1.181	1.349	1.494	1.650	2.050	2.276	2.813	2.978
.2000	.972	1.079	1.169	1.315	1.441	1.561	1.889	2.049	2.436	2.784
.2500	.984	1.078	1.172	1.284	1.391	1.506	1.793	1.924	2.140	2.524
.3000	.997	1.089	1.159	1.282	1.360	1.464	1.718	1.770	1.978	2.319
.3500	.997	1.085	1.153	1.241	1.317	1.424	1.601	1.667	1.841	2.120
.4000	1.003	1.082	1.144	1.231	1.292	1.396	1.533	1.591	1.744	1.984
.4500	1.006	1.091	1.141	1.210	1.280	1.366	1.486	1.539	1.667	1.879
.5000	1.031	1.104	1.147	1.225	1.280	1.326	1.455	1.500	1.626	1.796
.5500	1.009	1.085	1.113	1.188	1.230	1.265	1.402	1.433	1.564	1.713
.6000	1.012	1.075	1.110	1.167	1.227	1.253	1.380	1.442	1.564	1.666
.6500	1.015	1.072	1.089	1.139	1.193	1.217	1.372	1.434	1.545	1.614
.7000	1.000	1.060	1.070	1.129	1.174	1.189	1.372	1.415	1.498	1.552
.8000	1.015	1.040	1.064	1.096	1.124	1.189	1.294	1.273	1.358	1.450
.8500	1.003	1.050	1.043	1.065	1.084	1.149	1.195	1.179	1.265	1.295
.9000	.994	1.035	1.034	1.052	1.072	1.107	1.121	1.106	1.196	1.230
.9500	.978	1.009	1.003	1.029	1.025	1.049	1.037	1.039	1.121	1.250

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	1.246	.994	.751	.620	.568	.610	.706	.730	.716	.503
.0250	1.165	.981	.791	.628	.491	.418	.348	.348	.324	.238
.0500	1.131	.994	.898	.670	.525	.378	.288	.255	.218	.219
.0750	1.112	1.003	.856	.704	.559	.430	.328	.297	.287	.295
.1000	1.109	1.025	.868	.747	.596	.479	.395	.367	.352	.361
.1500	1.100	1.016	.889	.772	.642	.540	.492	.459	.445	.447
.2000	1.128	1.056	.932	.821	.708	.610	.542	.512	.495	.494
.2500	1.168	1.085	.969	.873	.770	.677	.601	.582	.564	.543
.3000	1.165	1.125	1.004	.901	.808	.713	.646	.618	.617	.596
.3500	1.208	1.160	1.041	.949	.879	.781	.715	.689	.660	.639
.4000	1.230	1.185	1.083	1.000	.916	.826	.765	.727	.707	.679
.4500	1.243	1.198	1.107	1.031	.954	.866	.802	.767	.735	.718
.5000	1.227	1.191	1.113	1.037	.963	.878	.839	.776	.760	.738
.5500	1.196	1.166	1.126	1.031	.975	.893	.832	.791	.791	.756
.6000	1.189	1.166	1.095	1.043	.981	.912	.841	.815	.807	.775
.6500	1.168	1.151	1.086	1.021	.972	.896	.873	.827	.807	.781
.7000	1.146	1.135	1.080	1.021	.989	.909	.876	.850	.829	.812
.8000	1.106	1.110	1.058	1.025	.981	.927	.907	.867	.878	.864
.8500	1.084	1.078	1.049	1.003	.991	.951	.920	.897	.894	.895
.9000	1.050	1.053	1.009	.994	.969	.942	.916	.891	.910	.917
.9500	1.037	1.047	1.024	1.009	.991	.980	.983	.936	.966	.997

TABLE IV.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(c) Plain wing configuration; $\beta = 0^\circ$; $\frac{y}{b/2} = 0.30$.

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	1.776	.909	1.619	2.438	3.149	4.028	4.554	4.430	4.050	3.154
.0125	.816	1.119	1.785	2.451	3.100	3.677	4.034	4.045	3.953	3.172
.0250	.869	1.104	1.840	2.549	3.205	3.826	4.155	4.039	3.965	3.172
.0500	.910	1.100	1.733	2.298	3.544	3.747	4.003	4.066	3.984	3.216
.0750	.919	1.085	1.276	2.811	4.591	5.040	4.659	4.348	4.054	3.194
.1000	.938	1.078	1.165	1.703	3.379	4.574	4.622	4.275	3.969	3.166
.1500	.950	1.078	1.154	1.191	1.744	2.820	3.870	3.809	3.679	3.101
.2000	.972	1.085	1.184	1.254	1.391	2.003	2.981	3.172	3.284	2.984
.2500	.987	1.075	1.169	1.237	1.317	1.474	2.449	2.691	2.950	2.842
.3000	.997	1.091	1.165	1.234	1.298	1.549	2.173	2.412	2.704	2.716
.3500	.991	1.085	1.150	1.219	1.264	1.445	1.923	2.153	2.436	2.558
.4000	1.003	1.091	1.147	1.222	1.275	1.403	1.805	1.982	2.274	2.432
.4500	1.006	1.100	1.156	1.216	1.264	1.393	1.752	1.912	2.193	2.364
.5000	1.009	1.100	1.158	1.197	1.242	1.357	1.650	1.775	2.006	2.176
.5500	1.009	1.082	1.129	1.173	1.224	1.323	1.594	1.494	1.903	2.092
.6000	1.025	1.075	1.123	1.167	1.208	1.302	1.545	1.621	1.832	2.003
.6500	1.028	1.085	1.116	1.165	1.180	1.284	1.495	1.591	1.748	1.916
.7500	.987	1.050	1.067	1.099	1.151	1.210	1.384	1.470	1.620	1.744
.8000	1.003	1.038	1.055	1.074	1.106	1.174	1.341	1.427	1.545	1.660
.8500	.991	1.016	1.037	1.055	1.072	1.137	1.303	1.376	1.464	1.583
.9000	.966	.997	1.006	1.021	1.047	1.085	1.231	1.303	1.384	1.484
.9500	.944	.969	.969	.981	1.003	1.018	1.155	1.185	1.237	1.379

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	1.246	.959	.761	.759	.876	1.079	1.272	1.291	1.377	1.299
.0250	1.258	1.003	.794	.716	.702	.762	.848	.848	.903	.867
.0500	1.190	1.013	.843	.719	.655	.616	.598	.582	.617	.596
.0750	1.184	1.028	.877	.762	.655	.595	.543	.515	.526	.512
.1000	1.171	1.044	.914	.790	.677	.588	.536	.491	.492	.478
.1500	1.148	1.085	.960	.833	.714	.637	.570	.512	.508	.481
.2000	1.190	1.113	.981	.880	.764	.668	.591	.535	.517	.481
.2500	1.204	1.138	1.024	.926	.804	.701	.628	.570	.554	.522
.3000	1.218	1.147	1.043	.950	.851	.747	.672	.612	.595	.555
.3500	1.234	1.172	1.067	.988	.891	.790	.715	.658	.635	.596
.4000	1.221	1.172	1.080	1.000	.910	.814	.755	.691	.673	.630
.4500	1.230	1.182	1.092	1.031	.944	.857	.793	.730	.713	.670
.5000	1.212	1.182	1.098	1.031	.997	.875	.814	.748	.744	.713
.5500	1.212	1.179	1.107	1.049	.975	.896	.839	.785	.773	.738
.6000	1.177	1.147	1.086	1.025	.966	.906	.848	.791	.794	.762
.6500	1.162	1.138	1.073	1.028	.978	.906	.879	.827	.816	.790
.7500	1.131	1.091	1.060	1.018	.981	.921	.895	.864	.857	.834
.8000	1.106	1.075	1.031	1.012	.969	.918	.907	.873	.878	.864
.8500	1.096	1.056	1.027	1.009	.988	.942	.938	.906	.913	.913
.9000	1.072	1.056	1.018	1.009	.994	.948	.966	.936	.959	.972
.9500	1.053	1.035	1.024	1.021	1.019	.979	1.022	.994	1.022	1.063

TABLE IV.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(d) Plain wing configuration; $\beta = 0^\circ$; $\frac{y}{b/2} = 0.48$.

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	1.520	.890	1.636	1.981	2.491	2.662	2.694	2.618	2.714	2.688
.0125	.791	1.094	1.638	1.975	2.454	2.637	2.774	2.730	2.794	2.713
.0250	.850	1.094	1.592	2.006	2.454	2.650	2.882	2.782	2.810	2.710
.0500	.903	1.082	1.583	2.074	2.994	2.738	2.827	2.766	2.838	2.691
.0750	.931	1.082	1.626	2.129	2.528	2.686	2.805	2.797	2.855	2.694
.1000	.944	1.075	1.589	2.339	2.475	2.576	2.783	2.827	2.872	2.694
.1500	.975	1.075	1.386	2.611	2.991	2.704	2.910	2.903	2.903	2.703
.2000	.984	1.082	1.211	2.092	3.288	3.144	3.051	2.918	2.875	2.691
.2500	.994	1.082	1.129	1.481	2.569	3.000	2.907	2.779	2.763	2.629
.3000	1.000	1.097	1.132	1.207	2.041	2.796	2.830	2.718	2.732	2.614
.3500	1.009	1.091	1.132	1.126	1.606	2.387	2.585	2.530	2.592	2.527
.4000	1.019	1.091	1.141	1.133	1.413	2.113	2.440	2.424	2.524	2.475
.4500	1.028	1.100	1.129	1.133	1.308	1.924	2.315	2.309	2.427	2.419
.5000	1.019	1.094	1.110	1.117	1.214	1.689	2.108	2.157	2.280	2.302
.5500	1.031	1.082	1.119	1.123	1.199	1.579	1.994	2.060	2.212	2.253
.6000	1.028	1.088	1.107	1.120	1.180	1.497	1.904	1.966	2.124	2.182
.7000	1.015	1.060	1.085	1.089	1.143	1.362	1.715	1.797	1.953	2.055
.7500	1.025	1.053	1.070	1.080	1.137	1.314	1.663	1.745	1.906	2.012
.8000	1.015	1.047	1.049	1.049	1.100	1.268	1.588	1.663	1.835	1.955
.8500	.994	1.025	1.021	1.025	1.078	1.204	1.489	1.576	1.748	1.892
.9000	.978	1.003	.988	.991	1.044	1.153	1.409	1.506	1.687	1.830
.9500	.941	.950	.935	.926	.963	1.059	1.285	1.376	1.581	1.762

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	1.558	1.047	.794	.809	.976	1.122	1.279	1.303	1.614	1.466
.0250	1.495	1.088	.834	.759	.792	.832	.913	.915	.975	1.012
.0500	1.452	1.107	.886	.778	.733	.698	.709	.706	.732	.747
.0750	1.383	1.104	.911	.802	.756	.662	.659	.627	.639	.660
.1000	1.339	1.116	.939	.827	.745	.659	.635	.600	.611	.628
.1500	1.283	1.129	.975	.861	.786	.671	.635	.603	.586	.593
.2000	1.262	1.138	.997	.886	.801	.704	.650	.609	.592	.589
.2500	1.258	1.151	1.021	.920	.834	.732	.675	.631	.611	.605
.3000	1.240	1.147	1.031	.932	.863	.744	.700	.654	.632	.628
.3500	1.242	1.163	1.055	.960	.895	.787	.734	.679	.667	.657
.4000	1.249	1.163	1.067	.979	.919	.814	.765	.715	.688	.685
.4500	1.227	1.157	1.070	.984	.929	.826	.783	.739	.720	.713
.5000	1.212	1.147	1.054	.984	.941	.845	.802	.751	.741	.731
.5500	1.194	1.129	1.054	.991	.960	.857	.824	.776	.773	.768
.6000	1.181	1.128	1.061	1.003	.960	.875	.851	.806	.797	.790
.7000	1.159	1.116	1.070	1.012	.988	.924	.910	.857	.843	.861
.7500	1.140	1.104	1.082	1.006	.994	.933	.929	.900	.897	.901
.8000	1.112	1.085	1.052	1.000	.997	.945	.941	.912	.922	.941
.8500	1.087	1.072	1.031	.994	1.000	.951	.975	.951	.972	.988
.9000	1.078	1.066	1.031	1.003	1.006	.982	1.019	.994	1.028	1.050
.9500	1.090	1.088	1.058	1.031	1.044	1.034	1.111	1.109	1.148	1.253

TABLE IV.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(e) Plain wing configuration; $\beta = 0^\circ$; $\frac{y}{b/2} = 0.55$.

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	1.237	.941	1.595	1.754	2.106	2.156	2.173	2.163	2.261	2.299
.0125	.844	1.116	1.598	1.808	2.140	2.188	2.224	2.227	2.311	2.321
.0250	.829	1.085	1.607	1.784	2.104	2.137	2.232	2.230	2.314	2.324
.0500	.846	1.085	1.564	1.811	2.090	2.185	2.279	2.263	2.327	2.305
.0750	.908	1.083	1.463	1.836	2.100	2.204	2.251	2.263	2.336	2.308
.1000	.925	1.091	1.402	1.864	2.134	2.211	2.283	2.285	2.336	2.296
.1500	.950	1.091	1.353	1.873	2.115	2.128	2.285	2.291	2.330	2.277
.2000	.959	1.088	1.307	2.009	2.072	2.095	2.307	2.300	2.327	2.277
.2500	.984	1.087	1.291	2.268	2.121	2.095	2.331	2.318	2.352	2.284
.3000	.997	1.107	1.285	2.305	2.404	2.183	2.345	2.330	2.349	2.287
.3500	1.003	1.104	1.261	2.102	2.375	2.290	2.341	2.306	2.333	2.277
.4000	1.009	1.097	1.230	1.821	2.497	2.317	2.300	2.260	2.296	2.250
.4500	1.028	1.110	1.202	1.565	2.326	2.290	2.254	2.215	2.274	2.225
.5000	1.028	1.100	1.187	1.361	2.140	2.223	2.204	2.179	2.233	2.197
.6000	1.000	1.059	1.135	1.096	1.770	1.997	2.062	2.060	2.137	2.134
.6500	1.025	1.082	1.126	1.052	1.609	1.903	1.991	2.003	2.093	2.105
.7000	1.028	1.072	1.104	1.025	1.494	1.808	1.929	1.939	2.037	2.074
.7500	1.022	1.053	1.095	1.009	1.407	1.714	1.885	1.903	1.994	2.046
.8500	1.006	1.016	1.052	.957	1.249	1.573	1.735	1.794	1.913	2.009
.9000	.973	1.000	1.012	.941	1.168	1.479	1.678	1.691	1.832	1.969
.9500	.959	.975	.995	.910	1.078	1.332	1.525	1.570	1.719	1.848

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	1.729	1.110	.828	.880	1.067	1.232	1.415	1.448	1.573	1.651
.0250	1.682	1.232	.846	.805	.863	.887	1.012	1.018	1.106	1.160
.0500	1.648	1.147	.826	.821	.795	.756	.793	.797	.835	.861
.0750	1.486	1.129	.939	.802	.758	.695	.703	.694	.723	.728
.1000	1.427	1.132	.951	.827	.758	.689	.678	.661	.670	.691
.1500	1.343	1.132	.981	.852	.783	.695	.669	.648	.632	.639
.2000	1.318	1.144	1.009	.880	.814	.707	.681	.642	.642	.639
.2500	1.299	1.141	1.015	.915	.829	.735	.700	.670	.640	.648
.3000	1.271	1.147	1.031	.925	.857	.753	.715	.688	.678	.660
.3500	1.250	1.154	1.035	.941	.895	.784	.737	.715	.701	.691
.4000	1.268	1.160	1.064	.940	.916	.802	.783	.739	.726	.716
.4500	1.237	1.141	1.055	.966	.913	.826	.789	.751	.735	.741
.5000	1.227	1.147	1.047	.981	.938	.842	.820	.779	.766	.768
.6000	1.199	1.122	1.047	.997	.969	.878	.873	.836	.822	.827
.6500	1.190	1.119	1.047	1.000	.981	.912	.892	.864	.860	.861
.7000	1.149	1.094	1.032	.991	.997	.927	.916	.882	.888	.901
.7500	1.140	1.094	1.061	.997	1.005	.936	.944	.921	.928	.957
.8000	1.137	1.094	1.052	1.009	1.022	.970	.991	.976	1.000	1.009
.8500	1.121	1.082	1.041	1.018	1.047	1.006	1.031	1.027	1.047	1.080
.9000	1.096	1.065	1.038	1.015	1.050	1.043	1.067	1.088	1.134	1.158
.9500	1.075	1.066	1.043	1.018	1.075	1.095	1.189	1.200	1.268	1.341

TABLE IV.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Concluded

(f) Plain wing configuration; $\beta = 0^\circ$; $\frac{y}{b/2} = 0.72$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	1.530	.958	1.635	1.687	1.777	1.841	1.821	1.839	1.877	1.897
.0125	.793	1.034	1.609	1.651	1.734	1.781	1.771	1.812	1.862	1.894
.0250	.815	1.063	1.593	1.691	1.734	1.803	1.796	1.833	1.883	1.897
.0500	.865	1.069	1.583	1.666	1.734	1.791	1.818	1.848	1.912	1.916
.0750	.890	1.063	1.608	1.684	1.743	1.825	1.842	1.867	1.908	1.912
.1000	.890	1.059	1.558	1.657	1.715	1.809	1.833	1.839	1.902	1.906
.1500	.921	1.069	1.561	1.654	1.715	1.834	1.858	1.873	1.914	1.909
.2000	.966	1.091	1.580	1.681	1.728	1.841	1.870	1.891	1.911	1.906
.2500	.988	1.100	1.529	1.687	1.734	1.822	1.870	1.891	1.902	1.897
.3000	.984	1.091	1.439	1.672	1.712	1.806	1.861	1.878	1.892	1.882
.3500	1.022	1.116	1.397	1.684	1.706	1.822	1.864	1.870	1.892	1.878
.4000	1.028	1.116	1.335	1.687	1.681	1.806	1.843	1.854	1.874	1.867
.4500	1.038	1.116	1.298	1.708	1.678	1.788	1.849	1.851	1.862	1.855
.5000	1.047	1.122	1.276	1.775	1.703	1.788	1.839	1.842	1.852	1.849
.5500	1.038	1.109	1.257	1.815	1.706	1.769	1.827	1.824	1.840	1.846
.6000	1.028	1.103	1.218	1.818	1.700	1.750	1.802	1.803	1.834	1.831
.6500	1.032	1.097	1.189	1.809	1.718	1.738	1.787	1.781	1.813	1.825
.7000	1.038	1.084	1.179	1.803	1.728	1.741	1.774	1.769	1.794	1.819
.7500	1.047	1.097	1.165	1.784	1.734	1.741	1.759	1.751	1.785	1.816
.8000	1.047	1.066	1.144	1.713	1.728	1.734	1.750	1.730	1.779	1.818
.8500	1.038	1.056	1.119	1.614	1.684	1.694	1.719	1.702	1.751	1.794
.9000	1.013	1.044	1.099	1.523	1.647	1.675	1.685	1.672	1.730	1.791

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	1.667	1.225	.872	.827	.892	1.031	1.105	1.155	1.227	1.284
.0250	1.661	1.219	.894	.775	.802	.872	.929	.970	1.009	1.057
.0500	1.588	1.175	.917	.760	.742	.759	.781	.800	.816	.861
.0750	1.525	1.219	.958	.781	.742	.738	.734	.742	.751	.773
.1000	1.519	1.181	.974	.793	.743	.719	.710	.699	.712	.737
.1500	1.469	1.165	.990	.812	.752	.722	.691	.681	.675	.689
.2000	1.437	1.150	1.003	.839	.786	.747	.710	.687	.684	.686
.2500	1.352	1.119	1.006	.864	.796	.766	.728	.702	.699	.698
.3000	1.296	1.119	1.015	.894	.833	.788	.756	.724	.718	.716
.3500	1.264	1.134	1.045	.915	.864	.825	.778	.748	.739	.740
.4000	1.203	1.116	1.051	.927	.876	.841	.802	.769	.764	.764
.4500	1.183	1.109	1.051	.948	.898	.859	.821	.790	.770	.782
.5000	1.134	1.106	1.064	.958	.916	.875	.833	.800	.785	.798
.5500	1.123	1.097	1.067	.958	.926	.881	.861	.824	.816	.813
.6000	1.110	1.106	1.077	.979	.944	.922	.889	.854	.843	.827
.6500	1.113	1.091	1.080	.991	.969	.950	.920	.888	.874	.876
.7000	1.088	1.072	1.064	.991	.978	.956	.941	.915	.902	.903
.7500	1.079	1.059	1.054	.997	.997	.991	.972	.942	.937	.967
.8000	1.067	1.044	1.051	1.006	1.016	1.025	1.012	1.003	1.009	1.012
.8500	1.032	1.028	1.056	1.024	1.039	1.076	1.080	1.044	1.075	1.085
.9000	1.016	1.028	1.054	1.038	1.115	1.153	1.151	1.134	1.162	1.184
.9500	.969	.997	1.038	1.076	1.192	1.241	1.256	1.249	1.288	1.326

TABLE V.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE

(a) Plain wing configuration; $\beta = 10^{\circ}$; $\frac{y}{b/2} = 0$

Upper surface

x/c	C_p for -									
	$\alpha = -4^{\circ}$	$\alpha = 0^{\circ}$	$\alpha = 4^{\circ}$	$\alpha = 8^{\circ}$	$\alpha = 12^{\circ}$	$\alpha = 16^{\circ}$	$\alpha = 20^{\circ}$	$\alpha = 22^{\circ}$	$\alpha = 24^{\circ}$	$\alpha = 26^{\circ}$
Fuselage										
.0000	.067	.076	.079	.080	.127	.148	.185	.231	.259	.306
.0500	.850	.948	.982	1.074	1.136	1.133	1.218	1.294	1.358	1.455
.1000	.932	1.028	1.064	1.126	1.175	1.163	1.159	1.160	1.183	1.237
.1500	1.006	1.076	1.119	1.166	1.202	1.202	1.185	1.181	1.201	1.244
.2000	1.041	1.122	1.143	1.191	1.211	1.196	1.200	1.193	1.210	1.284
.2500	1.095	1.167	1.180	1.222	1.224	1.187	1.206	1.208	1.230	1.287
.3000	1.138	1.176	1.184	1.203	1.199	1.193	1.223	1.219	1.239	1.320
.3500	1.123	1.158	1.164	1.173	1.160	1.196	1.241	1.267	1.309	1.383
.4000	1.089	1.122	1.125	1.126	1.154	1.205	1.276	1.314	1.382	1.464
.4500	1.084	1.122	1.128	1.136	1.196	1.278	1.412	1.421	1.481	1.590
.5000	1.047	1.114	1.128	1.172	1.266	1.393	1.550	1.552	1.638	1.778
.5500	1.067	1.113	1.148	1.215	1.341	1.480	1.626	1.745	1.886	1.982
.6000	1.086	1.140	1.164	1.255	1.378	1.511	1.665	1.813	1.915	2.033
.6500	1.086	1.137	1.183	1.246	1.369	1.474	1.689	1.759	1.877	1.994
.7000	1.077	1.140	1.161	1.215	1.341	1.426	1.515	1.694	1.825	1.940
.7500	1.092	1.134	1.140	1.200	1.284	1.326	1.403	1.517	1.726	1.871
.8000	1.098	1.140	1.155	1.168	1.227	1.257	1.332	1.504	1.627	1.778
.8500	1.110	1.140	1.155	1.172	1.175	1.202	1.250	1.403	1.519	1.662
.9000	1.101	1.128	1.137	1.158	1.157	1.124	1.194	1.264	1.382	1.506
.9500	1.129	1.137	1.155	1.172	1.130	1.115	1.141	1.181	1.262	1.374
.9940	1.199	1.219	1.204	1.215	1.202	1.157	1.182	1.198	1.227	1.284

Lower surface

x/c	C_p for -									
	$\alpha = -4^{\circ}$	$\alpha = 0^{\circ}$	$\alpha = 4^{\circ}$	$\alpha = 8^{\circ}$	$\alpha = 12^{\circ}$	$\alpha = 16^{\circ}$	$\alpha = 20^{\circ}$	$\alpha = 22^{\circ}$	$\alpha = 24^{\circ}$	$\alpha = 26^{\circ}$
Fuselage										
.0500	.932	.958	.869	.812	.755	.642	.576	.543	.519	.524
.1000	1.015	1.031	.955	.920	.864	.761	.700	.644	.609	.602
.1500	1.073	1.088	1.021	.988	.924	.840	.771	.721	.697	.683
.2000	1.104	1.122	1.067	1.040	.976	.897	.841	.780	.755	.743
.2500	1.123	1.143	1.104	1.077	1.036	.943	.882	.828	.802	.795
.3000	1.141	1.152	1.122	1.102	1.057	.982	.932	.890	.845	.829
.3500	1.123	1.134	1.113	1.117	1.078	1.003	.938	.899	.875	.841
.4000	1.064	1.094	1.082	1.068	1.018	.958	.900	.857	.828	.814
.4500	1.040	1.073	1.052	1.054	.997	.918	.859	.807	.784	.754
.5000	1.021	1.058	1.018	1.009	.946	.858	.788	.748	.732	.710
.5500	1.021	1.058	1.000	.969	.909	.819	.753	.718	.688	.677
.6000	.994	1.018	.958	.917	.856	.764	.700	.665	.656	.614
.7500	1.243	1.237	1.189	1.169	1.115	1.039	.997	.973	.982	.944
.9500	1.027	1.079	1.070	1.074	1.069	1.015	.962	.932	.927	.946
.9940	.989	1.125	1.187	1.188	1.187	1.145	1.132	1.086	1.108	1.141

TABLE V.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(b) Plain wing configuration; $\beta = 10^{\circ}$; $\frac{y}{b/2} = 0.21$

Upper surface

x/c	C_p for -									
	$\alpha = -4^{\circ}$	$\alpha = 0^{\circ}$	$\alpha = 4^{\circ}$	$\alpha = 8^{\circ}$	$\alpha = 12^{\circ}$	$\alpha = 16^{\circ}$	$\alpha = 20^{\circ}$	$\alpha = 22^{\circ}$	$\alpha = 24^{\circ}$	$\alpha = 26^{\circ}$
Wing										
.0000	1.159	.759	1.222	2.736	4.498	6.892	8.764	2.411	2.233	2.185
.0125	.754	1.043	1.480	2.392	5.564	9.806	10.438	2.326	2.084	2.057
.0250	.804	1.043	1.344	1.889	3.653	5.376	8.323	2.350	2.097	2.054
.0500	.850	1.046	1.272	1.577	1.786	2.431	4.443	2.208	2.069	2.043
.0750	.878	1.050	1.238	1.460	1.638	1.957	3.107	2.199	2.072	2.037
.1000	.903	1.053	1.213	1.429	1.588	1.834	2.749	2.181	2.075	2.060
.1500	.931	1.050	1.188	1.352	1.477	1.692	2.275	2.169	2.069	2.056
.2000	.944	1.056	1.188	1.312	1.424	1.603	2.049	2.148	2.072	2.066
.2500	.959	1.059	1.166	1.278	1.381	1.539	1.902	2.136	2.057	2.060
.3000	.944	1.062	1.156	1.247	1.337	1.477	1.792	2.097	2.033	2.054
.3500	.956	1.062	1.150	1.219	1.307	1.434	1.704	2.033	2.006	2.030
.4000	.956	1.056	1.134	1.213	1.279	1.400	1.640	1.976	1.974	2.027
.4500	.956	1.059	1.131	1.207	1.263	1.369	1.613	1.903	1.955	2.030
.5000	.978	1.068	1.141	1.188	1.248	1.360	1.584	1.861	1.946	2.009
.5500	.981	1.056	1.128	1.157	1.220	1.314	1.523	1.813	1.921	1.985
.6000	.981	1.037	1.106	1.139	1.207	1.308	1.505	1.785	1.903	1.955
.6500	.978	1.034	1.094	1.114	1.167	1.280	1.462	1.741	1.873	1.940
.7000	.950	1.025	1.063	1.074	1.130	1.231	1.419	1.714	1.837	1.910
.8000	.944	1.009	1.047	1.052	1.096	1.197	1.270	1.649	1.737	1.821
.8500	.953	1.031	1.066	1.058	1.105	1.194	1.285	1.607	1.710	1.789
.9000	.941	.994	1.019	1.018	1.053	1.123	1.281	1.548	1.640	1.725
.9500	.950	.926	.949	.950	.975	1.114	1.242	1.541	1.607	1.669

Lower surface

x/c	C_p for -									
	$\alpha = -4^{\circ}$	$\alpha = 0^{\circ}$	$\alpha = 4^{\circ}$	$\alpha = 8^{\circ}$	$\alpha = 12^{\circ}$	$\alpha = 16^{\circ}$	$\alpha = 20^{\circ}$	$\alpha = 22^{\circ}$	$\alpha = 24^{\circ}$	$\alpha = 26^{\circ}$
Wing										
.0125	1.156	.950	.713	.582	.517	.563	.605	.363	.336	.346
.0250	1.115	.947	.730	.580	.455	.391	.315	.272	.242	.213
.0500	1.063	.943	.606	.639	.498	.354	.257	.272	.233	.206
.0750	1.062	.949	.619	.670	.542	.397	.306	.311	.272	.242
.1000	1.072	.978	.650	.691	.563	.431	.342	.350	.311	.281
.1500	1.089	.978	.846	.731	.613	.498	.413	.424	.381	.382
.2000	1.050	1.006	.909	.775	.663	.554	.465	.468	.438	.400
.2500	1.078	1.037	.953	.818	.712	.597	.511	.505	.486	.442
.3000	1.068	1.062	1.016	.858	.746	.643	.566	.538	.520	.493
.3500	1.078	1.093	1.053	.904	.805	.698	.630	.622	.577	.543
.4000	1.118	1.127	1.075	.947	.848	.754	.673	.650	.619	.591
.4500	1.168	1.142	1.084	.969	.879	.788	.709	.695	.656	.633
.5000	1.143	1.142	1.186	.991	.904	.809	.746	.734	.701	.678
.5500	1.187	1.192	1.075	1.108	1.040	.969	.936	.918	.894	.890
.6000	1.153	1.124	1.056	.997	.932	.862	.801	.798	.761	.734
.6500	1.128	1.108	1.059	.997	.932	.868	.816	.819	.782	.764
.7000	1.103	1.087	1.044	1.009	.954	.880	.847	.843	.819	.803
.8000	1.072	1.062	1.028	1.000	.957	.920	.899	.909	.897	.890
.8500	1.044	1.037	1.059	.994	.960	.932	.930	.955	.949	.942
.9000	1.019	1.062	1.000	1.012	.994	.991	1.000	1.034	1.051	1.037
.9580	.989	1.006	1.025	.972	.954	.991	.985	1.085	1.097	1.116

TABLE V.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(c) Plain wing configuration; $\beta = 10^\circ$; $\frac{y}{b/2} = 0.30$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	1.642	.752	1.888	2.484	3.265	3.788	4.046	2.172	2.031	2.036
.0125	.729	1.153	1.713	2.555	3.183	3.643	3.804	2.163	2.040	2.048
.0250	.807	1.124	1.547	2.657	3.347	3.797	3.847	2.163	2.043	2.054
.0500	.866	1.108	1.369	3.234	3.399	3.894	3.798	2.154	2.040	2.054
.0750	.885	1.090	1.288	3.345	4.746	4.182	3.895	2.160	2.043	2.051
.1000	.900	1.087	1.209	2.398	4.353	4.403	3.838	2.169	2.049	2.057
.1500	.925	1.077	1.225	1.281	2.260	3.859	3.590	2.157	2.063	2.054
.2000	.947	1.090	1.194	1.176	1.505	2.828	3.199	2.148	2.072	2.060
.2500	.919	1.081	1.184	1.176	1.303	2.209	2.847	2.160	2.078	2.054
.3000	.950	1.090	1.159	1.191	1.260	1.917	2.618	2.124	2.057	2.054
.3500	.969	1.087	1.163	1.176	1.223	1.686	2.242	2.091	2.036	2.048
.4000	.969	1.087	1.189	1.182	1.226	1.588	2.196	2.063	2.024	2.036
.4500	.994	1.093	1.144	1.179	1.217	1.557	2.150	2.039	2.012	2.030
.5000	1.000	1.087	1.128	1.163	1.211	1.492	1.963	1.988	1.979	2.006
.5500	.975	1.071	1.122	1.162	1.183	1.409	1.878	1.970	1.967	1.994
.6000	.969	1.074	1.113	1.145	1.186	1.372	1.807	1.921	1.943	1.973
.6500	.997	1.068	1.083	1.129	1.176	1.342	1.723	1.873	1.909	1.943
.7500	.959	1.031	1.047	1.065	1.111	1.274	1.596	1.788	1.837	1.878
.8000	.923	1.028	1.041	1.058	1.102	1.240	1.544	1.737	1.779	1.839
.8500	.922	1.015	1.003	1.034	1.074	1.200	1.498	1.692	1.734	1.785
.9000	.903	.994	.969	.994	1.022	1.145	1.422	1.625	1.680	1.731
.9500	.897	.957	.863	.957	.972	1.071	1.321	1.589	1.625	1.681

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	1.263	.916	.719	.902	.678	.812	.730	.725	.740	.767
.0250	1.255	.960	.778	.589	.554	.569	.596	.526	.528	.528
.0500	1.184	.985	.838	.638	.520	.465	.443	.429	.417	.400
.0750	1.146	1.006	.863	.675	.554	.477	.413	.420	.396	.376
.1000	1.140	1.012	.916	.719	.598	.499	.431	.432	.408	.376
.1500	1.134	1.043	.954	.768	.650	.545	.468	.471	.441	.409
.2000	1.140	1.077	.991	.818	.700	.594	.508	.508	.477	.448
.2500	1.149	1.099	.991	.867	.739	.643	.563	.550	.520	.487
.3000	1.177	1.127	1.019	.892	.793	.695	.605	.598	.548	.528
.3500	1.194	1.149	1.047	.923	.830	.729	.648	.634	.607	.564
.4000	1.196	1.149	1.069	.947	.867	.786	.694	.671	.643	.603
.4500	1.199	1.184	1.078	.981	.907	.815	.734	.719	.692	.648
.5000	1.177	1.161	1.091	.991	.920	.825	.765	.732	.719	.690
.5500	1.153	1.158	1.106	1.019	.935	.862	.801	.795	.767	.734
.6000	1.115	1.127	1.079	.997	.944	.865	.826	.810	.785	.755
.6500	1.100	1.118	1.072	1.009	.954	.886	.847	.846	.821	.794
.7500	1.019	1.093	1.053	.997	.953	.917	.890	.888	.885	.846
.8000	.966	1.062	1.041	.991	.949	.914	.917	.921	.918	.901
.8500	1.012	1.065	1.044	1.000	.978	.957	.951	.976	.979	.961
.9000	.972	1.046	1.034	.991	.981	.972	.994	1.048	1.066	1.037
.9500	1.004	1.043	1.031	1.005	.997	1.009	1.076	1.175	1.187	1.284

TABLE V.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(d) Plain wing configuration; $\beta = 10^\circ$; $\frac{y}{b/2} = 0.45$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	1.371	.793	1.841	2.098	2.523	2.634	2.581	2.081	2.024	2.024
.0125	.679	1.136	1.831	2.092	2.483	2.594	2.618	2.072	2.024	2.030
.0250	.754	1.154	1.794	2.508	2.474	2.615	2.657	2.075	2.024	2.027
.0500	.816	1.121	1.709	2.145	2.911	2.695	2.691	2.073	2.030	2.034
.0750	.857	1.118	1.631	2.191	2.582	2.686	2.648	2.064	2.015	2.024
.1000	.878	1.115	1.531	2.262	2.551	2.643	2.673	2.063	2.027	2.030
.1500	.888	1.099	1.353	2.537	2.619	2.615	2.642	2.060	2.027	2.021
.2000	.916	1.096	1.263	2.484	3.108	2.745	2.615	2.048	2.018	2.021
.2500	.931	1.099	1.225	2.163	3.078	2.754	2.523	2.030	2.000	2.003
.3000	.978	1.099	1.238	1.864	2.796	2.729	2.489	2.030	2.000	2.015
.3500	.984	1.105	1.215	1.487	2.362	2.523	2.385	2.021	1.985	2.012
.4000	.978	1.099	1.222	1.293	2.065	2.409	2.318	1.994	1.976	2.003
.4500	1.003	1.102	1.206	1.173	1.842	2.286	2.245	1.973	1.961	1.991
.5000	.966	1.084	1.191	1.099	1.568	2.097	2.121	1.958	1.939	1.985
.5500	.984	1.084	1.178	1.089	1.492	2.006	2.073	1.924	1.936	1.944
.6000	1.012	1.090	1.149	1.077	1.469	1.892	1.997	1.891	1.912	1.949
.7000	.991	1.065	1.131	1.057	1.279	1.723	1.824	1.857	1.861	1.913
.7500	.997	1.053	1.118	1.037	1.248	1.677	1.835	1.828	1.846	1.890
.8000	1.004	1.031	1.084	1.028	1.211	1.600	1.774	1.779	1.816	1.846
.8500	.984	1.019	1.072	1.000	1.158	1.529	1.791	1.752	1.791	1.851
.9000	.966	.985	1.041	.978	1.115	1.448	1.670	1.725	1.761	1.824
.9500	.922	.938	.988	.910	1.031	1.348	1.578	1.646	1.737	1.797

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	1.589	.994	.706	.636	.721	.837	.945	.858	.885	.919
.0250	1.402	1.037	.781	.626	.548	.603	.690	.616	.619	.639
.0500	1.327	1.050	.859	.670	.582	.523	.514	.505	.501	.501
.0750	1.271	1.048	.891	.710	.616	.523	.483	.486	.462	.464
.1000	1.284	1.081	.922	.759	.650	.545	.492	.495	.471	.464
.1500	1.263	1.105	.966	.799	.684	.591	.523	.523	.492	.469
.2000	1.255	1.118	.991	.842	.737	.637	.563	.553	.520	.490
.2500	1.224	1.123	1.028	.889	.771	.668	.596	.589	.559	.522
.3000	1.237	1.130	1.038	.913	.805	.711	.633	.616	.589	.555
.3500	1.221	1.152	1.059	.954	.843	.751	.679	.656	.625	.603
.4000	1.243	1.161	1.072	.972	.879	.791	.716	.707	.666	.639
.4500	1.209	1.149	1.078	.988	.892	.803	.746	.737	.704	.672
.5000	1.206	1.142	1.088	.994	.913	.834	.771	.767	.728	.707
.5500	1.193	1.136	1.084	1.006	.929	.862	.804	.804	.764	.743
.6000	1.171	1.124	1.084	1.015	.941	.877	.829	.828	.804	.785
.7000	1.133	1.111	1.094	1.034	.981	.926	.902	.904	.891	.866
.7500	1.128	1.094	1.081	1.028	.994	.954	.936	.946	.930	.922
.8000	1.084	1.071	1.072	1.025	1.000	.957	.954	.991	.976	.944
.8500	1.078	1.062	1.059	1.018	1.006	.988	1.003	1.039	1.033	1.033
.9000	1.059	1.062	1.059	1.025	1.031	1.046	1.070	1.124	1.136	1.151
.9500	1.078	1.087	1.088	1.052	1.077	1.126	1.208	1.314	1.332	1.340

TABLE V.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(e) Plain wing configuration; $\beta = 10^\circ$; $\frac{y}{b/2} = 0.56$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	.959	.904	2.153	1.901	2.118	2.108	2.128	1.973	1.967	1.971
.0125	.698	1.201	1.938	1.915	2.136	2.142	2.162	1.970	1.964	1.979
.0250	.732	1.149	1.788	1.886	2.108	2.132	2.168	1.967	1.955	1.976
.0500	.807	1.118	1.572	1.898	2.099	2.157	2.183	1.955	1.949	1.970
.0750	.850	1.118	1.456	1.910	2.090	2.160	2.194	1.946	1.945	1.958
.1000	.897	1.124	1.400	1.923	2.105	2.169	2.194	1.934	1.933	1.952
.1500	.925	1.115	1.328	1.938	2.115	2.154	2.163	1.915	1.921	1.927
.2000	.941	1.108	1.291	1.990	2.096	2.148	2.134	1.909	1.915	1.925
.2500	.959	1.118	1.272	2.048	2.077	2.123	2.107	1.897	1.897	1.919
.3000	.997	1.130	1.253	2.135	2.124	2.123	2.089	1.882	1.891	1.910
.3500	1.000	1.130	1.241	2.126	2.232	2.120	2.061	1.870	1.885	1.904
.4000	1.025	1.130	1.234	2.071	2.248	2.092	2.024	1.867	1.882	1.904
.4500	1.025	1.130	1.216	1.994	2.211	2.062	2.009	1.843	1.882	1.895
.5000	1.031	1.130	1.203	1.907	2.149	2.040	1.988	1.843	1.870	1.892
.6000	1.037	1.131	1.156	1.650	1.944	1.911	1.914	1.810	1.840	1.869
.6500	1.025	1.087	1.150	1.552	1.861	1.862	1.884	1.801	1.843	1.859
.7000	1.034	1.081	1.141	1.454	1.789	1.819	1.853	1.788	1.819	1.846
.7500	1.025	1.068	1.125	1.367	1.728	1.772	1.813	1.776	1.813	1.837
.8500	1.003	1.037	1.049	1.222	1.604	1.695	1.741	1.746	1.791	1.827
.9000	.994	.991	1.031	1.126	1.529	1.643	1.706	1.728	1.776	1.815
.9500	.963	.991	1.006	1.065	1.393	1.514	1.636	1.686	1.713	1.752

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	1.878	1.053	.722	.685	.811	.932	1.088	1.048	1.109	1.149
.0250	1.885	1.096	.802	.663	.847	.655	.728	.740	.785	.782
.0500	1.542	1.118	.872	.704	.625	.572	.578	.607	.601	.594
.0750	1.448	1.108	.891	.722	.622	.560	.538	.558	.555	.522
.1000	1.402	1.111	.925	.739	.656	.563	.526	.550	.523	.501
.1500	1.327	1.127	.963	.805	.703	.594	.547	.553	.538	.504
.2000	1.296	1.146	.997	.846	.744	.634	.578	.592	.565	.516
.2500	1.284	1.185	1.019	.883	.789	.683	.618	.622	.598	.561
.3000	1.274	1.152	1.041	.907	.814	.714	.651	.683	.634	.603
.3500	1.277	1.164	1.063	.944	.861	.734	.697	.683	.665	.634
.4000	1.263	1.158	1.078	.966	.885	.785	.731	.731	.698	.663
.4500	1.240	1.158	1.078	.978	.901	.812	.759	.761	.734	.701
.5000	1.240	1.158	1.084	.994	.920	.834	.798	.795	.767	.734
.6000	1.224	1.149	1.091	1.018	.963	.892	.856	.870	.840	.803
.6500	1.206	1.136	1.100	1.031	.994	.920	.893	.900	.882	.842
.7000	1.171	1.211	1.081	1.031	1.000	.935	.924	.930	.921	.890
.7500	1.153	1.105	1.075	1.034	1.025	.972	.963	.982	.961	.937
.8000	1.134	1.103	1.078	1.037	1.040	1.009	1.015	1.033	1.024	1.006
.8500	1.125	1.087	1.078	1.052	1.071	1.058	1.088	1.106	1.106	1.084
.9000	1.115	1.084	1.066	1.074	1.099	1.092	1.141	1.187	1.199	1.173
.9500	1.084	1.071	1.078	1.062	1.184	1.209	1.278	1.333	1.365	1.370

TABLE V.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Concluded

(f) Plain wing configuration; $\beta = 10^\circ$; $\frac{y}{b/2} = 0.72$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	1.540	.757	2.043	1.739	1.770	1.764	1.732	1.727	1.717	1.784
.0125	.632	1.085	1.927	1.732	1.744	1.722	1.706	1.727	1.732	1.787
.0250	.708	1.113	1.824	1.735	1.749	1.731	1.709	1.718	1.723	1.784
.0500	.794	1.125	1.742	1.735	1.749	1.734	1.720	1.718	1.723	1.787
.0750	.828	1.119	1.678	1.735	1.749	1.752	1.738	1.700	1.720	1.787
.1000	.846	1.097	1.550	1.729	1.725	1.737	1.729	1.697	1.721	1.790
.1500	.880	1.115	1.426	1.729	1.734	1.746	1.738	1.688	1.708	1.775
.2000	.920	1.119	1.368	1.726	1.725	1.732	1.747	1.685	1.699	1.766
.2500	.945	1.128	1.298	1.723	1.728	1.749	1.735	1.679	1.691	1.763
.3000	.966	1.122	1.268	1.711	1.716	1.745	1.729	1.670	1.688	1.757
.3500	.988	1.125	1.252	1.705	1.707	1.722	1.706	1.670	1.679	1.748
.4000	1.006	1.128	1.234	1.680	1.701	1.704	1.691	1.662	1.679	1.748
.4500	1.012	1.128	1.216	1.677	1.689	1.692	1.673	1.659	1.682	1.746
.5000	1.009	1.131	1.216	1.677	1.680	1.648	1.662	1.656	1.676	1.748
.5500	1.024	1.122	1.189	1.632	1.659	1.654	1.641	1.647	1.679	1.754
.6000	1.006	1.113	1.170	1.612	1.637	1.632	1.632	1.644	1.670	1.748
.6500	1.024	1.110	1.149	1.588	1.610	1.607	1.609	1.635	1.673	1.757
.7000	1.021	1.104	1.137	1.579	1.592	1.589	1.603	1.641	1.679	1.763
.7500	1.021	1.085	1.128	1.584	1.580	1.577	1.594	1.638	1.682	1.766
.8000	1.015	1.073	1.094	1.526	1.559	1.548	1.588	1.644	1.685	1.769
.8500	1.012	1.061	1.078	1.486	1.532	1.533	1.570	1.647	1.685	1.760
.9000	.994	1.049	1.055	1.455	1.514	1.514	1.556	1.635	1.670	1.772

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	1.932	1.170	.696	.615	.677	.752	.806	.846	.892	.967
.0250	1.902	1.143	.751	.625	.613	.640	.671	.688	.732	.769
.0500	1.727	1.131	.800	.665	.595	.568	.573	.587	.600	.644
.0750	1.512	1.143	.865	.723	.625	.562	.553	.561	.565	.590
.1000	1.377	1.131	.876	.732	.644	.588	.550	.543	.557	.569
.1500	1.215	1.122	.912	.775	.686	.598	.565	.555	.560	.563
.2000	1.187	1.113	.921	.794	.722	.651	.594	.593	.580	.590
.2500	1.181	1.113	.942	.834	.752	.659	.623	.608	.615	.611
.3000	1.202	1.110	.970	.868	.788	.701	.662	.647	.641	.641
.3500	1.193	1.104	.973	.885	.816	.734	.688	.685	.679	.677
.4000	1.156	1.097	.979	.914	.843	.761	.715	.709	.714	.710
.4500	1.141	1.088	.988	.923	.867	.798	.747	.739	.752	.740
.5000	1.141	1.091	1.000	.951	.894	.831	.785	.788	.781	.775
.5500	1.119	1.097	1.006	.966	.921	.858	.815	.810	.807	.805
.6000	.981	1.091	1.015	.991	.945	.879	.847	.840	.845	.853
.6500	1.095	1.079	1.015	.997	.970	.912	.882	.893	.892	.898
.7000	1.085	1.067	1.009	1.000	.994	.949	.912	.929	.927	.934
.7500	1.073	1.061	1.003	1.025	1.012	.976	.950	.976	.971	.988
.8000	1.046	1.044	1.003	1.040	1.054	1.015	.991	1.035	1.035	1.057
.8500	1.012	1.040	1.006	1.055	1.094	1.069	1.053	1.101	1.096	1.144
.9000	.997	1.024	1.003	1.105	1.160	1.139	1.132	1.173	1.195	1.244
.9500	.975	1.015	1.003	1.154	1.229	1.227	1.241	1.294	1.318	1.392

TABLE VI.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE

(a) Plain wing configuration; $\beta = -10^\circ$; $\frac{y}{b/2} = 0$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Fuselage										
.0000	.000	.000	.006	.015	.000	.009	.033	.077	.103	.159
.0500	.894	.927	1.025	1.044	1.125	1.139	1.242	1.326	1.411	1.441
.1000	.977	1.009	1.094	1.116	1.181	1.151	1.173	1.169	1.196	1.204
.1500	1.055	1.073	1.169	1.164	1.197	1.187	1.197	1.187	1.205	1.192
.2000	1.094	1.113	1.175	1.173	1.209	1.196	1.215	1.213	1.250	1.214
.2500	1.139	1.149	1.213	1.192	1.223	1.196	1.215	1.219	1.220	1.219
.3000	1.165	1.159	1.218	1.186	1.200	1.190	1.230	1.231	1.272	1.272
.3500	1.169	1.165	1.203	1.161	1.181	1.190	1.251	1.284	1.317	1.311
.4000	1.139	1.116	1.159	1.125	1.150	1.214	1.289	1.329	1.342	1.358
.4500	1.119	1.116	1.153	1.128	1.194	1.274	1.418	1.465	1.495	1.447
.5000	1.094	1.119	1.159	1.161	1.289	1.376	1.545	1.598	1.622	1.580
.5500	1.100	1.119	1.175	1.201	1.321	1.473	1.645	1.693	1.728	1.663
.6000	1.107	1.122	1.194	1.237	1.378	1.506	1.673	1.746	1.770	1.687
.6500	1.100	1.134	1.181	1.243	1.375	1.479	1.673	1.752	1.794	1.719
.7000	1.113	1.134	1.197	1.216	1.322	1.431	1.605	1.698	1.779	1.748
.7500	1.119	1.131	1.191	1.195	1.269	1.352	1.491	1.627	1.740	1.719
.8000	1.126	1.140	1.181	1.173	1.222	1.280	1.394	1.515	1.707	1.672
.8500	1.142	1.140	1.181	1.152	1.191	1.214	1.288	1.459	1.652	1.613
.9000	1.123	1.119	1.159	1.154	1.153	1.163	1.224	1.388	1.596	1.586
.9500	1.158	1.151	1.175	1.151	1.134	1.109	1.102	1.332	1.492	1.492
.9940	1.261	1.226	1.256	1.216	1.200	1.172	1.197	1.302	1.372	1.343

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Fuselage										
.0060	1.003	.939	.913	.848	.794	.693	.627	.550	.541	.500
.1000	1.068	1.037	.997	.918	.879	.789	.703	.660	.643	.607
.1500	1.126	1.073	1.036	.997	.938	.858	.770	.731	.722	.657
.2000	1.145	1.104	1.097	1.028	1.000	.907	.836	.796	.773	.737
.2500	1.174	1.137	1.138	1.070	1.041	.985	.891	.852	.829	.796
.3000	1.177	1.146	1.147	1.100	1.069	.991	.939	.900	.885	.840
.3500	1.171	1.140	1.163	1.094	1.094	1.015	.984	.920	.894	.873
.4000	1.107	1.058	1.109	1.053	1.041	.967	.921	.879	.867	.826
.4500	1.084	1.061	1.058	1.024	.997	.919	.873	.823	.804	.760
.5000	1.084	1.055	1.053	.991	.963	.861	.806	.758	.740	.707
.5500	1.068	1.043	1.038	.964	.925	.816	.751	.698	.683	.651
.6000	1.068	1.015	1.003	.909	.859	.769	.709	.657	.646	.613
.7500	1.287	1.235	1.238	1.158	1.138	1.063	1.006	.979	.970	.944
.9500	1.071	1.064	1.094	1.058	1.063	1.003	.973	.958	.985	.932
.9940	.987	1.114	1.206	1.180	1.184	1.130	1.142	1.142	1.181	1.126

TABLE VI.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(b) Plain wing configuration; $\beta = -10^\circ$; $\frac{y}{b/2} = 0.21$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	.921	.771	1.121	2.043	3.762	5.988	8.063	9.777	10.954	7.538
.0125	.878	1.031	1.255	2.043	4.982	8.307	10.784	12.586	12.740	7.322
.0250	.900	1.034	1.221	1.518	2.186	3.814	5.716	7.245	8.731	6.460
.0500	.931	1.040	1.211	1.381	1.744	2.380	3.075	3.675	4.688	4.984
.0750	.944	1.034	1.190	1.345	1.637	2.081	2.609	2.915	3.399	4.107
.1000	.962	1.040	1.196	1.329	1.589	2.009	2.328	2.598	3.027	3.686
.1500	.982	1.034	1.183	1.311	1.515	1.728	1.953	2.137	2.432	2.898
.2000	.978	1.052	1.171	1.271	1.409	1.534	1.722	1.897	2.081	2.412
.2500	.994	1.058	1.165	1.244	1.305	1.412	1.541	1.684	1.852	2.087
.3000	1.006	1.064	1.155	1.220	1.293	1.344	1.472	1.587	1.719	1.934
.3500	1.009	1.064	1.140	1.183	1.229	1.365	1.459	1.590	1.662	1.815
.4000	1.022	1.070	1.137	1.180	1.223	1.284	1.513	1.620	1.680	1.779
.4500	1.022	1.064	1.140	1.171	1.222	1.325	1.408	1.477	1.547	1.698
.5000	1.041	1.076	1.140	1.168	1.207	1.243	1.250	1.298	1.390	1.591
.5500	1.038	1.067	1.109	1.128	1.156	1.211	1.097	1.149	1.245	1.484
.6000	1.041	1.067	1.109	1.119	1.110	1.093	1.088	1.161	1.227	1.397
.6500	1.038	1.067	1.100	1.082	1.092	1.102	1.109	1.170	1.217	1.334
.7000	1.022	1.046	1.072	1.082	1.095	1.103	1.151	1.183	1.199	1.316
.8000	1.031	1.046	1.062	1.067	1.092	1.108	1.134	1.176	1.202	1.263
.8500	1.025	1.052	1.059	1.073	1.092	1.121	1.134	1.183	1.199	1.260
.9000	1.013	1.018	1.031	1.037	1.046	1.056	1.066	1.100	1.124	1.158
.9500	.988	.967	.950	.973	.991	1.022	1.041	1.061	1.000	1.138

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	1.079	.979	.839	.468	.398	.404	.654	.772	.810	.729
.0250	1.085	.973	.848	.674	.534	.424	.343	.377	.369	.394
.0500	1.079	.979	.882	.701	.558	.469	.335	.344	.335	.400
.0750	1.088	.991	.888	.758	.592	.458	.400	.433	.459	.487
.1000	1.110	1.000	.898	.762	.643	.529	.419	.535	.534	.549
.1500	1.097	1.009	.919	.796	.704	.641	.628	.648	.631	.630
.2000	1.119	1.040	.960	.851	.784	.724	.697	.708	.692	.687
.2500	1.138	1.082	1.003	.909	.854	.811	.772	.784	.752	.746
.3000	1.154	1.104	1.034	.954	.906	.888	.831	.827	.804	.794
.3500	1.172	1.146	1.087	1.018	.963	.920	.889	.876	.848	.821
.4000	1.201	1.165	1.131	1.038	1.021	.972	.916	.904	.879	.840
.4500	1.229	1.192	1.140	1.082	1.046	.981	.941	.927	.903	.884
.5000	1.220	1.180	1.140	1.092	1.052	1.009	.959	.942	.903	.874
.5500	1.226	1.210	1.205	1.142	1.134	1.045	1.069	1.078	1.051	1.039
.6000	1.201	1.165	1.118	1.085	1.049	.991	.947	.950	.900	.887
.6500	1.172	1.128	1.096	1.092	1.018	.969	.919	.906	.873	.866
.7000	1.157	1.119	1.084	1.040	1.005	.957	.905	.894	.864	.857
.8000	1.113	1.079	1.053	1.018	.982	.938	.905	.894	.867	.857
.8500	1.091	1.058	1.034	1.021	.982	.941	.905	.900	.879	.887
.9000	1.091	1.092	1.078	1.064	1.028	.991	.959	.961	.933	.943
.9500	1.028	1.015	1.009	.988	.967	.932	.906	.912	.897	.907

TABLE VI.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(c) Plain wing configuration; $\beta = -10^{\circ}$; $\frac{y}{b/2} = 0.30$

Upper surface

x/c	C_p for -									
	$\alpha = -4^{\circ}$	$\alpha = 0^{\circ}$	$\alpha = 4^{\circ}$	$\alpha = 8^{\circ}$	$\alpha = 12^{\circ}$	$\alpha = 16^{\circ}$	$\alpha = 20^{\circ}$	$\alpha = 22^{\circ}$	$\alpha = 24^{\circ}$	$\alpha = 26^{\circ}$
Wing										
.0000	1.853	.934	1.687	2.195	2.976	3.938	4.363	4.323	4.241	3.898
.0125	.850	1.082	1.749	2.204	2.799	3.674	4.113	4.283	4.235	3.898
.0250	.878	1.079	1.923	2.287	2.805	3.437	4.005	4.119	4.166	3.940
.0500	.912	1.061	1.510	2.985	3.589	4.362	5.003	4.897	4.646	4.098
.0750	.925	1.058	1.137	2.076	3.546	4.672	5.441	5.414	4.947	4.125
.1000	.937	1.052	1.165	1.381	2.512	3.697	4.572	4.818	4.625	3.940
.1500	.947	1.046	1.149	1.214	1.695	2.415	3.191	3.624	3.834	3.606
.2000	.949	1.058	1.168	1.238	1.470	1.913	2.441	2.800	3.103	3.194
.2500	.956	1.055	1.159	1.210	1.387	1.697	2.034	2.335	2.637	2.872
.3000	.978	1.058	1.146	1.214	1.345	1.585	1.878	2.134	2.390	2.638
.3500	.984	1.058	1.140	1.204	1.302	1.498	1.728	1.927	2.136	2.376
.4000	.997	1.070	1.149	1.198	1.290	1.468	1.678	1.860	2.015	2.230
.4500	1.013	1.076	1.146	1.195	1.299	1.474	1.703	1.848	1.973	2.140
.5000	1.019	1.064	1.134	1.198	1.258	1.443	1.615	1.684	1.767	1.928
.5500	.994	1.052	1.127	1.180	1.265	1.368	1.416	1.420	1.429	1.552
.6000	1.006	1.047	1.124	1.186	1.281	1.378	1.393	1.368	1.458	1.698
.6500	1.019	1.070	1.118	1.174	1.271	1.234	1.228	1.289	1.387	1.594
.7500	.985	1.037	1.084	1.116	1.156	1.093	1.141	1.207	1.263	1.409
.8000	.956	1.024	1.072	1.088	1.098	1.090	1.122	1.176	1.224	1.352
.8500	.969	1.009	1.044	1.052	1.065	1.059	1.091	1.134	1.175	1.249
.9000	.925	.979	1.003	1.009	1.000	1.009	1.031	1.070	1.103	1.173
.9500	.915	.936	.972	.967	.954	.994	.969	.988	.987	1.089

Lower surface

x/c	C_p for -									
	$\alpha = -4^{\circ}$	$\alpha = 0^{\circ}$	$\alpha = 4^{\circ}$	$\alpha = 8^{\circ}$	$\alpha = 12^{\circ}$	$\alpha = 16^{\circ}$	$\alpha = 20^{\circ}$	$\alpha = 22^{\circ}$	$\alpha = 24^{\circ}$	$\alpha = 26^{\circ}$
Wing										
.0125	1.235	.948	.839	.881	1.049	1.265	1.525	1.654	1.713	1.719
.0250	1.246	.991	.856	.817	.857	.941	1.085	1.128	1.145	1.164
.0500	1.194	1.003	.882	.805	.759	.746	.750	.760	.761	.764
.0750	1.160	1.021	.910	.817	.744	.700	.672	.666	.643	.633
.1000	1.163	1.028	.935	.838	.747	.681	.628	.620	.577	.573
.1500	1.160	1.064	.969	.878	.793	.715	.650	.620	.574	.556
.2000	1.169	1.075	1.006	.909	.808	.712	.641	.602	.589	.528
.2500	1.194	1.104	1.041	.953	.842	.746	.672	.632	.577	.555
.3000	1.201	1.119	1.068	.957	.863	.768	.691	.663	.607	.582
.3500	1.224	1.137	1.084	.997	.890	.805	.725	.693	.643	.612
.4000	1.226	1.143	1.100	1.012	.921	.824	.747	.724	.671	.639
.4500	1.223	1.152	1.109	1.021	.942	.854	.784	.751	.704	.675
.5000	1.204	1.134	1.112	1.028	.960	.873	.816	.778	.728	.710
.5500	1.188	1.146	1.112	1.046	.973	.892	.838	.806	.795	.734
.6000	1.181	1.113	1.090	1.021	.960	.885	.828	.812	.770	.749
.6500	1.141	1.101	1.081	1.021	.957	.890	.836	.827	.792	.770
.7500	1.066	1.064	1.059	1.000	.951	.895	.863	.848	.813	.812
.8000	1.022	1.040	1.034	.991	.942	.892	.866	.851	.828	.812
.8500	1.050	1.046	1.034	1.003	.951	.910	.884	.872	.843	.842
.9000	1.028	1.034	1.028	.988	.957	.920	.900	.900	.870	.849
.9500	1.050	1.046	1.041	1.009	.973	.944	.934	.930	.909	.913

TABLE VI.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(d) Plain wing configuration; $\beta = -10^{\circ}$; $\frac{y}{b/2} = 0.43$

Upper surface

x/c	C_p for -									
	$\alpha = -4^{\circ}$	$\alpha = 0^{\circ}$	$\alpha = 4^{\circ}$	$\alpha = 8^{\circ}$	$\alpha = 12^{\circ}$	$\alpha = 16^{\circ}$	$\alpha = 20^{\circ}$	$\alpha = 22^{\circ}$	$\alpha = 24^{\circ}$	$\alpha = 26^{\circ}$
Wing										
.0000	1.508	.957	1.541	1.817	2.177	2.567	2.906	2.912	2.834	2.749
.0125	.884	1.082	1.806	1.802	2.171	2.615	2.963	2.985	2.973	2.818
.0250	.900	1.079	1.828	1.835	2.256	2.724	3.031	3.034	2.961	2.824
.0500	.919	1.076	1.575	1.851	2.211	2.847	3.903	2.915	2.936	2.860
.0750	.944	1.067	1.724	1.933	2.235	2.625	3.897	2.894	2.967	2.898
.1000	.950	1.070	1.631	2.232	2.467	2.805	3.041	3.022	3.075	2.943
.1500	.947	1.076	1.205	2.012	3.025	3.492	3.683	3.450	3.314	2.979
.2000	.969	1.076	1.121	1.439	2.518	3.294	3.609	3.487	3.266	2.925
.2500	.978	1.064	1.134	1.190	1.863	2.687	3.165	3.192	3.042	2.782
.3000	1.009	1.070	1.155	1.168	1.552	2.240	2.856	2.985	2.934	2.704
.3500	.997	1.070	1.157	1.149	1.354	1.870	2.413	2.627	2.671	2.546
.4000	1.003	1.079	1.152	1.156	1.278	1.672	2.153	2.417	2.523	2.466
.4500	1.025	1.079	1.149	1.156	1.247	1.548	1.991	2.231	2.359	2.361
.5000	.981	1.061	1.121	1.128	1.192	1.449	1.784	1.973	2.124	2.221
.5500	1.006	1.064	1.137	1.131	1.192	1.393	1.675	1.800	1.952	2.113
.6000	1.028	1.058	1.127	1.131	1.168	1.353	1.563	1.632	1.810	1.994
.7000	1.009	1.034	1.103	1.092	1.131	1.288	1.331	1.596	1.574	1.785
.7500	1.013	1.037	1.081	1.076	1.107	1.235	1.269	1.341	1.501	1.701
.8000	1.006	1.021	1.065	1.058	1.098	1.152	1.200	1.274	1.420	1.597
.8500	.991	1.006	1.037	1.037	1.073	1.081	1.150	1.195	1.332	1.504
.9000	.975	.973	1.006	1.000	1.044	1.028	1.075	1.128	1.289	1.400
.9500	.934	.921	.934	.970	1.004	.947	.991	1.009	1.103	1.257

Lower surface

x/c	C_p for -									
	$\alpha = -4^{\circ}$	$\alpha = 0^{\circ}$	$\alpha = 4^{\circ}$	$\alpha = 8^{\circ}$	$\alpha = 12^{\circ}$	$\alpha = 16^{\circ}$	$\alpha = 20^{\circ}$	$\alpha = 22^{\circ}$	$\alpha = 24^{\circ}$	$\alpha = 26^{\circ}$
Wing										
.0125	1.555	1.046	.941	.988	1.174	1.443	1.646	1.705	1.749	1.755
.0250	1.599	1.064	.934	.912	.970	1.084	1.216	1.228	1.260	1.263
.0500	1.643	1.076	.981	.896	.879	.904	.950	.927	.958	.952
.0750	1.282	1.082	.997	.909	.840	.842	.875	.827	.837	.827
.1000	1.210	1.088	1.003	.918	.860	.817	.813	.775	.767	.758
.1500	1.228	1.113	1.037	.933	.869	.799	.778	.733	.723	.687
.2000	1.223	1.119	1.041	.948	.875	.805	.769	.720	.692	.663
.2500	1.216	1.131	1.042	.967	.881	.817	.769	.727	.686	.660
.3000	1.213	1.120	1.048	.976	.893	.820	.772	.720	.698	.649
.3500	1.216	1.131	1.087	1.003	.927	.848	.813	.751	.728	.678
.4000	1.223	1.137	1.070	1.009	.942	.864	.828	.769	.743	.781
.4500	1.201	1.137	1.096	1.012	.939	.870	.825	.778	.752	.710
.5000	1.182	1.116	1.084	1.012	.959	.876	.841	.781	.767	.719
.5500	1.172	1.116	1.078	1.012	.951	.892	.850	.800	.785	.731
.6000	1.154	1.101	1.084	1.012	.954	.898	.864	.809	.795	.767
.7000	1.138	1.098	1.072	1.015	.963	.923	.891	.851	.840	.806
.7500	1.122	1.082	1.065	1.018	.973	.933	.900	.863	.849	.824
.8000	1.094	1.084	1.059	1.006	.963	.929	.916	.876	.867	.846
.8500	1.072	1.052	1.041	1.003	.963	.955	.925	.885	.885	.872
.9000	1.063	1.043	1.041	1.012	.982	.947	.938	.918	.918	.916
.9500	1.075	1.067	1.072	1.037	1.015	.991	.991	.985	.991	1.009

TABLE VI.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(e) Plain wing configuration; $\beta = -10^\circ$; $\frac{y}{b/2} = 0.66$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	1.334	.976	1.413	1.634	1.924	2.232	2.372	2.326	2.290	2.269
.0125	.962	1.095	1.438	1.683	1.942	2.263	2.413	2.374	2.378	2.328
.0250	.928	1.067	1.426	1.642	1.930	2.260	2.438	2.408	2.399	2.319
.0500	.937	1.073	1.419	1.677	1.997	2.348	2.497	2.420	2.396	2.319
.0750	.953	1.067	1.438	1.717	1.988	2.322	2.416	2.344	2.365	2.319
.1000	.959	1.070	1.463	1.734	1.961	2.257	2.349	2.307	2.356	2.322
.1500	.964	1.070	1.556	1.735	1.963	2.211	2.294	2.268	2.353	2.328
.2000	.964	1.070	1.482	2.000	2.235	2.341	2.284	2.355	2.390	2.344
.2500	.994	1.073	1.332	2.055	2.601	2.490	2.653	2.508	2.477	2.367
.3000	1.016	1.079	1.190	1.848	2.607	2.841	2.819	2.617	2.513	2.370
.3500	1.019	1.088	1.127	1.534	2.329	2.814	2.872	2.681	2.554	2.349
.4000	1.022	1.082	1.112	1.305	1.970	2.585	2.784	2.642	2.507	2.322
.4500	1.035	1.082	1.127	1.198	1.680	2.337	2.663	2.572	2.462	2.298
.5000	1.041	1.079	1.131	1.149	1.509	2.111	2.525	2.490	2.411	2.260
.6000	1.022	1.082	1.093	1.079	1.247	1.709	2.119	2.171	2.178	2.128
.6500	1.091	1.058	1.109	1.088	1.189	1.576	1.931	2.009	2.075	2.044
.7000	1.031	1.046	1.096	1.079	1.153	1.477	1.772	1.876	1.973	2.009
.7500	1.031	1.082	1.084	1.058	1.125	1.381	1.638	1.757	1.867	1.949
.8500	1.009	1.018	1.044	1.018	1.058	1.226	1.416	1.544	1.688	1.827
.9000	.981	.985	1.006	.982	1.018	1.130	1.284	1.401	1.568	1.719
.9500	.946	.963	.988	.997	.982	1.045	1.193	1.231	1.349	1.543

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	1.580	1.119	.991	1.088	1.314	1.591	1.769	1.803	1.852	1.894
.0250	1.649	1.146	.997	.988	1.082	1.211	1.334	1.325	1.381	1.391
.0500	1.655	1.143	1.031	.960	.949	.997	1.047	1.034	1.037	1.043
.0750	1.645	1.125	1.009	.955	.921	.898	.919	.897	.912	.910
.1000	1.648	1.125	1.028	.931	.867	.858	.869	.836	.849	.836
.1500	1.652	1.131	1.044	.942	.890	.836	.809	.778	.778	.755
.2000	1.238	1.131	1.056	.954	.899	.839	.794	.763	.752	.728
.2500	1.198	1.137	1.062	.982	.912	.842	.803	.757	.744	.719
.3000	1.191	1.137	1.065	.982	.918	.851	.803	.763	.758	.719
.3500	1.204	1.140	1.078	.997	.924	.854	.825	.772	.735	.740
.4000	1.210	1.134	1.093	1.012	.936	.867	.831	.790	.778	.743
.4500	1.207	1.128	1.084	1.012	.943	.873	.844	.808	.783	.732
.5000	1.188	1.128	1.075	1.012	.957	.882	.850	.812	.795	.755
.6000	1.169	1.113	1.081	1.021	.963	.913	.881	.848	.824	.806
.6500	1.164	1.107	1.087	1.034	.982	.923	.897	.897	.859	.821
.7000	1.147	1.095	1.068	1.024	.973	.916	.913	.889	.870	.836
.7500	1.122	1.082	1.081	1.021	.982	.932	.921	.891	.900	.866
.8000	1.116	1.088	1.062	1.021	.982	.956	.944	.909	.936	.907
.8500	1.110	1.070	1.065	1.028	1.009	.963	.972	.942	.967	.949
.9000	1.100	1.064	1.065	1.024	1.012	.978	.991	.976	1.012	1.033
.9500	1.072	1.049	1.065	1.021	1.015	1.000	1.028	1.024	1.072	1.151

TABLE VI.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Concluded

(f) Plain wing configuration; $\beta = -10^\circ$; $\frac{y}{b/2} = 0.72$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	1.474	.976	1.431	2.596	1.791	1.895	1.903	1.882	1.876	1.861
.0125	.929	1.049	1.405	1.841	1.731	1.828	1.866	1.843	1.846	1.861
.0250	.926	1.055	1.406	1.550	1.744	1.840	1.846	1.858	1.876	1.873
.0500	.932	1.067	1.419	1.560	1.744	1.858	1.900	1.905	1.912	1.882
.0750	.941	1.079	1.434	1.572	1.786	1.898	1.948	1.925	1.915	1.894
.1000	.939	1.052	1.406	1.535	1.733	1.889	1.927	1.891	1.891	1.882
.1500	.958	1.055	1.403	1.560	1.791	1.898	1.885	1.843	1.888	1.891
.2000	.977	1.067	1.422	1.587	1.784	1.852	1.827	1.832	1.909	1.903
.2500	1.007	1.085	1.428	1.575	1.739	1.807	1.803	1.829	1.924	1.914
.3000	1.003	1.073	1.431	1.568	1.751	1.768	1.779	1.832	1.936	1.912
.3500	1.024	1.085	1.469	1.581	1.725	1.738	1.766	1.852	1.958	1.917
.4000	1.016	1.082	1.463	1.636	1.719	1.714	1.766	1.882	1.961	1.909
.4500	1.032	1.076	1.431	1.734	1.794	1.732	1.818	1.923	1.979	1.926
.5000	1.039	1.092	1.416	1.873	1.941	1.768	1.903	1.983	2.012	1.941
.5500	1.039	1.079	1.344	1.867	2.025	1.816	1.959	2.003	2.009	1.923
.6000	1.033	1.067	1.288	1.791	2.075	1.886	1.973	2.000	2.003	1.920
.6500	1.023	1.058	1.219	1.687	2.054	1.940	1.997	2.003	1.991	1.917
.7000	1.026	1.055	1.169	1.596	2.025	1.976	2.018	2.004	1.985	1.916
.7500	1.016	1.055	1.134	1.477	1.928	1.958	1.997	1.988	1.967	1.909
.8000	1.029	1.043	1.088	1.380	1.838	1.931	1.976	1.971	1.967	1.903
.8500	1.007	1.031	1.044	1.256	1.684	1.846	1.906	1.909	1.903	1.861
.9000	1.007	1.012	1.029	1.195	1.538	1.744	1.842	1.840	1.870	1.859

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	1.452	1.110	.947	.989	1.153	1.301	1.409	1.420	1.459	1.484
.0250	1.452	1.104	.956	.921	1.028	1.111	1.197	1.210	1.242	1.261
.0500	1.442	1.098	.956	.876	.931	.961	1.009	1.000	1.039	1.050
.0750	1.436	1.113	.981	.888	.903	.895	.921	.900	.949	.962
.1000	1.455	1.110	.988	.885	.884	.861	.873	.861	.894	.891
.1500	1.481	1.095	.997	.888	.881	.825	.818	.793	.825	.820
.2000	1.523	1.085	1.000	.900	.881	.825	.806	.781	.807	.796
.2500	1.484	1.073	1.003	.909	.894	.828	.803	.778	.798	.790
.3000	1.416	1.079	1.025	.927	.919	.846	.818	.796	.801	.796
.3500	1.342	1.085	1.034	.942	.919	.858	.824	.802	.813	.805
.4000	1.209	1.076	1.041	.942	.934	.870	.839	.799	.825	.808
.4500	1.142	1.079	1.053	.967	.944	.880	.848	.814	.831	.825
.5000	1.094	1.076	1.053	.958	.941	.876	.839	.820	.837	.826
.5500	1.048	1.055	1.058	.955	.950	.880	.848	.811	.831	.817
.6000	.952	1.015	1.044	.955	.930	.889	.845	.808	.840	.814
.6500	1.058	1.040	1.034	.961	.939	.907	.876	.846	.861	.846
.7000	1.045	1.028	1.019	.958	.949	.928	.912	.873	.894	.897
.7500	1.045	1.028	1.019	.967	.981	.961	.964	.935	.982	.991
.8000	1.029	1.009	1.016	.973	1.009	1.006	1.051	1.046	1.097	1.121
.8500	1.028	1.003	1.016	.985	1.038	1.087	1.125	1.110	1.162	1.162
.9000	1.007	1.000	1.019	.997	1.091	1.130	1.153	1.124	1.169	1.184
.9500	.977	.976	1.006	.994	1.302	1.178	1.236	1.243	1.293	1.320

TABLE VII.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE

(a) Double slotted flap configuration; $\delta_f = 60^\circ$; $\frac{y}{b/2} = 0$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Fuselage										
.0000	.015	.009	.006	.000	.003	.030	.065	.096	.120	.136
.0500	.782	.831	.907	.930	.982	1.006	1.039	1.044	1.062	1.080
.1000	.856	.918	.985	.997	1.042	1.060	1.086	1.097	1.120	1.139
.1500	.951	.991	1.033	1.051	1.081	1.088	1.113	1.114	1.129	1.151
.2000	1.000	1.039	1.081	1.082	1.105	1.112	1.119	1.111	1.112	1.139
.2500	1.043	1.069	1.102	1.106	1.126	1.130	1.130	1.129	1.129	1.145
.3000	1.073	1.091	1.105	1.100	1.114	1.118	1.107	1.105	1.115	1.139
.3500	1.058	1.069	1.093	1.094	1.093	1.100	1.098	1.099	1.109	1.139
.4000	1.034	1.051	1.054	1.057	1.078	1.088	1.095	1.105	1.112	1.148
.4500	1.021	1.045	1.081	1.094	1.129	1.148	1.175	1.195	1.226	1.271
.5000	1.037	1.066	1.111	1.148	1.196	1.248	1.305	1.316	1.370	1.537
.5500	1.052	1.105	1.162	1.205	1.277	1.317	1.367	1.392	1.472	1.856
.6000	1.083	1.142	1.216	1.245	1.301	1.347	1.358	1.360	1.458	1.941
.6500	1.116	1.187	1.264	1.288	1.310	1.338	1.326	1.313	1.373	1.732
.7000	1.178	1.227	1.291	1.309	1.328	1.350	1.337	1.310	1.346	1.631
.7500	1.239	1.281	1.318	1.333	1.349	1.375	1.370	1.366	1.405	1.561
.8000	1.291	1.320	1.342	1.354	1.367	1.414	1.429	1.444	1.493	1.516
.8500	1.297	1.320	1.330	1.330	1.358	1.399	1.435	1.453	1.508	1.487
.9000	1.208	1.224	1.237	1.236	1.262	1.296	1.329	1.360	1.408	1.413
.9500	1.196	1.148	1.165	1.160	1.172	1.196	1.249	1.269	1.329	1.333
.9940	1.199	1.193	1.180	1.176	1.190	1.199	1.237	1.260	1.311	1.333

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Fuselage										
.0500	.862	.891	.757	.712	.636	.565	.488	.444	.402	.389
.1000	.939	.930	.850	.818	.729	.662	.607	.553	.513	.487
.1500	1.003	.982	.913	.876	.807	.734	.675	.626	.590	.572
.2000	1.024	1.015	.964	.921	.864	.798	.731	.684	.654	.625
.2500	1.037	1.033	.991	.967	.910	.849	.784	.743	.698	.684
.3000	1.061	1.057	1.018	.991	.928	.876	.820	.772	.742	.717
.3500	1.040	1.033	1.018	.985	.946	.885	.843	.789	.754	.743
.4000	.975	.979	.952	.927	.883	.828	.772	.737	.710	.684
.4500	.942	.943	.916	.888	.840	.785	.722	.675	.648	.614
.5000	.911	.906	.862	.842	.771	.707	.642	.599	.572	.543
.5500	.877	.873	.817	.782	.720	.646	.580	.538	.507	.487
.6000	.834	.816	.748	.715	.645	.583	.518	.471	.452	.437
.7500	.957	.946	.895	.882	.825	.779	.737	.702	.686	.681
.9500	.981	1.006	1.039	1.057	1.054	1.048	1.056	1.041	1.071	1.118
.9940	1.003	1.106	1.153	1.167	1.205	1.196	1.243	1.246	1.285	1.310

TABLE VII.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(b) Double slotted flap configuration; $\delta_f = 50^\circ$; $\frac{y}{b/2} = 0.21$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	.879	.944	1.818	3.291	5.269	7.821	10.369	9.877	4.156	2.900
.0125	.954	1.274	1.657	3.309	7.424	10.707	12.838	10.204	5.769	2.614
.0250	.991	1.224	1.518	2.542	3.552	6.245	8.939	8.654	5.748	2.649
.0500	1.013	1.196	1.420	1.600	2.003	2.716	3.980	5.172	3.454	2.508
.0750	1.031	1.184	1.373	1.521	1.803	2.179	2.864	3.598	3.248	2.519
.1000	1.050	1.184	1.349	1.506	1.728	2.006	2.572	3.085	3.170	2.478
.1500	1.062	1.187	1.393	1.460	1.618	1.788	2.186	2.499	2.922	2.407
.2000	1.090	1.206	1.335	1.430	1.567	1.710	2.015	2.232	2.653	2.360
.2500	1.112	1.224	1.321	1.412	1.516	1.648	1.906	2.055	2.361	2.310
.3000	1.124	1.230	1.324	1.406	1.484	1.597	1.829	1.902	2.153	2.260
.3500	1.149	1.243	1.327	1.394	1.457	1.570	1.705	1.783	1.988	2.207
.4000	1.174	1.262	1.327	1.382	1.442	1.555	1.646	1.719	1.875	2.159
.4500	1.205	1.290	1.352	1.394	1.439	1.528	1.617	1.661	1.788	2.133
.5000	1.252	1.324	1.383	1.406	1.454	1.507	1.611	1.629	1.740	2.109
.5500	1.270	1.349	1.389	1.406	1.457	1.475	1.572	1.586	1.696	2.041
.6000	1.317	1.383	1.413	1.427	1.472	1.493	1.584	1.603	1.685	1.997
.6500	1.367	1.424	1.441	1.457	1.490	1.507	1.617	1.644	1.685	1.965
.7000	1.447	1.483	1.500	1.503	1.534	1.558	1.676	1.667	1.676	1.888
.7815	1.428	1.648	1.624	1.642	1.669	1.728	1.761	1.694	1.717	1.841
.8179	1.817	1.813	1.774	1.770	1.800	1.925	1.906	1.841	1.835	1.900
.8325	1.923	1.900	1.855	1.854	1.890	2.015	1.997	1.925	1.919	1.935
.8362	1.957	1.922	1.870	1.866	1.913	2.033	2.024	1.966	1.939	1.953
.8398	1.918	1.872	1.821	1.824	1.863	1.976	1.965	1.908	1.907	1.926
.8434	2.019	1.962	1.904	1.900	1.937	2.078	2.077	2.021	1.988	1.977
Vane										
.0000	1.665	1.685	1.654	1.648	1.654	1.687	1.625	1.580	1.568	1.546
.0250	2.969	2.897	2.858	2.900	2.979	3.281	3.272	3.218	3.127	3.003
.0500	3.068	2.953	2.882	2.885	2.973	3.322	3.319	3.279	3.150	2.903
.1000	3.351	3.193	3.098	3.042	3.146	3.618	3.655	3.574	3.335	2.938
.1500	3.457	3.268	3.148	3.075	3.206	3.698	3.782	3.711	3.436	2.985
.2000	3.547	3.336	3.206	3.118	3.284	3.812	3.926	3.859	3.566	3.092
.3000	3.417	3.162	3.009	2.948	3.146	3.636	3.856	3.832	3.601	3.171
.4000	3.122	2.850	2.697	2.679	2.928	3.382	3.640	3.661	3.494	3.092
.5000	2.901	2.657	2.512	2.548	2.779	3.110	3.469	3.563	3.456	3.062
.6000	2.528	2.355	2.250	2.282	2.466	2.698	3.021	3.174	3.162	2.853
.7000	2.379	2.241	2.154	2.142	2.289	2.501	2.794	2.928	2.968	2.726
.8000	2.348	2.199	2.108	2.106	2.218	2.434	2.705	2.844	2.850	2.584
.9000	2.280	2.131	2.018	2.027	2.137	2.343	2.575	2.687	2.636	2.387
Flap										
.0000	1.876	1.673	1.463	1.218	1.122	1.221	1.316	1.339	1.318	1.266
.0125	2.419	2.258	2.074	1.951	1.979	2.200	2.392	2.467	2.355	2.165
.0250	2.625	2.495	2.352	2.315	2.385	2.642	2.873	2.974	2.852	2.569
.0500	2.718	2.585	2.463	2.457	2.564	2.835	3.086	3.209	3.198	3.015
.0750	2.516	2.414	2.290	2.297	2.403	2.642	2.876	2.980	3.020	2.856
.1000	2.255	2.190	2.114	2.135	2.218	2.427	2.640	2.737	2.798	2.667
.1500	1.916	1.869	1.811	1.806	1.872	2.045	2.201	2.264	2.318	2.263
.2000	1.845	1.807	1.747	1.739	1.788	1.970	2.121	2.183	2.217	2.151
.4000	1.836	1.810	1.744	1.779	1.794	1.931	2.074	2.181	2.162	2.071
.6000	1.733	1.710	1.651	1.675	1.701	1.842	1.985	2.021	1.997	1.953
.8000	1.652	1.629	1.592	1.600	1.639	1.749	1.850	1.873	1.798	1.782

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	1.062	.813	.657	.576	.582	.648	.743	.716	.503	.389
.0250	1.031	.841	.682	.545	.468	.388	.345	.322	.260	.209
.0500	1.019	.846	.719	.576	.457	.331	.251	.217	.223	.212
.0750	1.013	.882	.744	.606	.484	.361	.289	.258	.280	.266
.1000	1.013	.885	.753	.627	.519	.409	.351	.322	.350	.342
.1500	.997	.888	.768	.661	.567	.487	.434	.406	.422	.404
.2000	1.009	.906	.802	.697	.621	.540	.484	.455	.460	.440
.2500	1.022	.925	.821	.733	.663	.585	.528	.502	.503	.484
.3000	1.022	.931	.836	.748	.684	.609	.566	.536	.520	.519
.3500	1.031	.963	.880	.782	.725	.648	.602	.562	.558	.549
.4000	1.028	.969	.880	.806	.746	.672	.622	.586	.581	.566
.4500	1.019	.963	.886	.821	.767	.693	.634	.603	.584	.581
.5000	.975	.931	.864	.803	.749	.681	.628	.600	.584	.578
.5500	.938	.891	.824	.776	.716	.657	.620	.577	.578	.563
.6000	.863	.841	.793	.733	.693	.639	.590	.560	.558	.549
.6500	.789	.773	.716	.679	.639	.588	.546	.525	.520	.513
.7000	.736	.704	.651	.618	.576	.540	.502	.475	.483	.472
.7815	.665	.570	.506	.485	.457	.427	.401	.391	.393	.378
.8179	.612	.526	.472	.451	.433	.403	.395	.386	.381	.372
.8252	.599	.520	.478	.455	.445	.418	.404	.397	.393	.384
.8325	.593	.542	.500	.497	.487	.475	.469	.455	.462	.454
Vane										
.0250	1.087	1.025	.954	.864	.785	.725	.681	.661	.647	.634
.0500	.770	.623	.505	.373	.275	.209	.192	.188	.185	.171
.1000	.584	.383	.253	.124	.048	.018	.032	.032	.026	.018
.1500	.509	.318	.188	.073	.021	.012	.018	.017	.017	.009
.2000	.463	.283	.164	.064	.030	.021	.038	.029	.035	.030
.3000	.407	.249	.148	.064	.057	.045	.059	.055	.061	.059
.4000	.376	.246	.151	.079	.075	.081	.086	.084	.087	.077
.5000	.360	.259	.170	.103	.099	.099	.112	.107	.107	.103
.6000	.357	.252	.179	.121	.122	.128	.133	.136	.139	.133
.7000	.379	.290	.216	.167	.185	.188	.198	.217	.205	.180
.8000	.432	.368	.296	.247	.284	.299	.319	.330	.321	.295
.9200	.882	.807	.725	.712	.749	.809	.888	.910	.873	.797
Flap										
.0125	.534	.153	.043	.021	.030	.035	.044	.058	.058	.030
.0250	.236	.062	.012	.021	.024	.024	.030	.035	.040	.044
.0500	.174	.115	.080	.091	.093	.090	.091	.093	.101	.097
.0750	.168	.146	.117	.127	.128	.116	.124	.116	.124	.124
.1000	.196	.184	.167	.164	.173	.158	.156	.157	.156	.153
.1500	.221	.212	.210	.203	.206	.197	.198	.200	.185	.195
.4000	.485	.475	.454	.436	.427	.409	.401	.394	.379	.384
.6000	.637	.620	.586	.570	.552	.534	.510	.490	.486	.472
.8000	.755	.723	.701	.679	.654	.630	.602	.574	.552	.555
.9000	.839	.829	.799	.770	.740	.734	.696	.664	.642	.637

[redacted]

TABLE VII.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(c) Double slotted flap configuration; $\delta_f = 50^\circ$; $\frac{y}{b/2} = 0.30$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	.881	1.331	2.127	2.765	3.580	4.452	4.730	4.431	3.398	2.545
.0125	1.003	1.555	2.239	2.802	3.519	4.132	4.333	4.219	3.375	2.531
.0250	1.044	1.426	2.353	2.903	3.649	4.326	4.411	4.228	3.390	2.557
.0500	1.063	1.298	2.894	3.814	3.958	4.225	4.306	4.234	3.336	2.484
.0750	1.075	1.254	1.791	3.622	5.326	5.614	4.979	4.467	3.360	2.499
.1000	1.085	1.241	1.236	2.198	4.121	5.144	4.955	4.404	3.351	2.534
.1500	1.107	1.241	1.257	1.387	2.154	3.257	4.153	3.868	3.174	2.435
.2000	1.135	1.248	1.305	1.351	1.607	2.335	3.297	3.358	3.018	2.400
.2500	1.154	1.263	1.326	1.363	1.480	1.931	2.751	2.957	2.862	2.409
.3000	1.179	1.279	1.341	1.381	1.447	1.772	2.438	2.668	2.711	2.337
.3500	1.198	1.273	1.329	1.378	1.420	1.650	2.144	2.368	2.502	2.311
.4000	1.223	1.304	1.356	1.399	1.441	1.623	2.024	2.216	2.387	2.270
.4500	1.260	1.345	1.381	1.421	1.459	1.611	1.955	2.141	2.292	2.229
.5000	1.301	1.364	1.411	1.439	1.465	1.593	1.868	2.012	2.159	2.174
.5500	1.332	1.398	1.405	1.457	1.483	1.593	1.829	1.928	2.077	2.125
.6000	1.392	1.445	1.459	1.491	1.517	1.611	1.814	1.898	2.027	2.081
.6500	1.467	1.520	1.535	1.552	1.583	1.650	1.817	1.874	1.968	2.041
.7521	1.715	1.718	1.704	1.759	1.776	1.829	1.919	1.981	1.920	1.934
.7934	2.006	1.997	1.955	1.994	2.030	2.096	2.219	2.192	2.021	1.968
.8099	2.176	2.141	2.100	2.153	2.190	2.257	2.390	2.341	2.083	1.992
.8141	2.242	2.198	2.148	2.204	2.248	2.317	2.450	2.401	2.121	2.009
.8182	2.282	2.226	2.166	2.238	2.275	2.353	2.480	2.428	2.142	2.012
.8224	2.395	2.326	2.263	2.329	2.371	2.434	2.559	2.527	2.215	2.038
Vane										
.0000	2.872	2.746	2.619	2.698	2.743	2.841	2.901	2.847	2.558	2.444
.0250	1.881	3.662	3.686	3.845	3.849	4.126	4.333	4.081	3.242	2.963
.0500	4.488	4.264	4.130	4.272	4.353	4.512	4.835	4.629	3.673	3.090
.1000	4.953	4.681	4.516	4.685	4.743	4.919	5.303	5.024	3.826	3.009
.1500	5.204	4.906	4.725	4.900	4.942	5.129	5.544	5.242	3.959	3.041
.2000	5.420	5.101	4.894	5.067	5.102	5.308	5.760	5.467	4.160	3.163
.3000	5.326	5.003	4.776	4.936	4.957	5.228	5.670	5.437	4.227	3.227
.4000	4.941	4.627	4.393	4.540	4.556	4.889	5.309	5.072	3.962	3.047
.5000	4.536	4.220	4.003	4.125	4.133	4.533	4.949	4.716	3.711	2.890
.6000	4.013	3.737	3.516	3.619	3.634	4.048	4.426	4.231	3.416	2.734
.7000	3.492	3.260	3.060	3.140	3.166	3.566	3.916	3.731	3.109	2.348
.8000	2.981	2.765	2.613	2.653	2.683	3.036	3.363	3.234	2.744	2.328
.9000	2.564	2.405	2.278	2.308	2.332	2.620	2.898	2.844	2.493	2.189
Flap										
.0000	1.473	1.373	1.335	1.293	1.202	1.305	1.450	1.449	1.283	1.142
.0125	2.405	2.242	2.157	2.174	2.112	2.284	2.483	2.485	2.245	2.026
.0250	2.928	2.793	2.665	2.711	2.677	2.940	3.186	3.189	2.891	2.621
.0500	3.041	2.881	2.716	2.762	2.752	3.090	3.287	3.383	3.118	2.809
.0750	3.019	2.847	2.668	2.698	2.719	3.111	3.402	3.461	3.186	2.850
.1000	2.828	2.658	2.498	2.537	2.556	2.934	3.246	3.266	3.006	2.690
.1500	2.298	2.173	2.048	2.070	2.166	2.515	2.757	2.793	2.623	2.392
.2000	1.956	1.850	1.764	1.784	1.819	2.048	2.288	2.299	2.168	2.009
.4000	1.643	1.627	1.577	1.604	1.631	1.772	1.913	1.955	1.938	1.876
.6000	1.580	1.586	1.550	1.582	1.613	1.719	1.808	1.838	1.808	1.789
.8000	1.549	1.555	1.514	1.558	1.565	1.641	1.670	1.671	1.634	1.655

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	1.003	.784	.716	.796	.970	1.192	1.366	1.380	1.295	1.107
.0250	1.035	.825	.704	.698	.728	.805	.874	.889	.851	.765
.0500	1.022	.859	.725	.677	.604	.590	.583	.584	.578	.525
.0750	1.022	.878	.746	.692	.595	.545	.502	.497	.475	.444
.1000	1.028	.887	.767	.698	.598	.530	.474	.461	.448	.420
.1500	1.028	.906	.795	.732	.591	.548	.471	.461	.440	.415
.2000	1.031	.919	.822	.753	.643	.563	.495	.464	.434	.412
.2500	1.035	.944	.843	.771	.671	.581	.508	.485	.454	.435
.3000	1.028	.947	.846	.784	.686	.605	.526	.497	.469	.444
.3500	1.022	.944	.846	.808	.701	.614	.550	.533	.496	.467
.4000	.994	.928	.846	.802	.701	.632	.556	.539	.513	.484
.4500	.981	.912	.831	.796	.710	.635	.580	.551	.522	.496
.5000	.922	.881	.801	.777	.695	.623	.574	.551	.525	.502
.5500	.878	.831	.761	.747	.674	.611	.559	.545	.519	.496
.6000	.799	.746	.698	.686	.634	.575	.532	.515	.496	.467
.6500	.737	.690	.628	.628	.574	.542	.492	.473	.457	.444
.7521	.567	.508	.468	.454	.426	.401	.378	.362	.354	.333
.7934	.511	.448	.417	.412	.381	.353	.342	.329	.316	.301
.8017	.502	.439	.414	.406	.375	.356	.342	.329	.319	.304
.8099	.508	.445	.402	.418	.393	.386	.384	.383	.357	.345
Vane										
.0250	1.232	1.160	1.091	1.061	.985	.922	.865	.844	.770	.719
.0500	.602	.564	.526	.460	.344	.254	.198	.198	.162	.154
.1000	.282	.260	.269	.180	.076	.024	.003	.003	.000	.000
.1500	.210	.182	.187	.122	.024	.000	.000	.000	.000	.000
.2000	.163	.138	.145	.095	.030	.015	.018	.015	.003	.000
.3000	.138	.107	.115	.085	.033	.033	.033	.045	.032	.020
.4000	.119	.107	.112	.088	.057	.066	.066	.069	.053	.049
.5000	.132	.100	.109	.098	.082	.090	.090	.090	.083	.067
.6000	.144	.113	.127	.119	.115	.117	.117	.114	.103	.096
.7000	.198	.169	.166	.174	.166	.177	.180	.177	.159	.130
.8000	.317	.282	.278	.293	.287	.311	.330	.329	.277	.255
.9200	.865	.806	.782	.796	.807	.868	.928	.916	.811	.733
Flap										
.0125	.110	.091	.106	.091	.063	.093	.111	.117	.086	.061
.0250	.006	.000	.000	.000	.000	.000	.000	.000	.003	.000
.0500	.041	.025	.024	.040	.033	.021	.030	.021	.006	.014
.0750	.053	.041	.030	.049	.036	.036	.024	.027	.024	.023
.1000	.085	.075	.063	.088	.060	.060	.048	.048	.047	.038
.1500	.132	.119	.106	.119	.100	.102	.078	.084	.074	.075
.2000	.179	.154	.157	.168	.145	.132	.111	.111	.109	.093
.4000	.335	.317	.296	.290	.254	.246	.222	.216	.209	.200
.6000	.470	.448	.426	.421	.390	.371	.333	.332	.322	.319
.8000	.602	.599	.571	.573	.553	.530	.486	.479	.478	.475
.9000	.746	.740	.731	.726	.722	.689	.649	.629	.628	.629

TABLE VII.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(d) Double slotted flap configuration; $\delta_f = 50^\circ$; $\frac{y}{b/2} = 0.43$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	.814	1.825	2.043	2.466	2.910	2.997	2.814	2.687	2.592	2.354
.0125	1.075	1.772	2.021	2.454	2.875	2.988	2.941	2.800	2.613	2.390
.0250	1.109	1.701	2.074	2.494	2.884	3.027	3.036	2.815	2.610	2.363
.0500	1.119	1.626	2.098	2.569	3.009	3.128	2.982	2.847	2.607	2.401
.0750	1.127	1.551	2.345	2.633	2.976	2.994	2.971	2.867	2.601	2.376
.1000	1.149	1.442	2.639	2.860	2.943	2.916	2.994	2.885	2.613	2.369
.1500	1.165	1.311	2.203	3.600	3.654	3.188	3.136	2.919	2.613	2.364
.2000	1.186	1.311	1.518	2.088	3.979	3.576	3.180	2.873	2.598	2.366
.2500	1.211	1.327	1.241	1.942	3.116	3.415	3.047	2.763	2.549	2.345
.3000	1.242	1.352	1.290	1.533	2.430	3.179	2.950	2.702	2.517	2.325
.3500	1.258	1.358	1.312	1.339	1.925	2.782	2.723	2.554	2.439	2.295
.4000	1.298	1.389	1.370	1.321	1.701	2.504	2.599	2.461	2.381	2.272
.4500	1.342	1.430	1.410	1.345	1.606	2.304	2.466	2.371	2.321	2.236
.5000	1.379	1.442	1.429	1.367	1.501	2.078	2.295	2.232	2.231	2.198
.5500	1.447	1.511	1.497	1.430	1.519	1.976	2.198	2.163	2.170	2.159
.6000	1.528	1.582	1.577	1.503	1.546	1.887	2.103	2.084	2.110	2.127
.7000	1.783	1.814	1.799	1.739	1.716	1.878	2.018	1.988	2.014	2.056
.7500	2.115	2.137	2.120	2.051	2.003	2.060	2.112	2.041	2.014	2.047
.7700	2.367	2.383	2.355	2.303	2.239	2.242	2.224	2.128	2.046	2.056
.7750	2.429	2.433	2.419	2.354	2.292	2.287	2.248	2.102	2.058	2.053
.7800	2.503	2.501	2.478	2.430	2.370	2.331	2.286	2.148	2.046	2.056
Vane										
.0000	2.426	2.389	2.281	2.212	2.146	2.113	2.027	1.913	1.876	1.906
.0250	4.025	4.012	3.970	3.939	3.842	3.780	3.490	3.215	2.864	2.803
.0500	4.448	4.415	4.385	4.348	4.245	4.030	3.749	3.406	2.878	2.717
.1000	5.075	4.987	4.990	4.972	4.872	4.504	4.103	3.644	2.844	2.564
.1500	5.620	5.479	5.490	5.490	5.397	4.937	4.428	3.893	2.902	2.525
.2000	6.100	5.959	5.974	5.981	5.916	5.400	4.838	4.227	3.046	2.637
.3000	5.858	5.725	5.731	5.769	5.743	5.257	4.661	4.038	2.910	2.515
.4000	5.451	5.336	5.317	5.384	5.382	4.970	4.416	3.827	2.780	2.431
.5000	4.970	4.866	4.860	4.927	4.949	4.609	4.109	3.569	2.647	2.345
.6000	4.435	4.342	4.314	4.390	4.424	4.167	3.747	3.287	2.526	2.277
.7000	3.931	3.875	3.851	3.915	3.967	3.770	3.437	3.041	2.404	2.221
.8000	3.404	3.342	3.321	3.375	3.433	3.242	3.047	2.734	2.231	2.118
.9000	2.917	2.872	2.845	2.882	2.913	2.854	2.670	2.421	2.087	2.012
Flap										
.0000	1.898	1.897	1.984	2.000	1.967	1.901	1.800	1.641	1.419	1.381
.0125	2.634	2.604	2.611	2.618	2.621	2.576	2.428	2.229	1.962	1.923
.0250	3.382	3.333	3.311	3.321	3.355	3.325	3.139	2.896	2.561	2.522
.0500	3.538	3.476	3.435	3.463	3.534	3.504	3.342	3.070	2.682	2.602
.0750	3.485	3.439	3.398	3.493	3.516	3.516	3.351	3.108	2.673	2.544
.1000	3.370	3.286	3.228	3.285	3.430	3.391	3.254	2.998	2.578	2.472
.1500	2.662	2.607	2.564	2.627	2.722	2.788	2.732	2.560	2.243	2.183
.2000	2.233	2.193	2.163	2.227	2.257	2.281	2.274	2.145	1.959	1.912
.4000	1.750	1.704	1.673	1.703	1.740	1.848	1.918	1.870	1.833	1.853
.6000	1.497	1.486	1.457	1.491	1.540	1.645	1.738	1.702	1.784	1.794
.8000	1.288	1.290	1.268	1.300	1.373	1.448	1.507	1.490	1.639	1.699

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	1.062	.810	.805	.927	1.131	1.301	1.398	1.386	1.413	1.354
.0250	1.062	.847	.768	.770	.839	.907	.956	.945	.954	.950
.0500	1.062	.885	.775	.715	.704	.698	.708	.684	.696	.699
.0750	1.059	.897	.784	.718	.669	.633	.622	.588	.616	.611
.1000	1.062	.913	.802	.721	.660	.603	.584	.554	.564	.558
.1500	1.050	.928	.821	.742	.663	.594	.549	.513	.526	.513
.2000	1.028	.935	.818	.745	.683	.591	.549	.504	.517	.507
.2500	1.019	.925	.830	.754	.681	.603	.552	.510	.512	.504
.3000	.997	.916	.824	.759	.675	.609	.558	.513	.520	.504
.3500	.975	.903	.821	.751	.681	.612	.563	.519	.529	.510
.4000	.944	.875	.799	.742	.669	.609	.561	.519	.523	.510
.4500	.898	.838	.765	.709	.660	.591	.552	.507	.512	.502
.5000	.845	.797	.719	.676	.621	.570	.522	.493	.503	.472
.5500	.783	.738	.682	.648	.594	.543	.516	.475	.488	.469
.6000	.717	.667	.617	.588	.555	.513	.472	.438	.448	.440
.7000	.540	.505	.460	.442	.418	.385	.372	.333	.350	.336
.7500	.466	.436	.395	.385	.364	.334	.333	.293	.315	.301
.7600	.466	.424	.398	.376	.361	.331	.327	.296	.318	.295
.7700	.475	.455	.429	.412	.376	.358	.357	.313	.344	.316
Vane										
.0250	.752	.741	.734	.673	.603	.534	.484	.438	.436	.431
.0500	.370	.368	.410	.367	.287	.227	.186	.148	.150	.145
.1000	.217	.206	.256	.227	.152	.090	.077	.041	.078	.035
.1500	.112	.118	.173	.142	.096	.054	.080	.023	.052	.041
.2000	.096	.084	.133	.121	.087	.066	.053	.032	.075	.053
.3000	.087	.081	.123	.106	.090	.096	.109	.061	.095	.050
.4000	.087	.090	.105	.109	.090	.093	.094	.072	.098	.080
.5000	.112	.112	.120	.121	.119	.107	.109	.096	.101	.094
.6000	.137	.134	.136	.139	.146	.128	.150	.113	.142	.106
.7000	.208	.190	.170	.188	.188	.191	.183	.159	.165	.148
.8000	.342	.318	.321	.315	.337	.322	.310	.273	.275	.283
.9200	.981	.959	.950	.970	.976	.961	.900	.818	.765	.725
Flap										
.0125	.098	.112	.139	.155	.152	.149	.136	.116	.116	.089
.0250	.051	.016	.012	.036	.036	.033	.032	.003	.038	.012
.0500	.037	.094	.046	.041	.051	.051	.056	.023	.052	.030
.0750	.081	.062	.052	.073	.069	.063	.077	.058	.069	.053
.1000	.093	.081	.065	.067	.087	.072	.080	.061	.084	.056
.1500	.143	.125	.111	.100	.099	.090	.094	.084	.095	.080
.2000	.193	.178	.148	.145	.155	.131	.133	.110	.139	.118
.4000	.329	.299	.272	.255	.260	.251	.236	.194	.246	.224
.6000	.444	.421	.383	.376	.370	.337	.330	.299	.335	.327
.8000	.575	.548	.515	.503	.519	.484	.472	.461	.497	.528
.9000	.671	.654	.620	.621	.633	.615	.605	.588	.659	.667

TABLE VII.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(e) Double slotted flap configuration; $\delta_f = 50^\circ$; $\frac{y}{b/2} = 0.56$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	1.110	1.903	2.018	2.290	2.550	2.515	2.420	2.356	2.274	2.224
.0125	1.276	1.951	2.075	2.248	2.574	2.527	2.492	2.425	2.295	2.224
.0250	1.198	1.922	2.042	2.311	2.535	2.509	2.492	2.431	2.307	2.229
.0500	1.194	1.887	2.094	2.378	2.553	2.599	2.541	2.467	2.316	2.224
.0750	1.179	1.749	2.157	2.454	2.595	2.623	2.526	2.470	2.292	2.218
.1000	1.185	1.633	2.208	2.506	2.646	2.578	2.541	2.488	2.310	2.218
.1500	1.220	1.492	2.565	2.579	2.577	2.530	2.580	2.500	2.310	2.218
.2000	1.241	1.417	2.755	2.878	2.535	2.521	2.601	2.482	2.316	2.215
.2500	1.267	1.401	2.637	3.403	2.680	2.584	2.637	2.518	2.333	2.215
.3000	1.307	1.414	2.184	3.354	3.048	2.683	2.631	2.491	2.328	2.203
.3500	1.339	1.436	1.734	2.909	3.151	2.737	2.601	2.470	2.313	2.197
.4000	1.367	1.473	1.438	2.366	3.048	2.710	2.559	2.440	2.298	2.186
.4500	1.433	1.508	1.335	1.872	2.794	2.605	2.483	2.386	2.263	2.166
.5000	1.499	1.577	1.393	1.634	2.604	2.512	2.432	2.347	2.245	2.154
.6201	1.699	1.749	1.649	1.564	2.088	2.234	2.249	2.228	2.142	2.093
.6834	2.069	2.116	2.009	1.906	2.166	2.234	2.240	2.219	2.139	2.084
.7087	2.301	2.336	2.236	2.110	2.220	2.228	2.249	2.225	2.133	2.090
.7151	2.298	2.348	2.239	2.128	2.217	2.228	2.246	2.216	2.130	2.087
.7214	2.408	2.443	2.359	2.256	2.257	2.240	2.252	2.234	2.142	2.090
.7278	2.439	2.489	2.399	2.284	2.245	2.222	2.231	2.219	2.133	2.073
Vane										
.0000	1.890	1.937	1.713	1.604	1.613	1.587	1.595	1.572	1.549	1.536
.0250	5.878	3.963	3.752	3.534	3.365	3.201	3.180	3.081	2.882	2.800
.0500	4.270	4.320	4.157	3.912	3.628	3.344	3.264	3.120	2.817	2.702
.1000	5.123	5.098	5.000	4.744	4.166	3.593	3.417	3.213	2.782	2.632
.1500	5.436	5.380	5.332	5.080	4.326	3.665	3.414	3.201	2.703	2.554
.2000	5.677	5.599	5.586	5.360	4.507	3.784	3.508	3.234	2.687	2.519
.3000	5.709	5.602	5.649	5.519	4.556	3.754	3.444	3.171	2.599	2.386
.4000	5.276	5.195	5.250	5.202	4.269	3.473	3.189	2.958	2.451	2.267
.5000	4.906	4.834	4.897	4.912	4.051	3.290	3.042	2.835	2.387	2.221
.6000	4.345	4.301	4.362	4.403	3.655	3.009	2.820	2.662	2.298	2.177
.7000	3.822	3.787	3.822	3.897	3.284	2.769	2.631	2.506	2.227	2.128
.8000	3.354	3.331	3.362	3.436	2.970	2.581	2.474	2.383	2.159	2.087
.9000	2.884	2.878	2.879	2.930	2.631	2.350	2.273	2.213	2.056	2.035
Flap										
.0000	1.552	1.511	1.619	1.692	1.538	1.362	1.321	1.296	1.218	1.194
.0125	2.727	2.706	2.719	2.787	2.565	2.287	2.246	2.218	2.103	2.061
.0250	3.314	3.311	3.278	3.375	3.109	2.778	2.730	2.680	2.543	2.499
.0500	3.486	3.483	3.453	3.564	3.302	2.946	2.883	2.808	2.623	2.537
.0750	3.524	3.517	3.519	3.647	3.384	3.006	2.916	2.832	2.611	2.493
.1000	3.392	3.376	3.350	3.500	3.278	2.919	2.826	2.745	2.516	2.395
.1500	2.900	2.897	2.834	2.985	2.776	2.500	2.432	2.368	2.186	2.102
.2000	2.348	2.348	2.320	2.421	2.359	2.186	2.150	2.099	1.982	1.910
.4000	1.890	1.900	1.858	1.930	1.988	1.943	1.931	1.916	1.856	1.821
.6000	1.734	1.731	1.677	1.732	1.825	1.852	1.847	1.835	1.823	1.789
.8000	1.477	1.470	1.420	1.454	1.565	1.614	1.646	1.671	1.708	1.722

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	.950	.803	.921	1.116	1.381	1.521	1.619	1.638	1.661	1.673
.0250	.994	.815	.891	.869	.970	1.051	1.114	1.129	1.156	1.177
.0500	1.003	.834	.770	.759	.776	.787	.811	.811	.841	.852
.0750	.972	.837	.746	.717	.701	.677	.679	.686	.693	.704
.1000	.972	.843	.758	.698	.683	.641	.631	.632	.646	.644
.1500	.962	.846	.767	.701	.665	.608	.580	.563	.569	.565
.2000	.950	.846	.761	.710	.668	.599	.568	.554	.546	.542
.2500	.931	.846	.758	.704	.656	.605	.562	.542	.540	.528
.3000	.890	.825	.746	.695	.636	.599	.550	.530	.534	.516
.3500	.875	.815	.743	.683	.653	.584	.541	.524	.531	.507
.4000	.834	.781	.731	.677	.637	.581	.526	.512	.513	.499
.4500	.774	.746	.695	.656	.619	.560	.528	.509	.504	.481
.5000	.727	.693	.643	.619	.580	.527	.489	.476	.487	.464
.6201	.533	.517	.495	.476	.450	.407	.372	.371	.378	.351
.6834	.458	.429	.411	.399	.390	.347	.324	.320	.322	.310
.6961	.455	.433	.408	.393	.378	.350	.327	.320	.313	.310
.7087	.442	.423	.402	.384	.378	.347	.327	.326	.327	.313
Vane										
.0250	.834	.793	.764	.701	.671	.617	.577	.548	.537	.525
.0500	.451	.376	.435	.390	.372	.299	.249	.225	.221	.215
.1000	.254	.176	.284	.238	.202	.156	.120	.096	.103	.090
.1500	.172	.107	.224	.192	.166	.117	.087	.081	.077	.087
.2000	.125	.091	.193	.165	.148	.138	.102	.081	.083	.081
.3000	.103	.091	.163	.149	.133	.099	.102	.093	.100	.096
.4000	.119	.094	.160	.146	.142	.126	.114	.114	.115	.116
.5000	.107	.119	.166	.149	.151	.138	.123	.126	.139	.125
.6000	.135	.157	.184	.177	.187	.159	.156	.150	.165	.162
.7000	.194	.207	.230	.226	.239	.210	.204	.207	.204	.197
.8000	.317	.345	.350	.375	.353	.320	.321	.296	.307	.313
.9200	1.009	1.022	1.036	1.031	.979	.883	.865	.826	.817	.809
Flap										
.0125	.091	.075	.175	.177	.175	.162	.168	.153	.142	.136
.0250	.006	.034	.030	.027	.051	.054	.039	.060	.053	.038
.0500	.044	.053	.076	.085	.081	.087	.084	.069	.065	.075
.0750	.088	.078	.100	.079	.130	.096	.081	.087	.086	.084
.1000	.103	.125	.130	.116	.163	.111	.111	.105	.097	.107
.1500	.147	.166	.178	.159	.199	.153	.138	.141	.145	.133
.2000	.216	.201	.202	.189	.329	.177	.156	.159	.153	.139
.4000	.379	.367	.353	.329	.444	.311	.294	.278	.280	.290
.6000	.502	.492	.474	.448	.586	.422	.399	.395	.407	.403
.8000	.633	.636	.598	.585	.716	.575	.553	.548	.563	.577
.9000	.740	.734	.710	.689	.958	.698	.697	.695	.723	.728

TABLE VII - PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Concluded

(f) Double slotted flap configuration; $\delta_f = 60^\circ$; $\frac{y}{b/2} = 0.72$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	2.168	2.133	2.318	2.300	2.214	2.118	2.024	1.962	1.953	1.979
.0125	1.957	2.097	2.222	2.221	2.160	2.063	1.994	1.950	1.959	1.988
.0250	1.884	2.103	2.264	2.242	2.169	2.084	2.024	1.971	1.965	1.982
.0500	1.564	2.142	2.300	2.266	2.175	2.139	2.059	1.997	1.971	1.994
.0750	1.494	2.178	2.345	2.300	2.193	2.163	2.077	2.006	1.974	1.988
.1000	1.399	2.183	2.336	2.285	2.181	2.139	2.062	1.997	1.971	1.982
.1500	1.399	2.175	2.405	2.390	2.205	2.139	2.089	2.029	1.980	1.994
.2000	1.411	2.236	2.462	2.382	2.199	2.160	2.125	2.035	1.989	2.000
.2500	1.492	2.109	2.550	2.393	2.178	2.166	2.116	2.038	1.992	2.000
.3000	1.448	1.861	2.727	2.945	2.166	2.157	2.107	2.035	1.989	2.003
.3500	1.484	1.722	2.886	2.318	2.181	2.172	2.119	2.032	1.997	2.009
.4000	1.509	1.577	2.994	2.397	2.205	2.172	2.101	2.029	1.997	2.009
.4500	1.555	1.517	3.027	2.491	2.223	2.172	2.113	2.035	2.000	2.018
.5000	1.607	1.577	3.086	2.706	2.271	2.187	2.119	2.032	2.009	2.024
.5500	1.629	1.586	2.907	2.591	2.256	2.172	2.098	2.023	1.997	2.024
.6000	1.662	1.631	2.730	2.488	2.241	2.142	2.074	2.006	1.992	2.018
.6500	1.656	1.619	2.459	2.291	2.199	2.121	2.071	2.006	2.009	2.036
.7000	1.601	1.577	2.252	2.191	2.181	2.127	2.068	2.006	2.024	2.059
.7500	1.543	1.526	2.087	2.145	2.181	2.115	2.059	1.994	2.015	2.077
.8000	1.451	1.441	1.943	2.088	2.151	2.106	2.054	1.985	2.015	2.083
.8500	1.395	1.384	1.796	2.027	2.090	2.072	2.036	1.971	2.027	2.100
.9000	1.359	1.356	1.613	1.648	1.639	1.637	1.610	1.585	1.598	1.605

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	.797	.837	1.012	1.085	1.211	1.287	1.337	1.357	1.423	1.481
.0250	.828	.804	.865	.906	.988	1.051	1.092	1.117	1.164	1.215
.0500	.846	.792	.778	.791	.819	.840	.879	.889	.924	.953
.0750	.853	.810	.766	.754	.759	.764	.795	.795	.818	.850
.1000	.846	.819	.754	.736	.729	.722	.734	.737	.760	.791
.1500	.862	.834	.760	.730	.702	.683	.678	.673	.692	.714
.2000	.883	.846	.787	.751	.720	.677	.678	.667	.683	.696
.2500	.889	.858	.805	.773	.738	.701	.686	.667	.675	.684
.3000	.923	.906	.850	.809	.777	.734	.719	.699	.707	.708
.3500	.991	.973	.937	.903	.852	.798	.787	.766	.766	.767
.4000	1.080	1.075	1.063	1.036	.964	.906	.873	.860	.851	.856
.4500	1.251	1.269	1.240	1.206	1.120	1.039	1.000	.968	.968	.959
.5000	1.435	1.462	1.453	1.397	1.289	1.190	1.142	1.099	1.088	1.094
.5500	1.626	1.649	1.643	1.573	1.473	1.372	1.329	1.278	1.267	1.269
.6000	1.760	1.822	1.796	1.724	1.605	1.529	1.503	1.474	1.464	1.469
.6500	1.920	1.979	1.919	1.809	1.693	1.646	1.639	1.632	1.648	1.670
.7000	2.000	2.054	1.919	1.806	1.687	1.643	1.675	1.696	1.742	1.755
.7500	2.012	2.048	1.898	1.794	1.648	1.595	1.627	1.649	1.713	1.755
.8000	1.908	1.961	1.898	1.770	1.623	1.550	1.548	1.550	1.604	1.664
.8500	1.810	1.852	1.793	1.776	1.639	1.556	1.527	1.509	1.560	1.605
.9000	1.675	1.734	1.745	1.779	1.687	1.601	1.556	1.529	1.560	1.587
.9500	1.527	1.586	1.661	1.766	1.744	1.692	1.645	1.608	1.619	1.640

TABLE VIII.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE

(a) Double slotted flap configuration; $\delta_f = 60^\circ$; $\frac{y}{b/2} = 0$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Fuselage										
.0000	.012	.031	.000	.006	.000	.030	.064	.089	.126	.156
.0500	.767	.916	.870	.925	.964	1.006	1.041	1.065	1.064	1.097
.1000	.858	.976	.972	.985	1.036	1.050	1.085	1.113	1.123	1.153
.1500	.927	.988	1.025	1.045	1.072	1.092	1.117	1.142	1.132	1.162
.2000	.979	1.027	1.058	1.075	1.078	1.098	1.105	1.136	1.120	1.148
.2500	1.015	1.069	1.089	1.093	1.105	1.116	1.117	1.148	1.123	1.159
.3000	1.048	1.084	1.102	1.102	1.102	1.104	1.102	1.121	1.123	1.148
.3500	1.048	1.078	1.086	1.087	1.096	1.086	1.099	1.107	1.114	1.145
.4000	1.015	1.045	1.058	1.066	1.078	1.074	1.088	1.113	1.117	1.159
.4500	1.021	1.045	1.077	1.096	1.120	1.127	1.173	1.210	1.234	1.274
.5000	1.024	1.066	1.117	1.150	1.193	1.228	1.313	1.332	1.395	1.381
.5500	1.039	1.105	1.185	1.228	1.268	1.311	1.371	1.408	1.567	1.962
.6000	1.088	1.157	1.241	1.267	1.304	1.329	1.351	1.397	1.611	1.991
.6500	1.124	1.190	1.275	1.297	1.307	1.311	1.333	1.329	1.465	1.832
.7000	1.175	1.250	1.302	1.315	1.334	1.335	1.319	1.311	1.418	1.755
.7500	1.236	1.292	1.333	1.339	1.355	1.365	1.363	1.376	1.433	1.673
.8000	1.299	1.340	1.370	1.372	1.379	1.400	1.436	1.456	1.474	1.602
.8500	1.293	1.328	1.345	1.348	1.364	1.400	1.450	1.503	1.512	1.549
.9000	1.202	1.235	1.247	1.249	1.271	1.317	1.366	1.411	1.450	1.481
.9500	1.130	1.163	1.173	1.174	1.199	1.243	1.289	1.343	1.398	1.419
.9940	1.178	1.190	1.191	1.183	1.202	1.243	1.287	1.343	1.398	1.434

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Fuselage										
.0500	.852	.819	.762	.691	.614	.552	.494	.441	.406	.404
.1000	.930	.910	.855	.781	.717	.647	.594	.550	.515	.490
.1500	.985	.984	.926	.859	.795	.727	.667	.627	.591	.569
.2000	1.003	.994	.963	.901	.852	.780	.731	.692	.655	.622
.2500	1.021	1.021	.988	.937	.898	.837	.789	.737	.708	.679
.3000	1.033	1.033	1.025	.961	.922	.869	.819	.781	.749	.711
.3500	1.015	1.000	1.009	.967	.934	.881	.836	.802	.768	.740
.4000	.955	.964	.954	.910	.870	.822	.775	.743	.713	.681
.4500	.921	.925	.907	.865	.822	.763	.702	.686	.637	.625
.5000	.888	.889	.852	.796	.750	.688	.626	.601	.561	.537
.5500	.852	.849	.812	.754	.705	.623	.567	.539	.509	.472
.6000	.804	.774	.744	.673	.617	.552	.500	.479	.450	.434
.7500	.891	.889	.870	.823	.789	.745	.699	.678	.655	.664
.9500	1.033	1.069	1.117	1.123	1.153	1.130	1.140	1.178	1.205	1.257
.9940	.997	1.120	1.185	1.195	1.232	1.261	1.322	1.352	1.380	1.419

TABLE VIII.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(b) Double slotted flap configuration; $\delta_f = 60^\circ$; $\frac{y}{b/2} = 0.21$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	.786	.966	1.892	3.210	5.339	7.422	9.922	8.482	3.669	2.601
.0125	.906	1.296	1.712	2.213	7.513	10.278	11.945	8.508	3.313	2.372
.0250	.928	1.237	1.551	2.469	3.604	5.894	8.658	7.588	3.319	2.396
.0500	.978	1.202	1.455	1.629	2.021	2.631	4.202	5.177	2.997	2.309
.0750	.991	1.187	1.396	1.547	1.820	2.124	2.938	3.810	2.879	2.297
.1000	1.010	1.199	1.378	1.529	1.750	1.965	2.628	3.270	2.783	2.288
.1500	1.098	1.199	1.353	1.480	1.625	1.782	2.222	2.587	2.620	2.237
.2000	1.069	1.215	1.355	1.456	1.570	1.714	2.059	2.299	2.488	2.207
.2500	1.091	1.234	1.353	1.429	1.528	1.646	1.935	2.082	2.334	2.171
.3000	1.126	1.258	1.356	1.429	1.491	1.593	1.826	1.948	2.196	2.114
.3500	1.148	1.255	1.362	1.414	1.464	1.564	1.710	1.813	2.084	2.084
.4000	1.170	1.280	1.362	1.408	1.460	1.543	1.648	1.742	1.973	2.063
.4500	1.205	1.308	1.384	1.423	1.457	1.516	1.619	1.692	1.904	2.033
.5000	1.245	1.349	1.415	1.438	1.479	1.505	1.595	1.642	1.843	2.006
.5500	1.274	1.374	1.430	1.447	1.473	1.475	1.568	1.604	1.774	1.955
.6000	1.324	1.408	1.461	1.465	1.488	1.478	1.583	1.607	1.732	1.925
.6500	1.378	1.452	1.492	1.505	1.528	1.499	1.607	1.642	1.711	1.898
.7000	1.456	1.533	1.560	1.555	1.579	1.543	1.672	1.663	1.702	1.874
.7815	1.648	1.704	1.718	1.727	1.741	1.741	1.761	1.733	1.735	1.853
.8179	1.827	1.894	1.898	1.906	1.927	1.977	1.956	1.915	1.867	1.910
.8325	1.934	2.000	2.003	2.012	2.037	2.121	2.098	2.056	1.970	1.982
.8362	1.944	2.043	2.028	2.046	2.073	2.159	2.145	2.106	2.006	1.976
.8398	1.918	1.987	1.981	2.000	2.040	2.109	2.092	2.071	1.988	1.958
.8434	1.934	2.100	2.102	2.107	2.153	2.236	2.258	2.203	2.102	2.009
Vane										
.0000	2.359	2.380	2.399	2.429	2.488	2.572	2.509	2.484	2.394	2.159
.0250	4.016	4.078	4.099	4.125	4.323	4.776	4.802	4.743	4.262	3.631
.0500	3.827	3.822	3.833	3.833	4.040	4.508	4.530	4.487	3.985	3.351
.1000	3.711	3.741	3.721	3.681	3.891	4.390	4.489	4.452	3.888	3.087
.1500	3.576	3.613	3.573	3.566	3.790	4.266	4.421	4.394	3.855	3.042
.2000	3.425	3.504	3.464	3.420	3.720	4.165	4.376	4.382	3.901	3.081
.3000	2.928	3.071	3.025	3.061	3.381	3.782	4.075	4.112	3.792	3.054
.4000	2.576	2.666	2.653	2.727	3.015	3.322	3.693	3.772	3.357	2.898
.5000	2.349	2.411	2.406	2.496	2.756	2.991	3.353	3.446	3.340	2.775
.6000	2.126	2.193	2.198	2.222	2.390	2.608	2.864	2.942	2.913	2.329
.7000	2.091	2.146	2.133	2.168	2.308	2.510	2.702	2.739	2.699	2.381
.8000	2.110	2.103	2.096	2.146	2.296	2.481	2.675	2.734	2.599	2.279
.9000	2.079	2.081	2.056	2.095	2.241	2.425	2.631	2.707	2.494	2.156
Flap										
.0000	.975	1.483	1.551	1.435	1.354	1.389	1.509	1.557	1.494	1.340
.0125	2.189	2.255	2.176	2.098	2.177	2.360	2.539	2.599	2.357	2.030
.0250	2.390	2.436	2.393	2.399	2.531	2.752	2.962	3.044	2.753	2.465
.0500	2.453	2.523	2.498	2.520	2.674	2.885	3.107	3.185	3.012	2.700
.0750	2.283	2.352	2.313	2.323	2.473	2.661	2.858	2.918	2.828	2.577
.1000	2.098	2.143	2.108	2.128	2.247	2.413	2.583	2.651	2.620	2.429
.1500	1.821	1.819	1.774	1.781	1.875	2.027	2.169	2.238	2.265	2.138
.2000	1.771	1.766	1.740	1.736	1.825	1.968	2.104	2.156	2.169	2.057
.4000	1.837	1.832	1.783	1.794	1.842	1.959	2.098	2.165	2.148	2.015
.6000	1.764	1.763	1.731	1.736	1.781	1.897	2.039	2.091	2.051	1.970
.8000	1.679	1.688	1.678	1.672	1.717	1.820	1.944	1.965	1.852	1.826

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	.997	.794	.659	.575	.573	.640	.716	.675	.443	.386
.0250	.978	.835	.690	.550	.451	.389	.340	.317	.232	.207
.0500	.956	.860	.731	.584	.439	.330	.257	.220	.211	.207
.0750	.950	.866	.749	.605	.473	.369	.296	.267	.265	.279
.1000	.962	.872	.759	.632	.504	.407	.346	.334	.337	.345
.1500	.940	.869	.774	.657	.588	.478	.426	.411	.410	.405
.2000	.956	.885	.796	.684	.601	.531	.482	.463	.449	.441
.2500	.972	.903	.824	.717	.646	.569	.524	.502	.488	.474
.3000	.953	.903	.833	.739	.665	.605	.553	.537	.524	.511
.3500	.953	.928	.861	.772	.701	.637	.595	.566	.551	.533
.4000	.953	.931	.873	.787	.726	.661	.604	.584	.566	.559
.4500	.928	.922	.873	.793	.732	.673	.624	.598	.578	.574
.5000	.852	.875	.842	.766	.713	.655	.610	.584	.572	.568
.5500	.771	.822	.793	.748	.692	.614	.592	.569	.578	.553
.6000	.726	.773	.737	.681	.649	.590	.559	.546	.521	.526
.6500	.676	.695	.663	.617	.582	.534	.503	.487	.476	.477
.7000	.664	.598	.576	.532	.506	.475	.444	.440	.434	.435
.7815	.569	.502	.446	.410	.372	.351	.340	.328	.316	.315
.8179	.481	.439	.393	.362	.332	.319	.311	.311	.286	.291
.8252	.459	.433	.393	.368	.345	.327	.331	.326	.304	.306
.8325	.453	.433	.415	.398	.396	.389	.382	.375	.367	.360
Vane										
.0250	1.327	1.330	1.307	1.222	1.153	1.103	1.045	1.027	.964	.916
.0500	.811	.779	.724	.587	.470	.395	.370	.358	.334	.330
.1000	.500	.461	.378	.228	.122	.080	.071	.076	.057	.072
.1500	.393	.361	.266	.128	.037	.024	.021	.032	.006	.015
.2000	.315	.299	.223	.088	.021	.009	.003	.021	.003	.006
.3000	.242	.224	.158	.052	.021	.018	.021	.038	.012	.021
.4000	.198	.190	.130	.049	.030	.032	.044	.047	.039	.036
.5000	.164	.168	.130	.058	.049	.065	.068	.067	.057	.063
.6000	.148	.156	.124	.079	.076	.091	.107	.103	.087	.087
.7000	.164	.181	.152	.122	.140	.159	.163	.170	.148	.135
.8000	.230	.249	.235	.222	.250	.271	.299	.311	.265	.249
.9200	.670	.701	.697	.708	.762	.826	.894	.915	.822	.727
Flap										
.0125	.478	.318	.189	.103	.085	.100	.109	.129	.096	.081
.0250	.170	.069	.025	.000	.012	.021	.021	.029	.006	.018
.0500	.097	.069	.037	.033	.046	.047	.050	.059	.042	.037
.0750	.104	.075	.065	.058	.070	.062	.068	.079	.060	.078
.1000	.116	.112	.096	.094	.101	.100	.104	.111	.096	.111
.1500	.135	.140	.159	.125	.137	.136	.145	.144	.133	.147
.4000	.374	.380	.378	.359	.354	.348	.331	.340	.313	.309
.6000	.535	.542	.529	.486	.485	.466	.444	.440	.422	.420
.8000	.651	.660	.641	.608	.588	.572	.547	.528	.491	.508
.9000	.755	.757	.734	.714	.689	.679	.657	.622	.587	.613

TABLE VIII.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(c) Double slotted flap configuration; $\delta_f = 60^\circ$; $\frac{y}{b/2} = 0.30$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	.839	1.256	2.297	2.939	3.605	4.293	4.381	3.772	2.867	2.283
.0125	.994	1.466	2.370	2.954	3.551	4.003	4.074	3.737	2.859	2.271
.0250	1.037	1.336	2.501	3.049	3.675	4.136	4.101	3.749	2.870	2.277
.0500	1.062	1.271	3.101	3.939	4.019	4.067	4.059	3.792	2.850	2.259
.0750	1.074	1.241	1.917	4.080	5.328	5.354	4.499	3.830	2.853	2.259
.1000	1.068	1.222	1.269	2.561	4.110	4.924	4.446	3.722	2.853	2.274
.1500	1.108	1.231	1.306	1.442	2.168	3.354	3.788	3.431	2.782	2.241
.2000	1.136	1.250	1.342	1.366	1.620	2.433	3.127	3.084	2.708	2.229
.2500	1.158	1.259	1.364	1.387	1.496	1.994	2.652	2.787	2.605	2.223
.3000	1.189	1.281	1.379	1.396	1.468	1.826	2.387	2.577	2.528	2.193
.3500	1.201	1.284	1.373	1.387	1.455	1.680	2.121	2.303	2.381	2.170
.4000	1.238	1.318	1.413	1.424	1.459	1.653	1.997	2.189	2.292	2.146
.4500	1.282	1.355	1.434	1.436	1.477	1.646	1.935	2.093	2.213	2.122
.5000	1.313	1.389	1.465	1.460	1.490	1.619	1.838	1.976	2.092	2.068
.5500	1.356	1.413	1.480	1.473	1.514	1.625	1.785	1.903	2.027	2.050
.6000	1.421	1.466	1.526	1.506	1.553	1.631	1.758	1.854	1.974	2.021
.6500	1.511	1.562	1.605	1.576	1.614	1.674	1.752	1.819	1.912	1.991
.7521	1.681	1.802	1.826	1.848	1.854	1.875	1.900	1.898	1.882	1.931
.7934	2.146	2.154	2.159	2.143	2.177	2.195	2.192	2.122	1.962	1.953
.8099	2.390	2.385	2.373	2.357	2.389	2.409	2.392	2.277	2.039	1.970
.8141	2.505	2.472	2.456	2.451	2.493	2.500	2.469	2.347	2.074	1.991
.8182	2.560	2.543	2.520	2.515	2.563	2.579	2.534	2.396	2.097	2.003
.8224	2.703	2.725	2.678	2.683	2.718	2.726	2.699	2.530	2.183	2.030
Vane										
.0000	4.876	4.718	4.559	4.567	4.657	4.738	4.552	4.265	3.629	3.288
.0250	6.362	6.104	5.960	5.964	6.068	6.132	6.033	5.419	4.074	3.324
.0500	6.302	6.209	6.055	6.119	6.124	6.266	6.127	5.513	4.168	3.280
.1000	6.545	6.252	6.055	6.110	6.098	6.258	6.189	5.469	3.994	2.890
.1500	6.505	6.215	5.994	6.052	6.019	6.211	6.154	5.442	3.974	2.881
.2000	6.393	6.187	5.923	5.997	5.952	6.159	6.157	5.466	4.044	2.937
.3000	5.845	5.703	5.407	5.482	5.457	5.732	5.797	5.206	3.956	2.902
.4000	5.149	4.996	4.706	4.738	4.703	5.077	5.165	4.676	3.620	2.717
.5000	4.641	4.422	4.131	4.134	4.125	4.552	4.632	4.279	3.372	2.592
.6000	3.985	3.777	3.463	3.488	3.523	3.961	4.056	3.807	3.100	2.473
.7000	3.421	3.209	2.985	2.942	3.007	3.433	3.531	3.381	2.838	2.345
.8000	2.876	2.697	2.566	2.523	2.572	2.918	3.006	2.953	2.549	2.187
.9000	2.514	2.395	2.324	2.293	2.307	2.555	2.643	2.606	2.331	2.080
Flap										
.0000	1.576	1.599	1.578	1.476	1.426	1.518	1.590	1.551	1.381	1.277
.0125	2.235	2.315	2.242	2.214	2.183	2.323	2.425	2.355	2.151	2.012
.0250	2.737	2.821	2.706	2.683	2.666	2.903	3.024	2.962	2.711	2.539
.0500	2.697	2.796	2.660	2.640	2.678	2.967	3.130	3.119	2.909	2.678
.0750	2.632	2.716	2.566	2.543	2.611	2.933	3.124	3.159	2.923	2.669
.1000	2.554	2.352	2.410	2.384	2.475	2.787	2.955	2.985	2.776	2.521
.1500	2.003	2.037	1.979	1.982	2.119	2.372	2.505	2.556	2.440	2.253
.2000	1.882	1.802	1.764	1.738	1.815	2.037	2.115	2.172	2.065	1.932
.4000	1.656	1.691	1.679	1.683	1.757	1.860	1.915	1.950	1.906	1.884
.6000	1.641	1.707	1.691	1.723	1.781	1.860	1.906	1.892	1.832	1.839
.8000	1.610	1.685	1.660	1.686	1.736	1.823	1.829	1.758	1.696	1.756

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	.975	.796	.737	.817	.991	1.177	1.286	1.294	1.165	1.045
.0250	.997	.833	.709	.689	.786	.799	.835	.840	.802	.729
.0500	.985	.858	.728	.646	.608	.592	.569	.566	.549	.521
.0750	.985	.883	.755	.652	.596	.549	.499	.484	.475	.452
.1000	1.000	.898	.780	.665	.596	.534	.475	.458	.445	.432
.1500	1.000	.904	.807	.701	.620	.546	.472	.458	.437	.429
.2000	.994	.910	.823	.713	.632	.567	.487	.461	.443	.432
.2500	.997	.923	.835	.732	.657	.579	.499	.478	.451	.435
.3000	.981	.920	.841	.741	.675	.595	.522	.493	.475	.444
.3500	.963	.920	.838	.736	.693	.613	.540	.510	.493	.485
.4000	.941	.895	.829	.747	.690	.619	.552	.519	.504	.474
.4500	.901	.870	.813	.741	.678	.619	.552	.531	.516	.497
.5000	.842	.824	.774	.701	.657	.613	.546	.522	.504	.503
.5500	.789	.772	.725	.671	.632	.585	.528	.507	.493	.491
.6000	.712	.694	.651	.604	.568	.537	.487	.472	.457	.449
.6500	.653	.626	.578	.537	.508	.482	.443	.431	.416	.426
.7521	.498	.447	.398	.360	.347	.329	.301	.297	.292	.289
.7934	.430	.389	.358	.314	.292	.271	.248	.259	.245	.247
.8017	.424	.383	.352	.305	.298	.277	.257	.259	.245	.241
.8099	.409	.380	.349	.311	.316	.308	.298	.292	.271	.283
Vane										
.0250	1.768	1.722	1.660	1.540	1.477	1.418	1.307	1.236	1.094	1.024
.0500	.749	.759	.755	.610	.511	.439	.384	.361	.322	.301
.1000	.260	.284	.309	.171	.097	.058	.044	.044	.032	.039
.1500	.127	.145	.187	.070	.015	.000	.000	.006	.003	.000
.2000	.071	.080	.119	.024	.000	.000	.000	.000	.000	.003
.3000	.022	.037	.055	.000	.000	.000	.003	.012	.009	.003
.4000	.022	.022	.031	.012	.012	.015	.027	.032	.024	.027
.5000	.037	.031	.040	.030	.027	.043	.050	.052	.056	.045
.6000	.074	.065	.061	.055	.067	.070	.089	.082	.080	.065
.7000	.133	.117	.116	.110	.131	.131	.145	.140	.130	.116
.8000	.269	.259	.242	.244	.252	.268	.277	.277	.251	.229
.9200	.842	.812	.786	.777	.778	.848	.858	.842	.752	.717
Flap										
.0125	.426	.225	.239	.180	.164	.189	.209	.216	.171	.164
.0250	.019	.025	.034	.003	.000	.018	.018	.035	.021	.006
.0500	.003	.000	.000	.000	.000	.000	.000	.003	.006	.000
.0750	.015	.015	.015	.003	.006	.003	.006	.006	.003	.000
.1000	.028	.031	.021	.006	.012	.012	.018	.023	.012	.015
.1500	.056	.056	.064	.046	.053	.043	.032	.044	.044	.033
.2000	.087	.080	.074	.079	.079	.073	.056	.058	.059	.062
.4000	.214	.216	.202	.180	.188	.174	.150	.146	.150	.143
.6000	.331	.346	.336	.314	.310	.284	.260	.257	.245	.253
.8000	.495	.509	.489	.497	.480	.463	.428	.394	.392	.405
.9000	.644	.679	.661	.674	.663	.637	.593	.542	.543	.562

TABLE VIII.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(d) Double slotted flap configuration; $\delta_f = 60°$; $\frac{y}{b/2} = 0.43$

Upper surface

x/c	C_p for -									
	$\alpha = -4°$	$\alpha = 0°$	$\alpha = 4°$	$\alpha = 8°$	$\alpha = 12°$	$\alpha = 16°$	$\alpha = 20°$	$\alpha = 22°$	$\alpha = 24°$	$\alpha = 26°$
Wing										
.0000	.015	1.878	2.136	2.511	2.939	2.870	2.684	2.610	2.461	2.192
.0125	1.060	1.841	2.118	2.484	2.909	2.873	2.808	2.698	2.476	2.222
.0250	1.116	1.763	2.176	2.544	2.912	2.923	2.855	2.707	2.476	2.204
.0500	1.047	1.741	2.204	2.617	3.052	2.997	2.855	2.734	2.479	2.234
.0750	1.091	1.707	2.471	2.709	3.003	2.879	2.858	2.731	2.470	2.213
.1000	1.126	1.539	2.802	2.979	2.961	2.832	2.897	2.763	2.470	2.213
.1500	1.113	1.315	2.344	3.630	3.683	3.086	2.977	2.766	2.479	2.222
.2000	1.176	1.321	1.588	2.845	3.939	3.360	2.959	2.722	2.458	2.192
.2500	1.195	1.343	1.291	1.909	3.128	3.195	2.823	2.634	2.431	2.195
.3000	1.236	1.389	1.328	1.523	2.485	2.971	2.722	2.569	2.398	2.174
.3500	1.258	1.389	1.353	1.365	2.006	2.626	2.545	2.434	2.337	2.147
.4000	1.277	1.420	1.415	1.362	1.781	2.407	2.441	2.364	2.271	2.129
.4500	1.346	1.470	1.458	1.383	1.677	2.236	2.338	2.294	2.229	2.114
.5000	1.337	1.498	1.495	1.417	1.567	2.015	2.181	2.170	2.157	2.087
.5500	1.440	1.573	1.579	1.487	1.585	1.941	2.113	2.109	2.108	2.084
.6000	1.535	1.651	1.647	1.566	1.607	1.873	2.036	2.050	2.057	2.030
.7000	1.626	1.928	1.947	1.879	1.811	1.835	1.953	1.965	1.976	1.991
.7500	2.069	2.364	2.390	2.320	2.177	2.068	2.071	2.030	1.994	1.988
.7700	2.349	2.716	2.743	2.684	2.512	2.301	2.225	2.141	2.030	2.012
.7750	2.422	2.813	2.839	2.794	2.604	2.372	2.278	2.165	2.048	2.006
.7800	2.551	2.950	2.972	2.921	2.738	2.460	2.338	2.211	2.063	2.021
Vane										
.0000	4.101	4.199	4.177	4.052	3.830	3.398	3.122	2.904	2.617	2.498
.0250	6.639	6.819	6.851	6.797	6.354	5.375	4.776	4.247	3.382	3.039
.0500	6.878	6.981	7.043	6.989	6.534	5.496	4.853	4.256	3.241	2.838
.1000	7.105	7.149	7.201	7.177	6.735	5.558	4.797	4.121	2.964	2.535
.1500	7.312	7.330	7.396	7.390	6.952	5.714	4.894	4.159	2.907	2.438
.2000	7.513	7.613	7.694	7.694	7.269	6.024	5.137	4.356	3.033	2.588
.3000	6.859	6.959	7.000	7.025	6.699	5.549	4.723	3.998	2.816	2.378
.4000	6.120	6.205	6.229	6.275	6.034	5.056	4.338	3.701	2.663	2.285
.5000	5.441	5.457	5.489	5.542	5.336	4.557	3.959	3.405	2.521	2.216
.6000	4.699	4.730	4.762	4.818	4.653	4.030	3.557	3.145	2.404	2.139
.7000	4.098	4.159	4.164	4.213	4.086	3.587	3.252	2.883	2.313	2.102
.8000	3.491	3.557	3.557	3.587	3.503	3.151	2.894	2.607	2.163	2.042
.9000	2.994	3.056	3.077	3.083	2.985	2.726	2.554	2.329	2.027	1.952
Flap										
.0000	2.076	2.430	2.573	2.566	2.404	2.165	1.991	1.807	1.593	1.544
.0125	2.529	2.941	3.000	3.007	2.887	2.631	2.432	2.226	2.003	1.940
.0250	3.025	3.604	3.644	3.657	3.573	3.269	3.048	2.810	2.545	2.462
.0500	3.167	3.626	3.644	3.657	3.598	3.339	3.190	2.948	2.642	2.523
.0750	3.041	3.514	3.545	3.572	3.540	3.331	3.178	2.956	2.620	2.462
.1000	3.092	3.374	3.339	3.390	3.345	3.183	3.066	2.851	2.316	2.354
.1500	2.428	2.598	2.560	2.596	2.616	2.605	2.583	2.440	2.208	2.093
.2000	2.239	2.215	2.207	2.240	2.207	2.165	2.166	2.094	1.916	1.868
.4000	1.585	1.716	1.703	1.721	1.735	1.820	1.897	1.877	1.849	1.841
.6000	1.359	1.529	1.523	1.541	1.601	1.664	1.752	1.760	1.795	1.808
.8000	1.068	1.361	1.350	1.395	1.464	1.490	1.542	1.572	1.684	1.733

Lower surface

x/c	C_p for -									
	$\alpha = -4°$	$\alpha = 0°$	$\alpha = 4°$	$\alpha = 8°$	$\alpha = 12°$	$\alpha = 16°$	$\alpha = 20°$	$\alpha = 22°$	$\alpha = 24°$	$\alpha = 26°$
Wing										
.0125	.969	.797	.824	.945	1.134	1.269	1.349	1.373	1.352	1.273
.0250	.988	.832	.777	.775	.838	.891	.926	.933	.949	.901
.0500	.994	.866	.777	.708	.698	.693	.689	.689	.699	.679
.0750	.988	.885	.789	.702	.662	.631	.611	.610	.608	.601
.1000	.981	.894	.805	.711	.656	.596	.574	.560	.597	.544
.1500	.975	.897	.814	.717	.649	.587	.539	.528	.515	.503
.2000	.962	.903	.820	.717	.652	.584	.521	.513	.509	.492
.2500	.944	.894	.820	.717	.652	.581	.533	.522	.500	.480
.3000	.915	.875	.802	.708	.649	.572	.539	.513	.500	.486
.3500	.899	.854	.802	.708	.649	.587	.536	.522	.512	.489
.4000	.862	.832	.774	.687	.634	.575	.530	.507	.503	.480
.4500	.805	.788	.728	.669	.604	.581	.509	.502	.494	.474
.5000	.752	.720	.697	.617	.561	.528	.485	.478	.464	.453
.5500	.682	.670	.638	.544	.488	.487	.444	.443	.428	.396
.6000	.610	.589	.557	.505	.463	.443	.417	.393	.395	.378
.7000	.415	.414	.406	.356	.335	.316	.296	.293	.271	.279
.7500	.355	.368	.359	.304	.284	.266	.254	.252	.235	.231
.7600	.359	.358	.344	.307	.271	.263	.246	.241	.229	.216
.7700	.352	.386	.322	.331	.296	.289	.260	.217	.277	.216
Vane										
.0250	.981	1.037	1.081	.955	.854	.726	.654	.598	.560	.535
.0500	.418	.495	.548	.471	.372	.298	.231	.220	.214	.198
.1000	.167	.224	.325	.237	.159	.106	.065	.079	.046	.060
.1500	.016	.084	.152	.100	.058	.050	.050	.023	.021	.012
.2000	.013	.078	.105	.058	.040	.047	.009	.015	.018	.006
.3000	.028	.062	.080	.046	.044	.062	.024	.023	.042	.045
.4000	.038	.069	.040	.038	.073	.065	.030	.029	.048	.036
.5000	.047	.078	.068	.085	.076	.071	.068	.070	.051	.048
.6000	.082	.100	.108	.103	.091	.097	.092	.088	.072	.063
.7000	.145	.171	.158	.164	.162	.139	.151	.147	.123	.117
.8000	.315	.327	.325	.316	.317	.298	.260	.249	.214	.222
.9200	1.032	1.053	1.056	1.064	1.021	.912	.849	.798	.706	.700
Flap										
.0125	.230	.299	.381	.344	.302	.283	.254	.238	.190	.171
.0250	.013	.053	.065	.049	.012	.032	.030	.056	.036	.024
.0500	.022	.044	.031	.018	.000	.009	.009	.044	.018	.006
.0750	.016	.040	.022	.027	.030	.027	.033	.032	.042	.036
.1000	.006	.050	.022	.027	.027	.035	.038	.041	.036	.048
.1500	.031	.056	.046	.067	.067	.041	.030	.038	.039	.045
.2000	.057	.084	.071	.079	.079	.071	.074	.070	.063	.063
.4000	.182	.202	.183	.167	.165	.159	.148	.135	.139	.129
.6000	.270	.315	.285	.277	.274	.260	.234	.232	.235	.237
.8000	.450	.452	.430	.416	.418	.401	.373	.381	.395	.405
.9000	.528	.589	.554	.562	.561	.540	.521	.522	.566	.577

TABLE VIII.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(e) Double slotted flap configuration; $\delta_f = 60^\circ$; $\frac{y}{b/2} = 0.55$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	1.372	1.938	2.104	2.418	2.548	2.436	2.233	2.186	2.204	2.101
.0125	1.362	1.957	2.188	2.476	2.572	2.461	2.310	2.245	2.218	2.101
.0250	1.241	1.966	2.144	2.451	2.535	2.467	2.322	2.236	2.207	2.113
.0500	1.214	1.901	2.214	2.518	2.560	2.555	2.369	2.240	2.207	2.107
.0750	1.211	1.728	2.278	2.567	2.590	2.537	2.360	2.260	2.201	2.101
.1000	1.238	1.611	2.345	2.634	2.630	2.506	2.378	2.270	2.195	2.093
.1500	1.260	1.432	2.743	2.665	2.554	2.494	2.407	2.270	2.201	2.093
.2000	1.285	1.389	2.954	2.909	2.511	2.464	2.390	2.271	2.201	2.095
.2500	1.322	1.413	2.844	3.576	2.681	2.528	2.422	2.280	2.210	2.095
.3000	1.350	1.423	2.306	3.641	2.988	2.576	2.392	2.259	2.198	2.089
.3500	1.390	1.460	1.777	3.171	3.034	2.570	2.354	2.242	2.189	2.086
.4000	1.452	1.500	1.468	2.598	2.918	2.540	2.313	2.218	2.182	2.080
.4500	1.477	1.543	1.367	2.076	2.687	2.454	2.268	2.178	2.133	2.077
.5000	1.560	1.617	1.437	1.802	2.514	2.378	2.224	2.145	2.109	2.074
.6201	1.647	1.833	1.731	1.656	2.064	2.146	2.086	2.029	2.021	2.036
.6834	2.108	2.321	2.245	2.076	2.195	2.174	2.109	2.055	2.021	2.042
.7087	2.443	2.663	2.587	2.363	2.298	2.223	2.136	2.073	2.050	2.045
.7131	2.480	2.710	2.626	2.418	2.326	2.232	2.139	2.070	2.033	2.039
.7214	2.477	2.833	2.829	2.604	2.429	2.299	2.183	2.096	2.041	2.042
.7278	2.601	2.972	2.951	2.692	2.472	2.305	2.180	2.110	2.044	2.047
Vane										
.0000	3.381	3.379	3.205	2.823	2.493	2.326	2.210	2.140	2.059	2.039
.0250	6.588	6.619	6.575	5.824	4.657	4.006	3.682	3.428	3.103	2.899
.0500	6.793	6.749	6.758	5.991	4.721	3.985	3.611	3.329	2.938	2.732
.1000	7.260	7.187	7.296	6.513	4.894	3.939	3.466	3.148	2.714	2.521
.1500	7.353	7.221	7.382	6.604	4.840	3.842	3.348	3.061	2.590	2.428
.2000	7.409	7.246	7.465	6.729	4.867	3.842	3.363	3.037	2.537	2.375
.3000	7.012	6.888	7.128	6.540	4.688	3.705	3.230	2.892	2.410	2.277
.4000	6.276	6.191	6.382	5.924	4.232	3.345	2.953	2.673	2.272	2.184
.5000	5.684	5.595	5.798	5.439	3.928	3.147	2.791	2.584	2.218	2.131
.6000	4.963	4.873	5.046	4.759	3.453	2.854	2.599	2.419	2.145	2.089
.7000	4.288	4.246	4.361	4.156	3.143	2.637	2.431	2.291	2.083	2.047
.8000	3.743	3.716	3.792	3.628	2.852	2.470	2.304	2.189	2.030	2.027
.9000	3.207	3.166	3.232	3.092	2.551	2.265	2.193	2.052	1.947	1.982
Flap										
.0000	1.836	1.975	2.153	2.110	1.791	1.573	1.454	1.376	1.322	1.354
.0125	2.839	3.055	3.156	3.104	2.660	2.369	2.186	2.081	2.027	2.092
.0250	3.337	3.635	3.731	3.683	3.171	2.808	2.614	2.484	2.416	2.482
.0500	3.433	3.759	3.856	3.826	3.308	2.936	2.726	2.586	2.475	2.488
.0750	3.433	3.771	3.868	3.860	3.363	2.988	2.752	2.583	2.449	2.419
.1000	3.412	3.635	3.688	3.677	3.228	2.881	2.661	2.510	2.348	2.327
.1500	2.820	2.932	2.914	2.979	2.672	2.442	2.280	2.175	2.059	2.042
.2000	2.412	2.515	2.532	2.525	2.335	2.159	2.044	1.983	1.867	1.887
.4000	1.991	2.031	1.985	2.015	2.000	1.936	1.864	1.825	1.791	1.827
.6000	1.737	1.796	1.737	1.817	1.876	1.866	1.805	1.746	1.779	1.833
.8000	1.353	1.463	1.416	1.512	1.642	1.680	1.640	1.647	1.699	1.786

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	.889	.793	.948	1.183	1.386	1.488	1.519	1.362	1.608	1.595
.0250	.944	.799	.798	.887	.967	1.037	1.047	1.087	1.115	1.137
.0500	.957	.821	.761	.762	.763	.781	.785	.799	.805	.842
.0750	.935	.809	.740	.698	.690	.674	.658	.679	.681	.702
.1000	.938	.821	.734	.689	.657	.637	.608	.618	.620	.631
.1500	.920	.818	.734	.671	.638	.595	.563	.560	.572	.568
.2000	.895	.802	.734	.671	.632	.582	.531	.545	.528	.542
.2500	.882	.796	.725	.659	.620	.576	.525	.531	.510	.530
.3000	.842	.778	.716	.656	.611	.561	.510	.519	.502	.512
.3500	.817	.753	.697	.631	.605	.558	.510	.501	.490	.509
.4000	.768	.722	.667	.613	.584	.543	.487	.490	.472	.482
.4500	.721	.673	.627	.585	.530	.513	.472	.475	.454	.473
.5000	.666	.626	.587	.537	.517	.476	.434	.443	.425	.429
.6201	.452	.435	.431	.402	.380	.354	.319	.326	.301	.315
.6834	.393	.367	.379	.329	.316	.299	.268	.274	.254	.265
.6961	.387	.364	.361	.320	.310	.287	.266	.262	.239	.256
.7007	.375	.367	.361	.320	.295	.293	.268	.262	.251	.256
Vane										
.0250	1.198	1.129	1.147	.976	.885	.781	.699	.697	.658	.667
.0500	.498	.475	.563	.482	.432	.354	.295	.283	.263	.289
.1000	.180	.164	.312	.235	.201	.148	.121	.114	.106	.122
.1500	.096	.074	.165	.146	.128	.091	.080	.085	.083	.077
.2000	.056	.043	.119	.091	.097	.070	.080	.079	.065	.065
.3000	.028	.049	.095	.061	.079	.076	.089	.073	.077	.077
.4000	.056	.074	.092	.079	.088	.079	.089	.093	.083	.101
.5000	.080	.093	.104	.095	.103	.104	.109	.099	.097	.113
.6000	.108	.151	.147	.131	.134	.128	.121	.120	.124	.137
.7000	.195	.216	.211	.204	.188	.168	.177	.178	.162	.182
.8000	.368	.376	.385	.357	.316	.302	.292	.289	.280	.298
.9200	1.161	1.148	1.180	1.119	.939	.848	.805	.787	.767	.824
Flap										
.0125	.257	.293	.376	.366	.301	.274	.257	.259	.224	.250
.0250	.015	.034	.073	.085	.079	.070	.074	.099	.065	.080
.0500	.034	.025	.043	.049	.061	.052	.056	.093	.053	.071
.0750	.040	.034	.055	.070	.070	.067	.074	.082	.065	.086
.1000	.065	.071	.092	.073	.085	.079	.080	.093	.065	.095
.1500	.096	.096	.092	.110	.106	.101	.103	.099	.100	.113
.2000	.121	.117	.122	.140	.119	.122	.130	.120	.103	.128
.4000	.241	.253	.226	.226	.225	.213	.209	.210	.201	.226
.6000	.362	.361	.355	.335	.324	.317	.301	.309	.301	.327
.8000	.489	.494	.477	.470	.496	.482	.460	.446	.448	.482
.9000	.379	.608	.375	.592	.626	.625	.602	.598	.608	.652

TABLE VIII.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Concluded

(f) Double slotted flap configuration; $\delta_f = 60^\circ$; $\frac{y}{b/2} = 0.72$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	2.166	2.226	2.438	2.312	2.181	2.006	1.950	1.926	1.877	1.935
.0125	2.115	2.193	2.342	2.216	2.126	1.958	1.930	1.929	1.889	1.947
.0250	2.187	2.229	2.379	2.240	2.145	1.991	1.959	1.947	1.895	1.950
.0500	2.371	2.274	2.423	2.258	2.157	2.026	1.982	1.962	1.892	1.953
.0750	1.855	2.322	2.475	2.282	2.181	2.044	2.000	1.965	1.892	1.950
.1000	1.329	2.316	2.463	2.270	2.169	2.026	1.988	1.971	1.901	1.956
.1500	1.369	2.479	2.543	2.294	2.175	2.044	2.009	1.988	1.901	1.965
.2000	1.426	2.648	2.608	2.300	2.169	2.059	2.023	2.080	1.915	1.965
.2500	1.459	2.639	2.697	2.276	2.154	2.059	2.035	1.985	1.915	1.968
.3000	1.492	2.461	2.879	2.252	2.160	2.053	2.023	1.988	1.918	1.962
.3500	1.544	2.277	3.129	2.258	2.187	2.059	2.026	1.985	1.918	1.979
.4000	1.565	1.973	3.213	2.315	2.184	2.047	2.006	1.983	1.924	1.979
.4500	1.625	1.768	3.240	2.420	2.196	2.047	2.015	1.980	1.930	1.991
.5000	1.686	1.696	3.256	2.529	2.241	2.047	2.006	1.983	1.936	2.003
.5500	1.707	1.636	3.055	2.462	2.208	2.032	1.988	1.983	1.930	2.000
.6000	1.746	1.642	2.802	2.381	2.196	2.006	1.977	1.977	1.936	2.006
.6500	1.719	1.596	2.509	2.249	2.154	1.997	1.971	1.977	1.933	2.027
.7000	1.652	1.527	2.296	2.213	2.142	1.988	1.965	1.968	1.953	2.027
.7500	1.571	1.479	2.123	2.213	2.123	1.976	1.959	1.965	1.965	2.033
.8000	1.462	1.386	1.969	2.156	2.084	1.981	1.959	1.965	1.965	2.044
.8500	1.405	1.334	1.839	2.045	2.033	1.934	1.947	1.962	1.980	2.038
.9000	1.353	1.316	1.741	1.973	2.009	1.929	1.942	1.959	1.980	2.036

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	.779	.876	1.062	1.153	1.226	1.285	1.336	1.391	1.398	1.475
.0250	.767	.795	.901	.934	1.000	1.041	1.097	1.142	1.164	1.230
.0500	.788	.765	.790	.787	.810	.837	.866	.905	.924	.971
.0750	.788	.774	.762	.733	.741	.765	.778	.808	.822	.858
.1000	.795	.780	.753	.715	.711	.721	.731	.760	.769	.797
.1500	.813	.798	.753	.694	.675	.679	.673	.689	.690	.717
.2000	.834	.814	.787	.715	.687	.682	.675	.681	.675	.699
.2500	.856	.834	.796	.742	.705	.685	.667	.684	.684	.693
.3000	.897	.883	.870	.781	.738	.712	.696	.710	.713	.711
.3500	.991	.991	.981	.871	.816	.786	.766	.781	.772	.782
.4000	1.124	1.139	1.154	1.019	.940	.890	.871	.879	.874	.879
.4500	1.341	1.358	1.383	1.195	1.078	1.021	.994	1.000	.982	.982
.5000	1.592	1.633	1.645	1.381	1.250	1.172	1.135	1.151	1.117	1.118
.5500	1.825	1.883	1.895	1.607	1.449	1.359	1.333	1.349	1.298	1.301
.6000	2.036	2.111	2.105	1.805	1.654	1.573	1.553	1.568	1.544	1.558
.6500	2.226	2.292	2.210	1.916	1.795	1.751	1.766	1.793	1.751	1.758
.7000	2.314	2.352	2.210	1.931	1.852	1.834	1.851	1.885	1.830	1.811
.7500	2.275	2.310	2.154	1.934	1.852	1.834	1.830	1.852	1.807	1.820
.8000	2.121	2.157	2.055	1.886	1.804	1.756	1.749	1.778	1.751	1.788
.8500	1.952	1.991	1.960	1.862	1.780	1.730	1.719	1.746	1.725	1.764
.9000	1.779	1.891	1.878	1.820	1.765	1.724	1.708	1.740	1.722	1.767
.9500	1.592	1.651	1.765	1.805	1.777	1.736	1.731	1.752	1.737	1.770

TABLE IX.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE

(a) Double slotted flap configuration; $\delta_f = 65^\circ$; $\frac{y}{b/2} = 0$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 28^\circ$
Fuselage										
.0000	.009	.022	.000	.000	.009	.036	.054	.092	1.067	.145
.0500	.745	.840	.909	.936	.979	.979	1.030	1.029	1.120	1.078
.1000	.829	.929	.984	.997	1.039	1.030	1.074	1.081	1.140	1.128
.1500	.952	.997	1.044	1.046	1.082	1.059	1.107	1.101	1.137	1.136
.2000	.972	1.043	1.078	1.076	1.106	1.080	1.116	1.098	1.132	1.119
.2500	1.028	1.071	1.106	1.107	1.127	1.092	1.125	1.104	1.135	1.139
.3000	1.065	1.095	1.125	1.100	1.115	1.083	1.113	1.087	1.126	1.128
.3500	1.062	1.095	1.100	1.097	1.106	1.071	1.092	1.069	1.111	1.128
.4000	1.028	1.058	1.075	1.070	1.094	1.068	1.086	1.072	1.120	1.139
.4500	1.034	1.065	1.094	1.107	1.136	1.119	1.173	1.173	1.211	1.247
.5000	1.047	1.083	1.134	1.161	1.208	1.210	1.292	1.292	1.374	1.528
.5500	1.059	1.108	1.194	1.231	1.275	1.287	1.366	1.364	1.506	1.913
.6000	1.103	1.163	1.244	1.274	1.314	1.302	1.354	1.344	1.515	1.954
.6500	1.159	1.212	1.294	1.301	1.323	1.293	1.324	1.283	1.406	1.812
.7000	1.193	1.268	1.322	1.319	1.344	1.302	1.318	1.266	1.366	1.731
.7500	1.258	1.298	1.338	1.350	1.359	1.329	1.357	1.324	1.418	1.652
.8000	1.317	1.345	1.384	1.374	1.372	1.373	1.420	1.413	1.474	1.563
.8500	1.320	1.329	1.366	1.350	1.390	1.376	1.443	1.459	1.515	1.528
.9000	1.227	1.240	1.269	1.262	1.299	1.299	1.381	1.399	1.456	1.450
.9500	1.159	1.169	1.203	1.183	1.239	1.225	1.318	1.324	1.406	1.409
.9940	1.202	1.206	1.213	1.186	1.239	1.240	1.327	1.335	1.389	1.426

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 28^\circ$
Fuselage										
.0500	.842	.837	.766	.693	.640	.559	.485	.439	.421	.377
.1000	.907	.905	.872	.778	.734	.657	.580	.532	.518	.470
.1500	.981	.966	.931	.851	.810	.728	.664	.610	.608	.557
.2000	1.003	1.009	.994	.903	.864	.778	.735	.676	.667	.623
.2500	1.025	1.018	1.019	.948	.912	.820	.777	.728	.719	.675
.3000	1.053	1.040	1.038	.967	.946	.858	.821	.757	.757	.707
.3500	1.022	1.031	1.034	.976	.958	.870	.830	.777	.775	.728
.4000	.957	.972	.963	.915	.882	.826	.777	.725	.725	.670
.4500	.916	.938	.925	.866	.840	.769	.717	.662	.661	.609
.5000	.885	.886	.875	.809	.779	.689	.634	.584	.570	.531
.5500	.829	.846	.825	.754	.707	.633	.571	.523	.515	.475
.6000	.789	.772	.744	.672	.631	.562	.506	.471	.465	.435
.7500	.876	.877	.856	.818	.788	.734	.699	.662	.667	.644
.9500	1.084	1.126	1.175	1.161	1.187	1.157	1.205	1.199	1.275	1.305
.9940	1.020	1.157	1.219	1.222	1.275	1.278	1.354	1.358	1.427	1.432

TABLE IX.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(b) Double slotted flap configuration; $\delta_f = 65^\circ$; $\frac{y}{b/2} = 0.21$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	.797	1.000	2.037	3.243	5.430	7.693	9.691	8.026	3.012	2.456
.0125	.941	1.306	1.771	3.262	7.651	10.558	11.368	7.768	2.740	2.244
.0250	.969	1.245	1.599	2.487	5.678	6.181	8.587	7.026	2.755	2.257
.0500	1.006	1.199	1.478	1.623	2.003	2.675	4.264	4.973	2.645	2.205
.0750	1.022	1.190	1.416	1.549	1.794	2.151	2.968	3.756	2.613	2.199
.1000	1.027	1.194	1.395	1.524	1.722	1.994	2.625	3.190	2.577	2.184
.1500	1.036	1.199	1.364	1.491	1.606	1.799	2.190	2.584	2.486	2.135
.2000	1.096	1.217	1.358	1.463	1.549	1.719	2.015	2.282	2.391	2.111
.2500	1.115	1.232	1.352	1.441	1.501	1.651	1.900	2.065	2.273	2.085
.3000	1.140	1.251	1.355	1.435	1.472	1.610	1.811	1.908	2.160	2.058
.3500	1.162	1.251	1.361	1.426	1.436	1.571	1.693	1.786	2.068	2.041
.4000	1.181	1.278	1.364	1.416	1.436	1.559	1.630	1.721	1.997	2.029
.4500	1.218	1.300	1.379	1.429	1.442	1.530	1.601	1.664	1.941	2.018
.5000	1.265	1.336	1.404	1.457	1.445	1.509	1.580	1.641	1.894	1.991
.5500	1.296	1.370	1.413	1.450	1.448	1.480	1.542	1.593	1.814	1.947
.6000	1.336	1.416	1.447	1.478	1.460	1.488	1.542	1.599	1.749	1.930
.6500	1.396	1.449	1.487	1.509	1.487	1.494	1.568	1.614	1.710	1.895
.7000	1.464	1.514	1.546	1.565	1.537	1.536	1.633	1.650	1.690	1.863
.7815	1.651	1.667	1.682	1.707	1.666	1.707	1.707	1.709	1.719	1.825
.8179	1.841	1.826	1.821	1.839	1.824	1.926	1.879	1.868	1.811	1.848
.8325	1.950	1.905	1.895	1.916	1.919	2.030	1.980	1.955	1.888	1.886
.8362	1.978	1.942	1.929	1.947	1.955	2.071	2.027	2.006	1.912	1.895
.8398	1.938	1.893	1.864	1.898	1.907	2.012	1.971	1.964	1.891	1.877
.8434	2.000	1.988	1.973	1.984	2.009	2.142	2.116	2.095	1.971	1.913
Vane										
.0000	2.869	2.767	2.784	2.811	2.815	2.965	2.906	2.860	2.634	2.310
.0250	4.493	4.180	4.117	4.163	4.325	4.888	4.897	4.780	4.137	3.251
.0500	4.112	3.823	3.709	3.808	3.967	4.441	4.430	4.332	3.782	3.044
.1000	3.729	3.440	3.336	3.354	3.543	4.036	4.101	4.014	3.427	2.719
.1500	3.470	3.202	3.105	3.135	3.288	3.805	3.891	3.821	3.302	2.670
.2000	3.240	3.000	2.901	2.941	3.179	3.560	3.711	3.661	3.240	2.678
.3000	2.629	2.520	2.463	2.531	2.746	2.968	3.187	3.210	2.995	2.585
.4000	2.221	2.162	2.142	2.225	2.385	2.512	2.749	2.801	2.737	2.421
.5000	2.081	2.024	2.000	2.055	2.176	2.385	2.530	2.581	2.554	2.298
.6000	2.025	1.942	1.919	1.984	2.119	2.261	2.391	2.391	2.314	2.140
.7000	1.931	1.884	1.858	1.904	2.003	2.157	2.258	2.285	2.190	2.030
.8000	1.903	1.878	1.839	1.882	1.973	2.139	2.199	2.210	2.083	1.982
.9000	1.866	1.852	1.827	1.844	1.935	2.125	2.196	2.166	1.997	1.906
Flap										
.0000	1.003	1.440	1.460	1.349	1.261	1.979	1.453	1.448	1.367	1.298
.0125	1.997	2.046	1.990	1.959	2.038	2.213	2.284	2.267	2.089	1.959
.0250	2.205	2.229	2.206	2.240	2.349	2.551	2.651	2.635	2.438	2.301
.0500	2.311	2.358	2.376	2.404	2.528	2.767	2.867	2.896	2.740	2.585
.0750	2.162	2.214	2.210	2.259	2.375	2.601	2.725	2.739	2.636	2.491
.1000	2.025	2.076	2.068	2.129	2.200	2.418	2.548	2.563	2.500	2.374
.1500	1.760	1.746	1.725	1.805	1.866	2.024	2.128	2.130	2.163	2.105
.2000	1.713	1.682	1.676	1.737	1.788	1.947	2.033	2.044	2.057	2.023
.4000	1.763	1.740	1.713	1.788	1.824	1.971	2.054	2.041	2.024	1.962
.6000	1.744	1.734	1.694	1.744	1.782	1.917	1.988	1.979	1.977	1.933
.8000	1.701	1.682	1.660	1.703	1.728	1.864	1.926	1.929	1.909	1.848

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	.994	.780	.642	.586	.576	.639	.704	.656	.408	.363
.0250	.975	.813	.660	.559	.442	.385	.334	.303	.234	.216
.0500	.959	.835	.704	.589	.427	.328	.249	.228	.222	.222
.0750	.959	.850	.716	.614	.469	.361	.296	.270	.278	.289
.1000	.966	.859	.738	.639	.501	.405	.346	.326	.349	.354
.1500	.944	.862	.759	.663	.549	.479	.420	.404	.420	.424
.2000	.959	.878	.778	.694	.582	.530	.479	.451	.468	.453
.2500	.966	.896	.790	.722	.624	.577	.518	.498	.503	.491
.3000	.950	.887	.805	.744	.654	.604	.553	.528	.515	.518
.3500	.959	.902	.833	.778	.690	.639	.580	.561	.544	.541
.4000	.944	.911	.842	.793	.698	.660	.604	.576	.562	.558
.4500	.931	.899	.833	.787	.707	.666	.616	.587	.580	.567
.5000	.854	.853	.809	.762	.681	.648	.601	.576	.565	.558
.5500	.797	.798	.765	.713	.646	.630	.571	.543	.583	.558
.6000	.732	.740	.710	.676	.606	.580	.544	.519	.521	.513
.6500	.685	.664	.630	.602	.540	.521	.488	.472	.470	.474
.7000	.670	.590	.537	.518	.469	.450	.423	.415	.414	.415
.7815	.530	.465	.401	.380	.346	.331	.311	.300	.293	.301
.8179	.439	.410	.370	.349	.322	.317	.299	.282	.284	.298
.8252	.424	.413	.376	.361	.331	.340	.311	.303	.299	.301
.8325	.414	.422	.401	.383	.382	.388	.373	.371	.364	.366
Vane										
.0250	1.548	1.550	1.546	1.450	1.322	1.305	1.255	1.216	1.148	1.038
.0500	.866	.881	.839	.682	.552	.515	.476	.466	.444	.432
.1000	.473	.480	.395	.241	.149	.130	.118	.104	.107	.111
.1500	.340	.342	.247	.114	.042	.041	.033	.030	.033	.044
.2000	.262	.266	.182	.062	.015	.012	.012	.009	.012	.020
.3000	.168	.180	.105	.028	.009	.012	.015	.012	.012	.023
.4000	.118	.144	.059	.034	.021	.027	.030	.021	.030	.032
.5000	.081	.107	.032	.034	.030	.041	.044	.033	.053	.053
.6000	.075	.095	.052	.052	.057	.077	.071	.068	.068	.070
.7000	.100	.125	.089	.111	.122	.133	.133	.121	.130	.132
.8000	.178	.190	.179	.201	.230	.252	.257	.233	.240	.234
.9200	.626	.639	.633	.660	.696	.766	.781	.783	.728	.690
Flap										
.0125	.386	.321	.204	.127	.119	.133	.199	.139	.127	.132
.0250	.125	.083	.028	.019	.018	.030	.038	.024	.027	.041
.0500	.069	.061	.028	.037	.030	.041	.044	.033	.047	.058
.0750	.075	.070	.037	.046	.051	.062	.059	.050	.071	.076
.1000	.087	.086	.059	.077	.075	.086	.086	.071	.083	.091
.1500	.112	.119	.105	.108	.104	.115	.112	.098	.115	.123
.4000	.540	.542	.327	.327	.301	.305	.287	.282	.281	.281
.6000	.503	.502	.475	.466	.442	.444	.417	.398	.408	.395
.8000	.620	.630	.596	.583	.549	.536	.524	.495	.497	.497
.9000	.726	.713	.685	.667	.663	.675	.627	.620	.595	.579

TABLE IX.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(c) Double slotted flap configuration; $\delta_f = 65^\circ$; $\frac{y}{b/2} = 0.30$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	.830	1.537	2.281	2.972	3.583	4.249	4.034	3.797	2.503	2.169
.0125	1.018	1.728	2.369	2.988	3.822	3.961	3.783	3.757	2.489	2.166
.0250	1.039	1.642	2.491	3.094	3.652	4.111	3.808	3.761	2.503	2.178
.0500	1.076	1.398	3.137	4.004	3.961	4.015	3.794	3.773	2.459	2.166
.0750	1.073	1.299	1.964	4.137	5.320	5.210	4.119	3.815	2.471	2.163
.1000	1.076	1.271	1.284	2.556	4.121	4.856	4.040	3.719	2.491	2.183
.1500	1.109	1.271	1.296	1.472	2.190	3.378	3.478	3.373	2.426	2.145
.2000	1.136	1.296	1.339	1.395	1.622	2.471	2.921	3.057	2.394	2.140
.2500	1.157	1.312	1.366	1.404	1.486	2.027	2.509	2.767	2.352	2.134
.3000	1.185	1.318	1.381	1.423	1.453	1.841	2.283	2.543	2.308	2.122
.3500	1.209	1.321	1.372	1.423	1.438	1.703	2.011	2.304	2.246	2.090
.4000	1.233	1.358	1.412	1.460	1.459	1.667	1.915	2.185	2.184	2.075
.4500	1.291	1.392	1.439	1.463	1.474	1.652	1.848	2.092	2.148	2.075
.5000	1.327	1.426	1.473	1.485	1.495	1.619	1.763	1.970	2.054	2.029
.5500	1.367	1.460	1.482	1.500	1.498	1.625	1.703	1.904	1.988	1.994
.6000	1.439	1.521	1.543	1.550	1.544	1.640	1.675	1.851	1.941	1.976
.6500	1.521	1.583	1.616	1.624	1.604	1.670	1.675	1.838	1.891	1.944
.7521	1.703	1.879	1.872	1.910	1.835	1.865	1.768	1.892	1.840	1.860
.7934	2.206	2.253	2.217	2.252	2.205	2.231	2.068	2.131	1.897	1.906
.8099	2.479	2.509	2.451	2.510	2.462	2.492	2.277	2.322	1.959	1.921
.8141	2.603	2.611	2.570	2.631	2.568	2.607	2.384	2.412	1.988	1.933
.8182	2.669	2.700	2.647	2.696	2.632	2.688	2.463	2.472	2.015	1.947
.8224	2.839	2.898	2.833	2.898	2.849	2.880	2.656	2.657	2.086	1.965
Vane										
.0000	6.393	6.317	6.144	6.234	6.108	6.219	5.588	5.513	4.069	3.271
.0250	7.302	7.369	7.126	7.138	6.951	7.105	6.560	6.230	3.992	3.032
.0500	7.505	7.274	7.080	7.277	7.087	7.255	6.594	6.289	3.974	2.930
.1000	7.272	7.030	6.818	7.001	6.785	7.006	6.418	6.057	3.616	2.676
.1500	7.045	6.823	6.589	6.774	6.559	6.811	6.260	5.919	3.572	2.670
.2000	6.763	6.626	6.391	6.554	6.362	6.637	6.159	5.674	3.592	2.711
.3000	6.005	5.913	5.641	5.783	5.628	6.045	5.667	5.480	3.480	2.682
.4000	5.136	5.030	4.787	4.886	4.752	5.204	4.916	4.821	3.193	2.513
.5000	4.357	4.251	4.089	4.187	4.112	4.640	4.382	4.237	2.968	2.393
.6000	3.832	3.648	3.400	3.460	3.462	3.994	3.780	3.809	2.755	2.303
.7000	3.239	3.074	2.900	2.932	2.945	3.399	3.277	3.364	2.351	2.201
.8000	2.772	2.682	2.561	2.603	2.377	2.934	2.819	2.928	2.317	2.087
.9000	2.512	2.453	2.366	2.410	2.338	2.407	2.500	2.609	2.163	2.014
Flap										
.0000	1.673	1.716	1.662	1.621	1.523	1.652	1.613	1.664	1.376	1.323
.0125	2.257	2.364	2.268	2.277	2.199	2.402	2.348	2.421	2.051	1.971
.0250	2.736	2.827	2.711	2.727	2.665	2.973	2.896	3.013	2.574	2.460
.0500	2.603	2.753	2.619	2.634	2.634	2.985	2.963	3.194	2.710	2.531
.0750	2.515	2.642	2.506	2.534	2.574	2.946	2.938	3.134	2.725	2.524
.1000	2.448	2.481	2.351	2.382	2.444	2.796	2.785	2.964	2.574	2.384
.1500	1.935	2.009	1.961	2.006	2.103	2.357	2.362	2.549	2.281	2.131
.2000	1.821	1.787	1.726	1.777	1.849	2.048	2.009	2.158	1.956	1.889
.4000	1.636	1.710	1.680	1.746	1.776	1.886	1.848	1.979	1.843	1.828
.6000	1.624	1.734	1.698	1.761	1.791	1.865	1.831	1.934	1.784	1.781
.8000	1.600	1.703	1.665	1.724	1.737	1.802	1.729	1.800	1.687	1.711

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	.942	.753	.729	.832	.982	1.168	1.192	1.295	1.056	1.000
.0250	.970	.793	.704	.699	.722	.790	.780	.848	.737	.705
.0500	.954	.827	.707	.682	.598	.574	.534	.573	.512	.494
.0750	.954	.846	.729	.662	.589	.541	.475	.487	.459	.437
.1000	.961	.858	.753	.677	.589	.535	.446	.457	.429	.411
.1500	.964	.880	.781	.705	.613	.550	.449	.460	.423	.411
.2000	.957	.892	.799	.721	.622	.556	.455	.460	.426	.414
.2500	.957	.904	.814	.736	.650	.580	.469	.472	.441	.429
.3000	.948	.898	.811	.739	.689	.586	.492	.498	.456	.446
.3500	.921	.892	.811	.749	.665	.607	.506	.510	.473	.482
.4000	.894	.864	.799	.745	.662	.610	.509	.519	.482	.466
.4500	.861	.842	.784	.733	.683	.604	.520	.525	.488	.472
.5000	.785	.784	.735	.702	.619	.586	.506	.519	.470	.469
.5500	.734	.738	.692	.655	.595	.556	.489	.501	.453	.455
.6000	.673	.654	.613	.578	.535	.508	.444	.454	.417	.417
.6500	.612	.568	.527	.509	.471	.480	.393	.418	.376	.376
.7521	.461	.395	.369	.335	.305	.288	.260	.284	.246	.259
.7934	.391	.355	.323	.298	.275	.255	.226	.236	.198	.204
.8017	.379	.339	.320	.292	.272	.258	.226	.248	.204	.207
.8099	.367	.339	.314	.298	.281	.282	.260	.269	.234	.222
Vane										
.0250	2.109	2.142	2.076	1.988	1.852	1.814	1.593	1.582	1.240	1.081
.0500	.815	.883	.854	.733	.622	.574	.484	.490	.376	.332
.1000	.233	.296	.290	.199	.127	.105	.078	.078	.056	.041
.1500	.103	.142	.143	.068	.027	.027	.020	.030	.003	.000
.2000	.039	.062	.070	.025	.000	.000	.000	.010	.000	.000
.3000	.021	.009	.000	.000	.000	.000	.000	.009	.000	.000
.4000	.009	.000	.000	.000	.006	.015	.011	.021	.009	.006
.5000	.030	.012	.009	.012	.021	.039	.034	.039	.024	.017
.6000	.070	.043	.034	.030	.039	.063	.057	.066	.044	.041
.7000	.112	.105	.085	.096	.097	.123	.121	.131	.092	.090
.8000	.255	.247	.216	.230	.230	.270	.254	.249	.216	.207
.9200	.848	.824	.768	.792	.764	.868	.814	.848	.701	.688
Flap										
.0125	.312	.312	.302	.267	.245	.267	.271	.287	.216	.204
.0250	.064	.049	.040	.037	.021	.039	.037	.031	.024	.017
.0500	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000
.0750	.006	.000	.000	.000	.000	.000	.000	.000	.000	.000
.1000	.021	.006	.000	.000	.000	.009	.000	.006	.000	.000
.1500	.042	.031	.027	.031	.021	.030	.017	.024	.000	.006
.2000	.055	.049	.052	.047	.033	.039	.028	.039	.015	.029
.4000	.167	.164	.146	.143	.136	.129	.110	.116	.101	.102
.6000	.285	.296	.274	.276	.248	.237	.201	.209	.189	.198
.8000	.448	.475	.459	.447	.417	.390	.345	.346	.337	.341
.9000	.594	.654	.622	.618	.610	.562	.484	.496	.491	.507

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TABLE IX.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(d) Double slotted flap configuration; $\delta_f = 65^\circ$; $\frac{y}{b/2} = 0.43$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	.913	1.871	2.169	2.540	2.886	2.841	2.625	2.546	2.341	2.114
.0125	1.137	1.847	2.163	2.503	2.872	2.864	2.740	2.620	2.358	2.133
.0250	1.162	1.780	2.219	2.568	2.860	2.923	2.779	2.632	2.349	2.120
.0500	1.131	1.852	2.277	2.642	2.994	2.968	2.793	2.667	2.355	2.161
.0750	1.143	1.813	2.561	2.722	2.981	2.864	2.803	2.676	2.349	2.137
.1000	1.156	1.602	2.950	3.012	2.878	2.823	2.820	2.679	2.349	2.137
.1500	1.165	1.309	2.518	3.691	3.492	3.036	2.873	2.682	2.352	2.149
.2000	1.212	1.303	1.639	2.845	3.803	3.246	2.826	2.635	2.338	2.132
.2500	1.221	1.327	1.287	1.901	3.092	3.092	2.722	2.555	2.317	2.126
.3000	1.265	1.367	1.318	1.828	2.546	2.929	2.634	2.498	2.296	2.120
.3500	1.290	1.376	1.339	1.373	2.081	2.607	2.471	2.380	2.244	2.091
.4000	1.324	1.425	1.407	1.364	1.848	2.426	2.370	2.314	2.207	2.085
.4500	1.386	1.465	1.457	1.383	1.707	2.252	2.275	2.240	2.160	2.058
.5000	1.405	1.486	1.487	1.419	1.588	2.062	2.148	2.121	2.089	2.032
.5500	1.498	1.575	1.571	1.497	1.582	1.980	2.062	2.074	2.057	2.006
.6000	1.598	1.645	1.648	1.577	1.591	1.897	1.997	2.018	2.012	1.991
.7000	1.779	1.948	1.969	1.901	1.755	1.870	1.935	1.955	1.950	1.965
.7500	2.283	2.410	2.492	2.370	2.101	2.062	2.045	2.021	1.955	1.962
.7700	2.626	2.789	2.845	2.774	2.442	2.284	2.199	2.156	1.991	1.985
.7750	2.729	2.905	2.968	2.901	2.545	2.370	2.240	2.172	1.997	1.982
.7800	2.881	3.058	3.123	3.081	2.684	2.462	2.311	2.219	2.006	1.991
Vane										
.0000	3.791	5.734	5.783	5.457	5.024	4.436	3.983	3.689	3.036	2.529
.0250	8.417	8.262	8.957	8.408	7.907	6.195	5.194	4.810	3.412	2.556
.0500	8.386	8.177	8.274	8.243	7.274	6.152	5.294	4.664	3.261	2.500
.1000	8.208	7.978	8.054	8.054	7.131	5.903	5.021	4.339	2.811	2.231
.1500	8.158	7.914	7.993	8.014	7.131	5.897	4.974	4.272	2.740	2.202
.2000	8.224	7.991	8.110	8.129	7.277	6.078	5.119	4.412	2.841	2.272
.3000	7.429	7.202	7.249	7.289	6.543	5.477	4.616	3.979	2.616	2.184
.4000	6.437	6.244	6.283	6.326	5.770	4.912	4.184	3.696	2.486	2.126
.5000	5.635	5.428	5.450	5.509	5.030	4.382	3.796	3.341	2.367	2.073
.6000	4.831	4.667	4.675	4.725	4.331	3.847	3.412	3.038	2.267	2.032
.7000	4.190	4.043	4.055	4.089	3.755	3.429	3.104	2.816	2.193	2.006
.8000	3.548	3.437	3.432	3.456	3.209	3.024	2.796	2.578	2.092	1.968
.9000	3.081	2.985	3.018	3.024	2.776	2.651	2.459	2.291	1.953	1.901
Flap										
.0000	2.433	2.544	2.685	2.634	2.346	2.160	1.985	1.848	1.604	1.579
.0125	2.797	2.934	3.030	3.009	2.731	2.542	2.344	2.216	1.938	1.915
.0250	3.374	3.538	3.629	3.404	3.328	3.142	2.932	2.765	2.438	2.406
.0500	3.395	3.498	3.561	3.561	3.328	3.228	3.057	2.899	2.521	2.430
.0750	3.261	3.391	3.425	3.432	3.242	3.184	3.045	2.908	2.491	2.371
.1000	3.224	3.202	3.231	3.228	3.057	3.051	2.953	2.792	2.404	2.287
.1500	2.304	2.443	2.450	2.441	2.406	2.524	2.497	2.403	2.125	2.038
.2000	2.227	2.125	2.098	2.129	2.039	2.110	2.101	2.029	1.847	1.848
.4000	1.679	1.676	1.691	1.697	1.687	1.840	1.873	1.869	1.823	1.816
.6000	1.489	1.541	1.574	1.586	1.618	1.731	1.767	1.768	1.778	1.789
.8000	1.262	1.440	1.475	1.494	1.537	1.571	1.580	1.599	1.690	1.719

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	.959	.789	.833	.957	1.116	1.269	1.320	1.329	1.296	1.249
.0250	.987	.816	.768	.784	.812	.902	.905	.923	.914	.892
.0500	1.000	.844	.759	.716	.693	.698	.684	.674	.686	.673
.0750	.991	.859	.772	.719	.651	.630	.618	.590	.604	.594
.1000	1.000	.868	.784	.719	.639	.618	.562	.558	.556	.553
.1500	.984	.875	.799	.719	.639	.598	.541	.522	.524	.512
.2000	.966	.881	.799	.722	.633	.595	.533	.516	.509	.503
.2500	.944	.875	.799	.722	.633	.589	.533	.516	.509	.494
.3000	.919	.844	.781	.704	.633	.586	.527	.504	.500	.482
.3500	.891	.826	.775	.697	.630	.586	.527	.510	.503	.494
.4000	.857	.792	.753	.679	.615	.571	.524	.498	.488	.468
.4500	.801	.740	.697	.643	.591	.556	.497	.484	.468	.471
.5000	.738	.685	.639	.589	.546	.509	.448	.451	.459	.439
.5500	.660	.630	.593	.555	.481	.476	.420	.424	.452	.412
.6000	.583	.544	.518	.472	.418	.417	.382	.374	.361	.380
.7000	.389	.382	.376	.333	.301	.298	.266	.255	.260	.281
.7500	.340	.330	.339	.290	.266	.257	.213	.217	.216	.211
.7600	.327	.324	.327	.290	.260	.254	.231	.217	.201	.208
.7700	.336	.300	.352	.315	.266	.263	.219	.225	.210	.219
Vane										
.0250	1.290	1.237	1.321	1.222	1.039	.897	.793	.730	.657	.594
.0500	.508	.544	.633	.597	.489	.540	.293	.273	.260	.240
.1000	.165	.226	.302	.247	.173	.139	.107	.098	.086	.085
.1500	.059	.083	.139	.083	.084	.086	.056	.065	.047	.050
.2000	.034	.040	.074	.040	.057	.050	.071	.047	.027	.050
.3000	.034	.061	.052	.049	.057	.038	.044	.056	.030	.032
.4000	.019	.043	.040	.052	.042	.059	.024	.053	.030	.038
.5000	.059	.067	.071	.077	.040	.056	.083	.059	.047	.059
.6000	.084	.092	.093	.093	.090	.098	.083	.077	.059	.096
.7000	.153	.135	.164	.142	.137	.139	.133	.119	.112	.123
.8000	.321	.321	.312	.312	.293	.278	.263	.249	.213	.213
.9200	1.039	1.024	1.052	1.043	.987	.894	.825	.760	.692	.699
Flap										
.0125	.349	.364	.444	.420	.358	.334	.311	.279	.237	.237
.0250	.065	.052	.093	.074	.078	.068	.047	.063	.062	.058
.0500	.003	.043	.031	.031	.033	.039	.018	.027	.012	.041
.0750	.012	.034	.037	.025	.039	.041	.015	.036	.029	.050
.1000	.022	.040	.040	.025	.021	.024	.018	.036	.018	.056
.1500	.025	.034	.049	.022	.030	.030	.024	.042	.041	.044
.2000	.040	.044	.077	.056	.045	.050	.044	.062	.071	.050
.4000	.165	.153	.167	.136	.134	.127	.115	.122	.118	.126
.6000	.265	.263	.259	.231	.218	.215	.198	.196	.222	.199
.8000	.427	.422	.407	.383	.379	.355	.328	.338	.355	.374
.9000	.539	.547	.549	.534	.522	.500	.479	.498	.533	.541

TABLE IX.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(e) Double-slotted flap configuration; $\delta_f = 60^\circ$; $\frac{y}{b/2} = 0.56$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 24^\circ$	$\alpha = 28^\circ$	$\alpha = 32^\circ$
Wing										
.0000	1.542	1.947	2.150	2.466	2.489	2.363	2.083	2.176	2.114	2.041
.0125	1.421	1.961	2.211	2.528	2.513	2.402	2.147	2.224	2.130	2.038
.0250	1.270	1.966	2.186	2.488	2.465	2.399	2.147	2.218	2.128	2.041
.0500	1.233	1.909	2.250	2.559	2.498	2.483	2.204	2.245	2.119	2.029
.0750	1.224	2.102	2.320	2.631	2.529	2.453	2.198	2.260	2.116	2.035
.1000	1.233	2.154	2.390	2.684	2.568	2.438	2.215	2.260	2.119	2.029
.1500	1.267	2.088	2.805	2.727	2.483	2.447	2.232	2.260	2.110	2.041
.2000	1.288	1.488	3.043	3.010	2.447	2.444	2.218	2.245	2.116	2.041
.2500	1.321	1.450	2.903	3.662	2.595	2.465	2.223	2.242	2.113	2.032
.3000	1.348	1.389	2.369	3.702	2.837	2.486	2.195	2.242	2.110	2.035
.3500	1.394	1.438	1.848	3.212	2.867	2.477	2.161	2.215	2.086	2.028
.4000	1.442	1.503	1.509	2.684	2.755	2.423	2.124	2.191	2.074	2.020
.4500	1.488	1.565	1.393	2.093	2.568	2.336	2.074	2.161	2.051	2.011
.5000	1.579	1.660	1.470	1.620	2.420	2.273	2.028	2.119	2.039	1.997
.6201	1.674	1.947	1.835	1.746	2.024	2.051	1.929	2.006	1.971	1.953
.6834	2.179	2.506	2.384	2.199	2.142	2.087	1.949	2.042	1.971	1.968
.7087	2.369	2.907	2.811	2.544	2.245	2.141	1.978	2.066	1.977	1.973
.7151	2.612	2.969	2.857	2.594	2.269	2.141	1.980	2.075	1.980	1.973
.7214	2.639	3.172	3.113	2.642	2.323	2.201	2.026	2.101	1.988	1.973
.7278	2.809	3.339	3.272	2.979	2.441	2.222	2.037	2.110	1.988	1.976
Vane										
.0000	4.560	4.700	4.513	3.929	3.051	2.802	2.534	2.600	2.275	1.982
.0250	8.202	8.824	8.330	7.249	5.021	4.241	3.712	3.725	2.894	2.277
.0500	9.190	8.449	8.309	7.215	4.940	4.180	3.609	3.588	2.808	2.245
.1000	8.317	8.601	8.513	7.454	4.837	3.901	3.288	3.248	2.462	2.096
.1500	8.233	8.436	8.412	7.361	4.673	3.754	3.153	3.113	2.358	2.035
.2000	8.160	8.345	8.330	7.312	4.404	3.697	3.110	3.060	2.305	2.032
.3000	7.384	7.783	7.793	6.945	4.350	3.492	2.944	2.901	2.207	1.994
.4000	6.636	6.805	6.842	6.165	3.870	3.144	2.684	2.663	2.110	1.959
.5000	5.942	6.089	6.138	5.534	3.571	2.949	2.548	2.540	2.062	1.938
.6000	5.145	5.262	5.287	4.789	3.166	2.691	2.379	2.400	2.012	1.921
.7000	4.427	4.530	4.555	4.162	2.879	2.492	2.243	2.260	1.971	1.901
.8000	3.842	3.932	3.952	3.631	2.652	2.345	2.116	2.161	1.941	1.883
.9000	3.206	3.379	3.384	3.115	2.396	2.156	1.961	2.027	1.879	1.880
Flap										
.0000	2.009	2.296	2.360	2.288	1.755	1.520	1.381	1.424	1.332	1.329
.0125	2.957	3.345	3.350	3.212	2.501	2.216	2.020	2.081	1.974	1.968
.0250	3.433	3.925	3.942	3.780	2.982	2.628	2.390	2.475	2.338	2.333
.0500	3.913	4.009	4.016	3.876	3.078	2.736	2.486	2.561	2.376	2.332
.0750	3.500	4.018	4.019	3.876	3.115	2.778	2.503	2.573	2.329	2.253
.1000	3.460	3.830	3.814	3.671	2.988	2.688	2.430	2.484	2.243	2.183
.1500	2.833	3.049	2.988	2.924	2.483	2.285	2.110	2.167	1.983	1.950
.2000	2.466	2.657	2.610	2.534	2.184	2.036	1.870	1.946	1.811	1.813
.4000	2.030	2.129	2.058	2.028	1.912	1.868	1.740	1.815	1.761	1.769
.6000	1.766	1.873	1.793	1.814	1.834	1.799	1.669	1.773	1.758	1.778
.8000	1.340	1.534	1.457	1.537	1.634	1.655	1.557	1.642	1.701	1.717

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 24^\circ$	$\alpha = 28^\circ$	$\alpha = 32^\circ$
Wing										
.0125	.848	.827	.976	1.218	1.381	1.462	1.427	1.558	1.568	1.551
.0250	.900	.802	.811	.898	.961	1.006	.992	1.084	1.101	1.099
.0500	.906	.812	.759	.758	.758	.772	.782	.788	.811	.819
.0750	.891	.813	.732	.702	.680	.670	.616	.669	.681	.679
.1000	.888	.815	.732	.690	.650	.616	.576	.609	.618	.618
.1500	.876	.815	.735	.674	.616	.577	.523	.543	.559	.560
.2000	.864	.812	.717	.662	.601	.552	.503	.531	.527	.531
.2500	.853	.790	.713	.665	.601	.536	.494	.504	.512	.510
.3000	.800	.759	.695	.637	.583	.538	.477	.498	.497	.496
.3500	.767	.738	.677	.627	.571	.520	.472	.484	.482	.487
.4000	.718	.694	.640	.590	.530	.508	.452	.457	.476	.461
.4500	.670	.657	.598	.550	.517	.477	.424	.457	.441	.434
.5000	.615	.599	.561	.509	.480	.441	.396	.406	.408	.408
.6201	.415	.426	.409	.379	.341	.324	.285	.290	.281	.286
.6834	.364	.376	.354	.320	.281	.258	.232	.239	.237	.233
.6961	.351	.370	.354	.317	.278	.252	.232	.239	.234	.222
.7087	.351	.364	.335	.317	.281	.249	.232	.239	.237	.224
Vane										
.0250	1.430	1.457	1.476	1.249	.994	.892	.774	.797	.734	.679
.0500	.851	.580	.662	.547	.432	.366	.311	.313	.302	.294
.1000	.191	.201	.290	.230	.193	.153	.124	.125	.136	.134
.1500	.064	.108	.162	.137	.106	.087	.062	.087	.089	.087
.2000	.039	.068	.098	.093	.079	.081	.062	.075	.077	.070
.3000	.033	.065	.076	.056	.057	.087	.073	.069	.071	.070
.4000	.052	.148	.076	.087	.063	.117	.088	.078	.080	.087
.5000	.094	.136	.107	.096	.103	.120	.090	.099	.098	.122
.6000	.127	.154	.125	.130	.121	.129	.113	.122	.136	.128
.7000	.197	.235	.201	.205	.169	.156	.150	.167	.172	.169
.8000	.382	.417	.390	.357	.287	.267	.271	.275	.266	.277
.9200	1.206	1.237	1.238	1.115	.879	.802	.751	.770	.760	.747
Flap										
.0125	.391	.451	.503	.488	.338	.300	.288	.304	.278	.289
.0250	.073	.102	.140	.106	.118	.081	.079	.096	.083	.102
.0500	.036	.062	.049	.056	.045	.045	.065	.066	.065	.076
.0750	.042	.077	.052	.075	.088	.069	.079	.078	.065	.064
.1000	.065	.099	.095	.084	.079	.081	.079	.075	.074	.079
.1500	.100	.096	.098	.115	.100	.072	.088	.093	.089	.087
.2000	.091	.120	.101	.105	.106	.114	.090	.099	.101	.105
.4000	.213	.235	.198	.211	.199	.174	.164	.170	.178	.181
.6000	.312	.324	.284	.301	.296	.264	.254	.251	.275	.257
.8000	.421	.472	.418	.438	.432	.417	.384	.403	.414	.426
.9000	.536	.583	.546	.556	.571	.568	.528	.543	.562	.577

TABLE IX.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Concluded

(f) Double slotted flap configuration; $\delta_f = 55^\circ$; $\frac{y}{b/2} = 0.7\%$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	2.221	2.283	2.519	2.307	2.151	1.938	1.884	1.832	1.874	1.902
.0125	2.199	2.246	2.416	2.222	2.091	1.891	1.875	1.832	1.883	1.908
.0250	2.270	2.274	2.456	2.247	2.115	1.920	1.896	1.844	1.886	1.908
.0500	2.350	2.335	2.500	2.262	2.130	1.956	1.911	1.855	1.892	1.910
.0750	2.215	2.369	2.569	2.286	2.157	1.971	1.931	1.852	1.898	1.908
.1000	1.478	2.351	2.559	2.274	2.148	1.956	1.931	1.861	1.898	1.910
.1500	1.332	2.514	2.641	2.304	2.139	1.968	1.955	1.873	1.904	1.908
.2000	1.413	2.729	2.713	2.298	2.133	1.991	1.964	1.879	1.908	1.925
.2500	1.469	2.742	2.809	2.274	2.136	1.983	1.955	1.873	1.912	1.919
.3000	1.510	2.579	3.025	2.280	2.148	1.977	1.955	1.876	1.915	1.925
.3500	1.575	2.434	3.241	2.268	2.151	1.977	1.946	1.879	1.918	1.942
.4000	1.612	2.182	3.347	2.353	2.157	1.965	1.943	1.879	1.921	1.931
.4500	1.662	1.895	3.363	2.435	2.163	1.956	1.937	1.873	1.921	1.942
.5000	1.730	1.788	3.344	2.535	2.181	1.962	1.937	1.879	1.930	1.939
.5500	1.755	1.699	3.128	2.453	2.151	1.932	1.925	1.867	1.927	1.939
.6000	1.761	1.668	2.844	2.374	2.139	1.920	1.908	1.861	1.921	1.939
.6500	1.742	1.603	2.541	2.244	2.106	1.909	1.899	1.852	1.924	1.948
.7000	1.674	1.532	2.313	2.195	2.072	1.891	1.893	1.855	1.933	1.960
.7500	1.596	1.465	2.134	2.201	2.066	1.888	1.893	1.855	1.936	1.966
.8000	1.478	1.375	1.988	2.134	2.021	1.876	1.893	1.861	1.936	1.971
.8500	1.416	1.317	1.863	2.025	1.976	1.855	1.878	1.858	1.936	1.974
.9000	1.357	1.314	1.772	1.976	1.970	1.838	1.878	1.855	1.942	1.963

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	.767	.880	1.106	1.152	1.227	1.234	1.301	1.329	1.383	1.450
.0250	.758	.800	.931	.948	.994	1.015	1.074	1.104	1.158	1.200
.0500	.761	.757	.809	.784	.825	.820	.860	.873	.918	.962
.0750	.749	.751	.788	.742	.752	.749	.777	.783	.816	.852
.1000	.752	.766	.766	.724	.719	.704	.723	.728	.760	.783
.1500	.767	.772	.772	.699	.677	.663	.684	.665	.690	.710
.2000	.798	.797	.797	.705	.689	.663	.664	.659	.678	.690
.2500	.836	.825	.822	.724	.698	.686	.664	.656	.673	.681
.3000	.888	.877	.881	.766	.731	.695	.699	.688	.702	.704
.3500	.985	.991	1.003	.863	.819	.768	.771	.757	.772	.771
.4000	1.127	1.157	1.194	1.009	.937	.876	.881	.853	.871	.867
.4500	1.376	1.406	1.463	1.176	1.078	1.009	.997	.977	.994	.980
.5000	1.662	1.686	1.763	1.386	1.251	1.157	1.155	1.113	1.123	1.102
.5500	1.893	1.982	2.053	1.645	1.468	1.352	1.354	1.321	1.343	1.323
.6000	2.125	2.234	2.266	1.888	1.704	1.583	1.610	1.587	1.623	1.612
.6500	2.333	2.400	2.369	2.025	1.903	1.802	1.851	1.809	1.828	1.737
.7000	2.359	2.443	2.338	2.052	1.970	1.873	1.902	1.850	1.860	1.757
.7500	2.330	2.357	2.250	2.028	1.943	1.829	1.860	1.806	1.825	1.757
.8000	2.137	2.185	2.134	1.952	1.870	1.784	1.798	1.740	1.760	1.742
.8500	1.947	2.012	2.038	1.903	1.834	1.740	1.765	1.717	1.751	1.751
.9000	1.758	1.859	1.941	1.848	1.818	1.725	1.756	1.711	1.749	1.751
.9500	1.581	1.655	1.825	1.815	1.791	1.707	1.753	1.711	1.757	1.751

[redacted]

TABLE X.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE

(a) Extended double slotted flap configuration; $\delta_f = 45^\circ$; $\frac{y}{b/2} = 0$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Fuselage										
.0000	.006	.009	.015	.012	.003	.015	.072	.085	.109	.128
.0500	.782	.816	.905	.907	.967	1.015	1.009	1.047	1.047	1.070
.1000	.875	.906	.982	.970	1.021	1.065	1.052	1.085	1.112	1.125
.1500	.953	.964	1.046	1.021	1.058	1.107	1.084	1.112	1.129	1.136
.2000	1.012	1.009	1.074	1.057	1.088	1.116	1.099	1.115	1.120	1.119
.2500	1.053	1.048	1.111	1.081	1.104	1.125	1.107	1.126	1.129	1.131
.3000	1.084	1.069	1.117	1.072	1.104	1.125	1.096	1.115	1.115	1.125
.3500	1.096	1.075	1.114	1.078	1.119	1.149	1.136	1.150	1.173	1.188
.4000	1.044	1.027	1.068	1.045	1.076	1.098	1.093	1.126	1.144	1.165
.4500	1.037	1.030	1.080	1.072	1.122	1.161	1.171	1.226	1.232	1.281
.5000	1.037	1.048	1.114	1.120	1.192	1.271	1.305	1.362	1.405	1.516
.5500	1.050	1.066	1.154	1.180	1.268	1.354	1.374	1.438	1.475	1.780
.6000	1.072	1.103	1.212	1.219	1.304	1.384	1.363	1.391	1.443	1.821
.6500	1.094	1.127	1.234	1.237	1.313	1.351	1.339	1.347	1.387	1.629
.7000	1.143	1.166	1.265	1.255	1.316	1.357	1.328	1.362	1.370	1.536
.7500	1.187	1.196	1.268	1.270	1.319	1.363	1.363	1.394	1.428	1.548
.8000	1.230	1.233	1.311	1.279	1.328	1.387	1.383	1.426	1.461	1.519
.8500	1.293	1.281	1.351	1.303	1.368	1.405	1.389	1.447	1.481	1.513
.9000	1.296	1.289	1.320	1.273	1.319	1.357	1.342	1.385	1.423	1.455
.9500	1.243	1.214	1.271	1.231	1.259	1.280	1.249	1.300	1.340	1.380
.9940	1.265	1.214	1.252	1.207	1.237	1.253	1.209	1.235	1.296	1.380

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Fuselage										
.0500	.869	.822	.785	.685	.620	.562	.475	.471	.402	.371
.1000	.963	.897	.874	.787	.736	.667	.568	.559	.502	.473
.1500	1.019	.958	.932	.865	.793	.738	.661	.621	.581	.551
.2000	1.050	.997	.975	.904	.854	.792	.702	.694	.648	.615
.2500	1.062	1.009	1.012	.943	.891	.845	.745	.738	.692	.675
.3000	1.006	.976	.991	.940	.936	.914	.867	.868	.848	.844
.3500	1.040	1.003	1.022	.964	.933	.881	.800	.788	.748	.713
.4000	.987	.958	.963	.919	.869	.833	.742	.744	.692	.667
.4500	.972	.927	.926	.865	.824	.774	.690	.676	.639	.603
.5000	.947	.903	.883	.826	.780	.708	.615	.603	.548	.528
.5500	.913	.873	.849	.769	.711	.649	.551	.532	.490	.461
.6000	.869	.819	.782	.703	.638	.577	.496	.465	.431	.412
.7500	1.047	.991	.988	.907	.857	.809	.733	.712	.701	.670
.9500	.878	.861	.926	.871	.860	.848	.841	.859	.883	.887
.9940	1.000	1.045	1.185	1.129	1.149	1.155	1.142	1.171	1.205	1.249

[redacted]

TABLE X.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(b) Extended double slotted flap configuration; $\delta_f = 45^\circ$; $\frac{y}{b/2} = 0.21$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	.934	.906	1.846	3.119	5.175	7.934	11.010	12.980	10.251	3.168
.0125	.927	1.258	1.659	2.969	7.147	10.743	13.680	15.178	9.690	2.898
.0250	.981	1.209	1.526	2.343	3.482	6.312	8.345	11.602	9.101	2.974
.0500	1.006	1.181	1.406	1.613	1.994	2.744	3.804	4.769	6.089	2.686
.0750	1.022	1.168	1.369	1.512	1.820	2.196	2.778	3.328	4.482	2.648
.1000	1.054	1.168	1.345	1.503	1.739	2.045	2.508	2.959	3.829	2.607
.1500	1.067	1.159	1.317	1.460	1.646	1.827	2.156	2.444	2.988	2.569
.2000	1.092	1.181	1.311	1.432	1.584	1.723	2.006	2.237	2.618	2.522
.2500	1.108	1.184	1.314	1.411	1.534	1.640	1.921	2.114	2.318	2.470
.3000	1.130	1.193	1.311	1.405	1.494	1.583	1.848	1.950	2.141	2.414
.3500	1.137	1.206	1.298	1.386	1.466	1.548	1.722	1.828	1.994	2.352
.4000	1.156	1.215	1.308	1.383	1.451	1.527	1.640	1.760	1.902	2.305
.4500	1.175	1.234	1.320	1.380	1.447	1.503	1.622	1.711	1.826	2.264
.5000	1.210	1.258	1.329	1.386	1.438	1.485	1.604	1.684	1.780	2.235
.5500	1.222	1.258	1.329	1.380	1.435	1.440	1.572	1.649	1.743	2.170
.6000	1.251	1.286	1.335	1.380	1.435	1.420	1.569	1.661	1.751	2.100
.6500	1.270	1.296	1.354	1.395	1.426	1.411	1.610	1.693	1.748	2.033
.7000	1.295	1.315	1.363	1.395	1.435	1.420	1.598	1.640	1.685	1.965
.7824	1.318	1.339	1.363	1.392	1.429	1.428	1.493	1.491	1.540	1.842
.8176	1.375	1.383	1.406	1.435	1.454	1.479	1.493	1.474	1.526	1.813
.8323	1.422	1.430	1.449	1.466	1.491	1.518	1.522	1.512	1.555	1.824
.8913	1.511	1.489	1.517	1.537	1.537	1.556	1.516	1.503	1.552	1.734
.9281	1.632	1.607	1.609	1.619	1.634	1.628	1.581	1.582	1.636	1.763
.9904	1.721	1.670	1.665	1.688	1.677	1.678	1.628	1.632	1.705	1.789
.9804	1.937	1.838	1.815	1.803	1.808	1.809	1.780	1.833	1.907	1.915
Vane										
.0000	1.197	1.006	.948	.788	.662	.536	.519	.518	.552	.551
.0250	1.915	1.832	1.819	1.822	1.801	1.818	1.813	1.868	1.931	1.904
.0500	2.765	2.582	2.535	2.484	2.497	2.556	2.561	2.658	2.777	2.563
.1000	3.251	3.006	2.923	2.816	2.851	2.970	3.018	3.161	3.263	2.813
.1500	3.505	3.227	3.114	2.963	3.016	3.166	3.250	3.421	3.526	2.956
.2000	3.718	3.398	3.274	3.113	3.168	3.327	3.449	3.646	3.760	3.132
.3000	3.743	3.420	3.259	3.093	3.162	3.360	3.528	3.751	3.864	3.300
.4000	3.556	3.190	3.025	2.883	2.988	3.178	3.376	3.576	3.719	3.259
.5000	3.362	2.997	2.834	2.717	2.845	3.024	3.223	3.404	3.575	3.220
.6000	2.889	2.579	2.443	2.374	2.482	2.592	2.778	2.909	3.101	2.945
.7000	2.578	2.318	2.212	2.147	2.227	2.289	2.461	2.582	2.766	2.754
.8000	2.486	2.271	2.166	2.067	2.137	2.205	2.352	2.459	2.610	2.578
.9000	2.480	2.258	2.148	2.067	2.115	2.199	2.320	2.421	2.500	2.317
Flap										
.0000	1.924	1.498	1.314	1.129	1.115	1.170	1.235	1.292	1.335	1.123
.0125	2.569	2.330	2.179	2.036	2.044	2.134	2.214	2.295	2.370	2.071
.0250	2.832	2.435	2.514	2.401	2.454	2.559	2.651	2.763	2.852	2.528
.0500	2.921	2.731	2.668	2.558	2.615	2.738	2.827	2.924	3.061	2.895
.0750	2.694	2.551	2.489	2.414	2.466	2.559	2.637	2.734	2.864	2.816
.1000	2.407	2.283	2.249	2.214	2.274	2.351	2.405	2.456	2.621	2.657
.1500	1.962	1.850	1.806	1.794	1.842	1.869	1.950	2.003	2.132	2.291
.2000	1.984	1.873	1.831	1.782	1.811	1.869	1.965	2.023	2.115	2.203
.4000	1.988	1.857	1.803	1.825	1.839	1.875	1.942	1.988	2.101	2.226
.6000	1.857	1.754	1.726	1.724	1.758	1.815	1.880	1.909	1.980	1.989
.8000	1.724	1.670	1.652	1.638	1.668	1.690	1.786	1.825	1.858	1.742

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	1.118	.832	.668	.604	.578	.631	.798	.909	.824	.411
.0250	1.080	.863	.708	.574	.469	.375	.375	.368	.353	.214
.0500	1.070	.885	.748	.626	.469	.310	.264	.208	.220	.211
.0750	1.057	.891	.760	.456	.497	.339	.287	.254	.269	.270
.1000	1.054	.906	.794	.666	.528	.387	.346	.319	.327	.334
.1500	1.045	.919	.809	.702	.575	.455	.419	.406	.405	.408
.2000	1.057	.938	.834	.736	.621	.524	.466	.462	.451	.443
.2500	1.073	.953	.862	.767	.671	.559	.525	.500	.483	.475
.3000	1.070	.972	.883	.797	.696	.595	.554	.529	.514	.504
.3500	1.092	1.003	.926	.850	.745	.640	.595	.561	.546	.534
.4000	1.118	1.019	.945	.871	.773	.670	.622	.588	.564	.560
.4500	1.105	1.028	.960	.889	.801	.696	.642	.614	.587	.575
.5000	1.080	1.015	.951	.892	.814	.693	.648	.611	.587	.584
.5500	1.029	.997	.926	.889	.789	.679	.636	.614	.598	.592
.6000	1.006	.950	.917	.853	.783	.673	.636	.599	.584	.584
.6500	.959	.910	.862	.816	.752	.655	.604	.570	.564	.563
.7000	.908	.866	.822	.785	.721	.625	.590	.558	.543	.551
.7815	.794	.760	.714	.678	.634	.539	.516	.494	.486	.493
.8179	.772	.720	.685	.650	.603	.527	.507	.474	.477	.481
.8325	.756	.710	.686	.653	.603	.533	.516	.500	.500	.496
.8566	.749	.720	.692	.675	.627	.568	.543	.529	.525	.528
.8921	.683	.592	.545	.515	.453	.393	.375	.363	.364	.367
.9286	.641	.587	.532	.509	.466	.417	.408	.389	.390	.393
.9518	.616	.599	.508	.466	.432	.372	.372	.357	.355	.364
.9803	.603	.590	.477	.445	.398	.342	.343	.330	.344	.349
Vane										
.0250	.787	.511	.391	.289	.174	.107	.120	.108	.124	.117
.0500	.610	.368	.249	.181	.115	.095	.114	.120	.124	.117
.1000	.521	.315	.228	.184	.146	.134	.150	.155	.171	.147
.1500	.476	.305	.222	.190	.165	.152	.167	.170	.176	.170
.2000	.457	.312	.234	.199	.180	.149	.173	.170	.179	.164
.3000	.425	.299	.237	.202	.177	.167	.185	.184	.188	.173
.4000	.416	.305	.243	.212	.180	.173	.188	.184	.194	.176
.5000	.406	.305	.255	.227	.199	.185	.202	.190	.208	.191
.6000	.419	.324	.268	.239	.208	.205	.217	.208	.214	.205
.7000	.457	.355	.311	.279	.255	.241	.258	.249	.263	.243
.8000	.546	.452	.397	.362	.325	.339	.364	.363	.373	.340
.9200	1.016	.906	.834	.830	.801	.821	.862	.883	.918	.801
Flap										
.0125	.460	.106	.034	.025	.003	.003	.029	.023	.043	.035
.0250	.251	.103	.065	.067	.071	.048	.056	.058	.072	.062
.0500	.229	.181	.157	.166	.149	.131	.144	.132	.142	.158
.0750	.232	.218	.194	.190	.189	.167	.167	.164	.176	.188
.1000	.260	.271	.243	.239	.224	.196	.208	.196	.205	.217
.1500	.286	.305	.280	.282	.258	.241	.241	.231	.243	.252
.4000	.540	.536	.514	.506	.481	.440	.437	.421	.428	.437
.6000	.683	.673	.643	.626	.603	.542	.537	.518	.523	.507
.8000	.778	.757	.738	.715	.686	.610	.604	.599	.587	.563
.9000	.864	.832	.834	.807	.780	.720	.701	.693	.662	.636

TABLE X.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(c) Extended double slotted flap configuration; $\delta_f = 45^\circ$; $\frac{y}{b/2} = 0.30$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	1.016	1.167	1.909	2.730	3.447	4.465	5.272	5.276	4.250	2.520
.0125	.944	1.373	2.087	2.793	3.390	4.157	4.845	4.945	4.224	2.485
.0250	.988	1.271	2.171	2.875	3.323	4.347	4.982	4.959	4.241	2.526
.0500	1.025	1.222	2.495	3.818	3.841	4.388	4.787	4.892	4.293	2.419
.0750	1.041	1.200	1.586	3.522	5.144	5.891	6.123	5.811	4.543	2.442
.1000	1.041	1.182	1.224	2.183	3.964	5.900	6.091	5.861	4.264	2.497
.1500	1.076	1.188	1.265	1.384	2.123	3.261	4.570	4.866	3.977	2.376
.2000	1.091	1.194	1.302	1.340	1.595	2.204	3.313	3.770	3.598	2.347
.2500	1.098	1.207	1.311	1.371	1.462	1.793	2.637	3.073	3.250	2.370
.3000	1.126	1.216	1.315	1.368	1.435	1.675	2.316	2.660	2.986	2.318
.3500	1.123	1.213	1.302	1.378	1.393	1.556	2.026	2.311	2.673	2.306
.4000	1.157	1.228	1.311	1.384	1.405	1.539	1.921	2.163	2.519	2.283
.4500	1.192	1.253	1.333	1.403	1.414	1.533	1.877	2.084	2.395	2.251
.5000	1.208	1.265	1.339	1.409	1.411	1.515	1.795	1.962	2.238	2.208
.5500	1.220	1.281	1.346	1.406	1.405	1.494	1.751	1.881	2.128	2.173
.6000	1.252	1.293	1.358	1.418	1.420	1.509	1.728	1.840	2.058	2.144
.6500	1.283	1.324	1.380	1.434	1.435	1.512	1.711	1.799	1.997	2.104
.7516	1.258	1.321	1.374	1.412	1.399	1.480	1.611	1.712	1.855	1.959
.7940	1.293	1.352	1.405	1.453	1.429	1.491	1.635	1.735	1.809	1.928
.8099	1.267	1.330	1.439	1.437	1.420	1.488	1.611	1.698	1.768	1.881
.8429	1.352	1.438	1.477	1.513	1.486	1.556	1.696	1.776	1.800	1.890
.8767	1.365	1.457	1.501	1.532	1.526	1.571	1.702	1.762	1.742	1.821
.9108	1.466	1.552	1.604	1.626	1.610	1.669	1.789	1.829	1.760	1.795
.9437	1.648	1.734	1.763	1.793	1.763	1.838	1.962	1.974	1.838	1.818
.9768	2.010	2.074	2.121	2.151	2.096	2.207	2.351	2.320	2.108	1.913
Vane										
.0000	2.312	2.139	2.090	1.959	1.739	1.666	1.722	1.619	1.464	1.200
.0250	3.164	2.987	3.137	3.101	2.748	3.228	3.035	2.991	2.796	2.477
.0500	3.878	3.805	3.881	3.903	3.793	4.051	4.322	4.140	3.708	2.800
.1000	4.444	4.320	4.383	4.425	4.228	4.604	4.857	4.695	4.143	2.928
.1500	4.743	4.607	4.678	4.718	4.459	4.909	5.140	5.003	4.412	2.988
.2000	5.032	4.913	4.975	5.015	4.694	5.190	5.424	5.323	4.720	3.136
.3000	5.063	4.993	5.043	5.079	4.712	5.214	5.406	5.375	4.833	3.196
.4000	4.872	4.752	4.788	4.812	4.462	4.971	5.167	5.154	4.638	3.046
.5000	4.651	4.469	4.504	4.494	4.225	4.758	4.851	4.887	4.441	2.855
.6000	4.205	4.036	4.059	4.035	3.811	4.308	4.377	4.448	4.088	2.717
.7000	3.721	3.546	3.554	3.513	3.363	3.782	3.880	3.951	3.659	2.509
.8000	3.211	3.055	3.065	3.007	2.934	3.284	3.406	3.442	3.198	2.280
.9000	2.774	2.635	2.617	2.579	2.526	2.802	2.980	2.997	2.789	2.139
Flap										
.0000	1.299	1.241	1.212	1.157	1.060	1.187	1.301	1.297	1.220	.957
.0125	2.308	2.308	2.296	2.264	2.126	2.299	2.439	2.459	2.387	1.968
.0250	2.912	2.929	2.931	2.922	2.748	2.997	3.184	3.201	3.061	2.363
.0500	3.080	3.105	3.109	3.076	2.946	3.237	3.482	3.520	3.372	2.824
.0750	3.120	3.157	3.143	3.104	2.973	3.302	3.588	3.642	3.473	2.881
.1000	3.016	2.969	2.947	2.912	2.802	3.139	3.392	3.439	3.296	2.717
.1500	2.579	2.537	2.483	2.475	2.477	2.773	2.953	2.991	2.902	2.428
.2000	2.161	2.031	2.025	1.975	1.979	2.219	2.380	2.387	2.325	1.988
.4000	1.676	1.645	1.632	1.639	1.640	1.775	1.883	1.892	1.913	1.832
.6000	1.478	1.509	1.511	1.550	1.541	1.607	1.684	1.689	1.719	1.717
.8000	1.296	1.367	1.405	1.484	1.456	1.453	1.526	1.529	1.519	1.569

Lower surface

z/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	1.085	.815	.732	.815	.964	1.222	1.529	1.605	1.507	1.101
.0250	1.091	.861	.729	.701	.721	.823	.968	.997	.960	.772
.0500	1.066	.907	.763	.686	.610	.595	.623	.608	.591	.535
.0750	1.054	.926	.797	.704	.595	.544	.538	.497	.487	.462
.1000	1.063	.932	.810	.717	.598	.533	.500	.456	.446	.436
.1500	1.076	.957	.857	.745	.634	.541	.488	.445	.423	.428
.2000	1.082	.969	.872	.783	.658	.553	.488	.448	.417	.428
.2500	1.091	.988	.894	.802	.688	.583	.518	.456	.438	.448
.3000	1.085	1.003	.919	.827	.697	.604	.532	.483	.458	.468
.3500	1.082	1.015	.935	.862	.730	.633	.564	.509	.487	.483
.4000	1.076	1.006	.931	.868	.733	.648	.578	.526	.499	.509
.4500	1.063	1.006	.938	.874	.751	.657	.594	.541	.522	.529
.5000	1.010	.988	.925	.874	.748	.672	.611	.555	.536	.540
.5500	.981	.963	.910	.862	.742	.669	.623	.567	.548	.558
.6000	.931	.917	.863	.824	.715	.645	.596	.552	.536	.540
.6500	.890	.870	.829	.786	.691	.624	.582	.544	.528	.535
.7517	.742	.759	.729	.704	.625	.580	.544	.503	.487	.503
.7934	.708	.725	.685	.660	.598	.556	.526	.488	.484	.500
.8094	.692	.694	.654	.642	.574	.536	.515	.480	.473	.483
.8137	.616	.679	.654	.648	.580	.533	.532	.491	.487	.486
.8439	.648	.670	.620	.620	.544	.515	.491	.453	.446	.445
.8772	.572	.565	.526	.500	.435	.402	.374	.349	.336	.347
.9117	.535	.546	.492	.478	.423	.402	.386	.366	.359	.367
.9448	.497	.497	.461	.440	.384	.361	.354	.323	.319	.327
.9781	.472	.480	.427	.406	.348	.328	.325	.297	.290	.289
Vane										
.0250	.717	.599	.545	.415	.255	.189	.175	.148	.133	.104
.0500	.289	.222	.190	.113	.048	.027	.018	.009	.017	.023
.1000	.182	.157	.121	.101	.081	.071	.073	.064	.084	.087
.1500	.164	.164	.125	.116	.102	.109	.111	.102	.107	.116
.2000	.192	.175	.140	.138	.126	.127	.129	.113	.119	.116
.3000	.195	.179	.143	.145	.132	.136	.137	.125	.128	.133
.4000	.195	.185	.159	.160	.147	.157	.155	.137	.142	.142
.5000	.195	.188	.165	.160	.153	.157	.170	.148	.148	.147
.6000	.201	.207	.184	.182	.174	.178	.181	.160	.168	.153
.7000	.248	.253	.227	.230	.219	.231	.243	.221	.226	.202
.8000	.381	.376	.358	.352	.336	.361	.383	.363	.354	.306
.9200	.925	.901	.888	.877	.844	.914	.962	.945	.902	.749
Flap										
.0125	.038	.034	.003	.006	.021	.033	.050	.049	.055	.029
.0250	.000	.009	.000	.006	.000	.006	.006	.000	.009	.009
.0500	.069	.089	.059	.075	.066	.059	.084	.038	.058	.072
.0750	.094	.111	.081	.088	.090	.077	.076	.049	.064	.087
.1000	.113	.139	.118	.116	.114	.101	.102	.081	.093	.095
.1500	.170	.188	.165	.164	.159	.142	.135	.119	.128	.127
.2000	.214	.231	.206	.217	.198	.186	.173	.148	.157	.168
.4000	.371	.376	.336	.343	.306	.278	.266	.253	.252	.273
.6000	.472	.494	.458	.465	.417	.388	.377	.346	.359	.370
.8000	.582	.593	.595	.579	.541	.509	.485	.480	.464	.509
.9000	.657	.691	.707	.717	.647	.627	.611	.610	.597	.634

TABLE X.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(d) Extended double slotted flap configuration; $\delta_f = 45^\circ$; $\frac{y}{b/2} = 0.43$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	.840	1.894	1.985	2.377	2.972	3.405	3.487	3.395	3.075	2.517
.0125	.978	1.766	1.960	2.334	2.923	3.387	3.490	3.471	3.202	2.849
.0250	1.048	1.642	1.994	2.423	2.963	3.396	3.599	3.623	3.208	2.522
.0500	1.064	1.458	2.025	2.441	3.063	3.521	3.587	3.518	3.234	2.569
.0750	1.076	1.324	2.237	2.536	3.044	3.422	3.449	3.447	3.277	2.546
.1000	1.102	1.290	2.502	2.791	3.041	3.363	3.391	3.453	3.309	2.528
.1500	1.121	1.265	2.139	3.343	3.883	3.815	3.810	3.804	3.364	2.558
.2000	1.153	1.274	1.483	2.668	4.106	4.595	4.303	4.012	3.324	2.531
.2500	1.165	1.274	1.215	1.828	3.168	4.095	4.185	3.912	3.225	2.511
.3000	1.181	1.290	1.240	1.478	2.401	3.476	3.927	3.769	3.165	2.493
.3500	1.197	1.285	1.255	1.313	1.845	2.771	3.452	3.436	2.988	2.461
.4000	1.222	1.305	1.302	1.303	1.584	2.363	3.094	3.219	2.910	2.446
.4500	1.254	1.311	1.332	1.319	1.494	2.095	2.819	3.015	2.795	2.411
.5000	1.264	1.311	1.338	1.319	1.401	1.851	2.470	2.731	2.644	2.370
.5500	1.286	1.343	1.369	1.356	1.413	1.759	2.311	2.573	2.540	2.323
.6000	1.311	1.355	1.375	1.365	1.410	1.675	2.167	2.415	2.453	2.285
.6993	1.299	1.367	1.388	1.386	1.416	1.548	1.848	2.158	2.220	2.200
.7507	1.280	1.358	1.388	1.365	1.398	1.509	1.748	2.026	2.107	2.141
.7704	1.348	1.414	1.431	1.411	1.447	1.548	1.786	2.026	2.101	2.141
.8099	1.381	1.467	1.474	1.475	1.510	1.583	1.780	2.003	2.072	2.109
.8512	1.454	1.517	1.526	1.513	1.553	1.607	1.746	1.947	2.006	2.053
.8911	1.549	1.610	1.637	1.622	1.659	1.693	1.816	1.980	1.983	2.012
.9322	1.755	1.779	1.791	1.779	1.811	1.845	1.948	2.079	2.017	2.021
.9724	2.121	2.156	2.169	2.190	2.221	2.220	2.291	2.383	2.199	2.074
Vane										
.0000	2.270	2.085	2.200	2.122	2.034	1.890	1.842	1.839	1.595	1.437
.0250	3.694	3.595	2.640	3.665	3.761	3.768	3.810	3.792	3.248	2.827
.0500	3.978	3.859	3.917	3.966	4.069	4.059	4.121	4.084	3.474	2.918
.1000	4.549	4.377	4.455	4.533	4.681	4.613	4.687	4.538	3.745	2.945
.1500	5.023	4.803	4.902	4.999	5.175	5.050	5.133	4.906	3.977	2.942
.2000	5.318	5.280	5.403	5.505	5.718	5.556	5.643	5.354	4.341	3.065
.3000	5.439	5.190	5.311	5.426	5.640	5.488	5.558	5.216	4.205	2.892
.4000	5.153	4.915	5.037	5.140	5.355	5.220	5.300	4.974	4.029	2.769
.5000	4.829	4.607	4.714	4.818	5.035	4.946	5.030	4.716	3.847	2.651
.6000	4.299	4.118	4.203	4.300	4.504	4.482	4.558	4.275	3.543	2.525
.7000	3.880	3.716	3.775	3.864	4.044	4.030	4.094	3.898	3.283	2.449
.8000	3.372	3.236	3.283	3.355	3.494	3.533	3.590	3.412	2.974	2.305
.9000	2.864	2.744	2.785	2.846	2.951	2.991	3.036	2.942	2.595	2.115
Flap										
.0000	1.638	1.576	1.671	1.693	1.767	1.780	1.843	1.816	1.607	1.282
.0125	2.451	2.423	2.483	2.506	2.606	2.595	2.640	2.597	2.329	1.948
.0250	3.165	3.146	3.200	3.242	3.382	3.438	3.440	3.389	3.055	2.584
.0500	3.381	3.361	3.394	3.447	3.615	3.646	3.704	3.667	3.324	2.786
.0750	3.356	3.377	3.406	3.453	3.622	3.664	3.722	3.702	3.373	2.822
.1000	3.315	3.218	3.255	3.312	3.479	3.515	3.587	3.567	3.248	2.692
.1500	2.683	2.634	2.665	2.720	2.836	2.875	2.948	2.968	2.763	2.338
.2000	2.261	2.159	2.188	2.242	2.326	2.360	2.420	2.418	2.277	2.006
.4000	1.718	1.698	1.695	1.718	1.783	1.821	1.862	1.944	1.928	1.877
.6000	1.464	1.467	1.462	1.484	1.533	1.556	1.593	1.678	1.705	1.766
.8000	1.165	1.249	1.258	1.264	1.329	1.309	1.352	1.389	1.433	1.384

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	1.191	.822	.815	.926	1.146	1.402	1.628	1.681	1.633	1.440
.0250	1.162	.857	.778	.770	.848	.943	1.071	1.091	1.081	.994
.0500	1.137	.900	.794	.739	.717	.702	.757	.763	.763	.736
.0750	1.127	.925	.825	.742	.683	.651	.645	.646	.624	.625
.1000	1.137	.950	.837	.754	.680	.598	.598	.582	.566	.575
.1500	1.139	.963	.865	.773	.702	.589	.551	.526	.517	.522
.2000	1.121	.972	.874	.785	.696	.583	.543	.515	.497	.510
.2500	1.111	.981	.886	.807	.717	.595	.546	.518	.497	.510
.3000	1.095	.969	.886	.807	.708	.598	.548	.518	.509	.519
.3500	1.083	.975	.902	.819	.733	.607	.566	.532	.512	.528
.4000	1.073	.969	.895	.810	.733	.616	.566	.535	.523	.534
.4500	1.038	.950	.883	.810	.730	.613	.575	.541	.529	.540
.5000	1.010	.935	.858	.797	.721	.604	.563	.532	.520	.534
.5500	.968	.894	.840	.773	.714	.595	.569	.535	.526	.534
.6000	.909	.844	.797	.736	.677	.580	.548	.515	.503	.516
.6995	.822	.788	.754	.712	.665	.565	.534	.506	.500	.504
.7499	.787	.741	.705	.656	.637	.533	.513	.488	.474	.490
.7700	.765	.716	.698	.656	.621	.530	.513	.488	.488	.487
.7744	.746	.716	.689	.647	.618	.524	.513	.497	.480	.493
.8110	.714	.679	.646	.616	.568	.488	.460	.447	.439	.440
.8519	.613	.551	.538	.497	.460	.375	.372	.357	.344	.349
.8919	.565	.517	.508	.476	.447	.378	.370	.357	.353	.361
.9332	.521	.483	.468	.441	.416	.339	.343	.330	.312	.320
.9738	.467	.430	.437	.411	.385	.312	.320	.313	.295	.299
Vane										
.0250	.276	.196	.280	.215	.130	.062	.076	.067	.069	.073
.0500	.117	.078	.117	.086	.040	.000	.056	.038	.033	.044
.1000	.102	.079	.098	.080	.056	.039	.044	.053	.069	.067
.1500	.127	.115	.123	.110	.081	.057	.085	.091	.104	.100
.2000	.133	.140	.138	.138	.109	.083	.109	.111	.113	.106
.3000	.165	.185	.163	.144	.137	.113	.153	.132	.139	.138
.4000	.178	.184	.175	.166	.158	.119	.141	.132	.139	.150
.5000	.194	.190	.188	.178	.174	.143	.179	.161	.156	.155
.6000	.213	.209	.200	.215	.193	.161	.191	.199	.168	.164
.7000	.251	.252	.252	.258	.248	.208	.243	.237	.208	.197
.8000	.391	.358	.375	.393	.379	.357	.384	.357	.341	.299
.9200	.994	.956	.966	.975	1.000	.982	1.021	.991	.873	.789
Flap										
.0125	.016	.034	.058	.058	.056	.045	.109	.096	.081	.047
.0250	.016	.000	.018	.012	.003	.000	.004	.003	.029	.012
.0500	.084	.078	.071	.067	.056	.009	.044	.023	.046	.067
.0750	.111	.109	.095	.092	.087	.036	.067	.050	.069	.076
.1000	.137	.159	.132	.129	.118	.065	.094	.079	.081	.106
.1500	.200	.190	.175	.166	.155	.089	.117	.108	.121	.135
.2000	.254	.240	.218	.202	.196	.125	.155	.143	.147	.164
.4000	.397	.371	.348	.319	.295	.238	.241	.234	.249	.270
.6000	.492	.473	.465	.439	.404	.339	.340	.325	.341	.381
.8000	.622	.595	.583	.552	.534	.464	.460	.453	.477	.446
.9003	.714	.701	.668	.662	.646	.571	.584	.576	.592	.580

TABLE X.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(e) Extended double slotted flap configuration; $\delta_f = 45^\circ$; $\frac{y}{b/2} = 0.55$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	.799	1.842	1.891	2.242	2.577	2.796	2.848	2.762	2.522	2.228
.0125	1.069	1.778	1.934	2.299	2.619	2.826	2.886	2.837	2.586	2.294
.0250	1.028	1.697	1.900	2.249	2.553	2.784	2.866	2.866	2.598	2.251
.0500	1.054	1.416	1.878	2.261	2.562	2.808	2.962	2.916	2.609	2.269
.0750	1.063	1.305	1.950	2.365	2.655	2.894	2.927	2.878	2.612	2.234
.1000	1.094	1.278	1.981	2.434	2.733	2.935	2.880	2.866	2.626	2.234
.1500	1.129	1.281	2.233	2.488	2.685	2.814	2.839	2.895	2.635	2.214
.2000	1.126	1.281	2.327	2.730	2.649	2.764	2.862	2.916	2.635	2.214
.2500	1.161	1.284	2.215	3.239	2.820	2.805	3.015	3.015	2.655	2.217
.3000	1.173	1.284	1.919	3.246	3.435	3.228	3.234	3.099	2.658	2.202
.3500	1.205	1.302	1.598	2.846	3.646	3.515	3.427	3.145	2.638	2.205
.4000	1.223	1.315	1.361	2.302	3.384	3.536	3.342	3.099	2.603	2.196
.4500	1.230	1.318	1.277	1.837	2.949	3.409	3.313	3.073	2.583	2.202
.5000	1.264	1.349	1.290	1.591	2.565	3.199	3.225	3.015	2.545	2.188
.6194	1.173	1.302	1.321	1.311	1.712	2.379	2.792	2.770	2.473	2.228
.6844	1.173	1.296	1.352	1.289	1.523	2.083	2.526	2.558	2.339	2.168
.7066	1.214	1.327	1.367	1.321	1.508	2.054	2.482	2.512	2.302	2.150
.7594	1.261	1.358	1.442	1.333	1.459	1.906	2.301	2.334	2.168	2.043
.8101	1.302	1.423	1.551	1.403	1.468	1.855	2.263	2.326	2.197	2.089
.8622	1.400	1.515	1.691	1.500	1.517	1.832	2.187	2.250	2.137	2.086
.9142	1.528	1.660	2.062	1.670	1.634	1.888	2.199	2.250	2.142	2.055
.9652	1.837	1.994	1.592	2.069	1.976	2.157	2.371	2.381	2.206	2.087
Vane										
.0000	1.670	1.512	3.442	1.598	1.444	1.444	1.430	1.381	1.212	1.133
.0250	3.243	3.339	3.722	3.463	3.291	3.303	3.579	3.410	2.937	2.693
.0500	3.620	3.601	4.423	3.765	3.601	3.602	3.857	3.648	3.067	2.751
.1000	4.255	4.219	4.831	4.519	4.315	4.462	4.398	4.093	3.253	2.760
.1500	4.642	4.549	5.046	4.953	4.745	4.838	4.687	4.288	3.279	2.679
.2000	4.843	4.725	5.152	5.199	4.988	5.048	4.880	4.399	3.270	2.605
.3000	4.922	4.808	4.962	5.350	5.168	5.214	4.921	4.424	3.256	2.549
.4000	4.733	4.629	4.691	5.180	5.042	5.092	4.769	4.282	3.102	2.430
.5000	4.475	4.385	4.236	4.950	4.868	4.918	4.594	4.105	3.006	2.376
.6000	4.029	3.950	3.735	4.482	4.462	4.521	4.219	3.770	2.815	2.298
.7000	3.535	3.503	3.271	3.969	3.955	4.033	3.804	3.433	2.638	2.211
.8000	3.098	3.080	2.800	3.472	3.459	3.539	3.371	3.081	2.453	2.118
.9000	2.676	2.639	1.212	2.956	2.952	3.027	2.915	2.686	2.226	1.997
Flap										
.0000	1.085	1.123	2.486	1.315	1.336	1.406	1.412	1.323	1.105	.971
.0125	2.217	2.339	3.056	2.607	2.583	2.663	2.608	2.454	2.096	1.907
.0250	2.720	2.904	3.308	3.230	3.201	3.299	3.254	3.052	2.606	2.378
.0500	2.919	3.132	3.392	3.504	3.501	3.619	3.570	3.355	2.844	2.563
.0750	2.941	3.191	3.280	3.607	3.607	3.740	3.693	3.459	2.925	2.598
.1000	2.925	3.095	2.881	3.494	3.689	3.601	3.544	3.337	2.809	2.491
.1500	2.601	2.728	2.271	3.070	3.084	3.148	3.026	2.826	2.418	2.168
.2000	2.079	2.157	1.847	2.409	2.399	2.477	2.450	2.326	2.081	1.928
.4000	1.714	1.765	1.673	1.909	1.874	1.914	1.912	1.884	1.838	1.803
.6000	1.563	1.623	1.427	1.701	1.628	1.630	1.646	1.663	1.748	1.774
.8000	1.289	1.379	.863	1.425	1.342	1.323	1.339	1.387	1.551	1.642

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	1.170	.824	.797	1.082	1.384	1.663	1.880	1.884	1.812	1.647
.0250	1.181	.861	.813	.852	.979	1.119	1.243	1.267	1.238	1.162
.0500	1.116	.895	.807	.777	.781	.808	.871	.881	.867	.838
.0750	1.104	.892	.813	.745	.697	.692	.719	.715	.713	.699
.1000	1.091	.910	.832	.745	.673	.657	.664	.657	.655	.639
.1500	1.079	.923	.844	.749	.664	.610	.599	.581	.571	.569
.2000	1.063	.929	.847	.758	.658	.610	.573	.552	.542	.549
.2500	1.050	.935	.850	.758	.679	.615	.570	.547	.536	.535
.3000	1.035	.920	.854	.764	.667	.607	.570	.544	.531	.529
.3500	1.013	.920	.844	.780	.688	.607	.570	.544	.533	.532
.4000	.991	.901	.838	.774	.688	.610	.570	.541	.533	.532
.4500	.972	.892	.822	.764	.682	.607	.570	.538	.536	.526
.5000	.953	.870	.773	.758	.673	.604	.564	.538	.525	.532
.6195	.811	.799	.726	.720	.637	.592	.533	.520	.507	.503
.6834	.771	.756	.701	.670	.628	.589	.544	.517	.516	.514
.7080	.758	.744	.707	.670	.607	.556	.515	.494	.475	.480
.7137	.736	.738	.673	.670	.601	.562	.520	.500	.487	.491
.7603	.708	.697	.570	.638	.571	.539	.497	.471	.470	.468
.8119	.585	.574	.545	.538	.486	.462	.424	.401	.403	.405
.8643	.563	.552	.495	.519	.471	.444	.415	.395	.394	.393
.9162	.509	.488	.433	.469	.432	.402	.383	.366	.357	.355
.9669	.443	.438	.283	.421	.390	.373	.351	.340	.328	.324
Vane										
.0250	.324	.238	.109	.261	.213	.166	.164	.169	.139	.136
.0500	.142	.099	.112	.097	.090	.056	.064	.058	.075	.072
.1000	.123	.114	.134	.091	.105	.095	.096	.099	.101	.110
.1500	.154	.139	.146	.129	.135	.130	.132	.137	.133	.133
.2000	.167	.164	.168	.148	.153	.157	.155	.157	.151	.156
.3000	.182	.173	.187	.164	.174	.172	.167	.163	.157	.156
.4000	.195	.201	.199	.186	.192	.178	.175	.180	.180	.171
.5000	.204	.194	.215	.186	.195	.186	.181	.174	.183	.173
.6000	.220	.207	.255	.211	.222	.213	.211	.201	.194	.191
.7000	.267	.256	.389	.252	.273	.260	.260	.247	.232	.223
.8000	.377	.389	1.012	.390	.408	.397	.401	.372	.336	.321
.9200	.978	.966	.034	1.063	1.060	1.071	1.029	.953	.826	.769
Flap										
.0125	.028	.019	.031	.047	.093	.107	.126	.116	.101	.084
.0250	.044	.019	.097	.030	.048	.041	.056	.052	.061	.069
.0500	.104	.096	.125	.073	.094	.089	.085	.099	.104	.104
.0750	.126	.127	.159	.126	.126	.118	.123	.110	.125	.121
.1000	.176	.173	.224	.154	.156	.136	.137	.125	.136	.130
.1500	.236	.235	.255	.204	.198	.178	.175	.166	.177	.185
.2000	.274	.268	.421	.233	.228	.216	.202	.206	.206	.214
.4000	.436	.441	.545	.393	.346	.325	.313	.308	.322	.335
.6000	.582	.571	.667	.506	.468	.429	.409	.410	.432	.445
.8000	.676	.679	.751	.638	.583	.556	.532	.535	.580	.624
.9000	.736	.734	1.249	.726	.673	.636	.629	.634	.702	.740

[redacted]

TABLE X.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Concluded

(f) Extended double slotted flap configuration; $\delta_f = 45^\circ$; $\frac{y}{b/2} = 0.72$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 28^\circ$
Wing										
.0000	.916	1.801	1.997	2.132	2.353	2.458	2.308	2.282	2.150	2.052
.0125	1.165	1.779	1.960	2.069	2.295	2.399	2.258	2.235	2.159	2.084
.0250	1.184	1.776	1.963	2.078	2.301	2.408	2.290	2.294	2.179	2.081
.0500	1.181	1.801	1.988	2.093	2.295	2.408	2.354	2.350	2.197	2.087
.0750	1.174	1.831	2.012	2.126	2.310	2.422	2.406	2.394	2.211	2.087
.1000	1.168	1.776	2.006	2.105	2.274	2.408	2.395	2.370	2.220	2.099
.1500	1.193	1.752	2.052	2.144	2.298	2.470	2.389	2.368	2.226	2.093
.2000	1.218	1.656	2.105	2.188	2.353	2.500	2.374	2.376	2.223	2.090
.2500	1.230	1.438	2.145	2.228	2.365	2.455	2.363	2.382	2.226	2.090
.3000	1.249	1.293	2.206	2.234	2.338	2.399	2.354	2.385	2.214	2.084
.3500	1.262	1.272	2.354	2.261	2.301	2.369	2.368	2.400	2.209	2.081
.4000	1.274	1.296	2.415	2.306	2.274	2.339	2.360	2.403	2.209	2.079
.4500	1.290	1.311	2.434	2.417	2.271	2.330	2.366	2.409	2.209	2.087
.5000	1.318	1.347	2.468	2.649	2.301	2.366	2.403	2.453	2.238	2.102
.5500	1.305	1.320	2.379	2.727	2.329	2.366	2.400	2.438	2.214	2.093
.6000	1.321	1.338	2.280	2.775	2.417	2.399	2.419	2.441	2.226	2.102
.6500	1.339	1.362	2.145	2.724	2.517	2.440	2.421	2.429	2.209	2.102
.7000	1.358	1.365	2.031	2.670	2.581	2.506	2.444	2.423	2.179	2.093
.7500	1.399	1.417	1.902	2.547	2.650	2.571	2.479	2.420	2.182	2.087
.8000	1.392	1.408	1.794	2.405	2.624	2.601	2.453	2.403	2.173	2.084
.8500	1.417	1.423	1.665	2.237	2.544	2.553	2.400	2.341	2.129	2.070
.9000	1.445	1.459	1.591	2.069	2.456	2.518	2.366	2.303	2.121	2.070

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 28^\circ$
Wing										
.0125	1.006	.801	.886	.994	1.195	1.399	1.452	1.503	1.499	1.487
.0250	1.015	.804	.803	.829	.961	1.107	1.151	1.206	1.200	1.200
.0500	1.015	.828	.772	.733	.787	.881	.881	.918	.921	.933
.0750	1.000	.846	.778	.724	.733	.771	.774	.800	.818	.809
.1000	.994	.835	.788	.712	.702	.738	.719	.729	.745	.748
.1500	.984	.864	.791	.709	.684	.687	.655	.668	.663	.661
.2000	.978	.873	.809	.793	.690	.690	.641	.641	.645	.652
.2500	.984	.864	.845	.736	.705	.687	.641	.647	.645	.649
.3000	.987	.882	.840	.766	.736	.711	.664	.665	.666	.667
.3500	.984	.894	.868	.796	.772	.744	.675	.671	.675	.673
.4000	.978	.900	.874	.793	.760	.726	.646	.644	.631	.638
.4500	.975	.894	.883	.799	.766	.714	.641	.629	.616	.615
.5000	.975	.906	.889	.811	.786	.720	.644	.635	.619	.620
.5500	.972	.912	.908	.832	.824	.738	.661	.653	.628	.632
.6000	.975	.918	.926	.853	.854	.762	.684	.676	.648	.661
.6500	.975	.933	.945	.883	.885	.792	.719	.709	.695	.704
.7000	.994	.949	.975	.901	.955	.842	.760	.765	.734	.774
.7500	1.022	.982	1.022	.961	1.046	.917	.847	.841	.832	.852
.8000	1.075	1.039	1.086	1.036	1.198	1.021	.957	.968	.974	.997
.8500	1.168	1.130	1.197	1.162	1.404	1.178	1.119	1.138	1.150	1.174
.9000	1.274	1.275	1.372	1.357	1.666	1.411	1.342	1.359	1.364	1.397
.9500	1.408	1.426	1.535	1.539	1.660	1.667	1.589	1.588	1.584	1.612

[redacted]

TABLE XI.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE

(a) Extended double slotted flap configuration; $\delta_f = 60^\circ$; $\frac{y}{b/2} = 0$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 28^\circ$
Fuselage										
.0000	.003	.000	.000	.000	.006	.015	.039	.082	.116	.182
.0500	.771	.832	.882	.929	.961	1.009	1.015	1.038	1.052	1.055
.1000	.872	.921	.967	1.004	1.021	1.069	1.065	1.085	1.110	1.121
.1500	.963	.988	1.024	1.062	1.063	1.111	1.100	1.115	1.131	1.118
.2000	1.003	1.037	1.054	1.077	1.093	1.126	1.100	1.115	1.116	1.109
.2500	1.040	1.067	1.085	1.093	1.105	1.138	1.117	1.115	1.128	1.127
.3000	1.055	1.079	1.097	1.108	1.099	1.123	1.106	1.115	1.122	1.132
.3500	1.083	1.101	1.097	1.108	1.111	1.155	1.141	1.162	1.172	1.167
.4000	1.024	1.046	1.048	1.065	1.075	1.114	1.088	1.121	1.137	1.152
.4500	1.027	1.037	1.067	1.098	1.117	1.159	1.176	1.229	1.233	1.276
.5000	1.027	1.061	1.097	1.154	1.195	1.273	1.317	1.373	1.395	1.572
.5500	1.031	1.098	1.136	1.215	1.267	1.348	1.399	1.435	1.494	1.926
.6000	1.061	1.122	1.182	1.255	1.300	1.372	1.381	1.409	1.485	1.977
.6500	1.086	1.134	1.209	1.274	1.303	1.333	1.335	1.356	1.401	1.779
.7000	1.128	1.171	1.233	1.286	1.309	1.360	1.346	1.359	1.372	1.673
.7500	1.159	1.210	1.254	1.289	1.315	1.357	1.381	1.394	1.416	1.601
.8000	1.214	1.250	1.273	1.308	1.330	1.384	1.393	1.438	1.477	1.552
.8500	1.248	1.290	1.315	1.348	1.339	1.393	1.425	1.468	1.509	1.509
.9000	1.260	1.278	1.303	1.308	1.306	1.354	1.358	1.406	1.445	1.437
.9500	1.223	1.241	1.236	1.243	1.227	1.285	1.293	1.309	1.373	1.377
.9940	1.220	1.220	1.203	1.209	1.198	1.222	1.203	1.250	1.337	1.405

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 28^\circ$
Fuselage										
.0500	.902	.820	.754	.686	.616	.565	.484	.465	.398	.379
.1000	.951	.903	.848	.782	.721	.673	.604	.532	.512	.471
.1500	.994	.982	.921	.862	.793	.742	.651	.609	.578	.549
.2000	1.027	1.000	.967	.911	.844	.795	.707	.673	.645	.604
.2500	1.043	1.037	.997	.948	.901	.847	.766	.721	.695	.664
.3000	1.006	1.006	.976	.948	.934	.928	.874	.868	.860	.842
.3500	1.031	1.049	1.015	.978	.925	.892	.821	.776	.759	.719
.4000	.979	.985	.945	.908	.871	.844	.760	.726	.698	.667
.4500	.957	.933	.906	.874	.823	.781	.704	.662	.634	.592
.5000	.939	.909	.876	.822	.754	.709	.619	.588	.552	.517
.5500	.908	.878	.827	.769	.700	.649	.548	.518	.488	.460
.6000	.862	.811	.764	.702	.640	.586	.490	.456	.430	.417
.7500	1.024	.997	.951	.902	.850	.811	.736	.697	.686	.673
.8500	.933	.954	.961	.945	.922	.916	.915	.923	.936	.943
.9940	1.049	1.122	1.179	1.194	1.189	1.210	1.200	1.215	1.256	1.296

TABLE XL.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(b) Extended double slotted flap configuration; $\delta_f = 50^\circ$; $\frac{\bar{y}}{b/2} = 0.21$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	.924	.904	1.975	3.406	5.437	8.090	11.283	12.877	6.901	2.760
.0125	.906	1.240	1.705	3.418	7.759	11.010	14.040	14.997	6.532	2.525
.0250	.951	1.201	1.553	2.679	3.645	6.486	9.694	11.427	6.133	2.569
.0500	.985	1.168	1.432	1.630	2.036	2.790	3.889	4.880	5.142	2.458
.0750	.997	1.147	1.382	1.533	1.837	2.246	2.824	3.395	4.312	2.450
.1000	1.018	1.159	1.354	1.518	1.741	2.086	2.525	2.982	3.869	2.429
.1500	1.039	1.162	1.329	1.470	1.636	1.870	2.188	2.462	3.188	2.406
.2000	1.064	1.162	1.311	1.442	1.581	1.758	2.024	2.243	2.722	2.389
.2500	1.085	1.177	1.305	1.424	1.533	1.675	1.980	2.111	2.402	2.348
.3000	1.106	1.186	1.305	1.412	1.509	1.616	1.851	1.942	2.211	2.325
.3500	1.115	1.198	1.298	1.397	1.476	1.589	1.716	1.839	2.053	2.290
.4000	1.133	1.213	1.305	1.382	1.455	1.559	1.648	1.763	1.965	2.270
.4500	1.160	1.225	1.308	1.388	1.449	1.527	1.613	1.719	1.874	2.250
.5000	1.179	1.246	1.329	1.397	1.449	1.497	1.610	1.687	1.824	2.221
.5500	1.200	1.264	1.342	1.385	1.434	1.465	1.563	1.655	1.774	2.177
.6000	1.215	1.276	1.342	1.388	1.431	1.462	1.581	1.661	1.783	2.128
.6500	1.242	1.306	1.354	1.391	1.440	1.456	1.613	1.696	1.777	2.079
.7000	1.282	1.324	1.373	1.394	1.446	1.471	1.616	1.652	1.728	2.021
.7824	1.321	1.339	1.387	1.394	1.440	1.503	1.478	1.497	1.607	1.919
.8176	1.364	1.375	1.419	1.427	1.488	1.518	1.493	1.506	1.607	1.876
.8323	1.406	1.432	1.478	1.467	1.506	1.565	1.528	1.526	1.628	1.873
.8913	1.518	1.505	1.534	1.533	1.551	1.589	1.516	1.520	1.616	1.809
.9281	1.645	1.619	1.646	1.639	1.629	1.654	1.593	1.614	1.698	1.806
.9504	1.709	1.688	1.696	1.685	1.675	1.684	1.634	1.661	1.745	1.809
.9804	2.060	2.015	1.988	1.939	1.922	1.935	1.950	2.018	2.094	1.971
Vane										
.0000	1.288	1.136	.966	.815	.711	.678	.657	.655	.683	.623
.0250	2.491	2.417	2.398	2.315	2.298	2.379	2.443	2.523	2.628	2.267
.0500	3.412	3.261	3.187	2.966	2.988	3.131	3.282	3.433	3.569	2.821
.1000	3.863	3.629	3.507	3.194	3.261	3.391	3.652	3.846	3.965	2.940
.1500	4.000	3.736	3.591	3.236	3.301	3.442	3.751	3.971	4.109	2.998
.2000	4.121	3.817	3.650	3.285	3.340	3.495	3.813	4.056	4.229	3.096
.3000	3.948	3.601	3.417	3.063	3.114	3.255	3.622	3.854	4.136	3.169
.4000	3.563	3.216	3.019	2.754	2.771	2.915	3.229	3.439	3.866	3.073
.5000	3.206	2.874	2.702	2.536	2.515	2.645	2.910	3.070	3.628	3.009
.6000	2.706	2.429	2.295	2.200	2.190	2.329	2.496	2.605	3.156	2.795
.7000	2.306	2.297	2.168	2.066	2.084	2.225	2.349	2.453	2.889	2.685
.8000	2.448	2.255	2.137	2.045	2.054	2.201	2.317	2.418	2.748	2.551
.9000	2.372	2.174	2.078	1.982	2.024	2.175	2.308	2.392	2.605	2.360
Flap										
.0000	1.963	1.670	1.469	1.433	1.229	1.293	1.379	1.427	1.481	1.261
.0125	2.479	2.321	2.243	2.060	2.081	2.204	2.311	2.374	2.499	2.192
.0250	2.733	2.577	2.525	2.406	2.452	2.613	2.748	2.830	2.995	2.667
.0500	2.812	2.631	2.606	2.503	2.578	2.725	2.880	2.977	3.256	3.029
.0750	2.585	2.411	2.389	2.303	2.367	2.474	2.643	2.725	3.089	2.942
.1000	2.336	2.159	2.159	2.097	2.136	2.201	2.385	2.462	2.807	2.771
.1500	1.918	1.787	1.789	1.742	1.747	1.808	1.924	2.000	2.329	2.374
.2000	1.818	1.718	1.702	1.639	1.657	1.734	1.827	1.892	2.188	2.203
.4000	1.851	1.754	1.727	1.691	1.705	1.764	1.824	1.904	2.211	2.116
.6000	1.812	1.727	1.708	1.676	1.702	1.764	1.827	1.874	2.097	1.971
.8000	1.697	1.667	1.668	1.618	1.639	1.687	1.777	1.828	1.912	1.768

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	1.406	.808	.649	.573	.584	.637	.664	.695	.689	.583
.0250	1.045	.844	.677	.539	.464	.402	.387	.380	.317	.209
.0500	1.030	.868	.708	.576	.452	.331	.264	.225	.226	.215
.0750	1.015	.877	.739	.606	.494	.361	.299	.260	.264	.273
.1000	1.024	.898	.764	.636	.521	.402	.355	.333	.326	.336
.1500	1.012	.892	.780	.676	.572	.479	.419	.415	.393	.388
.2000	1.012	.910	.804	.709	.626	.536	.478	.459	.437	.441
.2500	1.036	.940	.848	.734	.669	.589	.519	.491	.481	.478
.3000	1.033	.946	.851	.776	.699	.613	.554	.523	.510	.510
.3500	1.057	.979	.888	.808	.741	.654	.587	.573	.543	.528
.4000	1.067	.988	.907	.842	.768	.684	.619	.579	.563	.551
.4500	1.057	.997	.926	.854	.789	.707	.636	.611	.590	.571
.5000	1.033	.979	.913	.848	.777	.704	.639	.608	.590	.580
.5500	1.003	.994	.891	.830	.763	.692	.610	.608	.581	.577
.6000	.961	.928	.873	.812	.756	.684	.619	.591	.578	.577
.6500	.900	.868	.829	.782	.729	.654	.601	.564	.554	.560
.7000	.836	.826	.789	.751	.702	.624	.572	.547	.537	.542
.7815	.754	.721	.680	.645	.602	.547	.502	.485	.475	.481
.8179	.715	.694	.652	.630	.596	.539	.493	.471	.478	.484
.8325	.703	.679	.640	.627	.599	.553	.513	.506	.493	.502
.8366	.706	.694	.658	.645	.626	.577	.528	.523	.519	.528
.8921	.636	.532	.475	.451	.422	.379	.355	.348	.337	.333
.9286	.597	.508	.457	.433	.410	.385	.358	.351	.340	.351
.9518	.570	.474	.413	.385	.358	.351	.317	.310	.302	.304
.9813	.539	.447	.379	.345	.319	.287	.267	.260	.261	.253
Vane										
.0250	.773	.565	.363	.242	.154	.130	.123	.114	.103	.101
.0500	.597	.402	.239	.158	.127	.101	.109	.099	.085	.101
.1000	.512	.339	.230	.179	.157	.142	.153	.146	.138	.139
.1500	.470	.327	.230	.191	.172	.169	.173	.158	.161	.162
.2000	.442	.324	.233	.194	.181	.166	.179	.167	.161	.168
.3000	.409	.315	.230	.200	.181	.186	.182	.181	.176	.171
.4000	.388	.321	.236	.206	.202	.195	.199	.190	.185	.186
.5000	.376	.321	.242	.209	.202	.204	.202	.193	.191	.200
.6000	.382	.330	.255	.218	.211	.219	.220	.219	.214	.209
.7000	.412	.366	.286	.264	.247	.257	.270	.278	.267	.255
.8000	.488	.453	.391	.351	.355	.364	.372	.392	.396	.362
.9200	.948	.895	.829	.794	.807	.861	.895	.924	.962	.855
Flap										
.0125	.576	.261	.093	.055	.054	.074	.079	.088	.073	.075
.0250	.297	.135	.053	.045	.051	.065	.070	.061	.053	.075
.0500	.236	.156	.109	.121	.127	.130	.120	.114	.114	.133
.0750	.221	.177	.140	.158	.163	.157	.150	.146	.144	.159
.1000	.230	.207	.183	.194	.187	.183	.185	.184	.179	.191
.1500	.248	.246	.230	.234	.229	.219	.223	.216	.223	.220
.4000	.476	.480	.469	.464	.443	.420	.414	.415	.402	.412
.6000	.630	.625	.609	.588	.557	.533	.513	.503	.502	.487
.8000	.721	.712	.702	.674	.645	.615	.587	.585	.554	.545
.9000	.785	.784	.777	.774	.741	.710	.692	.675	.634	.615

TABLE XL - PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(c) Extended double slotted flap configuration; $\delta_f = 50^\circ$; $\frac{y}{b/2} = 0.30$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	.935	1.174	1.991	2.872	3.569	4.607	5.173	5.147	3.632	2.546
.0125	.944	1.393	2.154	2.884	3.494	4.265	4.754	4.850	3.602	2.519
.0250	.991	1.282	2.235	2.975	3.638	4.476	4.880	4.835	3.617	2.558
.0500	1.034	1.234	2.723	3.881	3.925	4.470	4.711	4.806	3.566	2.458
.0750	1.034	1.215	1.708	3.878	5.289	6.009	5.980	5.485	3.599	2.478
.1000	1.040	1.208	1.228	2.445	4.129	5.464	5.959	5.503	3.602	2.528
.1500	1.065	1.209	1.240	1.386	2.131	3.303	4.509	4.709	3.416	2.399
.2000	1.087	1.215	1.283	1.346	1.599	2.226	3.292	3.773	3.217	2.379
.2500	1.096	1.228	1.292	1.358	1.470	1.836	2.617	3.150	3.048	2.399
.3000	1.128	1.234	1.308	1.371	1.428	1.693	2.289	2.741	2.885	2.346
.3500	1.134	1.238	1.302	1.361	1.404	1.577	2.023	2.370	2.672	2.341
.4000	1.159	1.260	1.320	1.388	1.416	1.559	1.906	2.235	2.548	2.314
.4500	1.184	1.272	1.338	1.392	1.416	1.556	1.851	2.126	2.449	2.282
.5000	1.209	1.304	1.354	1.399	1.410	1.530	1.792	1.997	2.295	2.238
.5500	1.218	1.294	1.338	1.396	1.407	1.527	1.737	1.932	2.205	2.206
.6000	1.243	1.329	1.386	1.402	1.407	1.524	1.708	1.873	2.139	2.156
.6500	1.277	1.348	1.388	1.427	1.431	1.524	1.673	1.829	2.066	2.121
.7516	1.249	1.358	1.364	1.414	1.407	1.482	1.579	1.732	1.886	1.962
.7940	1.290	1.402	1.397	1.442	1.449	1.500	1.599	1.741	1.840	1.901
.8099	1.277	1.408	1.409	1.458	1.440	1.476	1.567	1.703	1.795	1.868
.8429	1.367	1.481	1.468	1.520	1.506	1.550	1.655	1.779	1.813	1.865
.8767	1.386	1.516	1.508	1.545	1.527	1.559	1.655	1.744	1.747	1.789
.9108	1.489	1.617	1.594	1.642	1.617	1.646	1.737	1.815	1.765	1.783
.9437	1.685	1.798	1.775	1.797	1.756	1.803	1.901	1.938	1.822	1.795
.9768	2.087	2.209	2.157	2.171	2.102	2.155	2.275	2.291	2.069	1.924
Vane										
.0000	3.015	2.867	2.686	2.486	2.211	2.092	2.164	2.112	1.807	1.604
.0250	3.884	3.760	3.622	3.694	3.334	3.446	3.719	3.706	3.229	3.006
.0500	4.725	4.643	4.468	4.417	4.129	4.346	4.632	4.600	3.792	3.144
.1000	5.146	5.064	4.846	4.797	4.416	4.759	4.988	4.997	4.027	3.212
.1500	5.380	5.279	5.056	4.978	4.548	4.931	5.152	5.197	4.166	3.264
.2000	5.563	5.517	5.297	5.174	4.678	5.107	5.292	5.417	4.367	3.393
.2000	5.420	5.431	5.196	5.053	4.527	4.952	5.120	5.320	4.388	3.411
.4000	5.068	5.032	4.748	4.660	4.190	4.631	4.787	5.014	4.138	3.229
.5000	4.738	4.621	4.345	4.224	3.851	4.262	4.368	4.644	3.867	3.039
.6000	4.218	4.108	3.837	3.716	3.392	3.765	3.901	4.164	3.521	2.863
.7000	3.685	3.545	3.326	3.202	2.976	3.280	3.453	3.682	3.175	2.643
.8000	3.152	3.016	2.823	2.718	2.572	2.815	3.047	3.197	2.786	2.387
.9000	2.704	2.592	2.434	2.349	2.259	2.437	2.690	2.791	2.488	2.223
Flap										
.0000	1.358	1.374	1.314	1.202	1.075	1.146	1.237	1.297	1.163	1.039
.0125	2.286	2.374	2.262	2.162	1.994	2.113	2.208	2.323	2.187	1.987
.0250	2.863	2.966	2.806	2.723	2.539	2.690	2.848	2.991	2.831	2.596
.0500	2.975	3.083	2.905	2.810	2.666	2.833	3.064	3.250	3.102	2.848
.0750	3.003	3.099	2.905	2.804	2.675	2.869	3.129	3.323	3.193	3.000
.1000	2.881	2.912	2.714	2.620	2.518	2.714	2.956	3.147	3.015	2.763
.1500	2.349	2.347	2.188	2.124	2.146	2.294	2.462	2.630	2.632	2.452
.2000	2.056	1.988	1.877	1.825	1.828	1.928	2.102	2.215	2.187	2.056
.4000	1.648	1.674	1.619	1.648	1.654	1.687	1.766	1.859	1.958	1.889
.6000	1.523	1.595	1.572	1.635	1.629	1.640	1.705	1.785	1.849	1.804
.8000	1.436	1.535	1.524	1.601	1.572	1.577	1.632	1.685	1.708	1.660

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	1.047	.813	.717	.601	.961	1.229	1.500	1.566	1.389	1.109
.0250	1.050	.867	.723	.688	.720	.821	.953	.985	.886	.771
.0500	1.034	.908	.748	.657	.596	.592	.605	.603	.566	.537
.0750	1.040	.918	.775	.673	.584	.539	.515	.497	.470	.458
.1000	1.037	.931	.791	.688	.593	.518	.474	.453	.434	.422
.1500	1.047	.953	.837	.723	.617	.536	.468	.444	.419	.416
.2000	1.053	.968	.858	.757	.636	.548	.480	.444	.419	.416
.2500	1.059	.987	.877	.779	.675	.565	.494	.456	.434	.434
.3000	1.056	.994	.889	.791	.687	.583	.509	.476	.449	.446
.3500	1.059	1.013	.911	.816	.717	.616	.538	.506	.482	.478
.4000	1.050	1.003	.908	.832	.729	.628	.556	.526	.500	.490
.4500	1.031	.997	.905	.835	.735	.634	.573	.538	.521	.510
.5000	.991	.975	.889	.825	.732	.649	.585	.553	.533	.519
.5500	.953	.946	.880	.810	.726	.643	.585	.556	.534	.531
.6000	.897	.896	.834	.773	.696	.622	.573	.544	.524	.522
.6500	.857	.855	.788	.732	.672	.607	.550	.526	.524	.516
.7517	.701	.731	.692	.654	.590	.542	.503	.485	.467	.472
.7934	.667	.690	.643	.604	.560	.509	.482	.462	.449	.458
.8094	.640	.655	.612	.589	.536	.494	.465	.456	.446	.440
.8137	.586	.649	.612	.583	.536	.500	.477	.462	.440	.449
.8439	.617	.633	.588	.561	.518	.470	.442	.444	.425	.425
.8772	.548	.541	.498	.455	.407	.372	.339	.324	.301	.317
.9117	.511	.510	.458	.427	.380	.351	.342	.332	.313	.326
.9448	.467	.478	.418	.386	.340	.315	.301	.294	.268	.285
.9781	.427	.437	.397	.346	.313	.277	.263	.268	.229	.243
Vane										
.0250	.978	.912	.840	.635	.473	.348	.322	.306	.232	.214
.0500	.356	.317	.302	.181	.087	.036	.029	.029	.000	.006
.1000	.146	.142	.142	.075	.045	.027	.035	.024	.018	.038
.1500	.112	.114	.120	.069	.060	.042	.056	.056	.030	.059
.2000	.109	.108	.120	.075	.072	.048	.061	.071	.054	.070
.3000	.112	.114	.120	.090	.090	.089	.094	.091	.072	.088
.4000	.121	.127	.132	.100	.102	.095	.102	.109	.084	.103
.5000	.128	.133	.138	.118	.111	.113	.126	.124	.102	.114
.6000	.153	.158	.166	.137	.133	.131	.137	.144	.111	.126
.7000	.209	.218	.203	.190	.184	.185	.199	.212	.163	.176
.8000	.340	.339	.338	.312	.292	.310	.327	.338	.289	.290
.9200	.903	.886	.852	.819	.765	.804	.860	.881	.807	.760
Flap										
.0125	.062	.051	.065	.031	.030	.033	.058	.068	.024	.038
.0250	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000
.0500	.031	.038	.049	.037	.042	.036	.035	.032	.009	.035
.0750	.056	.054	.058	.056	.042	.036	.038	.038	.018	.047
.1000	.078	.082	.083	.072	.075	.060	.064	.042	.045	.053
.1500	.118	.133	.138	.118	.120	.101	.105	.091	.075	.097
.2000	.168	.184	.185	.168	.163	.134	.129	.115	.096	.126
.4000	.315	.326	.305	.287	.268	.232	.225	.221	.205	.228
.6000	.411	.443	.434	.411	.386	.357	.342	.329	.313	.328
.8000	.570	.582	.563	.583	.545	.500	.494	.485	.485	.466
.9000	.663	.696	.695	.738	.699	.679	.652	.544	.642	.607

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TABLE XI.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(d) Extended double slotted flap configuration; $\delta_f = 50^\circ$; $\frac{y}{b/2} = 0.43$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	.797	1.862	2.010	2.463	2.973	3.403	3.484	3.371	2.930	2.374
.0125	.957	1.742	1.988	2.439	2.976	3.421	3.490	3.421	2.995	2.395
.0250	1.012	1.661	2.019	2.485	2.967	3.418	3.564	3.564	3.003	2.374
.0500	1.030	1.423	2.075	2.582	3.108	3.574	3.584	3.474	3.021	2.406
.0750	1.039	1.291	2.305	2.612	3.084	3.444	3.440	3.421	3.039	2.386
.1000	1.061	1.261	2.615	2.788	3.036	3.406	3.385	3.427	3.062	2.386
.1500	1.079	1.243	2.264	3.624	3.840	3.847	3.795	3.757	3.091	2.386
.2000	1.106	1.249	1.506	2.909	4.178	4.391	4.232	3.924	3.050	2.380
.2500	1.118	1.243	1.183	1.966	3.256	4.101	4.136	3.804	2.989	2.366
.3000	1.142	1.267	1.230	1.533	2.461	3.508	3.918	3.684	2.939	2.357
.3500	1.154	1.255	1.242	1.312	1.861	2.805	3.487	3.363	2.801	2.339
.4000	1.176	1.273	1.286	1.306	1.633	2.418	3.127	3.164	2.745	2.337
.4500	1.200	1.291	1.317	1.321	1.485	2.148	2.839	2.965	2.666	2.308
.5000	1.224	1.276	1.320	1.309	1.395	1.882	2.522	2.713	2.543	2.284
.5500	1.233	1.306	1.345	1.336	1.404	1.775	2.358	2.550	2.478	2.255
.6000	1.273	1.345	1.373	1.373	1.407	1.693	2.209	2.412	2.373	2.224
.6993	1.315	1.348	1.385	1.370	1.398	1.586	1.927	2.146	2.211	2.151
.7507	1.303	1.348	1.370	1.370	1.398	1.551	1.821	2.023	2.121	2.093
.7704	1.351	1.390	1.429	1.424	1.446	1.583	1.854	2.029	2.118	2.084
.8099	1.412	1.439	1.485	1.467	1.506	1.619	1.845	2.018	2.091	2.067
.8512	1.473	1.514	1.534	1.518	1.539	1.651	1.821	1.959	2.015	2.029
.8911	1.606	1.643	1.665	1.648	1.663	1.740	1.883	1.988	2.000	2.000
.9322	1.800	1.820	1.842	1.830	1.843	1.920	2.018	2.097	2.006	1.986
.9724	2.260	2.258	2.311	2.297	2.313	2.361	2.429	2.450	2.153	2.035
Vane										
.0000	3.085	2.925	3.112	2.957	2.798	2.669	2.590	2.512	2.058	1.826
.0250	4.633	4.538	4.690	4.672	4.729	4.720	4.751	4.591	3.555	3.064
.0500	4.866	4.751	4.929	4.930	4.964	4.977	5.015	4.816	3.678	3.041
.1000	5.339	5.186	5.411	5.421	5.476	5.430	5.485	5.181	3.763	2.942
.1500	5.709	5.520	5.752	5.805	5.879	5.782	5.843	5.453	3.860	2.882
.2000	6.160	5.934	6.215	6.263	6.349	6.235	6.309	5.868	4.094	2.960
.3000	5.902	5.682	5.939	5.987	6.091	5.995	6.045	5.573	3.872	2.783
.4000	5.484	5.291	5.516	5.584	5.687	5.628	5.684	5.234	3.672	2.641
.5000	5.042	4.865	5.057	5.121	5.238	5.240	5.297	4.880	3.490	2.531
.6000	4.448	4.288	4.448	4.503	4.602	4.640	4.690	4.363	3.203	2.412
.7000	3.963	3.844	3.963	4.018	4.096	4.169	4.215	3.968	3.003	2.348
.8000	3.406	3.315	3.410	3.469	3.530	3.616	3.652	3.421	2.791	2.215
.9000	2.891	2.808	2.876	2.930	2.979	3.054	3.080	2.971	2.408	2.044
Flap										
.0000	1.873	1.817	1.938	1.945	1.976	2.045	2.097	2.026	1.993	1.939
.0125	2.609	2.553	2.640	2.634	2.699	2.740	2.786	2.690	2.183	1.913
.0250	3.351	3.279	3.370	3.394	3.455	3.518	3.572	3.471	2.860	2.505
.0500	3.512	3.438	3.507	3.524	3.608	3.696	3.751	3.673	3.039	2.661
.0750	3.488	3.402	3.466	3.506	3.584	3.693	3.780	3.890	3.103	2.679
.1000	3.360	3.267	3.330	3.363	3.410	3.521	3.593	3.547	3.003	2.577
.1500	2.639	2.559	2.600	2.624	2.648	2.773	2.833	2.871	2.561	2.238
.2000	2.224	2.168	2.199	2.248	2.265	2.361	2.393	2.398	2.173	1.954
.4000	1.691	1.664	1.680	1.706	1.729	1.817	1.845	1.912	1.912	1.847
.6000	1.454	1.441	1.457	1.479	1.518	1.583	1.610	1.670	1.754	1.757
.8000	1.218	1.228	1.242	1.282	1.337	1.385	1.403	1.409	1.543	1.621

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	1.121	.811	.789	.924	1.160	1.447	1.637	1.681	1.560	1.368
.0250	1.100	.847	.758	.782	.849	.976	1.066	1.062	1.038	.957
.0500	1.079	.889	.770	.727	.711	.737	.757	.760	.733	.693
.0750	1.070	.907	.780	.724	.693	.640	.651	.629	.616	.609
.1000	1.070	.916	.808	.739	.681	.621	.598	.576	.557	.554
.1500	1.064	.937	.829	.758	.687	.610	.551	.532	.502	.502
.2000	1.054	.955	.845	.764	.690	.610	.543	.509	.498	.493
.2500	1.048	.955	.857	.776	.699	.618	.554	.509	.498	.490
.3000	1.036	.949	.854	.782	.711	.618	.540	.503	.493	.493
.3500	1.030	.937	.860	.800	.723	.636	.554	.523	.507	.510
.4000	1.006	.931	.863	.800	.735	.639	.557	.532	.513	.504
.4500	.976	.910	.839	.785	.723	.639	.569	.538	.519	.507
.5000	.939	.880	.814	.767	.711	.630	.560	.526	.510	.507
.5500	.909	.856	.798	.751	.699	.621	.554	.520	.504	.499
.6000	.848	.799	.745	.709	.640	.592	.528	.506	.484	.484
.6995	.776	.739	.683	.684	.614	.559	.515	.488	.478	.473
.7499	.721	.685	.637	.618	.578	.544	.484	.462	.460	.452
.7700	.706	.655	.603	.597	.566	.533	.481	.456	.455	.438
.7744	.694	.655	.615	.600	.566	.527	.478	.450	.455	.449
.8110	.661	.619	.572	.558	.530	.491	.449	.421	.414	.406
.8519	.542	.511	.460	.455	.416	.391	.352	.330	.323	.313
.8919	.497	.468	.438	.418	.401	.376	.346	.319	.311	.319
.9332	.446	.417	.391	.385	.370	.334	.308	.292	.282	.264
.9738	.406	.384	.360	.351	.337	.317	.296	.275	.273	.244
Vane										
.0250	.421	.336	.432	.370	.268	.207	.176	.167	.123	.110
.0500	.127	.090	.118	.097	.069	.047	.053	.041	.044	.006
.1000	.061	.048	.040	.045	.036	.036	.044	.029	.023	.020
.1500	.061	.060	.037	.067	.057	.059	.050	.056	.056	.046
.2000	.076	.084	.050	.079	.078	.089	.073	.067	.067	.067
.3000	.103	.114	.081	.100	.114	.138	.106	.094	.088	.087
.4000	.124	.120	.081	.127	.133	.142	.123	.117	.109	.099
.5000	.152	.141	.118	.142	.154	.154	.144	.132	.117	.116
.6000	.167	.174	.143	.176	.184	.192	.182	.158	.144	.138
.7000	.218	.222	.196	.218	.238	.246	.232	.222	.182	.180
.8000	.367	.366	.342	.376	.389	.394	.387	.383	.299	.267
.9200	1.006	.976	.997	1.018	1.039	1.059	1.056	1.015	.836	.725
Flap										
.0125	.073	.078	.075	.100	.120	.160	.164	.158	.094	.055
.0250	.009	.012	.000	.000	.024	.053	.021	.009	.009	.000
.0500	.027	.027	.000	.024	.036	.059	.029	.029	.018	.026
.0750	.070	.057	.022	.058	.060	.047	.047	.044	.044	.052
.1000	.097	.093	.056	.079	.090	.080	.070	.047	.070	.052
.1500	.130	.126	.090	.109	.117	.112	.094	.088	.088	.093
.2000	.176	.174	.127	.152	.184	.139	.123	.114	.117	.104
.4000	.306	.288	.261	.267	.259	.249	.202	.202	.194	.203
.6000	.412	.402	.354	.364	.361	.337	.305	.301	.296	.310
.8000	.539	.538	.485	.497	.488	.465	.428	.418	.428	.481
.9000	.633	.628	.590	.594	.602	.580	.546	.544	.587	.615

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TABLE XI.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(e) Extended double slotted flap configuration; $\delta_f = 50^\circ$; $\frac{y}{b/2} = 0.55$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	.829	1.921	1.899	2.314	2.632	2.818	2.789	2.650	2.440	2.267
.0125	1.103	1.848	1.945	2.361	2.675	2.869	2.833	2.744	2.482	2.267
.0250	1.068	1.766	1.905	2.324	2.626	2.809	2.822	2.756	2.497	2.279
.0500	1.072	1.469	1.905	2.267	2.642	2.854	2.912	2.806	2.512	2.285
.0750	1.096	1.348	1.988	2.436	2.717	2.934	2.874	2.767	2.485	2.264
.1000	1.115	1.320	2.034	2.498	2.798	2.961	2.825	2.797	2.488	2.252
.1500	1.149	1.317	2.292	2.523	2.720	2.838	2.801	2.844	2.497	2.244
.2000	1.159	1.317	2.462	2.741	2.684	2.783	2.830	2.836	2.491	2.244
.2500	1.159	1.326	2.372	3.333	2.852	2.821	2.965	2.932	2.512	2.244
.3000	1.202	1.336	2.049	3.414	3.482	3.229	3.140	2.973	2.500	2.232
.3500	1.212	1.348	1.680	3.031	3.705	3.512	3.257	2.976	2.497	2.226
.4000	1.243	1.351	1.400	2.461	3.443	3.544	3.237	2.920	2.467	2.223
.4500	1.249	1.358	1.271	2.012	3.051	3.428	3.199	2.879	2.452	2.223
.5000	1.286	1.386	1.258	1.695	2.654	3.256	3.105	2.832	2.434	2.200
.6194	1.187	1.342	1.295	1.336	1.828	2.533	2.725	2.650	2.398	2.206
.6844	1.199	1.355	1.302	1.296	1.611	2.226	2.491	2.462	2.289	2.180
.7066	1.246	1.386	1.348	1.308	1.593	2.184	2.442	2.429	2.265	2.132
.7594	1.290	1.421	1.360	1.321	1.530	2.018	2.263	2.262	2.139	2.033
.8101	1.346	1.500	1.443	1.408	1.533	1.961	2.237	2.262	2.169	2.079
.8622	1.458	1.605	1.560	1.514	1.578	1.931	2.187	2.197	2.136	2.038
.9142	1.604	1.791	1.732	1.701	1.723	1.991	2.184	2.215	2.126	2.038
.9652	1.997	2.216	2.182	2.190	2.114	2.303	2.392	2.362	2.196	2.071
Vane										
.0000	2.364	2.285	2.323	2.293	2.099	2.074	1.971	1.832	1.575	1.499
.0250	4.308	4.406	4.357	4.423	4.250	4.366	4.199	3.841	3.166	2.951
.0500	4.857	4.646	4.625	4.697	4.527	4.619	4.412	4.000	3.175	2.915
.1000	5.155	5.210	5.249	5.389	5.187	5.184	4.830	4.264	3.166	2.839
.1500	5.473	5.491	5.582	5.772	5.587	5.491	5.035	4.353	3.090	2.736
.2000	5.685	5.675	5.800	6.009	5.810	5.708	5.205	4.482	3.132	2.710
.3000	5.624	5.650	5.763	6.046	5.873	5.744	5.175	4.414	3.018	2.610
.4000	5.203	5.225	5.357	5.631	5.515	5.413	4.845	4.123	2.825	2.464
.5000	4.875	4.912	5.031	5.352	5.274	5.172	4.632	3.932	2.738	2.396
.6000	4.324	4.352	4.471	4.772	4.765	4.720	4.208	3.588	2.587	2.308
.7000	3.791	3.839	3.929	4.187	4.160	4.137	3.757	3.267	2.464	2.220
.8000	3.302	3.355	3.409	3.660	3.614	3.622	3.316	2.938	2.319	2.123
.9000	2.822	2.871	2.905	3.103	3.081	3.092	2.848	2.579	2.132	1.994
Flap										
.0000	1.262	1.361	1.418	1.570	1.569	1.586	1.515	1.403	1.160	1.047
.0125	2.380	2.583	2.619	2.810	2.771	2.774	2.608	2.432	2.081	1.918
.0250	2.891	3.156	3.206	3.442	3.395	3.470	3.196	2.976	2.546	2.352
.0500	3.078	3.390	3.440	3.694	3.684	3.690	3.488	3.244	2.771	2.528
.0750	3.112	3.447	3.511	3.800	3.780	3.788	3.588	3.338	2.852	2.569
.1000	3.078	3.336	3.382	3.660	3.635	3.634	3.436	3.203	2.723	2.473
.1500	2.738	2.921	2.935	3.149	3.111	3.077	2.901	2.726	2.349	2.150
.2000	2.171	2.323	2.323	2.498	2.476	2.470	2.357	2.270	2.051	1.915
.4000	1.788	1.883	1.889	1.975	1.913	1.887	1.848	1.870	1.852	1.795
.6000	1.607	1.712	1.683	1.733	1.654	1.610	1.602	1.676	1.789	1.757
.8000	1.318	1.446	1.418	1.442	1.355	1.318	1.316	1.486	1.617	1.634

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	1.100	.829	.865	1.112	1.389	1.670	1.828	1.826	1.762	1.666
.0250	1.112	.861	.778	.872	.976	1.122	1.216	1.235	1.214	1.170
.0500	1.090	.899	.778	.776	.774	.815	.854	.862	.864	.845
.0750	1.056	.899	.766	.744	.684	.696	.705	.721	.705	.686
.1000	1.063	.908	.775	.741	.672	.646	.640	.647	.651	.625
.1500	1.031	.918	.785	.741	.654	.604	.573	.573	.581	.554
.2000	1.028	.927	.806	.754	.663	.595	.556	.547	.554	.528
.2500	1.015	.921	.803	.754	.672	.601	.550	.544	.548	.528
.3000	.994	.912	.803	.760	.666	.604	.544	.538	.542	.516
.3500	.981	.908	.806	.773	.675	.601	.550	.541	.548	.513
.4000	.963	.899	.797	.769	.675	.601	.550	.529	.545	.513
.4500	.925	.874	.788	.754	.666	.595	.544	.529	.542	.513
.5000	.906	.855	.772	.744	.664	.589	.529	.523	.536	.499
.6195	.769	.788	.717	.704	.614	.559	.506	.506	.512	.478
.6834	.723	.724	.680	.663	.599	.548	.506	.494	.494	.460
.7080	.710	.715	.643	.642	.578	.521	.480	.471	.464	.455
.7137	.707	.715	.652	.642	.572	.524	.491	.468	.470	.446
.7603	.663	.671	.622	.617	.545	.509	.459	.456	.458	.431
.8119	.533	.541	.514	.517	.467	.432	.395	.365	.373	.358
.8643	.514	.519	.489	.492	.437	.417	.383	.371	.361	.343
.9162	.458	.462	.440	.445	.404	.366	.342	.338	.322	.314
.9669	.389	.408	.382	.402	.370	.339	.310	.300	.289	.273
Vane										
.0250	.495	.396	.458	.439	.358	.310	.269	.265	.220	.208
.0500	.150	.108	.132	.125	.087	.074	.064	.074	.078	.062
.1000	.065	.060	.068	.078	.066	.057	.064	.068	.069	.067
.1500	.081	.092	.074	.093	.093	.092	.091	.103	.084	.082
.2000	.100	.101	.095	.115	.120	.113	.105	.124	.102	.103
.3000	.131	.123	.120	.131	.130	.134	.126	.135	.127	.114
.4000	.137	.146	.142	.159	.157	.155	.146	.162	.142	.125
.5000	.153	.161	.151	.174	.172	.164	.158	.162	.148	.135
.6000	.181	.190	.185	.196	.202	.196	.173	.188	.169	.158
.7000	.240	.247	.237	.274	.253	.250	.243	.238	.211	.191
.8000	.374	.380	.378	.414	.407	.399	.374	.350	.310	.285
.9200	1.034	1.029	1.055	1.121	1.096	1.095	1.012	.944	.801	.766
Flap										
.0125	.040	.063	.080	.140	.142	.167	.158	.165	.108	.091
.0250	.000	.009	.015	.044	.042	.042	.041	.056	.033	.038
.0500	.059	.066	.055	.078	.060	.071	.061	.079	.063	.065
.0750	.090	.082	.077	.100	.087	.095	.094	.091	.081	.073
.1000	.121	.130	.117	.131	.111	.107	.099	.121	.096	.103
.1500	.171	.174	.163	.165	.151	.143	.135	.144	.135	.126
.2000	.206	.215	.203	.196	.181	.170	.155	.165	.169	.161
.4000	.389	.383	.351	.346	.304	.277	.263	.268	.283	.264
.6000	.508	.503	.462	.445	.401	.369	.351	.362	.389	.378
.8000	.607	.627	.585	.576	.533	.494	.471	.491	.539	.531
.9000	.688	.722	.674	.670	.630	.595	.573	.603	.681	.689

TABLE XL.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Concluded

(f) Extended double slotted flap configuration; $\delta_f = 60^\circ$; $\frac{y}{b/2} = 0.72$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	.963	1.896	2.003	2.252	2.363	2.474	2.326	2.229	2.084	2.009
.0125	1.190	1.881	1.957	2.182	2.294	2.414	2.285	2.200	2.093	2.003
.0250	1.190	1.896	1.976	2.197	2.297	2.423	2.323	2.241	2.110	2.018
.0500	1.190	1.936	1.997	2.222	2.300	2.423	2.402	2.282	2.128	2.023
.0750	1.196	1.991	2.030	2.237	2.312	2.444	2.449	2.303	2.128	2.020
.1000	1.187	1.985	2.024	2.212	2.270	2.429	2.431	2.282	2.131	2.023
.1500	1.202	2.028	2.069	2.265	2.306	2.486	2.426	2.303	2.142	2.018
.2000	1.214	1.921	2.121	2.317	2.348	2.523	2.408	2.323	2.137	2.012
.2500	1.229	1.610	2.182	2.345	2.366	2.468	2.399	2.332	2.137	2.015
.3000	1.248	1.342	2.257	2.351	2.330	2.423	2.393	2.329	2.128	2.020
.3500	1.263	1.296	2.409	2.379	2.309	2.390	2.402	2.344	2.125	2.020
.4000	1.275	1.305	2.476	2.397	2.276	2.375	2.408	2.341	2.142	2.023
.4500	1.290	1.335	2.497	2.505	2.288	2.357	2.411	2.350	2.128	2.032
.5000	1.318	1.381	2.536	2.751	2.327	2.390	2.455	2.379	2.151	2.041
.5500	1.312	1.372	2.424	2.871	2.372	2.399	2.455	2.350	2.140	2.038
.6000	1.324	1.384	2.306	2.986	2.486	2.438	2.475	2.362	2.151	2.041
.6500	1.336	1.396	2.142	2.939	2.577	2.501	2.470	2.335	2.140	2.043
.7000	1.361	1.421	2.048	2.865	2.613	2.562	2.481	2.309	2.119	2.035
.7500	1.398	1.467	1.891	2.739	2.649	2.625	2.493	2.309	2.113	2.029
.8000	1.401	1.473	1.782	2.606	2.631	2.697	2.481	2.294	2.110	2.025
.8500	1.431	1.485	1.636	2.403	2.544	2.607	2.429	2.232	2.067	2.032
.9000	1.462	1.537	1.582	2.246	2.459	2.568	2.390	2.209	2.073	2.029

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	.966	.796	.879	1.043	1.189	1.414	1.481	1.500	1.474	1.469
.0250	.979	.787	.794	.874	.967	1.117	1.170	1.203	1.183	1.196
.0500	.976	.802	.751	.760	.799	.877	.909	.921	.924	.928
.0750	.966	.832	.764	.732	.733	.787	.789	.812	.802	.825
.1000	.969	.835	.761	.720	.706	.739	.724	.738	.741	.750
.1500	.966	.854	.770	.717	.688	.688	.663	.662	.666	.684
.2000	.963	.857	.791	.735	.697	.676	.645	.644	.651	.664
.2500	.957	.866	.794	.748	.715	.691	.645	.644	.640	.664
.3000	.957	.890	.827	.788	.748	.721	.678	.659	.669	.684
.3500	.960	.903	.851	.815	.781	.742	.689	.653	.677	.675
.4000	.957	.909	.854	.809	.763	.730	.654	.621	.634	.629
.4500	.957	.906	.864	.815	.760	.712	.634	.618	.631	.629
.5000	.951	.915	.882	.834	.775	.724	.642	.615	.616	.629
.5500	.957	.927	.891	.855	.790	.739	.660	.632	.631	.641
.6000	.954	.945	.915	.874	.817	.760	.683	.662	.657	.667
.6500	.972	.963	.939	.902	.841	.799	.722	.703	.701	.719
.7000	.976	.979	.970	.938	.889	.838	.789	.768	.776	.790
.7500	1.009	1.015	1.027	1.015	.970	.919	.865	.844	.872	.882
.8000	1.070	1.092	1.106	1.098	1.078	1.036	.991	.991	.997	1.043
.8500	1.183	1.204	1.242	1.274	1.243	1.213	1.176	1.168	1.201	1.239
.9000	1.324	1.393	1.454	1.508	1.498	1.465	1.417	1.406	1.424	1.469
.9500	1.477	1.558	1.624	1.760	1.778	1.745	1.672	1.641	1.657	1.707

[redacted]

TABLE XII.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE

(a) Extended double slotted flap configuration; $\delta_f = 55^\circ$; $\frac{y}{b/2} = 0$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Fuselage										
.0000	.000	.000	.000	.000	.009	.018	.050	.104	.114	.149
.0500	.792	.829	.885	.915	.976	1.063	1.036	1.061	1.056	1.046
.1000	.880	.938	.969	.985	1.028	1.057	1.083	1.124	1.091	1.123
.1500	.962	.997	1.040	1.037	1.079	1.081	1.121	1.139	1.114	1.134
.2000	1.013	1.046	1.068	1.070	1.097	1.117	1.116	1.136	1.120	1.111
.2500	1.060	1.071	1.090	1.091	1.113	1.126	1.136	1.147	1.135	1.134
.3000	1.088	1.093	1.115	1.091	1.110	1.123	1.121	1.130	1.126	1.131
.3500	1.114	1.108	1.118	1.082	1.122	1.150	1.169	1.170	1.158	1.174
.4000	1.047	1.044	1.072	1.058	1.082	1.096	1.124	1.139	1.135	1.154
.4500	1.051	1.059	1.081	1.082	1.116	1.165	1.204	1.243	1.240	1.274
.5000	1.051	1.068	1.115	1.140	1.213	1.273	1.337	1.387	1.398	1.557
.5500	1.073	1.102	1.153	1.207	1.280	1.360	1.423	1.454	1.494	1.981
.6000	1.095	1.133	1.215	1.253	1.316	1.360	1.397	1.425	1.471	2.000
.6500	1.126	1.155	1.237	1.268	1.316	1.357	1.385	1.373	1.392	1.797
.7000	1.161	1.195	1.268	1.280	1.325	1.360	1.373	1.373	1.368	1.686
.7500	1.208	1.235	1.290	1.289	1.328	1.381	1.411	1.405	1.415	1.611
.8000	1.256	1.276	1.324	1.316	1.383	1.396	1.453	1.459	1.488	1.551
.8500	1.341	1.328	1.352	1.344	1.374	1.423	1.468	1.506	1.523	1.531
.9000	1.338	1.316	1.327	1.301	1.335	1.369	1.406	1.451	1.465	1.463
.9500	1.268	1.257	1.268	1.234	1.274	1.270	1.305	1.364	1.398	1.417
.9940	1.268	1.232	1.240	1.213	1.216	1.231	1.237	1.309	1.344	1.440

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Fuselage										
.0500	.868	.820	.773	.687	.623	.547	.539	.442	.409	.383
.1000	.959	.898	.878	.784	.748	.670	.601	.529	.506	.486
.1500	1.022	.969	.953	.854	.815	.736	.663	.616	.582	.543
.2000	1.047	1.003	.984	.897	.863	.796	.713	.688	.658	.597
.2500	1.070	1.028	1.012	.942	.921	.841	.763	.731	.711	.657
.3000	1.025	.997	1.003	.961	.945	.931	.888	.893	.871	.837
.3500	1.066	1.028	1.015	.970	.942	.886	.814	.789	.757	.706
.4000	1.000	.972	.969	.912	.888	.823	.769	.725	.699	.663
.4500	.959	.938	.931	.857	.836	.778	.701	.670	.635	.611
.5000	.931	.901	.891	.809	.778	.700	.624	.581	.558	.523
.5500	.905	.879	.841	.760	.720	.631	.550	.512	.497	.457
.6000	.861	.811	.785	.696	.641	.565	.497	.451	.427	.406
.7500	1.032	.978	.956	.891	.869	.805	.731	.702	.699	.666
.9500	.962	.985	1.023	.982	1.000	.979	.994	1.014	1.003	1.000
.9940	1.082	1.161	1.224	1.297	1.268	1.258	1.261	1.318	1.322	1.394

[redacted]

TABLE XII.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(b) Extended double slotted flap configuration; $\delta_f = 55^\circ$; $\frac{y}{b/2} = 0.21$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	.902	.919	2.006	3.317	5.485	8.289	11.268	12.211	4.786	2.768
.0125	.949	1.259	1.715	3.225	7.849	11.202	14.015	14.030	4.404	2.530
.0250	.975	1.203	1.572	2.529	3.705	6.606	9.695	10.753	4.302	2.565
.0500	1.000	1.159	1.444	1.628	2.057	2.867	3.872	5.055	3.896	2.479
.0750	1.019	1.156	1.389	1.542	1.855	2.301	2.805	3.416	3.586	2.479
.1000	1.041	1.169	1.362	1.520	1.759	2.143	2.526	2.991	3.586	2.467
.1500	1.060	1.169	1.319	1.474	1.653	1.912	2.201	2.449	2.974	2.434
.2000	1.082	1.181	1.319	1.440	1.596	1.797	2.038	2.280	2.687	2.405
.2500	1.107	1.200	1.304	1.425	1.542	1.705	1.936	2.099	2.400	2.345
.3000	1.123	1.191	1.304	1.409	1.500	1.648	1.855	1.934	2.218	2.303
.3500	1.129	1.206	1.298	1.391	1.476	1.617	1.733	1.816	2.061	2.259
.4000	1.145	1.215	1.301	1.378	1.461	1.596	1.669	1.749	1.951	2.229
.4500	1.177	1.234	1.307	1.385	1.449	1.564	1.628	1.699	1.876	2.199
.5000	1.208	1.269	1.332	1.397	1.440	1.529	1.605	1.667	1.818	2.178
.5500	1.218	1.266	1.325	1.397	1.425	1.499	1.576	1.632	1.754	2.113
.6000	1.234	1.275	1.358	1.388	1.416	1.493	1.564	1.650	1.734	2.080
.6500	1.278	1.306	1.339	1.391	1.431	1.493	1.605	1.685	1.716	2.036
.7000	1.309	1.328	1.365	1.394	1.440	1.484	1.610	1.635	1.690	1.988
.7824	1.328	1.341	1.362	1.394	1.428	1.496	1.497	1.492	1.597	1.890
.8176	1.376	1.388	1.411	1.431	1.461	1.556	1.500	1.458	1.583	1.869
.8323	1.423	1.438	1.459	1.480	1.494	1.396	1.526	1.498	1.618	1.866
.8913	1.503	1.506	1.523	1.520	1.527	1.617	1.503	1.481	1.612	1.809
.9281	1.606	1.616	1.614	1.582	1.584	1.687	1.561	1.557	1.658	1.818
.9504	1.690	1.647	1.642	1.606	1.596	1.712	1.596	1.583	1.705	1.824
.9804	1.792	1.797	1.763	1.680	1.681	1.847	1.747	1.790	1.896	1.946
Vane										
.0000	1.417	1.241	1.058	.914	.843	.903	.837	.851	.890	.875
.0250	2.521	2.484	2.481	2.265	2.328	2.690	2.616	2.676	2.754	2.568
.0500	3.265	3.206	3.137	2.775	2.895	3.450	3.390	3.498	3.569	3.149
.1000	3.461	3.363	3.274	2.865	2.985	3.618	3.587	3.720	3.777	3.256
.1500	3.448	3.331	3.222	2.822	2.934	3.581	3.579	3.714	3.809	3.280
.2000	3.373	3.263	3.152	2.797	2.907	3.536	3.547	3.682	3.890	3.330
.3000	2.997	2.878	2.766	2.511	2.617	3.104	3.134	3.265	3.583	3.253
.4000	2.609	2.513	2.456	2.265	2.404	2.575	2.587	2.717	3.238	3.065
.5000	2.319	2.250	2.207	2.166	2.241	2.262	2.241	2.373	2.957	2.943
.6000	2.136	2.141	2.085	2.031	2.081	2.234	2.192	2.256	2.638	2.681
.7000	2.085	2.081	2.043	1.972	2.033	2.198	2.189	2.247	2.534	2.533
.8000	2.054	2.044	1.991	1.945	1.988	2.149	2.166	2.239	2.482	2.428
.9000	2.022	1.984	1.927	1.895	1.934	2.088	2.108	2.189	2.406	2.292
Flap										
.0000	1.776	1.638	1.362	1.175	1.184	1.234	1.250	1.320	1.406	1.292
.0125	2.186	2.169	2.061	1.955	1.958	2.082	2.105	2.186	2.337	2.195
.0250	2.357	2.356	2.313	2.234	2.204	2.456	2.491	2.580	2.786	2.658
.0500	2.467	2.444	2.414	2.354	2.434	2.599	2.640	2.740	2.971	2.899
.0750	2.294	2.263	2.234	2.185	2.259	2.408	2.451	2.527	2.757	2.789
.1000	2.123	2.075	2.052	2.025	2.066	2.210	2.244	2.329	2.534	2.649
.1500	1.906	1.856	1.800	1.788	1.762	1.915	1.962	2.128	2.255	2.324
.2000	1.887	1.803	1.727	1.726	1.696	1.818	1.895	1.982	2.143	2.205
.4000	1.868	1.788	1.748	1.766	1.765	1.815	1.878	1.959	2.108	2.131
.6000	1.802	1.744	1.708	1.735	1.723	1.766	1.829	1.892	2.009	2.036
.8000	1.710	1.663	1.629	1.646	1.645	1.687	1.738	1.804	1.881	1.857

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	1.063	.819	.641	.578	.587	.666	.724	.837	.542	.390
.0250	1.041	.834	.672	.554	.470	.404	.372	.367	.278	.205
.0500	1.032	.869	.711	.594	.458	.344	.259	.222	.229	.211
.0750	1.016	.869	.733	.618	.503	.368	.291	.262	.275	.274
.1000	1.025	.891	.766	.640	.530	.423	.349	.332	.339	.336
.1500	1.013	.891	.766	.677	.572	.496	.419	.414	.406	.408
.2000	1.019	.919	.803	.714	.620	.544	.474	.455	.446	.440
.2500	1.038	.944	.839	.748	.663	.593	.520	.493	.475	.476
.3000	1.041	.938	.839	.775	.693	.629	.549	.531	.513	.500
.3500	1.063	.969	.891	.818	.738	.678	.587	.566	.545	.530
.4000	1.063	.981	.906	.843	.765	.696	.605	.583	.574	.554
.4500	1.070	.991	.918	.858	.786	.720	.634	.603	.583	.583
.5000	1.032	.963	.900	.855	.777	.711	.628	.600	.591	.589
.5500	.972	.922	.863	.834	.759	.720	.616	.621	.580	.583
.6000	.937	.888	.848	.818	.753	.690	.623	.580	.571	.577
.6500	.861	.809	.772	.751	.699	.635	.567	.554	.548	.542
.7000	.937	.913	.857	.806	.729	.660	.596	.571	.588	.568
.7815	.735	.703	.660	.634	.584	.532	.488	.481	.467	.476
.8179	.691	.673	.654	.628	.596	.550	.494	.490	.487	.497
.8325	.691	.706	.699	.686	.657	.626	.581	.560	.562	.577
.8366	.716	.741	.748	.738	.726	.705	.654	.653	.667	.684
.8921	.571	.488	.438	.406	.380	.356	.320	.306	.310	.304
.9286	.552	.450	.416	.394	.380	.356	.331	.332	.328	.330
.9518	.527	.419	.358	.351	.318	.301	.279	.277	.284	.286
.9819	.492	.397	.319	.277	.250	.240	.230	.233	.255	.247
Vane										
.0250	.820	.581	.356	.234	.154	.122	.116	.122	.136	.125
.0500	.615	.419	.210	.129	.090	.070	.070	.070	.070	.071
.1000	.495	.344	.195	.142	.120	.119	.105	.105	.099	.101
.1500	.445	.322	.196	.151	.133	.131	.126	.125	.133	.113
.2000	.410	.303	.204	.154	.145	.146	.134	.122	.128	.128
.3000	.372	.288	.201	.166	.154	.155	.145	.140	.145	.140
.4000	.338	.281	.207	.169	.169	.167	.163	.163	.157	.158
.5000	.325	.281	.219	.178	.172	.173	.172	.169	.174	.164
.6000	.325	.288	.228	.194	.193	.185	.192	.187	.188	.182
.7000	.363	.316	.255	.225	.235	.240	.238	.233	.232	.232
.8000	.429	.394	.353	.332	.325	.344	.337	.338	.357	.336
.9200	.823	.806	.772	.738	.762	.806	.797	.834	.887	.848
Flap										
.0125	.666	.313	.119	.062	.054	.064	.081	.076	.087	.083
.0250	.360	.131	.058	.049	.040	.052	.052	.047	.061	.065
.0500	.252	.131	.103	.105	.120	.116	.116	.105	.113	.116
.0750	.230	.150	.134	.126	.145	.131	.128	.128	.136	.140
.1000	.224	.172	.170	.160	.178	.167	.172	.157	.165	.179
.1500	.237	.203	.201	.200	.202	.210	.192	.189	.194	.202
.4000	.416	.431	.426	.418	.401	.380	.360	.367	.359	.372
.6000	.590	.584	.572	.548	.518	.496	.474	.472	.467	.461
.8000	.713	.691	.663	.652	.608	.587	.552	.557	.536	.536
.9000	.805	.788	.751	.732	.717	.699	.660	.656	.620	.592

TABLE XII.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(c) Extended double slotted flap configuration; $\delta_f = 55^\circ$; $\frac{y}{b/2} = 0.30$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	.965	1.264	2.012	2.776	3.660	4.565	5.099	4.934	3.663	2.429
.0125	.961	1.472	2.162	2.808	3.578	4.247	4.690	4.634	3.628	2.408
.0250	1.016	1.383	2.257	2.898	3.718	4.440	4.820	4.631	3.640	2.429
.0500	1.048	1.271	2.739	3.787	4.040	4.521	4.658	4.614	3.577	2.379
.0750	1.048	1.233	1.687	3.748	5.393	6.032	5.863	5.222	3.619	2.385
.1000	1.055	1.217	1.224	2.339	4.171	5.416	5.836	5.225	3.619	2.420
.1500	1.081	1.211	1.245	1.381	2.213	3.327	4.462	4.424	3.370	2.343
.2000	1.094	1.220	1.300	1.330	1.681	2.262	3.304	3.611	3.175	2.332
.2500	1.123	1.223	1.288	1.326	1.526	1.851	2.692	3.015	2.992	2.335
.3000	1.145	1.227	1.313	1.351	1.493	1.705	2.316	2.698	2.823	2.302
.3500	1.155	1.230	1.297	1.339	1.450	1.592	2.029	2.349	2.613	2.267
.4000	1.177	1.252	1.319	1.360	1.452	1.577	1.906	2.190	2.483	2.258
.4500	1.203	1.274	1.331	1.372	1.474	1.586	1.874	2.113	2.379	2.235
.5000	1.229	1.289	1.343	1.378	1.459	1.563	1.789	1.986	2.222	2.188
.5500	1.242	1.289	1.345	1.375	1.456	1.536	1.734	1.899	2.145	2.173
.6000	1.255	1.313	1.368	1.384	1.459	1.542	1.716	1.865	2.068	2.132
.6500	1.300	1.346	1.383	1.396	1.480	1.533	1.693	1.830	2.000	2.097
.7516	1.258	1.349	1.359	1.393	1.462	1.488	1.588	1.686	1.840	1.971
.7940	1.310	1.384	1.399	1.435	1.503	1.515	1.632	1.706	1.790	1.933
.8099	1.297	1.378	1.405	1.435	1.496	1.512	1.597	1.669	1.764	1.901
.8429	1.394	1.459	1.466	1.495	1.575	1.571	1.687	1.749	1.790	1.901
.8767	1.390	1.488	1.491	1.523	1.608	1.601	1.687	1.715	1.722	1.836
.9108	1.507	1.576	1.576	1.613	1.702	1.708	1.787	1.784	1.755	1.810
.9437	1.681	1.755	1.748	1.784	1.882	1.896	1.974	1.934	1.838	1.839
.9768	2.081	2.164	2.135	2.177	2.286	2.327	2.409	2.320	2.119	1.986
Vane										
.0000	3.874	3.643	3.444	3.276	3.159	3.113	3.102	2.934	2.574	2.170
.0250	4.923	4.787	4.579	4.661	4.779	4.982	5.149	4.827	4.261	3.317
.0500	5.239	5.123	4.959	5.060	5.141	5.476	5.518	5.300	4.498	3.446
.1000	5.484	5.321	5.131	5.267	5.284	5.702	5.699	5.523	4.628	3.367
.1500	5.352	5.378	5.183	5.327	5.296	5.759	5.757	5.617	4.693	3.358
.2000	5.587	5.460	5.245	5.372	5.347	5.794	5.807	5.741	4.663	3.449
.3000	5.268	5.164	4.939	5.045	5.028	5.452	5.494	5.493	4.776	3.423
.4000	4.768	4.636	4.416	4.532	4.566	4.997	4.994	5.052	4.421	3.218
.5000	4.326	4.151	3.941	4.030	4.086	4.512	4.488	4.591	4.072	3.015
.6000	3.739	3.592	3.407	3.468	3.572	3.940	3.965	4.069	3.675	2.833
.7000	3.226	3.092	2.941	2.982	3.113	3.408	3.480	3.571	3.279	2.631
.8000	2.710	2.613	2.503	2.538	2.687	2.916	3.038	3.081	2.890	2.382
.9000	2.345	2.290	2.202	2.225	2.347	2.524	2.670	2.677	2.548	2.220
Flap										
.0000	1.300	1.321	1.307	1.246	1.240	1.301	1.368	1.380	1.308	1.164
.0125	2.090	2.132	2.073	2.072	2.110	2.208	2.278	2.297	2.246	2.038
.0250	2.571	2.626	2.536	2.550	2.617	2.744	2.860	2.902	2.879	2.607
.0500	2.626	2.680	2.582	2.592	2.712	2.851	3.032	3.104	3.116	2.826
.0750	2.607	2.645	2.543	2.556	2.705	2.845	3.047	3.130	3.181	2.889
.1000	2.500	2.497	2.389	2.402	2.541	2.696	2.868	2.983	3.015	2.731
.1500	2.068	2.076	1.994	1.997	2.155	2.220	2.330	2.441	2.613	2.426
.2000	1.848	1.812	1.745	1.751	1.857	1.925	2.029	2.115	2.196	2.053
.4000	1.700	1.698	1.641	1.649	1.654	1.658	1.734	1.827	1.950	1.906
.6000	1.687	1.705	1.662	1.652	1.648	1.622	1.722	1.790	1.867	1.816
.8000	1.645	1.689	1.645	1.631	1.611	1.583	1.664	1.718	1.737	1.686

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	1.055	.799	.724	.796	1.018	1.238	1.480	1.802	1.361	1.062
.0250	1.045	.849	.712	.697	.737	.824	.939	.942	.894	.757
.0500	1.048	.887	.745	.685	.632	.595	.611	.597	.577	.528
.0750	1.048	.912	.770	.670	.617	.548	.515	.490	.488	.460
.1000	1.055	.925	.782	.682	.620	.533	.485	.452	.444	.422
.1500	1.061	.944	.816	.721	.648	.536	.480	.447	.432	.425
.2000	1.068	.969	.846	.748	.660	.554	.485	.455	.432	.419
.2500	1.074	.978	.868	.769	.684	.577	.503	.464	.444	.448
.3000	1.071	.981	.880	.781	.702	.592	.523	.484	.459	.455
.3500	1.068	1.000	.899	.817	.723	.625	.547	.496	.485	.484
.4000	1.052	.988	.902	.814	.733	.637	.564	.522	.509	.499
.4500	1.042	.981	.889	.829	.748	.658	.585	.533	.521	.522
.5000	.990	.956	.880	.814	.745	.661	.599	.545	.530	.525
.5500	.955	.951	.838	.802	.730	.652	.596	.550	.550	.537
.6000	.897	.881	.807	.760	.699	.622	.582	.527	.527	.519
.6500	.848	.850	.773	.718	.669	.601	.550	.516	.509	.504
.7517	.706	.717	.656	.628	.584	.542	.506	.467	.468	.468
.7934	.668	.642	.604	.574	.544	.494	.474	.444	.441	.437
.8094	.652	.623	.577	.553	.523	.470	.459	.438	.429	.428
.8137	.590	.604	.577	.553	.538	.485	.448	.432	.432	.428
.8439	.619	.591	.558	.532	.505	.464	.447	.412	.408	.405
.8772	.552	.509	.466	.426	.401	.354	.333	.308	.296	.299
.9117	.506	.472	.429	.387	.345	.330	.325	.303	.290	.302
.9448	.465	.434	.386	.348	.328	.292	.284	.262	.252	.261
.9781	.419	.403	.362	.318	.286	.256	.249	.233	.219	.223
Vane										
.0250	1.342	1.277	1.187	.943	.745	.625	.605	.545	.473	.393
.0500	.448	.443	.420	.267	.143	.086	.079	.061	.047	.044
.1000	.139	.160	.130	.075	.036	.000	.015	.026	.006	.012
.1500	.081	.107	.092	.042	.030	.015	.035	.032	.021	.029
.2000	.065	.079	.080	.045	.046	.039	.050	.040	.038	.044
.3000	.071	.072	.074	.060	.061	.045	.061	.055	.056	.067
.4000	.074	.082	.085	.069	.082	.074	.088	.089	.077	.073
.5000	.081	.088	.086	.084	.097	.089	.105	.095	.089	.097
.6000	.113	.120	.104	.111	.112	.110	.132	.124	.112	.111
.7000	.171	.157	.153	.165	.176	.167	.190	.176	.169	.161
.8000	.290	.283	.273	.285	.298	.312	.327	.314	.305	.276
.9200	.819	.799	.770	.778	.796	.839	.866	.853	.837	.754
Flap										
.0125	.100	.097	.098	.075	.079	.089	.120	.112	.095	.079
.0250	.000	.000	.000	.000	.000	.000	.006	.000	.000	.009
.0500	.026	.028	.018	.027	.030	.006	.032	.017	.015	.015
.0750	.032	.028	.025	.030	.036	.021	.038	.029	.018	.029
.1000	.048	.062	.052	.051	.040	.036	.050	.037	.038	.044
.1500	.090	.104	.092	.087	.085	.068	.073	.061	.062	.070
.2000	.129	.142	.129	.129	.122	.095	.108	.095	.089	.100
.4000	.281	.286	.258	.243	.225	.196	.208	.182	.183	.188
.6000	.403	.415	.390	.369	.347	.318	.325	.288	.293	.293
.8000	.574	.582	.540	.538	.508	.470	.453	.438	.441	.434
.9000	.719	.752	.712	.697	.675	.643	.614	.602	.598	.575

TABLE XII.- PRESSURE COEFFICIENT C_p* AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(d) Extended double slotted flap configuration; $\delta_f = 56^\circ$; $\frac{y}{b/2} = 0.43$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	.808	1.941	2.028	2.449	3.003	3.435	3.360	3.163	2.708	2.405
.0125	1.010	1.856	2.009	2.412	3.000	3.493	3.366	3.262	2.748	2.414
.0250	1.060	1.697	2.049	2.486	2.985	3.469	3.488	3.270	2.745	2.393
.0500	1.073	1.469	2.122	2.557	3.154	3.627	3.483	3.317	2.760	2.440
.0750	1.073	1.394	2.353	2.619	3.102	3.493	3.346	3.259	2.763	2.419
.1000	1.101	1.300	2.669	2.825	3.063	3.368	3.360	3.329	2.766	2.417
.1500	1.111	1.266	2.329	3.545	3.870	3.903	3.680	3.527	2.780	2.419
.2000	1.136	1.281	1.523	2.822	4.217	4.606	4.009	3.606	2.748	2.408
.2500	1.136	1.275	1.210	1.920	3.286	4.171	3.901	3.463	2.690	2.393
.3000	1.180	1.294	1.240	1.508	2.479	3.593	3.721	3.344	2.673	2.375
.3500	1.186	1.288	1.240	1.326	1.879	2.897	3.302	3.096	2.569	2.357
.4000	1.208	1.306	1.307	1.314	1.626	2.499	3.052	2.963	2.528	2.345
.4500	1.243	1.325	1.338	1.329	1.503	2.222	2.814	2.810	2.455	2.318
.5000	1.256	1.313	1.341	1.326	1.413	1.949	2.512	2.589	2.368	2.292
.5500	1.281	1.347	1.374	1.360	1.404	1.839	2.358	2.469	2.302	2.262
.6000	1.319	1.372	1.389	1.378	1.419	1.754	2.221	2.338	2.226	2.217
.6993	1.328	1.391	1.398	1.409	1.425	1.623	1.965	2.093	2.105	2.149
.7507	1.300	1.394	1.398	1.406	1.413	1.596	1.855	1.973	2.015	2.107
.7704	1.376	1.438	1.453	1.458	1.464	1.629	1.878	1.982	2.032	2.098
.8099	1.429	1.513	1.520	1.520	1.518	1.672	1.869	1.973	2.000	2.077
.8512	1.514	1.569	1.566	1.572	1.569	1.696	1.840	1.930	1.942	2.036
.8911	1.637	1.694	1.702	1.693	1.687	1.818	1.892	1.953	1.913	2.006
.9322	1.868	1.906	1.906	1.902	1.886	2.000	2.017	2.064	1.923	2.000
.9724	2.395	2.416	2.417	2.440	2.413	2.511	2.459	2.422	2.052	2.053
Vane										
.0000	4.404	4.225	4.382	4.194	3.961	3.906	3.651	3.428	2.563	2.467
.0250	6.070	5.881	5.943	5.942	5.928	6.059	5.753	5.332	3.772	3.485
.0500	6.177	5.975	6.062	6.059	6.057	6.180	5.872	5.410	3.728	3.354
.1000	6.427	6.181	6.281	6.314	6.331	6.396	6.064	5.469	3.549	3.125
.1500	6.632	6.363	6.490	6.526	6.551	6.591	6.233	5.582	3.537	3.018
.2000	6.995	6.706	6.855	6.896	6.925	6.974	6.645	5.961	3.717	3.104
.3000	6.512	6.225	6.360	6.391	6.431	6.503	6.116	5.463	3.430	2.872
.4000	5.950	5.691	5.776	5.831	5.910	6.016	5.663	5.072	3.232	2.720
.5000	5.335	5.113	5.159	5.197	5.256	5.411	5.111	4.620	3.041	2.601
.6000	4.638	4.456	4.487	4.523	4.567	4.721	4.456	4.087	2.818	2.464
.7000	4.102	3.947	3.964	3.991	4.020	4.207	3.971	3.705	2.667	2.387
.8000	3.499	3.384	3.390	3.412	3.398	3.627	3.436	3.253	2.435	2.268
.9000	2.947	2.853	2.879	2.862	2.907	3.046	2.901	2.787	2.192	2.077
Flap										
.0000	2.079	2.056	2.164	2.160	2.154	2.230	2.172	2.102	1.577	1.470
.0125	2.738	2.694	2.754	2.754	2.723	2.852	2.753	2.629	2.061	1.967
.0250	3.426	3.391	3.426	3.452	3.416	3.596	3.445	3.329	2.676	2.550
.0500	3.530	3.475	3.490	3.503	3.500	3.684	3.544	3.457	2.844	2.702
.0750	3.483	3.438	3.429	3.452	3.410	3.642	3.512	3.463	2.867	2.720
.1000	3.373	3.269	3.253	3.265	3.238	3.441	3.311	3.320	2.760	2.625
.1500	2.638	2.544	2.523	2.535	2.518	2.681	2.634	2.702	2.346	2.286
.2000	2.237	2.156	2.149	2.139	2.142	2.295	2.209	2.259	2.013	1.979
.4000	1.716	1.678	1.663	1.677	1.702	1.812	1.799	1.869	1.888	1.878
.6000	1.499	1.491	1.480	1.542	1.578	1.657	1.683	1.708	1.734	1.803
.8200	1.237	1.309	1.328	1.412	1.506	1.544	1.552	1.548	1.565	1.681

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	1.104	.800	.800	.926	1.175	1.468	1.587	1.592	1.467	1.393
.0250	1.088	.844	.751	.769	.864	.988	1.044	1.041	.989	.973
.0500	1.076	.875	.764	.729	.726	.751	.744	.732	.719	.714
.0750	1.073	.903	.800	.724	.690	.672	.642	.621	.609	.622
.1000	1.066	.922	.818	.742	.678	.635	.584	.568	.557	.565
.1500	1.063	.938	.830	.748	.678	.602	.541	.525	.516	.524
.2000	1.060	.938	.845	.766	.681	.611	.544	.510	.507	.506
.2500	1.054	.941	.854	.785	.696	.620	.529	.510	.499	.506
.3000	1.029	.938	.854	.775	.702	.620	.541	.510	.504	.509
.3500	1.016	.931	.843	.797	.714	.638	.547	.531	.516	.527
.4000	1.003	.919	.837	.791	.717	.644	.552	.536	.519	.527
.4500	.972	.894	.839	.775	.708	.638	.538	.528	.522	.530
.5000	.940	.869	.824	.754	.696	.632	.541	.525	.510	.527
.5500	.899	.844	.800	.738	.684	.617	.529	.525	.510	.515
.6000	.836	.784	.751	.692	.645	.590	.512	.501	.493	.500
.6995	.738	.706	.681	.637	.602	.559	.494	.484	.470	.467
.7499	.685	.653	.617	.603	.563	.523	.465	.455	.446	.461
.7700	.636	.619	.593	.569	.554	.511	.462	.443	.441	.446
.7744	.647	.622	.593	.569	.539	.514	.459	.443	.438	.452
.8110	.618	.597	.565	.538	.509	.471	.422	.405	.406	.408
.8519	.511	.469	.462	.434	.395	.371	.317	.315	.298	.315
.8919	.461	.428	.426	.403	.367	.359	.302	.297	.284	.301
.9332	.416	.388	.392	.366	.340	.310	.282	.268	.252	.285
.9732	.369	.356	.359	.329	.304	.295	.262	.251	.238	.241
Vane										
.0250	.584	.584	.717	.606	.476	.395	.343	.324	.244	.238
.0500	.120	.141	.225	.163	.105	.082	.061	.052	.052	.048
.1000	.000	.025	.082	.003	.027	.030	.023	.012	.017	.024
.1500	.003	.003	.040	.000	.030	.024	.049	.017	.035	.039
.2000	.006	.019	.030	.003	.045	.052	.035	.038	.038	.054
.3000	.057	.056	.067	.062	.081	.088	.076	.067	.075	.071
.4000	.073	.072	.076	.086	.090	.100	.096	.082	.087	.086
.5000	.095	.103	.106	.108	.123	.125	.113	.108	.096	.095
.6000	.129	.131	.146	.135	.160	.146	.142	.137	.119	.128
.7000	.199	.200	.192	.200	.226	.228	.209	.187	.162	.167
.8000	.359	.344	.347	.354	.373	.386	.352	.335	.278	.280
.9200	1.029	.988	1.000	1.012	1.030	1.064	1.003	.962	.760	.744
Flap										
.0125	.133	.131	.176	.175	.202	.222	.215	.210	.136	.131
.0250	.003	.000	.006	.000	.018	.043	.052	.032	.012	.018
.0500	.009	.006	.018	.009	.006	.036	.003	.017	.020	.024
.0750	.019	.013	.027	.028	.042	.024	.026	.038	.023	.042
.1000	.047	.041	.043	.058	.060	.052	.032	.044	.055	.057
.1500	.076	.078	.082	.077	.090	.073	.064	.064	.064	.077
.2000	.120	.119	.116	.111	.120	.103	.084	.087	.099	.104
.4000	.252	.244	.225	.218	.220	.188	.174	.166	.180	.199
.6000	.337	.328	.334	.317	.328	.295	.267	.268	.273	.283
.8000	.517	.481	.474	.477	.470	.429	.401	.394	.409	.449
.9000	.618	.597	.599	.597	.614	.590	.547	.536	.562	.607

TABLE XII.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(e) Extended double slotted flap configuration; $\delta_f = 55^\circ$; $\frac{y}{b/2} = 0.55$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	.858	1.953	1.902	2.261	2.703	2.866	2.754	2.533	2.409	2.214
.0125	1.123	1.915	1.948	2.318	2.748	2.899	2.784	2.628	2.435	2.217
.0250	1.087	1.922	1.911	2.276	2.696	2.845	2.787	2.649	2.480	2.235
.0500	1.090	1.632	1.920	2.309	2.712	2.881	2.871	2.709	2.474	2.238
.0750	1.116	1.896	1.987	2.396	2.803	2.991	2.830	2.672	2.450	2.217
.1000	1.126	1.840	2.030	2.441	2.882	3.003	2.787	2.689	2.450	2.220
.1500	1.152	1.830	2.316	2.474	2.800	2.884	2.769	2.712	2.450	2.214
.2000	1.168	1.827	2.472	2.703	2.776	2.824	2.792	2.718	2.450	2.220
.2500	1.187	1.843	2.374	3.285	2.991	2.896	2.933	2.784	2.474	2.214
.3000	1.207	1.855	2.043	3.348	3.639	3.365	3.085	2.813	2.462	2.211
.3500	1.226	1.862	1.662	2.955	3.821	3.634	3.167	2.801	2.441	2.209
.4000	1.248	1.881	1.862	2.390	3.542	3.628	3.137	2.758	2.418	2.197
.4500	1.271	1.881	1.248	1.928	3.098	3.479	3.114	2.744	2.418	2.194
.5000	1.310	1.906	1.251	1.631	2.681	3.271	3.032	2.686	2.388	2.185
.6194	1.210	1.274	1.297	1.288	1.800	2.419	2.661	2.536	2.346	2.185
.6844	1.236	1.387	1.310	1.276	1.617	2.134	2.409	2.349	2.240	2.121
.7066	1.268	1.418	1.340	1.303	1.605	2.089	2.366	2.311	2.216	2.115
.7594	1.326	1.450	1.368	1.315	1.553	1.952	2.196	2.162	2.086	2.021
.8101	1.394	1.538	1.460	1.411	1.566	1.914	2.178	2.162	2.116	2.071
.8622	1.519	1.657	1.583	1.558	1.651	1.908	2.117	2.107	2.071	2.044
.9142	1.703	1.852	1.788	1.756	1.815	2.012	2.161	2.127	2.077	2.047
.9652	2.174	2.352	2.306	2.300	2.323	2.419	2.444	2.323	2.184	2.071
Vane										
.0000	3.410	3.274	3.349	3.252	3.125	3.038	2.740	2.418	2.051	1.915
.0250	5.607	5.595	5.594	5.661	5.706	5.657	5.047	4.352	3.509	3.203
.0500	5.746	5.718	5.757	5.844	5.901	5.818	5.155	4.412	3.421	3.077
.1000	6.197	6.123	6.226	6.378	6.448	6.276	5.412	4.510	3.308	2.874
.1500	6.400	6.284	6.435	6.624	6.715	6.503	5.494	4.496	3.208	2.742
.2000	6.507	6.369	6.557	6.781	6.904	6.663	5.564	4.519	3.160	2.678
.3000	6.284	6.164	6.339	6.610	6.776	6.544	5.465	4.438	3.092	2.572
.4000	5.739	5.639	5.803	6.096	6.278	6.080	5.067	4.095	2.879	2.426
.5000	5.278	5.205	5.334	5.676	5.922	5.741	4.760	3.862	2.764	2.367
.6000	4.600	4.557	4.671	4.961	5.217	5.092	4.257	3.487	2.592	2.297
.7000	4.026	4.010	4.082	4.345	4.551	4.446	3.801	3.176	2.459	2.206
.8000	3.494	3.507	3.539	3.766	3.943	3.884	3.316	2.842	2.308	2.109
.9000	2.968	2.978	2.987	3.171	3.332	3.300	2.877	2.502	2.113	1.994
Flap										
.0000	1.484	1.268	1.622	1.784	1.876	1.878	1.719	1.539	1.287	1.159
.0125	2.387	2.736	2.766	2.931	3.055	3.021	2.772	2.473	2.136	1.956
.0250	3.110	3.305	3.337	3.523	3.681	3.637	3.307	2.986	2.589	2.370
.0500	3.268	3.500	3.533	3.748	3.934	3.899	3.599	3.234	2.784	2.531
.0750	3.297	3.563	3.585	3.817	4.013	3.973	3.670	3.306	2.861	2.563
.1000	3.252	3.441	3.488	3.667	3.830	3.788	3.500	3.167	2.722	2.467
.1500	2.852	2.978	2.941	3.054	3.122	3.062	2.892	2.649	2.338	2.156
.2000	2.287	2.387	2.368	2.505	2.590	2.550	2.389	2.219	2.039	1.918
.4000	1.884	1.944	1.895	1.964	1.967	1.920	1.857	1.816	1.846	1.816
.6000	1.697	1.764	1.705	1.715	1.669	1.610	1.594	1.628	1.761	1.789
.8000	1.387	1.475	1.423	1.402	1.359	1.309	1.313	1.369	1.592	1.886

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	1.113	.818	.862	1.004	1.459	1.717	1.796	1.741	1.744	1.887
.0250	1.126	.852	.875	.844	1.002	1.149	1.196	1.187	1.201	1.159
.0500	1.097	.896	.767	.748	.790	.827	.883	.827	.861	.842
.0750	1.065	.884	.761	.715	.714	.711	.690	.686	.710	.695
.1000	1.068	.899	.779	.718	.693	.658	.629	.620	.642	.636
.1500	1.045	.903	.776	.712	.672	.613	.576	.559	.580	.572
.2000	1.036	.912	.791	.721	.672	.604	.544	.542	.568	.537
.2500	1.023	.903	.800	.727	.663	.610	.547	.527	.550	.534
.3000	1.013	.903	.797	.727	.672	.610	.541	.525	.553	.522
.3500	.987	.903	.797	.730	.672	.610	.529	.527	.547	.528
.4000	.965	.881	.788	.730	.672	.604	.538	.525	.547	.522
.4500	.923	.862	.776	.709	.663	.598	.525	.522	.539	.516
.5000	.907	.843	.758	.694	.654	.592	.523	.510	.539	.513
.6195	.777	.758	.687	.652	.605	.559	.503	.476	.500	.478
.6834	.719	.720	.656	.631	.587	.542	.494	.467	.488	.464
.7080	.703	.692	.626	.601	.562	.503	.459	.432	.456	.437
.7137	.690	.689	.626	.604	.562	.512	.465	.438	.465	.437
.7603	.655	.649	.586	.553	.541	.491	.453	.429	.450	.428
.8119	.536	.532	.494	.477	.453	.414	.386	.349	.376	.346
.8643	.490	.497	.460	.444	.423	.387	.354	.343	.355	.334
.9162	.445	.431	.420	.399	.392	.354	.313	.300	.323	.290
.9669	.387	.381	.380	.372	.353	.318	.295	.274	.281	.264
Vane										
.0250	.755	.664	.748	.670	.590	.545	.468	.418	.374	.340
.0500	.206	.170	.215	.177	.140	.131	.114	.104	.127	.097
.1000	.071	.053	.064	.060	.061	.045	.058	.055	.077	.062
.1500	.058	.053	.025	.066	.076	.068	.070	.063	.092	.076
.2000	.077	.072	.071	.084	.097	.086	.085	.084	.109	.100
.3000	.077	.091	.074	.102	.112	.116	.102	.098	.124	.109
.4000	.103	.120	.110	.129	.134	.128	.120	.115	.136	.126
.5000	.135	.138	.132	.156	.152	.158	.135	.130	.157	.126
.6000	.165	.167	.166	.171	.204	.176	.178	.161	.169	.155
.7000	.219	.236	.215	.255	.274	.253	.222	.208	.219	.188
.8000	.374	.390	.383	.414	.438	.420	.368	.323	.328	.314
.9200	1.090	1.079	1.104	1.153	1.201	1.181	1.018	.902	.817	.780
Flap										
.0125	.110	.126	.159	.207	.246	.253	.240	.208	.189	.130
.0250	.003	.009	.015	.036	.058	.060	.056	.086	.068	.050
.0500	.032	.041	.040	.054	.061	.045	.044	.073	.077	.067
.0750	.061	.060	.067	.072	.082	.074	.082	.075	.109	.082
.1000	.090	.091	.080	.093	.088	.083	.088	.081	.101	.094
.1500	.135	.135	.123	.138	.128	.116	.108	.104	.139	.120
.2000	.158	.167	.166	.162	.161	.137	.120	.118	.163	.135
.4000	.329	.321	.291	.276	.264	.235	.202	.225	.260	.238
.6000	.445	.431	.402	.378	.365	.304	.292	.300	.358	.343
.8000	.577	.566	.537	.508	.483	.446	.424	.427	.500	.507
.9000	.641	.670	.632	.601	.587	.539	.520	.536	.636	.651

TABLE XII.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Concluded

(f) Extended double slotted flap configuration; $\delta_f = 55^\circ$; $\frac{y}{b/2} = 0.72$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	1.202	1.963	2.071	2.244	2.432	2.477	2.281	2.194	2.085	2.006
.0125	1.341	1.944	2.025	2.168	2.356	2.420	2.243	2.176	2.088	2.020
.0250	1.319	1.957	2.034	2.174	2.368	2.423	2.281	2.194	2.097	2.017
.0500	1.309	1.994	2.056	2.204	2.368	2.432	2.344	2.222	2.126	2.014
.0750	1.305	2.056	2.100	2.240	2.383	2.459	2.382	2.237	2.129	2.020
.1000	1.265	2.050	2.090	2.207	2.350	2.441	2.358	2.214	2.137	2.020
.1500	1.278	2.139	2.146	2.265	2.380	2.498	2.370	2.243	2.149	2.020
.2000	1.284	2.105	2.212	2.323	2.435	2.520	2.376	2.272	2.143	2.017
.2500	1.297	1.786	2.252	2.341	2.447	2.471	2.379	2.286	2.143	2.017
.3000	1.316	1.421	2.290	2.356	2.414	2.420	2.379	2.269	2.137	2.017
.3500	1.328	1.313	2.473	2.392	2.383	2.393	2.391	2.292	2.129	2.026
.4000	1.341	1.316	2.501	2.447	2.356	2.378	2.388	2.280	2.137	2.026
.4500	1.366	1.334	2.517	2.579	2.377	2.381	2.400	2.280	2.135	2.031
.5000	1.398	1.396	2.501	2.839	2.417	2.420	2.441	2.292	2.149	2.040
.5500	1.379	1.381	2.386	2.924	2.478	2.420	2.423	2.274	2.137	2.037
.6000	1.395	1.406	2.202	2.958	2.721	2.474	2.432	2.257	2.149	2.023
.6500	1.417	1.430	2.047	2.891	2.703	2.523	2.426	2.240	2.126	2.043
.7000	1.442	1.455	1.975	2.824	2.786	2.577	2.418	2.240	2.108	2.028
.7500	1.492	1.508	1.838	2.681	2.765	2.616	2.418	2.231	2.102	2.034
.8000	1.502	1.505	1.748	2.538	2.721	2.640	2.406	2.217	2.097	2.040
.8500	1.527	1.542	1.654	2.359	2.624	2.595	2.344	2.176	2.076	2.037
.9000	1.581	1.598	1.610	2.186	2.544	2.547	2.311	2.147	2.064	2.040

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	.912	.805	.891	1.046	1.237	1.432	1.462	1.491	1.482	1.471
.0250	.943	.777	.804	.866	.997	1.129	1.166	1.208	1.196	1.191
.0500	.962	.799	.766	.754	.812	.877	.905	.939	.921	.937
.0750	.937	.808	.779	.733	.760	.787	.781	.821	.819	.817
.1000	.931	.824	.776	.720	.717	.739	.725	.769	.749	.757
.1500	.940	.830	.782	.720	.708	.685	.660	.676	.667	.680
.2000	.956	.845	.794	.733	.708	.676	.648	.656	.652	.651
.2500	.962	.858	.810	.742	.714	.685	.660	.659	.649	.654
.3000	.975	.870	.841	.772	.731	.706	.675	.685	.673	.674
.3500	.978	.892	.869	.796	.775	.727	.675	.685	.673	.680
.4000	.962	.892	.875	.800	.769	.700	.651	.647	.623	.637
.4500	.972	.892	.875	.812	.757	.697	.654	.630	.617	.611
.5000	.969	.898	.888	.818	.784	.700	.633	.624	.617	.606
.5500	.969	.920	.903	.842	.809	.718	.651	.647	.629	.629
.6000	.981	.935	.931	.866	.827	.739	.684	.670	.652	.654
.6500	.997	.966	.956	.900	.860	.781	.725	.708	.705	.717
.7000	1.022	.994	.991	.930	.915	.823	.778	.795	.775	.794
.7500	1.073	1.037	1.044	1.006	.985	.925	.879	.887	.880	.909
.8000	1.142	1.118	1.149	1.116	1.107	1.048	1.003	1.032	1.026	1.060
.8500	1.268	1.269	1.311	1.289	1.292	1.240	1.201	1.248	1.240	1.280
.9000	1.429	1.468	1.533	1.556	1.584	1.511	1.456	1.500	1.491	1.523
.9500	1.600	1.647	1.710	1.800	1.882	1.823	1.734	1.774	1.768	1.828

TABLE XIII.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE

(a) Extended double slotted flap configuration; $\delta_f = 60^\circ$; $\frac{y}{b/2} = 0$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Fuselage										
.0000	.000	.000	.000	.000	.003	.021	.056	.085	.104	.139
.0500	.727	.826	.886	.942	.991	1.009	1.033	1.056	1.065	1.073
.1000	.815	.917	.960	1.012	1.082	1.051	1.077	1.109	1.116	1.131
.1500	.918	.982	1.021	1.064	1.104	1.105	1.119	1.129	1.142	1.142
.2000	.959	1.027	1.058	1.095	1.122	1.114	1.119	1.135	1.133	1.122
.2500	1.003	1.070	1.089	1.113	1.131	1.130	1.127	1.129	1.142	1.142
.3000	1.038	1.086	1.101	1.116	1.131	1.117	1.121	1.121	1.130	1.136
.3500	1.069	1.107	1.095	1.125	1.138	1.142	1.148	1.171	1.169	1.180
.4000	1.016	1.034	1.052	1.073	1.089	1.096	1.110	1.129	1.139	1.157
.4500	1.016	1.046	1.082	1.107	1.147	1.169	1.187	1.218	1.237	1.278
.5000	1.019	1.064	1.095	1.168	1.229	1.268	1.335	1.368	1.394	1.565
.5500	1.028	1.092	1.138	1.238	1.300	1.346	1.394	1.423	1.500	1.913
.6000	1.069	1.131	1.175	1.278	1.330	1.364	1.379	1.400	1.482	1.963
.6500	1.097	1.162	1.211	1.294	1.342	1.355	1.355	1.338	1.382	1.760
.7000	1.135	1.190	1.233	1.306	1.339	1.358	1.355	1.359	1.361	1.670
.7500	1.179	1.229	1.257	1.318	1.355	1.370	1.379	1.397	1.429	1.600
.8000	1.235	1.281	1.303	1.333	1.358	1.383	1.420	1.453	1.488	1.551
.8500	1.295	1.321	1.319	1.364	1.385	1.401	1.447	1.488	1.539	1.528
.9000	1.279	1.309	1.297	1.321	1.342	1.346	1.391	1.441	1.494	1.473
.9500	1.207	1.253	1.248	1.278	1.281	1.280	1.320	1.356	1.429	1.432
.9940	1.210	1.226	1.215	1.229	1.248	1.238	1.275	1.326	1.411	1.467

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Fuselage										
.0500	.831	.820	.791	.706	.636	.542	.473	.459	.414	.377
.1000	.903	.911	.846	.804	.737	.654	.586	.562	.500	.470
.1500	.975	.960	.908	.881	.813	.726	.666	.644	.589	.551
.2000	1.006	1.003	.986	.917	.875	.786	.725	.706	.648	.620
.2500	1.025	1.031	.994	.972	.911	.846	.772	.753	.710	.675
.3000	1.000	.994	.978	.985	.963	.928	.897	.900	.885	.858
.3500	1.016	1.031	1.009	.991	.957	.870	.817	.806	.763	.710
.4000	.944	.972	.942	.936	.896	.822	.778	.753	.710	.670
.4500	.915	.933	.911	.881	.847	.765	.704	.700	.642	.603
.5000	.884	.890	.862	.835	.768	.687	.615	.612	.565	.522
.5500	.853	.868	.825	.783	.725	.630	.568	.547	.503	.447
.6000	.815	.810	.751	.706	.645	.554	.497	.488	.432	.415
.6500	.975	.972	.932	.899	.853	.780	.728	.729	.689	.673
.9500	.966	1.052	1.083	1.092	1.083	1.060	1.086	1.106	1.121	1.122
.9940	1.050	1.199	1.282	1.330	1.336	1.298	1.314	1.365	1.388	1.415

TABLE XIII.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(b) Extended double slotted flap configuration; $\delta_f = 60^\circ$; $\frac{y}{b/2} = 0.21$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	.874	.900	1.892	3.463	5.495	8.126	10.464	11.172	4.110	2.670
.0125	.928	1.227	1.663	3.448	7.809	11.009	13.459	11.898	3.799	2.453
.0250	.972	1.177	1.517	2.712	3.710	6.511	9.407	9.898	3.808	2.465
.0500	1.016	1.159	1.424	1.631	2.045	2.817	3.838	5.404	3.389	2.421
.0750	1.013	1.140	1.353	1.550	1.840	2.273	2.823	3.680	3.230	2.418
.1000	1.041	1.159	1.350	1.553	1.761	2.090	2.565	3.160	3.145	2.418
.1500	1.060	1.159	1.328	1.475	1.646	1.862	2.228	2.547	2.901	2.383
.2000	1.079	1.174	1.322	1.447	1.586	1.757	2.051	2.299	2.669	2.354
.2500	1.104	1.190	1.322	1.432	1.538	1.685	1.959	2.131	2.404	2.316
.3000	1.120	1.206	1.337	1.407	1.514	1.625	1.876	1.974	2.233	2.260
.3500	1.133	1.206	1.316	1.404	1.471	1.589	1.757	1.840	2.093	2.225
.4000	1.145	1.221	1.319	1.393	1.456	1.577	1.678	1.776	1.988	2.167
.4500	1.171	1.240	1.316	1.404	1.447	1.526	1.648	1.721	1.904	2.158
.5000	1.202	1.274	1.359	1.388	1.441	1.505	1.624	1.689	1.878	2.132
.5500	1.227	1.283	1.350	1.395	1.429	1.462	1.589	1.640	1.829	2.079
.6000	1.230	1.296	1.347	1.395	1.417	1.459	1.589	1.681	1.786	2.030
.6500	1.268	1.318	1.378	1.401	1.420	1.462	1.613	1.692	1.773	2.009
.7000	1.284	1.330	1.384	1.413	1.432	1.463	1.619	1.666	1.718	1.950
.7824	1.306	1.349	1.378	1.413	1.432	1.462	1.494	1.509	1.634	1.874
.8176	1.357	1.396	1.433	1.438	1.435	1.483	1.491	1.500	1.625	1.848
.8323	1.410	1.439	1.477	1.475	1.465	1.535	1.518	1.526	1.645	1.851
.8919	1.477	1.489	1.529	1.516	1.495	1.532	1.482	1.494	1.622	1.798
.9281	1.574	1.558	1.588	1.575	1.532	1.568	1.536	1.552	1.686	1.810
.9504	1.619	1.592	1.607	1.578	1.532	1.553	1.538	1.599	1.721	1.825
.9804	1.719	1.679	1.672	1.631	1.586	1.616	1.645	1.750	1.866	1.927
Vane										
.0000	1.396	1.486	1.384	1.224	1.091	1.090	1.136	1.224	1.302	1.295
.0250	2.343	2.483	2.461	2.311	2.211	2.408	2.604	2.829	2.907	2.772
.0500	3.161	2.978	2.963	2.684	2.746	2.883	3.211	3.526	3.587	3.319
.1000	3.212	2.969	2.910	2.621	2.716	2.883	3.240	3.587	3.628	3.260
.1500	3.130	2.866	2.802	2.525	2.619	2.796	3.175	3.520	3.593	3.242
.2000	3.073	2.772	2.703	2.426	2.547	2.721	3.122	3.480	3.605	3.219
.3000	2.742	2.442	2.359	2.159	2.239	2.402	2.799	3.157	3.407	3.243
.4000	2.467	2.209	2.167	1.997	2.033	2.174	2.503	2.860	3.192	3.094
.5000	2.322	2.124	2.111	1.982	2.009	2.144	2.412	2.738	3.081	2.997
.6000	2.139	2.025	2.009	1.904	1.924	2.039	2.246	2.520	2.791	2.740
.7000	2.085	2.000	1.994	1.898	1.915	2.012	2.207	2.430	2.628	2.576
.8000	2.081	1.984	1.975	1.879	1.900	1.988	2.187	2.407	2.535	2.483
.9000	2.026	1.922	1.916	1.842	1.846	1.934	2.128	2.340	2.401	2.304
Flap										
.0000	1.688	1.620	1.390	1.211	1.160	1.222	1.337	1.442	1.439	1.348
.0125	2.126	2.093	2.028	1.898	1.876	1.964	2.139	2.302	2.323	2.211
.0250	2.335	2.296	2.263	2.177	2.181	2.297	2.515	2.709	2.738	2.632
.0500	2.448	2.386	2.393	2.339	2.323	2.450	2.666	2.887	2.971	2.909
.0750	2.306	2.227	2.245	2.184	2.186	2.282	2.491	2.698	2.802	2.769
.1000	2.139	2.075	2.087	2.034	2.021	2.126	2.311	2.497	2.610	2.617
.1500	1.887	1.813	1.793	1.761	1.740	1.820	1.980	2.119	2.224	2.260
.2000	1.861	1.769	1.731	1.696	1.671	1.763	1.903	2.041	2.122	2.132
.4000	1.884	1.829	1.789	1.739	1.731	1.820	1.962	2.081	2.122	2.117
.6000	1.802	1.776	1.759	1.724	1.693	1.778	1.894	1.977	2.029	2.032
.8000	1.735	1.704	1.700	1.671	1.619	1.676	1.784	1.884	1.910	1.889

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	1.087	.819	.664	.581	.577	.662	.740	.748	.448	.377
.0250	1.032	.847	.690	.555	.456	.390	.370	.355	.256	.194
.0500	1.013	.873	.728	.584	.435	.327	.257	.227	.227	.205
.0750	1.003	.885	.759	.612	.474	.354	.299	.270	.282	.260
.1000	1.010	.900	.780	.646	.508	.402	.352	.331	.343	.325
.1500	1.003	.894	.793	.680	.583	.477	.429	.413	.413	.392
.2000	1.000	.922	.817	.714	.613	.529	.485	.459	.456	.433
.2500	1.025	.935	.845	.749	.650	.580	.524	.503	.488	.471
.3000	1.019	.941	.864	.767	.674	.601	.559	.541	.520	.491
.3500	1.047	.969	.913	.798	.716	.655	.598	.567	.558	.523
.4000	1.047	.987	.916	.836	.746	.679	.624	.593	.576	.547
.4500	1.038	.987	.935	.848	.767	.697	.636	.608	.596	.567
.5000	1.000	.959	.913	.836	.755	.703	.639	.610	.593	.567
.5500	.950	.938	.920	.829	.761	.700	.618	.610	.593	.564
.6000	.921	.903	.870	.798	.734	.673	.615	.587	.584	.558
.6500	.858	.838	.824	.755	.677	.631	.580	.567	.552	.532
.7000	.801	.794	.771	.711	.646	.598	.559	.541	.526	.518
.7815	.697	.683	.666	.612	.539	.517	.479	.465	.459	.450
.8179	.669	.654	.659	.596	.559	.520	.500	.483	.480	.465
.8325	.663	.673	.649	.640	.598	.565	.550	.541	.538	.526
.8366	.681	.698	.703	.680	.646	.625	.595	.599	.613	.585
.8921	.577	.467	.437	.363	.311	.303	.299	.297	.291	.266
.9286	.511	.408	.361	.329	.317	.291	.293	.291	.302	.260
.9518	.480	.386	.350	.295	.275	.252	.257	.250	.262	.234
.9813	.451	.374	.322	.267	.233	.223	.225	.221	.247	.219
Vane										
.0250	.890	.738	.560	.370	.272	.237	.246	.250	.282	.249
.0500	.603	.458	.288	.158	.106	.078	.069	.073	.090	.061
.1000	.432	.305	.186	.106	.076	.072	.068	.073	.070	.044
.1500	.369	.268	.155	.096	.076	.075	.077	.090	.081	.056
.2000	.322	.249	.132	.093	.079	.081	.092	.084	.087	.054
.3000	.263	.224	.134	.106	.094	.096	.095	.108	.105	.067
.4000	.227	.199	.142	.115	.100	.108	.115	.113	.105	.099
.5000	.208	.193	.158	.118	.118	.111	.124	.131	.131	.102
.6000	.194	.190	.170	.127	.124	.135	.142	.154	.157	.132
.7000	.215	.224	.207	.183	.187	.186	.198	.201	.209	.181
.8000	.297	.312	.300	.280	.272	.282	.311	.320	.334	.295
.9200	.713	.720	.728	.690	.686	.713	.787	.837	.860	.801
Flap										
.0125	.498	.287	.155	.078	.057	.066	.080	.093	.099	.061
.0250	.215	.075	.040	.012	.015	.033	.041	.041	.038	.029
.0500	.158	.067	.074	.053	.040	.078	.074	.081	.084	.067
.0750	.148	.090	.096	.081	.100	.084	.107	.099	.099	.083
.1000	.151	.118	.127	.112	.118	.132	.127	.125	.131	.126
.1500	.164	.168	.164	.152	.148	.156	.160	.163	.169	.149
.4000	.385	.386	.390	.363	.359	.348	.349	.343	.348	.319
.6000	.555	.548	.545	.516	.488	.442	.465	.451	.448	.418
.8000	.666	.648	.659	.603	.585	.544	.541	.538	.523	.488
.9000	.770	.754	.734	.711	.674	.649	.633	.651	.605	.602

TABLE XIII.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(c) Extended double slotted flap configuration; $\delta_f = 60^\circ$; $\frac{y}{b/2} = 0.30$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	.920	1.234	2.136	2.897	3.619	4.749	5.044	4.763	3.514	2.522
.0125	.969	1.438	2.260	2.903	3.556	4.358	4.629	4.615	3.493	2.484
.0250	1.009	1.321	2.372	3.006	3.683	4.576	4.759	4.495	3.509	2.510
.0500	1.040	1.249	2.923	3.903	3.997	4.570	4.576	4.544	3.508	2.428
.0750	1.040	1.209	1.817	3.951	5.359	6.113	5.626	4.888	3.525	2.440
.1000	1.053	1.190	1.260	2.494	4.169	5.591	5.641	4.836	3.493	2.490
.1500	1.065	1.199	1.260	1.403	2.205	3.391	4.459	4.250	3.329	2.375
.2000	1.102	1.212	1.316	1.384	1.643	2.316	3.329	3.577	3.147	2.345
.2500	1.115	1.212	1.328	1.367	1.504	1.884	2.712	3.064	2.930	2.369
.3000	1.139	1.234	1.337	1.379	1.465	1.746	2.370	2.737	2.792	2.328
.3500	1.142	1.224	1.331	1.370	1.441	1.633	2.097	2.419	2.583	2.316
.4000	1.161	1.246	1.344	1.391	1.441	1.603	1.968	2.256	2.446	2.286
.4500	1.201	1.271	1.362	1.412	1.453	1.600	1.918	2.160	2.352	2.263
.5000	1.220	1.283	1.365	1.406	1.456	1.576	1.832	2.038	2.209	2.221
.5500	1.229	1.293	1.365	1.406	1.438	1.561	1.776	1.962	2.109	2.195
.6000	1.263	1.315	1.390	1.412	1.450	1.567	1.753	1.903	2.047	2.174
.6500	1.294	1.346	1.409	1.427	1.456	1.561	1.726	1.851	1.983	2.127
.7516	1.263	1.327	1.406	1.418	1.450	1.504	1.612	1.729	1.836	1.988
.7940	1.300	1.386	1.449	1.442	1.474	1.525	1.641	1.734	1.801	1.938
.8099	1.282	1.389	1.424	1.448	1.480	1.507	1.618	1.711	1.760	1.912
.8429	1.387	1.458	1.517	1.512	1.538	1.585	1.703	1.772	1.795	1.903
.8767	1.381	1.489	1.545	1.548	1.568	1.603	1.685	1.734	1.736	1.841
.9108	1.505	1.595	1.644	1.630	1.671	1.681	1.779	1.793	1.763	1.841
.9437	1.681	1.779	1.830	1.797	1.819	1.851	1.970	1.938	1.842	1.873
.9768	2.124	2.227	2.257	2.206	2.220	2.269	2.412	2.347	2.153	2.038
Vane										
.0000	5.012	4.875	4.564	4.172	3.948	3.949	4.147	4.055	3.479	2.923
.0250	6.204	5.869	5.793	5.445	5.278	5.677	6.141	6.101	5.048	3.974
.0500	6.272	6.040	5.923	5.660	5.423	5.833	6.226	6.206	5.133	3.986
.1000	6.254	6.028	5.892	5.621	5.322	5.794	6.198	6.206	5.062	3.797
.1500	6.173	5.965	5.808	5.521	5.205	5.701	6.047	6.177	5.021	3.738
.2000	6.074	5.925	5.749	5.442	5.118	5.606	5.967	6.183	5.103	3.820
.3000	5.573	5.451	5.269	4.960	4.704	5.161	5.506	5.789	4.895	3.749
.4000	4.947	4.794	4.588	4.290	4.103	4.507	4.850	5.230	4.484	3.478
.5000	4.443	4.221	4.022	3.733	3.637	4.003	4.359	4.725	4.089	3.224
.6000	3.827	3.626	3.440	3.188	3.154	3.454	3.829	4.154	3.684	3.012
.7000	3.288	3.087	2.950	2.736	2.764	2.988	3.335	3.641	3.285	2.761
.8000	2.768	2.610	2.542	2.351	2.411	2.567	2.947	3.145	2.898	2.493
.9000	2.402	2.314	2.257	2.091	2.133	2.254	2.576	2.728	2.569	2.313
Flap										
.0000	1.582	1.620	1.613	1.376	1.287	1.940	1.438	1.530	1.461	1.330
.0125	2.186	2.249	2.226	2.031	1.988	2.081	2.253	2.396	2.311	2.159
.0250	2.616	2.679	2.650	2.434	2.426	2.549	2.797	2.994	2.915	2.729
.0500	2.625	2.698	2.641	2.446	2.501	2.630	2.944	3.166	3.112	2.941
.0750	2.610	2.648	2.585	2.412	2.480	2.618	2.944	3.204	3.176	2.977
.1000	2.820	2.489	2.430	2.285	2.344	2.475	2.791	3.037	2.998	2.817
.1500	2.077	2.062	2.019	1.931	2.018	2.098	2.288	2.536	2.593	2.490
.2000	1.882	1.807	1.783	1.727	1.746	1.839	2.018	2.201	2.200	2.112
.4000	1.690	1.688	1.694	1.670	1.646	1.716	1.823	1.936	1.971	1.986
.6000	1.669	1.707	1.712	1.694	1.680	1.728	1.835	1.924	1.904	1.891
.8000	1.644	1.695	1.702	1.673	1.646	1.687	1.776	1.851	1.780	1.749

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	1.028	.773	.743	.616	1.000	1.281	1.476	1.463	1.334	1.112
.0250	1.046	.822	.728	.694	.737	.857	.929	.936	.886	.770
.0500	1.025	.850	.743	.654	.616	.603	.603	.589	.572	.537
.0750	1.015	.885	.771	.670	.610	.552	.512	.498	.481	.437
.1000	1.019	.885	.783	.697	.607	.537	.479	.452	.440	.428
.1500	1.028	.910	.820	.715	.637	.546	.471	.440	.434	.428
.2000	1.025	.925	.851	.739	.653	.558	.485	.443	.431	.428
.2500	1.044	.938	.876	.770	.683	.585	.494	.455	.449	.440
.3000	1.037	.944	.876	.779	.698	.603	.521	.484	.463	.454
.3500	1.028	.953	.898	.806	.725	.624	.541	.507	.487	.487
.4000	1.022	.950	.898	.812	.719	.636	.559	.525	.513	.496
.4500	.997	.941	.895	.809	.737	.654	.568	.536	.528	.513
.5000	.947	.913	.876	.794	.728	.666	.582	.545	.540	.528
.5500	.910	.885	.848	.776	.713	.651	.579	.545	.540	.528
.6000	.839	.825	.802	.739	.677	.606	.553	.528	.525	.516
.6500	.805	.769	.762	.703	.640	.585	.535	.510	.504	.499
.7517	.644	.638	.635	.600	.562	.513	.476	.455	.449	.448
.7934	.613	.586	.563	.533	.508	.472	.438	.431	.419	.425
.8094	.616	.554	.545	.506	.477	.460	.424	.414	.416	.407
.8137	.557	.548	.545	.503	.483	.460	.426	.408	.411	.404
.8439	.601	.542	.529	.491	.471	.433	.415	.399	.393	.392
.8772	.573	.495	.455	.400	.369	.343	.300	.292	.287	.280
.9117	.529	.467	.418	.358	.326	.307	.282	.268	.284	.268
.9448	.480	.430	.387	.321	.296	.269	.250	.242	.229	.221
.9781	.424	.396	.353	.297	.269	.242	.229	.222	.208	.209
Vane										
.0250	1.808	1.841	1.647	1.315	1.097	.976	.947	.901	.780	.643
.0500	.632	.707	.622	.412	.290	.221	.176	.166	.138	.100
.1000	.204	.240	.223	.109	.066	.039	.012	.006	.006	.000
.1500	.108	.115	.142	.067	.045	.015	.012	.009	.018	.000
.2000	.071	.084	.099	.055	.027	.027	.015	.017	.032	.015
.3000	.068	.044	.071	.036	.039	.042	.029	.038	.032	.038
.4000	.071	.057	.074	.048	.039	.057	.056	.055	.056	.050
.5000	.096	.050	.074	.073	.066	.075	.076	.082	.076	.045
.6000	.105	.072	.090	.094	.097	.093	.097	.108	.109	.089
.7000	.167	.128	.142	.130	.142	.161	.153	.160	.153	.142
.8000	.300	.249	.240	.251	.269	.266	.297	.300	.296	.268
.9200	.839	.782	.786	.730	.731	.776	.838	.886	.848	.770
Flap										
.0125	.260	.227	.241	.148	.124	.146	.165	.178	.164	.133
.0250	.053	.000	.025	.000	.000	.006	.000	.009	.021	.000
.0500	.019	.000	.003	.001	.000	.012	.000	.006	.000	.006
.0750	.022	.000	.012	.018	.018	.015	.015	.009	.018	.003
.1000	.025	.000	.022	.024	.024	.036	.018	.023	.023	.015
.1500	.042	.034	.050	.048	.048	.069	.050	.044	.050	.041
.2000	.084	.062	.080	.088	.085	.084	.071	.070	.059	.059
.4000	.220	.196	.201	.188	.187	.170	.153	.152	.147	.150
.6000	.328	.330	.344	.324	.305	.287	.271	.242	.258	.248
.8000	.493	.517	.502	.494	.456	.442	.418	.402	.381	.384
.9000	.647	.692	.687	.676	.637	.624	.597	.577	.540	.531

TABLE XIII.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(d) Extended double slotted flap configuration; $\delta_f = 60°$; $\frac{y}{b/2} = 0.43$

Upper surface

x/c	C_p for -									
	$\alpha = -4°$	$\alpha = 0°$	$\alpha = 4°$	$\alpha = 8°$	$\alpha = 12°$	$\alpha = 16°$	$\alpha = 20°$	$\alpha = 22°$	$\alpha = 24°$	$\alpha = 26°$
Wing										
.0000	.782	1.913	2.034	2.519	3.027	3.345	3.211	2.951	2.622	2.368
.0125	1.006	1.788	2.015	2.500	3.012	3.360	3.234	3.067	2.657	2.383
.0250	1.057	1.785	2.059	2.541	3.018	3.360	3.273	3.099	2.645	2.363
.0500	1.066	1.436	2.068	2.668	3.142	3.492	3.323	3.119	2.669	2.374
.0750	1.082	1.308	2.294	2.687	3.115	3.351	3.246	3.142	2.660	2.374
.1000	1.098	1.290	2.619	2.876	3.090	3.300	3.213	3.166	2.651	2.357
.1500	1.117	1.268	2.201	3.730	3.921	3.797	3.500	3.241	2.677	2.344
.2000	1.136	1.280	1.514	2.954	4.238	4.384	3.719	3.224	2.842	2.340
.2500	1.139	1.268	1.223	1.973	3.302	4.039	3.895	3.108	2.602	2.348
.3000	1.164	1.305	1.260	1.558	2.501	3.517	3.435	3.023	2.567	2.334
.3500	1.189	1.299	1.263	1.342	1.906	2.889	3.101	2.826	2.500	2.313
.4000	1.215	1.324	1.310	1.320	1.631	2.508	2.900	2.727	2.451	2.295
.4500	1.243	1.336	1.359	1.339	1.511	2.237	2.725	2.619	2.398	2.275
.5000	1.242	1.330	1.341	1.342	1.406	1.958	2.462	2.451	2.308	2.243
.5500	1.281	1.361	1.387	1.373	1.408	1.844	2.338	2.355	2.265	2.216
.6000	1.316	1.389	1.409	1.398	1.411	1.742	2.207	2.247	2.192	2.193
.6993	1.319	1.377	1.409	1.413	1.408	1.604	1.938	2.055	2.067	2.114
.7507	1.309	1.389	1.402	1.416	1.405	1.562	1.820	1.936	1.983	2.073
.7704	1.388	1.449	1.438	1.434	1.447	1.607	1.852	1.962	2.003	2.067
.8099	1.448	1.511	1.526	1.525	1.517	1.634	1.838	1.948	1.974	2.041
.8512	1.524	1.576	1.591	1.587	1.556	1.655	1.826	1.890	1.930	2.018
.8911	1.673	1.718	1.734	1.718	1.695	1.760	1.873	1.919	1.904	1.985
.9322	1.912	1.938	1.941	1.941	1.891	1.952	2.000	2.006	1.919	1.988
.9724	2.505	2.508	2.536	2.531	2.477	2.480	2.444	2.334	2.058	2.041
Vane										
.0000	5.890	5.781	5.960	5.700	5.362	5.125	4.838	4.273	3.244	3.061
.0250	7.539	7.298	7.440	7.408	7.229	7.033	6.702	5.846	4.169	3.787
.0500	7.452	7.183	7.344	7.315	7.172	6.943	6.628	5.718	3.545	3.567
.1000	7.389	7.087	7.283	7.271	7.136	6.859	6.489	5.491	3.643	3.231
.1500	7.389	7.093	7.283	7.249	7.139	6.853	6.468	5.442	3.500	3.038
.2000	7.366	7.245	7.438	7.445	7.202	7.033	6.720	5.677	3.694	3.114
.3000	6.935	6.616	6.774	6.768	6.646	6.432	6.087	5.160	3.394	2.887
.4000	6.247	5.922	6.016	6.066	5.997	5.877	5.533	4.706	3.108	2.687
.5000	5.439	5.211	5.288	5.283	5.163	5.087	4.888	4.262	2.910	2.570
.6000	4.714	4.501	4.545	4.538	4.447	4.426	4.243	3.782	2.712	2.442
.7000	4.124	3.950	3.966	3.960	3.879	3.919	3.776	3.433	2.576	2.354
.8000	3.508	3.358	3.375	3.358	3.293	3.348	3.255	3.032	2.366	2.225
.9000	2.953	2.872	2.889	2.876	2.797	2.847	2.779	2.645	2.134	2.050
Flap										
.0000	2.322	2.286	2.387	2.367	2.296	2.303	2.246	2.110	1.689	1.570
.0125	2.832	2.804	2.855	2.854	2.767	2.769	2.707	2.567	2.116	1.985
.0250	3.496	3.423	3.484	3.491	3.384	3.399	3.332	3.186	2.640	2.520
.0500	3.553	3.467	3.481	3.465	3.356	3.405	3.376	3.291	2.805	2.652
.0750	3.464	3.383	3.368	3.358	3.251	3.357	3.335	3.270	2.817	2.638
.1000	3.347	3.205	3.198	3.165	3.057	3.186	3.178	3.151	2.693	2.564
.1500	2.590	2.480	2.443	2.426	2.341	2.480	2.533	2.593	2.320	2.257
.2000	2.199	2.115	2.102	2.073	2.024	2.135	2.160	2.193	1.980	1.971
.4000	1.701	1.682	1.659	1.674	1.674	1.766	1.820	1.864	1.829	1.851
.6000	1.518	1.520	1.529	1.600	1.634	1.700	1.743	1.744	1.744	1.789
.8000	1.278	1.347	1.409	1.513	1.589	1.622	1.630	1.564	1.584	1.670

Lower surface

x/c	C_p for -									
	$\alpha = -4°$	$\alpha = 0°$	$\alpha = 4°$	$\alpha = 8°$	$\alpha = 12°$	$\alpha = 16°$	$\alpha = 20°$	$\alpha = 22°$	$\alpha = 24°$	$\alpha = 26°$
Wing										
.0125	1.066	.807	.799	.932	1.190	1.429	1.542	1.803	1.416	1.243
.0250	1.066	.847	.789	.783	.858	.964	1.021	.994	.962	.962
.0500	1.066	.885	.765	.730	.715	.724	.728	.709	.698	.711
.0750	1.057	.910	.793	.724	.680	.646	.636	.625	.605	.617
.1000	1.057	.922	.805	.733	.665	.613	.589	.570	.549	.558
.1500	1.054	.938	.827	.745	.671	.601	.553	.529	.515	.513
.2000	1.041	.941	.839	.755	.683	.601	.536	.520	.497	.497
.2500	1.038	.935	.854	.773	.680	.604	.541	.526	.494	.500
.3000	1.013	.935	.851	.767	.683	.607	.544	.515	.494	.497
.3500	1.010	.928	.858	.786	.704	.622	.556	.535	.503	.509
.4000	.984	.913	.845	.780	.698	.622	.565	.532	.517	.512
.4500	.952	.885	.833	.767	.685	.619	.545	.538	.506	.518
.5000	.905	.857	.808	.749	.674	.601	.550	.529	.500	.515
.5500	.871	.829	.774	.730	.659	.589	.541	.532	.500	.500
.6000	.808	.760	.721	.674	.625	.565	.512	.494	.471	.480
.6995	.710	.673	.641	.609	.565	.523	.494	.477	.448	.453
.7499	.650	.601	.585	.556	.514	.483	.447	.433	.419	.424
.7700	.625	.579	.548	.531	.498	.465	.444	.427	.413	.418
.7744	.625	.579	.542	.537	.498	.465	.444	.433	.410	.424
.8110	.596	.554	.529	.509	.462	.435	.405	.392	.375	.377
.8919	.473	.442	.430	.398	.349	.330	.302	.285	.262	.287
.8919	.429	.392	.375	.363	.329	.297	.287	.270	.244	.260
.9332	.375	.343	.337	.326	.296	.270	.249	.247	.215	.237
.9738	.328	.318	.313	.301	.278	.255	.246	.232	.195	.208
Vane										
.0250	.839	.910	1.043	.879	.725	.643	.577	.509	.378	.348
.0500	.192	.262	.334	.252	.205	.150	.133	.119	.076	.079
.1000	.019	.059	.098	.065	.045	.021	.047	.029	.000	.023
.1500	.000	.000	.009	.000	.027	.024	.059	.015	.006	.018
.2000	.000	.000	.003	.000	.033	.036	.033	.023	.009	.026
.3000	.000	.025	.006	.034	.060	.054	.056	.044	.038	.056
.4000	.057	.044	.034	.047	.076	.069	.068	.076	.058	.053
.5000	.088	.062	.065	.071	.103	.090	.086	.090	.073	.067
.6000	.107	.103	.102	.121	.127	.117	.130	.119	.084	.105
.7000	.170	.159	.152	.189	.196	.192	.189	.177	.137	.143
.8000	.334	.327	.310	.339	.353	.345	.337	.331	.253	.257
.9200	1.047	1.015	1.029	1.025	1.012	1.003	.976	.922	.759	.725
Flap										
.0125	.215	.218	.245	.261	.266	.270	.275	.264	.163	.167
.0250	.000	.000	.000	.016	.018	.045	.044	.044	.009	.015
.0500	.006	.000	.000	.000	.027	.030	.015	.015	.012	.015
.0750	.000	.000	.000	.000	.027	.021	.021	.032	.000	.000
.1000	.013	.012	.012	.019	.018	.027	.044	.041	.023	.018
.1500	.044	.047	.040	.053	.057	.036	.044	.049	.041	.028
.2000	.082	.072	.065	.062	.088	.078	.071	.061	.052	.079
.4000	.196	.187	.176	.177	.175	.156	.145	.148	.140	.143
.6000	.312	.287	.285	.276	.272	.258	.243	.238	.221	.231
.8000	.464	.452	.430	.441	.429	.402	.376	.366	.363	.389
.9000	.574	.570	.560	.578	.586	.559	.530	.503	.500	.544

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TABLE XIII.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(a) Extended double slotted flap configuration; $\delta_f = 60^\circ$; $\frac{y}{b/2} = 0.55$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 28^\circ$
Wing										
.0000	.898	1.088	1.278	2.324	2.713	2.863	2.659	2.486	2.352	2.245
.0125	1.156	1.275	2.028	2.279	2.734	2.895	2.723	2.571	2.387	2.284
.0250	1.084	1.880	1.991	2.334	2.655	2.854	2.732	2.568	2.390	2.263
.0500	1.096	1.642	2.022	2.406	2.704	2.910	2.806	2.621	2.399	2.266
.0750	1.118	1.433	2.087	2.472	2.782	2.988	2.753	2.624	2.382	2.248
.1000	1.130	1.371	2.136	2.524	2.876	2.997	2.744	2.647	2.387	2.242
.1500	1.146	1.352	2.409	2.542	2.807	2.857	2.756	2.670	2.405	2.239
.2000	1.170	1.339	2.601	2.769	2.770	2.803	2.776	2.670	2.399	2.236
.2500	1.192	1.333	2.545	3.397	2.988	2.866	2.879	2.702	2.420	2.248
.3000	1.204	1.349	2.167	3.500	3.643	3.254	2.976	2.693	2.411	2.239
.3500	1.226	1.364	1.765	3.054	3.831	3.501	3.003	2.670	2.396	2.230
.4000	1.251	1.377	1.437	2.469	3.544	3.510	2.959	2.618	2.352	2.224
.4500	1.266	1.377	1.294	1.997	3.090	3.403	2.923	2.583	2.335	2.224
.5000	1.300	1.414	1.288	1.694	2.665	3.257	2.844	2.536	2.317	2.213
.6194	1.189	1.371	1.344	1.524	1.797	2.543	2.579	2.405	2.261	2.193
.6844	1.220	1.380	1.356	1.291	1.607	2.254	2.353	2.253	2.156	2.136
.7066	1.257	1.424	1.396	1.312	1.592	2.215	2.323	2.227	2.141	2.127
.7594	1.313	1.445	1.424	1.321	1.541	2.054	2.165	2.093	2.015	2.021
.8101	1.396	1.561	1.529	1.415	1.574	2.018	2.159	2.114	2.062	2.077
.8622	1.517	1.693	1.659	1.548	1.668	2.009	2.120	2.078	2.021	2.056
.9142	1.724	1.909	1.904	1.794	1.849	2.107	2.156	2.093	2.006	2.050
.9652	2.233	2.470	2.520	2.442	2.414	2.528	2.420	2.280	2.118	2.100
Vane										
.0000	4.567	4.704	4.848	4.609	4.377	4.131	3.541	3.055	2.838	2.449
.0250	7.025	7.093	7.359	7.290	7.145	6.773	5.685	4.749	3.781	3.531
.0500	7.006	7.009	7.319	7.284	7.145	6.752	5.676	4.699	3.625	3.307
.1000	7.207	7.183	7.560	7.593	7.436	6.922	5.629	4.518	3.541	3.015
.1500	7.245	7.189	7.622	7.663	7.586	6.970	5.576	4.428	3.212	2.826
.2000	7.238	7.168	7.622	7.696	7.616	7.003	5.553	4.390	3.171	2.761
.3000	6.805	6.760	7.189	7.350	7.281	6.716	5.347	4.209	3.030	2.649
.4000	6.193	6.137	6.551	6.721	6.719	6.170	4.867	3.827	2.801	2.490
.5000	5.492	5.489	5.824	6.033	6.136	5.677	4.517	3.591	2.692	2.422
.6000	4.817	4.816	5.105	5.269	5.308	4.961	4.006	3.259	2.531	2.325
.7000	4.186	4.221	4.440	4.597	4.634	4.364	3.600	2.985	2.387	2.239
.8000	3.601	3.654	3.817	3.948	3.991	3.785	3.185	2.699	2.229	2.133
.9000	3.068	3.112	3.232	3.318	3.350	3.197	2.747	2.393	2.059	2.018
Flap										
.0000	1.690	1.853	2.006	2.088	2.105	2.033	1.788	1.565	1.329	1.260
.0125	2.700	2.928	3.090	3.191	3.193	3.084	2.723	2.393	2.085	2.003
.0250	3.178	3.476	3.663	3.784	3.797	3.672	3.259	2.863	2.511	2.425
.0500	3.325	3.641	3.817	3.960	3.985	3.866	3.464	3.058	2.675	2.564
.0750	3.331	3.682	3.864	3.997	4.018	3.898	3.514	3.125	2.731	2.584
.1000	3.294	3.551	3.697	3.812	3.809	3.701	3.338	2.994	2.422	2.505
.1500	2.805	2.990	3.016	3.003	2.952	2.895	2.759	2.536	2.253	2.189
.2000	2.328	2.435	2.539	2.588	2.559	2.489	2.315	2.154	1.989	1.930
.4000	1.920	1.997	2.025	1.991	1.936	1.881	1.847	1.834	1.816	1.844
.6000	1.709	1.804	1.786	1.724	1.640	1.606	1.641	1.685	1.757	1.814
.8000	1.362	1.483	1.480	1.394	1.351	1.334	1.365	1.434	1.601	1.496

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 28^\circ$
Wing										
.0125	1.040	.822	.901	1.139	1.447	1.724	1.768	1.734	1.719	1.682
.0250	1.059	.884	.808	.879	.991	1.158	1.188	1.189	1.185	1.192
.0500	1.050	.885	.799	.767	.788	.833	.841	.848	.839	.853
.0750	1.022	.869	.774	.724	.701	.719	.697	.697	.704	.714
.1000	1.019	.885	.786	.724	.680	.666	.641	.641	.639	.643
.1500	1.003	.891	.793	.718	.653	.615	.576	.574	.563	.567
.2000	.983	.897	.802	.724	.650	.612	.556	.542	.543	.534
.2500	.975	.894	.811	.733	.656	.609	.553	.539	.534	.534
.3000	.957	.878	.811	.733	.659	.606	.550	.533	.534	.531
.3500	.935	.882	.811	.742	.656	.606	.541	.539	.531	.531
.4000	.916	.863	.808	.730	.650	.597	.541	.528	.531	.519
.4500	.882	.841	.774	.718	.650	.597	.541	.528	.525	.519
.5000	.858	.816	.765	.706	.634	.588	.532	.513	.504	.507
.6195	.718	.723	.681	.639	.584	.540	.485	.475	.466	.466
.6834	.663	.682	.650	.600	.562	.519	.479	.461	.460	.457
.7080	.650	.637	.625	.573	.523	.478	.441	.420	.425	.416
.7137	.635	.654	.619	.576	.523	.490	.444	.434	.431	.434
.7603	.616	.607	.505	.545	.498	.478	.429	.423	.414	.413
.8119	.493	.511	.498	.470	.420	.403	.365	.350	.355	.345
.8643	.453	.473	.458	.433	.390	.379	.329	.326	.323	.319
.9162	.415	.430	.433	.403	.353	.328	.294	.286	.279	.280
.9669	.359	.386	.399	.370	.341	.310	.282	.268	.255	.254
Vane										
.0250	1.034	1.134	1.195	1.057	.943	.839	.700	.615	.534	.499
.0500	.276	.343	.372	.318	.263	.239	.194	.187	.170	.156
.1000	.050	.081	.111	.094	.079	.075	.062	.058	.070	.068
.1500	.032	.040	.062	.067	.063	.075	.062	.064	.079	.068
.2000	.031	.053	.062	.073	.076	.087	.065	.076	.088	.091
.3000	.053	.062	.080	.085	.094	.107	.091	.093	.100	.115
.4000	.080	.093	.108	.115	.115	.122	.109	.117	.117	.106
.5000	.108	.125	.133	.142	.142	.143	.124	.125	.132	.130
.6000	.146	.153	.173	.165	.172	.180	.159	.152	.153	.153
.7000	.217	.209	.238	.261	.248	.242	.215	.192	.194	.186
.8000	.381	.380	.409	.427	.417	.415	.350	.321	.293	.301
.9200	1.127	1.140	1.201	1.233	1.233	1.182	.997	.892	.810	.788
Flap										
.0125	.211	.246	.303	.342	.341	.349	.297	.262	.220	.207
.0250	.012	.012	.053	.088	.079	.093	.100	.073	.082	.071
.0500	.009	.019	.050	.058	.048	.066	.059	.058	.079	.059
.0750	.043	.054	.077	.076	.069	.066	.068	.070	.082	.074
.1000	.068	.069	.084	.094	.076	.075	.074	.065	.100	.089
.1500	.090	.106	.105	.112	.100	.101	.091	.099	.109	.118
.2000	.118	.134	.149	.136	.124	.122	.112	.120	.132	.124
.4000	.237	.268	.263	.251	.214	.215	.191	.198	.220	.215
.6000	.372	.386	.362	.327	.293	.293	.259	.283	.296	.301
.8000	.489	.505	.492	.451	.417	.415	.382	.405	.446	.463
.9000	.585	.626	.591	.551	.523	.525	.494	.519	.592	.658

TABLE XIII.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Concluded

(f) Extended double slotted flap configuration; $\delta_f = 60^\circ$; $\frac{y}{b/2} = 0.72$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	1.276	1.972	2.064	2.326	2.489	2.443	2.207	2.176	2.033	2.012
.0125	1.317	1.954	2.015	2.248	2.419	2.382	2.178	2.159	2.048	2.029
.0250	1.257	1.969	2.033	2.263	2.419	2.389	2.216	2.194	2.062	2.029
.0500	1.229	2.012	2.064	2.294	2.428	2.416	2.278	2.255	2.068	2.035
.0750	1.229	2.058	2.092	2.324	2.443	2.440	2.299	2.253	2.083	2.035
.1000	1.169	2.055	2.089	2.294	2.419	2.431	2.278	2.244	2.098	2.035
.1500	1.194	2.159	2.150	2.361	2.443	2.479	2.296	2.268	2.092	2.029
.2000	1.235	2.190	2.205	2.410	2.501	2.488	2.311	2.276	2.086	2.032
.2500	1.251	1.905	2.260	2.428	2.505	2.497	2.320	2.282	2.086	2.035
.3000	1.273	1.817	2.346	2.443	2.471	2.404	2.308	2.270	2.083	2.032
.3500	1.301	1.370	2.343	2.459	2.431	2.382	2.314	2.270	2.077	2.041
.4000	1.304	1.327	2.628	2.526	2.401	2.379	2.332	2.279	2.092	2.044
.4500	1.354	1.327	2.665	2.636	2.422	2.382	2.338	2.276	2.083	2.047
.5000	1.343	1.388	2.702	2.936	2.462	2.428	2.367	2.300	2.104	2.050
.5500	1.342	1.388	2.574	3.030	2.517	2.440	2.346	2.276	2.098	2.050
.6000	1.339	1.419	2.482	3.104	2.664	2.482	2.358	2.291	2.116	2.085
.6500	1.367	1.427	2.242	3.037	2.758	2.515	2.338	2.259	2.089	2.052
.7000	1.408	1.459	2.104	2.954	2.792	2.554	2.323	2.238	2.074	2.052
.7500	1.455	1.538	1.951	2.801	2.823	2.584	2.314	2.223	2.071	2.047
.8000	1.473	1.523	1.834	2.651	2.810	2.590	2.314	2.206	2.071	2.056
.8500	1.517	1.546	1.696	2.459	2.697	2.533	2.237	2.159	2.042	2.055
.9000	1.561	1.633	1.650	2.303	2.627	2.500	2.219	2.159	2.051	2.052

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	.843	.867	.905	1.095	1.297	1.434	1.453	1.482	1.465	1.487
.0250	.875	.777	.791	.905	1.057	1.130	1.154	1.194	1.181	1.220
.0500	.884	.786	.742	.780	.838	.870	.894	.932	.926	.948
.0750	.865	.804	.751	.749	.765	.771	.787	.818	.814	.829
.1000	.862	.813	.751	.737	.731	.726	.728	.753	.749	.771
.1500	.862	.820	.748	.723	.706	.672	.666	.688	.675	.684
.2000	.884	.841	.770	.737	.706	.666	.648	.673	.657	.667
.2500	.903	.850	.782	.758	.709	.672	.645	.676	.654	.664
.3000	.912	.862	.813	.789	.753	.696	.675	.691	.678	.693
.3500	.928	.881	.834	.820	.777	.711	.678	.694	.678	.693
.4000	.909	.881	.834	.813	.768	.690	.645	.656	.636	.641
.4500	.909	.893	.853	.813	.755	.678	.633	.641	.615	.615
.5000	.909	.899	.862	.829	.761	.684	.627	.635	.615	.608
.5500	.903	.917	.877	.850	.786	.699	.648	.653	.636	.632
.6000	.922	.936	.905	.875	.813	.723	.678	.685	.651	.658
.6500	.941	.960	.935	.905	.853	.759	.719	.735	.716	.725
.7000	.972	.991	.978	.951	.908	.828	.781	.806	.793	.820
.7500	1.022	1.046	1.043	1.040	.991	.913	.882	.912	.911	.939
.8000	1.122	1.144	1.190	1.162	1.125	1.054	1.030	1.071	1.071	1.113
.8500	1.229	1.315	1.346	1.379	1.339	1.256	1.243	1.273	1.302	1.354
.9000	1.379	1.514	1.570	1.667	1.667	1.548	1.503	1.559	1.580	1.638
.9500	1.561	1.697	1.751	1.884	1.951	1.870	1.835	1.873	1.906	1.986

TABLE XIV.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE

(a) Extended double slotted flap configuration; $\delta_f = 65^\circ$; $\frac{y}{b/2} = 0$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Fuselage										
.0000	.000	.000	.000	.000	.006	.021	.072	.102	.106	.130
.0500	.713	.836	.889	.933	.964	1.012	1.032	1.056	1.065	1.087
.1000	.786	.918	.960	1.000	1.039	1.060	1.070	1.097	1.106	1.115
.1500	.896	.997	1.031	1.051	1.081	1.090	1.096	1.117	1.117	1.110
.2000	.936	1.028	1.065	1.075	1.099	1.101	1.099	1.117	1.103	1.105
.2500	.982	1.073	1.095	1.094	1.114	1.131	1.110	1.126	1.117	1.119
.3000	1.040	1.088	1.098	1.097	1.108	1.119	1.093	1.117	1.109	1.119
.3500	1.070	1.094	1.108	1.100	1.130	1.143	1.136	1.161	1.159	1.162
.4000	1.009	1.043	1.055	1.051	1.078	1.101	1.093	1.111	1.123	1.145
.4500	1.006	1.046	1.071	1.085	1.136	1.167	1.177	1.213	1.220	1.247
.5000	1.015	1.073	1.108	1.151	1.214	1.275	1.302	1.360	1.402	1.551
.5500	1.034	1.088	1.157	1.220	1.289	1.358	1.386	1.421	1.566	1.887
.6000	1.070	1.128	1.209	1.266	1.331	1.385	1.368	1.386	1.598	1.896
.6500	1.116	1.173	1.249	1.281	1.331	1.264	1.342	1.348	1.464	1.734
.7000	1.158	1.207	1.265	1.293	1.334	1.364	1.331	1.345	1.402	1.670
.7500	1.187	1.246	1.295	1.305	1.346	1.379	1.371	1.395	1.434	1.615
.8000	1.242	1.286	1.332	1.326	1.355	1.406	1.423	1.447	1.487	1.571
.8500	1.281	1.344	1.357	1.362	1.383	1.430	1.447	1.500	1.525	1.529
.9000	1.272	1.316	1.329	1.314	1.331	1.370	1.392	1.439	1.478	1.493
.9500	1.211	1.259	1.271	1.260	1.265	1.295	1.319	1.383	1.443	1.444
.9940	1.202	1.240	1.231	1.290	1.220	1.251	1.273	1.342	1.437	1.484

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Fuselage										
.0500	.807	.809	.754	.683	.630	.546	.473	.499	.402	.377
.1000	.872	.909	.846	.788	.729	.657	.583	.532	.502	.467
.1500	.945	.970	.920	.864	.804	.719	.658	.620	.598	.542
.2000	.972	1.009	.963	.912	.864	.779	.713	.684	.639	.609
.2500	1.003	1.031	.994	.952	.904	.833	.757	.731	.701	.658
.3000	1.003	1.006	.988	.976	.955	.934	.899	.877	.874	.847
.3500	.994	1.031	1.003	.970	.937	.878	.803	.787	.763	.710
.4000	.927	.961	.948	.915	.873	.821	.760	.737	.692	.655
.4500	.893	.936	.911	.867	.819	.773	.704	.684	.628	.603
.5000	.856	.897	.865	.825	.756	.701	.629	.588	.546	.522
.5500	.816	.854	.812	.764	.690	.635	.562	.515	.490	.461
.6000	.783	.790	.738	.683	.620	.558	.493	.462	.443	.403
.7500	.942	.958	.920	.867	.828	.779	.719	.690	.675	.649
.9500	1.037	1.122	1.166	1.139	1.136	1.167	1.145	1.173	1.173	1.160
.9940	1.034	1.237	1.317	1.338	1.352	1.364	1.357	1.380	1.428	1.464

TABLE XIV. — PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(b) Extended double slotted flap configuration; $\delta_f = 65^\circ$; $\frac{y}{b/2} = 0.21$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	.859	.914	1.981	3.508	5.442	8.063	10.887	10.675	4.377	2.681
.0125	.920	1.235	1.733	3.594	7.740	10.904	13.486	11.082	4.056	2.466
.0250	.952	1.187	1.583	2.739	5.695	6.471	9.377	9.503	4.021	2.472
.0500	.994	1.156	1.459	1.625	2.045	2.805	3.813	5.386	3.465	2.406
.0750	.994	1.162	1.411	1.560	1.842	2.246	2.819	3.716	2.266	2.403
.1000	1.019	1.175	1.382	1.542	1.761	2.077	2.566	3.172	3.132	2.400
.1500	1.038	1.175	1.360	1.486	1.651	1.849	2.211	2.544	2.874	2.364
.2000	1.067	1.191	1.344	1.446	1.591	1.752	2.038	2.272	2.664	2.322
.2500	1.089	1.210	1.338	1.440	1.537	1.669	1.939	2.114	2.430	2.295
.3000	1.102	1.222	1.344	1.422	1.516	1.624	1.842	1.954	2.260	2.257
.3500	1.121	1.222	1.338	1.406	1.490	1.592	1.713	1.842	2.099	2.233
.4000	1.144	1.245	1.341	1.400	1.466	1.574	1.645	1.762	1.980	2.209
.4500	1.169	1.267	1.344	1.400	1.463	1.539	1.616	1.710	1.871	2.191
.5000	1.205	1.289	1.373	1.415	1.451	1.509	1.607	1.673	1.839	2.167
.5500	1.227	1.299	1.376	1.400	1.445	1.474	1.569	1.636	1.775	2.119
.6000	1.236	1.314	1.382	1.403	1.439	1.462	1.569	1.642	1.740	2.093
.6500	1.272	1.349	1.405	1.418	1.442	1.459	1.610	1.676	1.722	2.066
.7000	1.304	1.368	1.414	1.434	1.448	1.468	1.625	1.636	1.687	2.012
.7824	1.329	1.387	1.430	1.422	1.439	1.480	1.496	1.493	1.629	1.940
.8176	1.399	1.435	1.471	1.465	1.460	1.524	1.499	1.487	1.614	1.907
.8323	1.428	1.480	1.516	1.508	1.495	1.571	1.531	1.521	1.640	1.916
.8913	1.524	1.568	1.580	1.557	1.537	1.586	1.510	1.521	1.640	1.875
.9281	1.652	1.673	1.682	1.625	1.603	1.645	1.587	1.613	1.711	1.884
.9504	1.722	1.727	1.729	1.668	1.639	1.663	1.628	1.670	1.749	1.913
.9804	1.978	1.946	1.949	1.809	1.800	1.855	1.874	1.991	2.056	2.075
Vane										
.0000	2.463	2.356	2.335	1.932	1.848	1.882	1.968	2.089	2.117	1.943
.0250	3.697	3.480	3.433	2.929	3.078	3.287	3.599	3.879	3.883	3.540
.0500	4.492	4.191	4.118	3.412	3.633	3.891	4.350	4.713	4.725	3.922
.1000	4.166	3.886	3.812	3.092	3.328	3.589	4.097	4.478	4.570	3.734
.1500	3.802	3.559	3.500	2.825	3.006	3.276	3.813	4.174	4.371	3.630
.2000	3.428	3.223	3.201	2.548	2.704	2.953	3.528	3.879	4.208	3.570
.3000	2.665	2.515	2.542	2.105	2.152	2.326	2.801	3.103	3.681	3.373
.4000	2.278	2.203	2.175	2.019	2.048	2.175	2.414	2.696	3.278	3.137
.5000	2.150	2.083	2.067	1.957	1.997	2.148	2.349	2.570	3.035	2.994
.6000	2.083	2.016	2.026	1.908	1.943	2.098	2.279	2.475	2.763	2.716
.7000	2.061	1.988	1.987	1.865	1.913	2.048	2.226	2.432	2.635	2.573
.8000	2.042	1.969	1.987	1.846	1.881	2.006	2.191	2.384	2.553	2.472
.9000	1.978	1.921	1.930	1.815	1.830	1.962	2.091	2.284	2.439	2.346
Flap										
.0000	1.633	1.699	1.838	1.591	1.463	1.459	1.516	1.645	1.687	1.585
.0125	2.080	2.115	2.204	2.068	2.005	2.104	2.253	2.381	2.484	2.370
.0250	2.300	2.286	2.287	2.206	2.200	2.364	2.475	2.485	2.833	2.719
.0500	2.390	2.375	2.331	2.280	2.272	2.428	2.584	2.799	2.968	2.934
.0750	2.221	2.191	2.137	2.095	2.092	2.237	2.382	2.581	2.769	2.788
.1000	2.061	2.016	1.943	1.926	1.904	2.065	2.194	2.358	2.564	2.627
.1500	1.815	1.768	1.710	1.671	1.657	1.775	1.921	2.028	2.196	2.269
.2000	1.789	1.759	1.691	1.646	1.627	1.743	1.857	1.968	2.097	2.149
.4000	1.828	1.797	1.742	1.705	1.681	1.784	1.889	1.985	2.049	2.122
.6000	1.812	1.781	1.726	1.689	1.651	1.749	1.829	1.940	2.041	2.089
.8000	1.764	1.740	1.685	1.637	1.615	1.681	1.783	1.920	1.909	1.922

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	1.054	.826	.653	.578	.588	.649	.784	.771	.506	.367
.0250	1.019	.857	.682	.548	.460	.402	.364	.347	.244	.197
.0500	1.003	.892	.717	.563	.449	.351	.258	.232	.222	.206
.0750	.994	.899	.736	.603	.484	.364	.287	.269	.281	.278
.1000	1.003	.914	.771	.628	.519	.420	.340	.341	.342	.334
.1500	.987	.908	.780	.668	.561	.485	.425	.415	.406	.406
.2000	1.000	.933	.815	.692	.603	.536	.472	.461	.450	.445
.2500	1.016	.944	.841	.735	.639	.592	.513	.507	.491	.469
.3000	1.006	.953	.850	.754	.675	.607	.554	.539	.503	.501
.3500	1.016	.978	.889	.794	.716	.651	.590	.564	.544	.525
.4000	1.019	.984	.901	.812	.749	.672	.607	.587	.556	.552
.4500	1.016	.991	.920	.834	.761	.695	.628	.613	.588	.573
.5000	.965	.959	.901	.822	.752	.692	.625	.616	.591	.579
.5500	.891	.921	.879	.794	.737	.672	.634	.599	.705	.567
.6000	.863	.889	.831	.772	.713	.663	.595	.590	.567	.558
.6500	.808	.832	.793	.723	.672	.615	.566	.556	.532	.519
.7000	.751	.781	.742	.689	.636	.586	.531	.533	.518	.510
.7815	.703	.641	.612	.572	.537	.491	.449	.450	.442	.424
.8179	.690	.591	.551	.532	.504	.460	.431	.427	.418	.418
.8325	.690	.575	.535	.514	.487	.468	.437	.435	.424	.418
.8366	.655	.565	.532	.517	.496	.476	.440	.441	.430	.424
.8921	.620	.514	.417	.363	.349	.323	.285	.301	.284	.275
.9286	.537	.483	.382	.326	.307	.290	.270	.269	.266	.248
.9518	.473	.429	.376	.302	.278	.263	.243	.241	.234	.209
.9813	.384	.384	.363	.283	.266	.249	.214	.223	.211	.200
Vane										
.0250	1.157	1.127	1.080	.742	.630	.541	.519	.550	.541	.496
.0500	.661	.683	.621	.394	.299	.231	.199	.209	.193	.176
.1000	.293	.438	.292	.237	.176	.130	.103	.112	.102	.096
.1500	.297	.337	.315	.191	.152	.101	.088	.109	.088	.072
.2000	.243	.292	.271	.166	.134	.101	.082	.103	.094	.081
.3000	.173	.229	.217	.151	.122	.098	.082	.103	.096	.084
.4000	.137	.184	.194	.135	.113	.098	.085	.112	.102	.087
.5000	.115	.143	.178	.126	.116	.112	.100	.129	.114	.101
.6000	.099	.139	.175	.142	.128	.124	.123	.152	.137	.128
.7000	.128	.168	.188	.157	.144	.172	.167	.203	.187	.173
.8000	.211	.248	.258	.255	.260	.272	.282	.321	.313	.299
.9200	.663	.673	.678	.655	.675	.725	.751	.828	.866	.806
Flap										
.0125	.498	.448	.478	.255	.182	.163	.170	.198	.196	.158
.0250	.201	.139	.111	.049	.039	.038	.038	.049	.041	.030
.0500	.137	.098	.038	.028	.048	.059	.050	.060	.053	.051
.0750	.121	.095	.035	.049	.054	.065	.059	.077	.076	.069
.1000	.131	.108	.067	.074	.090	.101	.088	.097	.096	.099
.1500	.137	.140	.102	.092	.116	.124	.117	.129	.129	.116
.4000	.342	.337	.325	.292	.304	.302	.287	.301	.295	.290
.6000	.502	.508	.481	.455	.433	.429	.414	.418	.401	.388
.8000	.610	.622	.589	.548	.531	.512	.496	.507	.468	.457
.9000	.712	.714	.694	.662	.627	.601	.567	.590	.550	.555

TABLE XIV.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(c) Extended double slotted flap configuration; $\delta_f = 66^\circ$; $\frac{y}{b/2} = 0.30$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	.885	1.206	2.065	2.788	3.691	4.714	4.891	4.462	3.104	2.422
.0125	.969	1.448	2.205	2.843	3.618	4.339	4.482	4.237	3.083	2.431
.0250	.994	1.330	2.311	2.929	3.749	4.556	4.566	4.254	3.113	2.431
.0500	1.034	1.268	2.851	3.846	4.055	4.550	4.435	4.271	3.068	2.422
.0750	1.034	1.227	1.767	3.754	5.465	6.086	5.305	4.534	3.077	2.417
.1000	1.044	1.193	1.767	2.906	4.257	5.559	5.311	4.482	3.074	2.424
.1500	1.072	1.215	1.258	1.883	2.251	3.402	4.314	3.928	2.938	2.411
.2000	1.100	1.234	1.314	1.240	1.670	2.297	3.282	3.350	2.835	2.375
.2500	1.112	1.234	1.320	1.353	1.523	1.878	2.725	2.945	2.757	2.339
.3000	1.128	1.249	1.339	1.362	1.483	1.753	2.377	2.667	2.622	2.327
.3500	1.140	1.249	1.329	1.365	1.456	1.643	2.096	2.355	2.483	2.268
.4000	1.165	1.245	1.331	1.377	1.468	1.616	1.963	2.211	2.376	2.241
.4500	1.196	1.299	1.367	1.386	1.468	1.607	1.905	2.130	2.311	2.214
.5000	1.215	1.324	1.376	1.402	1.465	1.586	1.812	1.989	2.181	2.172
.5500	1.246	1.330	1.376	1.402	1.453	1.574	1.789	1.913	2.101	2.137
.6000	1.258	1.352	1.401	1.414	1.465	1.565	1.760	1.864	2.045	2.101
.6500	1.308	1.386	1.419	1.423	1.477	1.577	1.705	1.832	1.985	2.062
.7516	1.280	1.380	1.416	1.429	1.477	1.521	1.644	1.708	1.843	1.984
.7940	1.336	1.436	1.457	1.472	1.474	1.548	1.667	1.696	1.811	1.911
.8099	1.324	1.461	1.463	1.457	1.514	1.530	1.638	1.665	1.778	1.884
.8429	1.430	1.514	1.544	1.543	1.596	1.604	1.722	1.725	1.793	1.899
.8767	1.445	1.567	1.578	1.570	1.627	1.628	1.728	1.714	1.731	1.845
.9108	1.564	1.682	1.687	1.672	1.712	1.723	1.821	1.774	1.769	1.842
.9437	1.772	1.885	1.885	1.859	1.911	1.902	2.018	1.993	1.838	1.884
.9768	2.302	2.408	2.376	2.334	2.336	2.378	2.508	2.396	2.113	2.024
Vane										
.0000	6.454	6.420	6.049	5.628	5.260	5.381	5.363	5.306	4.288	3.690
.0250	8.002	7.442	7.040	6.674	6.324	6.812	7.105	6.806	5.226	4.467
.0500	7.376	7.473	7.016	6.744	6.400	6.955	7.210	6.953	5.347	4.282
.1000	7.242	7.149	6.690	6.407	6.024	6.589	6.871	6.687	5.069	3.928
.1500	7.024	6.984	6.476	6.180	5.801	6.378	6.694	6.560	4.980	3.839
.2000	6.756	6.710	6.282	5.956	5.584	6.122	6.468	6.410	4.962	3.815
.3000	6.034	6.034	5.569	5.248	4.963	5.455	5.847	5.884	4.699	3.684
.4000	5.196	5.143	4.727	4.429	4.241	4.660	5.070	5.173	4.246	3.375
.5000	4.373	4.486	4.044	3.772	3.697	4.092	4.473	4.604	3.873	3.155
.6000	3.875	3.791	3.395	3.162	3.196	3.491	3.911	4.011	3.483	2.928
.7000	3.265	3.187	2.895	2.730	2.786	3.030	3.430	3.508	3.128	2.708
.8000	2.807	2.757	2.538	2.371	2.422	2.625	2.954	3.043	2.767	2.464
.9000	2.326	2.489	2.326	2.144	2.156	2.330	2.592	2.670	2.489	2.274
Flap										
.0000	1.912	1.863	1.854	1.583	1.446	1.536	1.673	1.696	1.533	1.399
.0125	2.405	2.458	2.392	2.190	2.107	2.235	2.458	2.474	2.323	2.152
.0250	2.819	2.891	2.786	2.592	2.535	2.705	3.000	3.043	2.885	2.693
.0500	2.744	2.844	2.712	2.543	2.578	2.717	3.093	3.153	3.066	2.878
.0750	2.679	2.763	2.618	2.457	2.517	2.669	3.058	3.147	3.110	2.899
.1000	2.582	2.607	2.454	2.300	2.373	2.527	2.887	2.948	2.926	2.747
.1500	2.081	2.131	2.006	1.938	2.012	2.092	2.311	2.445	2.530	2.425
.2000	1.919	1.897	1.783	1.714	1.764	1.872	2.044	2.139	2.148	2.077
.4000	1.732	1.766	1.705	1.678	1.673	1.744	1.879	1.933	1.959	1.925
.6000	1.704	1.779	1.733	1.702	1.709	1.753	1.902	1.913	1.870	1.857
.8000	1.676	1.751	1.727	1.699	1.682	1.708	1.835	1.850	1.725	1.717

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	1.006	.794	.734	.731	1.004	1.271	1.426	1.387	1.243	1.095
.0250	1.025	.857	.717	.687	.752	.839	.910	.890	.829	.762
.0500	1.003	.882	.742	.653	.618	.595	.588	.572	.559	.518
.0750	1.000	.903	.758	.662	.602	.548	.499	.474	.479	.449
.1000	1.003	.906	.786	.678	.596	.527	.473	.448	.453	.417
.1500	1.006	.922	.820	.712	.636	.536	.461	.436	.432	.408
.2000	1.012	.941	.848	.730	.639	.542	.464	.431	.444	.417
.2500	1.019	.953	.854	.748	.670	.577	.484	.454	.456	.417
.3000	1.012	.956	.863	.764	.679	.586	.504	.471	.468	.446
.3500	1.003	.966	.888	.785	.709	.610	.533	.497	.494	.464
.4000	.978	.953	.876	.785	.709	.619	.536	.512	.509	.479
.4500	.953	.928	.863	.788	.728	.634	.551	.529	.524	.503
.5000	.897	.878	.842	.776	.713	.628	.554	.532	.518	.509
.5500	.860	.847	.814	.754	.700	.634	.560	.535	.524	.515
.6000	.825	.816	.786	.715	.670	.607	.533	.512	.509	.497
.6500	.816	.807	.761	.678	.636	.574	.516	.494	.506	.476
.7517	.698	.701	.624	.570	.538	.500	.444	.434	.435	.417
.7934	.642	.648	.553	.503	.483	.449	.417	.402	.405	.387
.8094	.632	.632	.512	.466	.459	.420	.391	.379	.388	.378
.8197	.573	.604	.544	.475	.459	.420	.397	.384	.382	.378
.8439	.604	.601	.503	.460	.446	.417	.383	.376	.379	.363
.8772	.570	.551	.460	.386	.346	.307	.281	.266	.263	.256
.9117	.523	.502	.432	.344	.303	.274	.252	.251	.243	.226
.9448	.461	.445	.382	.304	.275	.238	.215	.205	.213	.190
.9781	.371	.377	.354	.273	.252	.220	.209	.197	.192	.161
Vane										
.0250	2.221	2.165	2.196	1.813	1.520	1.426	1.394	1.318	1.083	.911
.0500	.766	.701	.614	.558	.401	.333	.299	.280	.234	.179
.1000	.246	.181	.273	.123	.067	.033	.023	.020	.015	.000
.1500	.137	.087	.143	.046	.021	.003	.000	.000	.003	.000
.2000	.097	.072	.090	.021	.009	.003	.000	.000	.009	.000
.3000	.073	.059	.059	.006	.012	.006	.012	.026	.009	.003
.4000	.084	.053	.043	.028	.024	.039	.038	.035	.028	.018
.5000	.081	.072	.065	.028	.046	.051	.049	.058	.056	.039
.6000	.115	.103	.071	.058	.061	.089	.078	.090	.077	.060
.7000	.168	.153	.134	.110	.131	.134	.142	.147	.133	.119
.8000	.315	.302	.261	.239	.254	.277	.293	.303	.272	.241
.9200	.897	.875	.820	.748	.761	.821	.878	.893	.820	.750
Flap										
.0125	.433	.380	.379	.248	.208	.217	.258	.251	.222	.173
.0250	.118	.081	.071	.018	.015	.021	.029	.038	.050	.003
.0500	.034	.016	.000	.000	.000	.000	.000	.000	.000	.000
.0750	.037	.028	.000	.000	.000	.000	.000	.003	.003	.000
.1000	.034	.028	.009	.000	.006	.009	.009	.009	.009	.000
.1500	.047	.040	.034	.021	.037	.021	.026	.023	.021	.012
.2000	.059	.059	.050	.043	.049	.051	.035	.035	.044	.018
.4000	.171	.159	.146	.144	.144	.128	.113	.107	.121	.089
.6000	.262	.274	.280	.261	.251	.235	.212	.205	.213	.190
.8000	.424	.433	.450	.439	.422	.393	.371	.393	.328	.345
.9000	.592	.617	.624	.616	.596	.565	.542	.526	.488	.500

TABLE XIV.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(d) Extended double slotted flap configuration; $\delta_f = 65^\circ$; $\frac{y}{b/2} = 0.43$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	.794	1.943	2.086	2.520	2.997	3.293	3.132	2.848	2.844	2.321
.0125	1.042	1.803	2.073	2.489	2.979	3.320	3.174	2.965	2.493	2.343
.0250	1.089	1.800	2.121	2.545	2.988	3.305	3.294	2.997	2.670	2.349
.0500	1.077	1.435	2.140	2.643	3.131	3.432	3.253	3.020	2.684	2.367
.0750	1.102	1.308	2.386	2.659	3.092	3.305	3.185	3.051	2.687	2.358
.1000	1.121	1.305	2.761	2.828	3.066	3.246	3.174	3.066	2.696	2.392
.1500	1.118	1.280	2.341	3.723	3.889	3.725	3.437	3.134	2.702	2.367
.2000	1.160	1.311	1.570	3.006	4.215	4.294	3.428	3.094	2.673	2.355
.2500	1.166	1.308	1.229	2.022	3.272	3.956	3.502	2.974	2.629	2.343
.3000	1.195	1.334	1.258	1.560	2.472	3.465	3.349	2.885	2.588	2.340
.3500	1.198	1.327	1.287	1.357	1.881	2.855	3.036	2.699	2.509	2.322
.4000	1.230	1.343	1.335	1.392	1.615	2.489	2.845	2.630	2.456	2.313
.4500	1.275	1.365	1.370	1.335	1.498	2.228	2.678	2.515	2.389	2.287
.5000	1.265	1.359	1.373	1.317	1.397	1.953	2.426	2.372	2.304	2.266
.5500	1.313	1.394	1.421	1.369	1.409	1.846	2.317	2.283	2.237	2.239
.6000	1.342	1.419	1.440	1.394	1.413	1.749	2.182	2.203	2.175	2.206
.6993	1.288	1.419	1.446	1.400	1.400	1.610	1.930	2.028	2.064	2.143
.7507	1.268	1.419	1.459	1.403	1.406	1.574	1.824	1.925	1.988	2.107
.7704	1.367	1.495	1.507	1.462	1.457	1.610	1.860	1.940	1.991	2.107
.8099	1.422	1.559	1.583	1.539	1.510	1.636	1.854	1.925	1.968	2.095
.8312	1.530	1.638	1.649	1.591	1.570	1.684	1.830	1.871	1.918	2.080
.8911	1.649	1.794	1.793	1.732	1.707	1.781	1.880	1.888	1.906	2.039
.9322	1.955	2.045	2.045	1.985	1.954	2.003	2.024	1.974	1.918	2.030
.9724	2.569	2.702	2.695	2.619	2.564	2.586	2.508	2.286	2.088	2.090
Vane										
.0000	8.112	8.068	8.294	7.806	7.394	7.155	6.576	5.484	4.228	3.875
.0250	9.690	9.331	9.291	9.111	8.889	8.738	8.119	6.664	4.795	4.191
.0500	9.294	8.896	8.861	8.705	8.528	8.371	7.720	6.226	4.482	3.946
.1000	8.834	8.411	8.385	8.280	8.101	7.918	7.262	5.816	3.971	3.316
.1500	8.583	8.144	8.106	8.022	7.851	7.685	7.057	5.598	3.743	3.098
.2000	8.614	8.182	8.154	8.037	7.868	7.761	7.201	5.756	3.877	3.200
.3000	7.729	7.299	7.122	7.105	7.000	6.945	6.359	5.111	3.480	2.901
.4000	6.663	6.283	6.109	6.077	5.949	6.004	5.614	4.898	3.234	2.734
.5000	5.844	5.496	5.252	5.256	5.128	5.249	4.930	4.131	3.000	2.600
.6000	4.971	4.667	4.383	4.391	4.278	4.483	4.232	3.647	2.769	2.466
.7000	4.291	4.039	3.781	3.757	3.697	3.897	3.725	3.321	2.614	2.379
.8000	3.633	3.445	3.316	3.240	3.146	3.329	3.244	2.957	2.406	2.257
.9000	3.134	3.045	3.067	2.951	2.872	2.909	2.833	2.613	2.178	2.095
Flap										
.0000	2.858	2.753	2.905	2.785	2.713	2.663	2.534	2.258	1.884	1.743
.0125	3.042	3.156	3.293	3.151	3.081	3.012	2.889	2.619	2.222	2.095
.0250	3.604	3.731	3.793	3.686	3.612	3.583	3.449	3.180	2.749	2.603
.0500	3.546	3.642	3.586	3.462	3.376	3.477	3.417	3.232	2.845	2.698
.0750	3.380	3.483	3.370	3.268	3.194	3.323	3.308	3.192	2.848	2.698
.1000	3.384	3.315	3.144	3.074	2.988	3.157	3.138	3.066	2.725	2.603
.1500	2.582	2.524	2.401	2.329	2.260	2.412	2.481	2.527	2.328	2.281
.2000	2.281	2.162	2.073	2.052	1.994	2.116	2.135	2.129	1.988	1.997
.4000	1.693	1.711	1.704	1.671	1.651	1.737	1.789	1.839	1.848	1.919
.6000	1.518	1.578	1.589	1.591	1.597	1.663	1.716	1.713	1.751	1.848
.8000	1.392	1.391	1.465	1.505	1.552	1.589	1.596	1.530	1.602	1.743

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	1.074	.819	.806	.935	1.170	1.417	1.496	1.473	1.430	1.382
.0250	1.058	.848	.784	.775	.848	.956	.994	.985	.982	.973
.0500	1.058	.889	.777	.717	.704	.719	.719	.699	.716	.722
.0750	1.042	.905	.793	.705	.666	.642	.628	.607	.620	.639
.1000	1.051	.924	.812	.714	.654	.610	.569	.556	.570	.585
.1500	1.038	.933	.838	.729	.660	.583	.534	.513	.525	.534
.2000	1.038	.943	.841	.735	.657	.583	.528	.507	.506	.519
.2500	1.029	.937	.847	.748	.672	.595	.531	.510	.506	.519
.3000	1.006	.930	.838	.742	.669	.592	.531	.504	.509	.519
.3500	.994	.930	.847	.760	.690	.604	.540	.519	.520	.534
.4000	.978	.899	.844	.760	.681	.604	.540	.521	.526	.531
.4500	.930	.876	.822	.748	.672	.601	.534	.524	.512	.534
.5000	.891	.848	.799	.735	.648	.580	.531	.516	.520	.525
.5500	.859	.813	.764	.692	.639	.565	.516	.504	.503	.510
.6000	.789	.743	.704	.643	.588	.530	.490	.470	.474	.484
.6995	.674	.657	.612	.566	.534	.485	.446	.483	.447	.460
.7499	.629	.600	.554	.508	.472	.441	.411	.407	.412	.424
.7700	.597	.575	.519	.483	.460	.423	.408	.395	.395	.409
.7744	.594	.559	.526	.474	.460	.423	.405	.393	.395	.412
.8110	.588	.543	.506	.455	.448	.405	.378	.367	.368	.382
.8519	.444	.448	.433	.382	.349	.317	.287	.275	.275	.287
.8919	.390	.387	.385	.329	.313	.275	.252	.246	.251	.257
.9332	.335	.337	.340	.298	.275	.246	.214	.218	.211	.224
.9738	.291	.289	.328	.274	.269	.246	.220	.215	.196	.203
Vane										
.0250	1.217	1.337	1.608	1.326	1.149	1.027	.918	.774	.611	.573
.0500	.326	.391	.561	.409	.346	.281	.241	.213	.175	.170
.1000	.054	.098	.182	.108	.090	.062	.056	.046	.050	.048
.1500	.003	.000	.032	.000	.027	.003	.059	.014	.015	.027
.2000	.000	.000	.000	.000	.021	.000	.000	.017	.012	.030
.3000	.003	.000	.000	.000	.027	.024	.032	.043	.020	.042
.4000	.022	.000	.003	.012	.039	.044	.044	.034	.056	.066
.5000	.054	.041	.048	.049	.075	.071	.065	.080	.082	.073
.6000	.089	.086	.080	.095	.113	.109	.106	.106	.105	.099
.7000	.173	.136	.153	.160	.182	.169	.170	.164	.149	.146
.8000	.348	.346	.338	.355	.352	.340	.323	.301	.266	.251
.9200	1.128	1.089	1.124	1.098	1.095	1.089	1.003	.914	.766	.749
Flap										
.0125	.348	.371	.446	.409	.400	.397	.372	.335	.273	.254
.0250	.000	.029	.057	.052	.081	.093	.070	.057	.044	.054
.0500	.000	.000	.067	.000	.030	.009	.056	.011	.009	.024
.0750	.000	.000	.000	.000	.015	.003	.003	.003	.003	.027
.1000	.000	.000	.000	.000	.015	.021	.003	.023	.015	.033
.1500	.000	.003	.009	.006	.030	.021	.026	.032	.018	.051
.2000	.029	.035	.035	.034	.048	.036	.035	.052	.053	.046
.4000	.141	.127	.124	.123	.119	.107	.094	.109	.117	.119
.6000	.230	.232	.236	.206	.215	.201	.182	.186	.193	.215
.8000	.396	.394	.385	.369	.373	.346	.308	.315	.327	.367
.9000	.508	.524	.545	.526	.531	.497	.460	.447	.471	.522

TABLE XIV.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(a) Extended double slotted flap configuration; $\delta_f = 55^\circ$; $\frac{y}{b/2} = 0.55$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	.966	1.844	1.997	2.325	2.767	2.830	2.531	2.376	2.320	2.226
.0125	1.190	1.838	2.041	2.286	2.810	2.857	2.600	2.448	2.332	2.226
.0250	1.134	1.788	2.003	2.337	2.758	2.815	2.618	2.457	2.344	2.235
.0500	1.115	1.645	2.034	2.383	2.789	2.878	2.655	2.509	2.344	2.232
.0750	1.125	1.539	2.112	2.469	2.868	2.952	2.632	2.509	2.338	2.226
.1000	1.143	1.458	2.152	2.524	2.948	2.943	2.635	2.520	2.323	2.229
.1500	1.159	1.396	2.463	2.558	2.853	2.800	2.667	2.537	2.326	2.220
.2000	1.177	1.361	2.659	2.840	2.823	2.762	2.682	2.526	2.314	2.217
.2500	1.196	1.364	2.544	2.417	3.055	2.827	2.763	2.552	2.338	2.223
.3000	1.218	1.377	2.134	3.435	3.737	3.178	2.803	2.535	2.314	2.223
.3500	1.240	1.380	1.721	2.984	3.908	3.384	2.809	2.503	2.302	2.223
.4000	1.265	1.402	1.416	2.395	3.599	3.393	2.760	2.451	2.278	2.202
.4500	1.271	1.389	1.305	1.908	3.147	3.294	2.693	2.436	2.273	2.193
.5000	1.315	1.433	1.314	1.632	2.725	3.143	2.629	2.393	2.243	2.184
.6194	1.227	1.396	1.376	1.337	1.865	2.521	2.418	2.295	2.175	2.152
.6844	1.252	1.405	1.382	1.313	1.670	2.232	2.253	2.165	2.110	2.104
.7066	1.208	1.452	1.432	1.349	1.654	2.193	2.226	2.159	2.095	2.098
.7594	1.361	1.489	1.484	1.362	1.599	2.039	2.084	2.003	1.983	2.003
.8101	1.436	1.598	1.578	1.478	1.636	2.024	2.108	2.040	2.024	2.071
.8622	1.598	1.744	1.721	1.638	1.719	2.018	2.055	1.994	1.988	2.042
.9142	1.832	1.994	1.994	1.908	1.939	2.104	2.084	2.011	1.991	2.036
.9632	2.442	2.663	2.690	2.631	2.590	2.559	2.325	2.202	2.068	2.077
Vane										
.0000	6.308	6.479	6.690	6.481	6.186	5.494	4.366	3.771	3.071	2.988
.0250	9.009	9.090	9.271	9.226	9.030	7.952	6.091	5.130	3.876	3.672
.0500	8.594	8.747	8.945	8.931	8.764	7.761	6.004	4.962	3.711	3.524
.1000	8.541	8.569	8.815	8.876	8.740	7.563	5.621	4.572	3.293	3.098
.1500	8.379	8.379	8.650	8.729	8.636	7.416	5.465	4.364	3.042	2.797
.2000	8.214	8.220	8.501	8.600	8.535	7.315	5.372	4.257	2.971	2.762
.3000	7.622	7.638	7.889	8.020	8.009	6.904	5.018	4.032	2.835	2.628
.4000	6.847	6.669	6.895	7.180	7.303	6.184	4.482	3.641	2.631	2.467
.5000	5.962	6.003	6.175	6.324	6.373	5.589	4.137	3.410	2.530	2.399
.6000	5.162	5.205	5.349	5.496	5.569	4.854	3.685	3.055	2.391	2.312
.7000	4.454	4.514	4.597	4.735	4.810	4.250	3.302	2.800	2.267	2.205
.8000	3.828	3.881	3.951	4.048	4.113	3.690	2.942	2.558	2.136	2.113
.9000	3.258	3.314	3.370	3.426	3.462	3.149	2.595	2.283	1.991	2.003
Flap										
.0000	2.022	2.199	2.328	2.398	2.407	2.232	1.783	1.616	1.358	1.327
.0125	3.025	3.227	3.351	3.432	3.471	3.190	2.577	2.347	2.036	1.991
.0250	3.517	3.788	3.910	4.009	4.055	3.741	3.035	2.774	2.415	2.381
.0500	3.607	3.875	3.988	4.088	4.131	3.833	3.186	2.919	2.568	2.509
.0750	3.576	3.894	3.991	4.067	4.110	3.839	3.232	2.968	2.598	2.521
.1000	3.514	3.784	3.808	3.871	3.881	3.616	3.102	2.844	2.494	2.434
.1500	2.856	3.009	2.972	2.938	2.838	2.786	2.595	2.404	2.172	2.143
.2000	2.467	2.589	2.609	2.613	2.578	2.428	2.192	2.069	1.914	1.922
.4000	2.015	2.087	2.062	2.012	1.923	1.866	1.823	1.789	1.790	1.843
.6000	1.769	1.850	1.805	1.730	1.693	1.586	1.650	1.656	1.758	1.830
.8000	1.389	1.498	1.463	1.402	1.339	1.301	1.377	1.428	1.636	1.717

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	.997	.835	.910	1.138	1.489	1.694	1.693	1.654	1.681	1.673
.0250	1.019	.844	.811	.874	1.015	1.140	1.151	1.127	1.169	1.193
.0500	1.022	.885	.786	.761	.801	.821	.815	.806	.840	.857
.0750	.997	.878	.777	.718	.713	.702	.687	.679	.695	.714
.1000	.991	.885	.789	.718	.679	.652	.623	.618	.642	.655
.1500	.984	.885	.795	.715	.684	.601	.565	.558	.571	.583
.2000	.972	.897	.792	.724	.651	.589	.542	.529	.539	.562
.2500	.953	.885	.804	.718	.687	.589	.542	.532	.536	.542
.3000	.935	.878	.798	.724	.657	.592	.542	.529	.524	.533
.3500	.922	.875	.808	.712	.664	.592	.536	.520	.521	.533
.4000	.897	.860	.792	.718	.667	.589	.531	.520	.521	.530
.4500	.860	.835	.773	.693	.639	.577	.525	.517	.515	.530
.5000	.838	.816	.749	.687	.623	.565	.519	.509	.503	.512
.6193	.720	.726	.677	.626	.569	.512	.455	.460	.447	.470
.6834	.663	.673	.643	.595	.547	.497	.452	.442	.435	.455
.7080	.629	.639	.590	.546	.523	.464	.408	.402	.389	.422
.7137	.623	.632	.593	.555	.523	.461	.420	.416	.399	.405
.7603	.601	.611	.568	.531	.498	.437	.406	.393	.394	.405
.8119	.489	.495	.503	.457	.434	.381	.336	.335	.325	.330
.8643	.439	.452	.450	.426	.382	.327	.301	.295	.293	.301
.9162	.392	.392	.419	.393	.355	.304	.270	.254	.252	.283
.9669	.336	.361	.367	.365	.321	.280	.249	.246	.243	.232
Vane										
.0250	1.445	1.548	1.715	1.576	1.428	1.208	.974	.844	.719	.687
.0500	.399	.458	.572	.503	.431	.357	.301	.272	.231	.241
.1000	.093	.112	.152	.141	.125	.098	.119	.087	.074	.092
.1500	.025	.053	.084	.089	.098	.080	.070	.084	.065	.080
.2000	.031	.044	.062	.104	.083	.071	.064	.092	.065	.107
.3000	.034	.053	.075	.092	.101	.074	.064	.107	.065	.092
.4000	.072	.081	.099	.098	.107	.095	.096	.116	.089	.110
.5000	.084	.097	.118	.126	.125	.104	.096	.110	.107	.107
.6000	.131	.146	.165	.169	.168	.182	.128	.133	.148	.137
.7000	.215	.218	.236	.245	.235	.226	.186	.171	.195	.173
.8000	.392	.402	.432	.448	.434	.393	.342	.303	.287	.295
.9200	1.221	1.246	1.289	1.316	1.306	1.176	.965	.850	.772	.792
Flap										
.0125	.352	.374	.447	.463	.492	.437	.359	.335	.257	.265
.0250	.050	.062	.096	.113	.125	.098	.099	.095	.077	.083
.0500	.019	.037	.053	.067	.064	.039	.058	.087	.071	.068
.0750	.037	.037	.053	.058	.058	.048	.099	.081	.074	.077
.1000	.047	.050	.073	.098	.076	.065	.099	.087	.077	.080
.1500	.072	.075	.094	.092	.092	.086	.107	.095	.089	.092
.2000	.087	.090	.106	.132	.116	.101	.101	.098	.101	.110
.4000	.212	.227	.208	.215	.199	.161	.165	.171	.175	.173
.6000	.315	.324	.295	.288	.260	.235	.229	.240	.263	.274
.8000	.430	.445	.432	.414	.382	.345	.348	.341	.397	.417
.9000	.542	.551	.544	.509	.492	.458	.461	.462	.533	.562

TABLE XIV.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Concluded

(f) Extended double slotted flap configuration; $\delta_f = 65^{\circ}$; $\frac{y}{b/2} = 0.72$

Upper surface

x/c	C_p for -									
	$\alpha = -4^{\circ}$	$\alpha = 0^{\circ}$	$\alpha = 4^{\circ}$	$\alpha = 8^{\circ}$	$\alpha = 12^{\circ}$	$\alpha = 16^{\circ}$	$\alpha = 20^{\circ}$	$\alpha = 22^{\circ}$	$\alpha = 24^{\circ}$	$\alpha = 26^{\circ}$
Wing										
.0000	1.511	1.994	2.154	2.341	2.491	2.427	2.145	2.088	1.959	1.963
.0125	1.391	1.994	2.086	2.266	2.419	2.364	2.119	2.073	1.971	1.963
.0250	1.294	2.003	2.099	2.275	2.428	2.376	2.148	2.102	1.977	1.974
.0500	1.248	2.046	2.139	2.305	2.425	2.397	2.197	2.126	2.000	1.986
.0750	1.238	2.088	2.182	2.347	2.449	2.439	2.221	2.140	1.994	1.974
.1000	1.162	2.095	2.160	2.311	2.413	2.427	2.203	2.137	1.992	1.974
.1500	1.187	2.225	2.228	2.384	2.467	2.469	2.226	2.164	1.994	1.974
.2000	1.229	2.320	2.283	2.435	2.515	2.454	2.235	2.175	1.989	1.974
.2500	1.242	2.161	2.354	2.453	2.509	2.421	2.250	2.170	1.994	1.977
.3000	1.266	1.769	2.455	2.459	2.473	2.391	2.235	2.164	1.989	1.983
.3500	1.303	1.520	2.677	2.483	2.425	2.385	2.253	2.173	1.992	1.989
.4000	1.303	1.356	2.779	2.547	2.395	2.382	2.258	2.167	2.003	1.992
.4500	1.327	1.319	2.825	2.689	2.419	2.385	2.253	2.164	2.000	1.997
.5000	1.336	1.380	2.883	3.000	2.449	2.433	2.284	2.175	2.012	2.000
.5500	1.339	1.374	2.739	3.097	2.500	2.439	2.261	2.158	2.006	2.000
.6000	1.318	1.414	2.594	3.166	2.648	2.487	2.264	2.158	2.009	2.000
.6500	1.349	1.435	2.385	3.069	2.744	2.501	2.241	2.137	2.003	1.995
.7000	1.407	1.468	2.219	2.976	2.795	2.513	2.224	2.123	1.989	2.003
.7500	1.459	1.550	2.062	2.797	2.822	2.540	2.215	2.117	1.994	2.006
.8000	1.489	1.563	1.939	2.655	2.804	2.540	2.200	2.105	1.994	2.009
.8500	1.523	1.602	1.797	2.456	2.699	2.484	2.142	2.067	1.980	2.006
.9000	1.563	1.675	1.799	2.305	2.623	2.448	2.131	2.050	1.989	2.009

Lower surface

x/c	C_p for -									
	$\alpha = -4^{\circ}$	$\alpha = 0^{\circ}$	$\alpha = 4^{\circ}$	$\alpha = 8^{\circ}$	$\alpha = 12^{\circ}$	$\alpha = 16^{\circ}$	$\alpha = 20^{\circ}$	$\alpha = 22^{\circ}$	$\alpha = 24^{\circ}$	$\alpha = 26^{\circ}$
Wing										
.0125	.786	.815	.945	1.100	1.304	1.448	1.415	1.442	1.431	1.458
.0250	.820	.781	.809	.912	1.036	1.140	1.139	1.164	1.161	1.191
.0500	.832	.769	.754	.779	.831	.887	.881	.901	.915	.933
.0750	.807	.793	.745	.731	.756	.779	.780	.798	.798	.820
.1000	.810	.787	.738	.716	.714	.737	.725	.740	.745	.762
.1500	.807	.803	.745	.710	.681	.681	.658	.667	.675	.681
.2000	.826	.821	.763	.725	.690	.666	.649	.652	.657	.664
.2500	.853	.827	.775	.728	.699	.678	.646	.652	.651	.658
.3000	.884	.848	.809	.779	.729	.704	.681	.675	.680	.687
.3500	.890	.885	.840	.804	.750	.713	.681	.678	.678	.684
.4000	.859	.860	.831	.795	.738	.687	.646	.635	.639	.641
.4500	.875	.872	.840	.788	.735	.675	.632	.626	.625	.612
.5000	.887	.879	.858	.804	.741	.672	.629	.620	.619	.606
.5500	.865	.900	.877	.831	.768	.698	.644	.637	.628	.629
.6000	.878	.930	.902	.861	.795	.722	.678	.664	.660	.664
.6500	.920	.958	.942	.897	.831	.764	.716	.719	.727	.742
.7000	.951	.997	.988	.952	.889	.830	.794	.813	.812	.841
.7500	1.024	1.067	1.074	1.039	.988	.934	.896	.924	.939	.968
.8000	1.113	1.170	1.209	1.190	1.139	1.084	1.058	1.099	1.126	1.157
.8500	1.229	1.359	1.431	1.435	1.389	1.322	1.296	1.333	1.381	1.418
.9000	1.388	1.575	1.671	1.737	1.741	1.669	1.626	1.681	1.742	1.792
.9500	1.575	1.778	1.886	1.985	2.000	1.955	1.913	1.974	2.038	2.081

TABLE XV.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE

(a) Extended double slotted flap configuration; $\delta_f = 60^\circ$; $\delta_c = 20^\circ$; $\frac{y}{b/2} = 0$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Fuselage										
.0000	.978	.874	.737	.713	[illegible]	.003	.085	.111	.135	
.0500	.595	.564	.532	.448	.392	.352	.697	1.451	1.709	
.1000	.720	.751	.737	.696	.659	.628	.522	.843	1.271	
.1500	.840	.877	.900	.868	.861	.865	.778	.760	1.038	
.2000	.917	.966	.982	.969	.997	1.009	1.003	.903	.994	
.2500	.978	1.021	1.043	1.031	1.089	1.133	1.134	1.077	1.075	
.3000	1.031	1.064	1.084	1.083	1.136	1.176	1.230	1.211	1.196	
.3500	1.071	1.101	1.146	1.129	1.181	1.238	1.303	1.306	1.294	
.4000	1.012	1.034	1.084	1.104	1.136	1.197	1.256	1.277	1.333	
.4500	1.009	1.031	1.084	1.135	1.172	1.247	1.303	1.326	1.378	
.5000	1.003	1.027	1.105	1.175	1.240	1.343	1.399	1.431	1.499	
.5500	1.015	1.037	1.125	1.221	1.317	1.428	1.516	1.546	1.600	
.6000	1.043	1.073	1.166	1.282	1.377	1.481	1.571	1.591	1.640	
.6500	1.074	1.107	1.228	1.328	1.439	1.475	1.562	1.586	1.634	
.7000	1.129	1.162	1.269	1.346	1.403	1.481	1.545	1.563	1.594	
.7500	1.189	1.211	1.331	1.349	1.392	1.440	1.507	1.520	1.548	
.8000	1.222	1.254	1.371	1.349	1.383	1.431	1.476	1.471	1.510	
.8500	1.280	1.316	1.391	1.349	1.368	1.414	1.451	1.443	1.444	
.9000	1.280	1.285	1.371	1.288	1.303	1.337	1.347	1.357	1.352	
.9500	1.243	1.254	1.309	1.215	1.234	1.241	1.242	1.243	1.248	
.9940	1.237	1.239	1.269	1.236	1.222	1.226	1.221	1.226	1.248	

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Fuselage										
.0500	.886	.896	.900	.846	.813	.777	.717	.697	.677	
.1000	.978	1.000	.982	.923	.881	.856	.860	.906	.908	
.1500	.997	1.018	1.023	.963	.932	.889	.834	.809	.810	
.2000	1.034	1.064	1.064	.997	.944	.874	.796	.763	.741	
.2500	1.052	1.080	1.084	1.018	.952	.883	.810	.786	.741	
.3000	1.018	1.027	1.064	1.012	.958	.933	.933	.894	.867	
.3500	1.046	1.067	1.084	1.015	.949	.895	.834	.800	.767	
.4000	.989	1.009	1.002	.945	.890	.833	.772	.743	.700	
.4500	.954	.978	.982	.896	.831	.780	.717	.677	.634	
.5000	.935	.951	.962	.859	.765	.710	.641	.600	.553	
.5500	.889	.923	.941	.813	.718	.657	.574	.517	.490	
.6000	.825	.853	.859	.730	.629	.575	.496	.451	.429	
.7500	.997	1.021	1.023	.905	.849	.798	.732	.691	.666	
.9500	1.095	1.165	1.105	1.132	1.086	1.082	1.076	1.071	1.055	
.9940	1.074	1.233	1.289	1.331	1.359	1.352	1.315	1.297	1.294	

TABLE XV.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(b) Extended double slotted flap configuration; $\delta_f = 60^\circ$; $\delta_c = 20^\circ$; $\frac{y}{b/2} = 0.21$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	2.374	2.279	2.056	1.090	2.969	5.314	8.585	10.473	11.662	
.0125	.649	.702	.808	1.149	2.830	7.489	11.528	13.300	14.282	
.0250	.716	.758	.848	1.152	2.136	3.055	6.257	8.104	9.361	
.0500	.802	.825	.879	1.193	1.633	2.078	2.971	3.595	4.108	
.0750	.850	.862	.898	1.208	1.579	1.924	2.447	2.769	3.125	
.1000	.888	.899	.935	1.246	1.579	1.866	2.291	2.584	2.827	
.1500	.949	.960	.994	1.311	1.573	1.772	2.078	2.269	2.466	
.2000	.994	1.021	1.074	1.360	1.576	1.743	1.963	2.092	2.199	
.2500	1.035	1.064	1.130	1.398	1.573	1.699	1.859	1.942	1.957	
.3000	1.080	1.092	1.183	1.423	1.573	1.685	1.775	1.832	1.905	
.3500	1.086	1.113	1.220	1.441	1.551	1.644	1.709	1.763	1.787	
.4000	1.105	1.141	1.257	1.441	1.545	1.609	1.663	1.711	1.736	
.4500	1.137	1.187	1.285	1.441	1.545	1.592	1.657	1.682	1.696	
.5000	1.173	1.215	1.319	1.463	1.545	1.595	1.651	1.670	1.696	
.5500	1.195	1.236	1.337	1.451	1.527	1.565	1.625	1.639	1.673	
.6000	1.214	1.248	1.353	1.454	1.515	1.551	1.605	1.636	1.665	
.6500	1.252	1.267	1.363	1.447	1.513	1.554	1.605	1.633	1.659	
.7000	1.265	1.285	1.362	1.447	1.506	1.539	1.594	1.616	1.653	
.7824	1.288	1.297	1.372	1.423	1.482	1.513	1.562	1.587	1.625	
.8176	1.329	1.340	1.399	1.441	1.500	1.536	1.582	1.616	1.642	
.8323	1.374	1.389	1.443	1.482	1.542	1.574	1.608	1.650	1.673	
.8913	1.425	1.445	1.474	1.513	1.557	1.600	1.634	1.662	1.682	
.9281	1.518	1.518	1.554	1.569	1.612	1.641	1.677	1.717	1.724	
.9504	1.572	1.537	1.573	1.584	1.606	1.647	1.680	1.717	1.750	
.9804	1.677	1.616	1.647	1.677	1.685	1.699	1.744	1.798	1.835	
Vane										
.0000	1.214	1.190	1.356	1.342	1.197	1.166	1.179	1.217	1.239	
.0250	2.540	2.319	2.452	2.497	2.463	2.466	2.536	2.621	2.685	
.0500	3.099	2.812	2.954	3.013	2.897	2.915	3.000	3.104	3.188	
.1000	3.035	2.803	2.916	2.991	2.889	2.857	2.937	3.040	3.125	
.1500	2.933	2.730	2.820	2.895	2.739	2.746	2.847	2.942	3.026	
.2000	2.808	2.656	2.706	2.811	2.660	2.670	2.767	2.867	2.952	
.3000	2.460	2.374	2.384	2.510	2.366	2.367	2.458	2.575	2.688	
.4000	2.316	2.190	2.183	2.295	2.148	2.151	2.236	2.352	2.469	
.5000	2.275	2.076	2.121	2.224	2.097	2.110	2.179	2.272	2.389	
.6000	2.099	1.935	1.985	2.100	2.018	2.014	2.040	2.121	2.207	
.7000	2.067	1.905	1.950	2.069	1.997	1.982	2.006	2.075	2.142	
.8000	2.115	1.902	1.972	2.069	1.994	1.988	2.000	2.078	2.139	
.9000	2.061	1.868	1.935	2.013	1.954	1.950	1.963	2.017	2.091	
Flap										
.0000	1.534	1.245	1.498	1.472	1.242	1.224	1.233	1.260	1.273	
.0125	2.191	1.880	2.071	2.125	1.976	1.953	1.983	2.025	2.041	
.0250	2.377	2.138	2.285	2.392	2.297	2.297	2.337	2.396	2.449	
.0500	2.467	2.291	2.409	2.541	2.476	2.460	2.502	2.581	2.636	
.0750	2.323	2.150	2.263	2.367	2.321	2.303	2.349	2.422	2.466	
.1000	2.160	1.990	2.099	2.190	2.154	2.140	2.185	2.243	2.298	
.1500	1.869	1.745	1.805	1.876	1.842	1.819	1.847	1.905	1.926	
.2000	1.821	1.696	1.749	1.801	1.766	1.749	1.778	1.832	1.841	
.4000	1.888	1.739	1.802	1.845	1.806	1.799	1.833	1.884	1.898	
.6000	1.834	1.711	1.746	1.801	1.794	1.775	1.784	1.826	1.838	
.8000	1.764	1.653	1.675	1.702	1.721	1.694	1.695	1.740	1.747	

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	1.463	1.475	1.505	.845	.612	.600	.712	.806	.864	
.0250	1.316	1.349	1.378	.848	.585	.455	.395	.396	.389	
.0500	1.192	1.236	1.310	.854	.594	.426	.303	.275	.241	
.0750	1.137	1.159	1.266	.832	.606	.443	.329	.303	.284	
.1000	1.134	1.119	1.238	.823	.615	.472	.375	.364	.335	
.1500	1.064	1.049	1.155	.783	.630	.525	.447	.436	.403	
.2000	1.054	1.021	1.099	.777	.658	.563	.499	.480	.452	
.2500	1.051	1.015	1.059	.777	.688	.600	.530	.512	.466	
.3000	1.022	.988	1.015	.783	.697	.630	.562	.535	.500	
.3500	1.019	1.003	.997	.814	.739	.665	.585	.572	.523	
.4000	1.016	1.009	.975	.823	.761	.688	.608	.595	.543	
.4500	1.003	.997	.944	.845	.779	.705	.625	.610	.565	
.5000	.949	.978	.929	.829	.776	.705	.625	.610	.568	
.5500	.927	.945	.895	.808	.754	.694	.614	.598	.557	
.6000	.882	.920	.867	.792	.745	.676	.602	.590	.548	
.6500	.847	.880	.811	.739	.700	.638	.571	.555	.526	
.7000	.805	.843	.762	.708	.667	.609	.539	.535	.503	
.7815	.703	.758	.663	.606	.570	.522	.470	.460	.438	
.8179	.693	.770	.650	.596	.576	.538	.478	.474	.443	
.8325	.725	.819	.687	.621	.624	.586	.545	.529	.497	
.8566	.757	.886	.728	.655	.664	.630	.582	.581	.551	
.8921	.492	.567	.446	.370	.389	.315	.288	.289	.270	
.9286	.489	.549	.430	.339	.327	.312	.280	.277	.256	
.9518	.450	.506	.387	.311	.285	.262	.245	.243	.227	
.9813	.431	.469	.356	.292	.245	.239	.216	.220	.207	
Vane										
.0250	.383	.426	.551	.481	.291	.248	.239	.249	.247	
.0500	.265	.513	.519	.255	.100	.085	.075	.081	.090	
.1000	.278	.316	.223	.165	.076	.073	.069	.072	.057	
.1500	.288	.322	.201	.137	.085	.073	.072	.075	.057	
.2000	.291	.325	.223	.143	.091	.085	.072	.081	.063	
.3000	.297	.340	.211	.140	.103	.090	.084	.092	.091	
.4000	.294	.344	.207	.134	.113	.111	.101	.110	.091	
.5000	.284	.340	.235	.134	.118	.122	.115	.124	.108	
.6000	.294	.347	.257	.162	.145	.140	.130	.147	.128	
.7000	.348	.402	.266	.199	.188	.187	.176	.197	.182	
.8000	.431	.478	.341	.292	.282	.286	.274	.298	.290	
.9200	.869	.846	.743	.733	.721	.726	.712	.754	.756	
Flap										
.0125	.383	.304	.245	.152	.067	.064	.061	.078	.080	
.0250	.240	.273	.080	.012	.015	.026	.029	.046	.040	
.0500	.224	.304	.096	.043	.061	.076	.044	.092	.068	
.0750	.230	.325	.111	.062	.088	.090	.089	.098	.088	
.1000	.243	.353	.130	.081	.118	.122	.112	.142	.125	
.1500	.259	.380	.176	.124	.158	.146	.144	.168	.159	
.4000	.498	.604	.365	.363	.364	.332	.320	.338	.321	
.6000	.697	.718	.517	.506	.500	.443	.429	.436	.429	
.8000	.799	.788	.619	.615	.600	.554	.522	.523	.497	
.9000	.895	.862	.728	.717	.703	.650	.628	.598	.588	

TABLE XV.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(a) Extended double slotted flap configuration; $\delta_f = 60^\circ$; $\delta_0 = 20^\circ$; $\frac{y}{b/2} = 0.30$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	2.067	2.222	2.189	1.484	2.779	4.998	5.324	5.447	5.024	4.897
.0125	.730	.705	.654	1.283	2.903	3.806	4.712	4.937	4.897	4.897
.0250	.810	.796	.737	1.236	3.098	3.899	4.856	4.923	4.747	
.0500	.899	.900	.838	1.268	3.652	5.094	5.424	5.504	5.309	
.0750	.926	.945	.918	1.289	2.484	4.978	6.479	6.561	6.018	
.1000	.954	.988	1.000	1.298	1.794	3.459	5.415	5.812	5.663	
.1500	1.009	1.043	1.111	1.392	1.596	2.168	3.407	4.034	4.444	
.2000	1.043	1.100	1.200	1.449	1.575	1.841	2.476	3.034	3.302	
.2500	1.073	1.137	1.267	1.455	1.566	1.725	2.092	2.547	2.974	
.3000	1.095	1.167	1.311	1.485	1.575	1.679	1.945	2.316	2.680	
.3500	1.123	1.176	1.308	1.485	1.543	1.635	1.844	2.094	2.398	
.4000	1.144	1.204	1.356	1.494	1.552	1.626	1.793	2.003	2.265	
.4500	1.175	1.240	1.365	1.509	1.566	1.623	1.784	1.963	2.187	
.5000	1.202	1.256	1.378	1.524	1.558	1.606	1.741	1.886	2.061	
.5500	1.224	1.259	1.372	1.512	1.531	1.594	1.715	1.838	1.994	
.6000	1.251	1.295	1.387	1.521	1.531	1.592	1.697	1.809	1.940	
.6500	1.285	1.328	1.403	1.530	1.543	1.600	1.693	1.783	1.885	
.7516	1.282	1.325	1.384	1.461	1.502	1.577	1.620	1.678	1.781	
.7940	1.328	1.368	1.429	1.479	1.531	1.597	1.634	1.681	1.775	
.8099	1.310	1.362	1.416	1.464	1.496	1.577	1.614	1.675	1.746	
.8429	1.402	1.438	1.486	1.530	1.566	1.635	1.672	1.721	1.798	
.8767	1.429	1.474	1.511	1.550	1.575	1.647	1.672	1.704	1.770	
.9108	1.524	1.581	1.616	1.617	1.646	1.728	1.758	1.775	1.836	
.9437	1.727	1.751	1.788	1.795	1.808	1.896	1.919	1.929	1.983	
.9768	2.171	2.195	2.229	2.217	2.210	2.308	2.349	2.342	2.332	
Vane										
.0000	4.475	3.944	4.382	4.434	4.071	3.998	3.980	3.909	3.752	
.0250	5.711	5.548	5.810	5.873	5.549	5.859	5.923	5.886	5.551	
.0500	5.987	5.648	6.026	5.997	5.741	5.937	6.121	5.997	5.626	
.1000	6.036	5.715	6.033	5.997	5.729	5.920	6.098	5.954	5.556	
.1500	5.984	5.654	5.975	5.946	5.684	5.839	6.012	5.872	5.478	
.2000	5.916	5.639	5.918	5.910	5.626	5.781	5.946	5.818	5.444	
.3000	5.392	5.186	5.464	5.488	5.213	5.352	5.490	5.387	5.087	
.4000	4.735	4.524	4.825	4.867	4.605	4.699	4.801	4.729	4.510	
.5000	4.202	3.940	4.293	4.343	4.086	4.160	4.237	4.188	4.023	
.6000	3.573	3.344	3.693	3.756	3.531	3.583	3.643	3.627	3.536	
.7000	3.055	2.842	3.162	3.232	3.030	3.093	3.150	3.151	3.113	
.8000	2.662	2.450	2.718	2.771	2.626	2.655	2.700	2.718	2.712	
.9000	2.447	2.274	2.394	2.446	2.310	2.334	2.343	2.370	2.401	
Flap										
.0000	2.098	1.666	1.749	1.669	1.483	1.418	1.415	1.410	1.404	
.0125	2.337	2.280	2.339	2.322	2.207	2.186	2.199	2.182	2.182	
.0250	2.662	2.678	2.769	2.786	2.670	2.670	2.692	2.692	2.698	
.0500	2.619	2.624	2.731	2.807	2.720	2.731	2.752	2.783	2.842	
.0750	2.546	2.548	2.670	2.768	2.690	2.690	2.721	2.769	2.842	
.1000	2.429	2.365	2.521	2.626	2.519	2.537	2.571	2.615	2.698	
.1500	2.064	1.994	2.070	2.154	2.089	2.093	2.113	2.162	2.271	
.2000	1.923	1.794	1.854	1.910	1.841	1.852	1.876	1.906	1.954	
.4000	1.734	1.699	1.715	1.741	1.717	1.737	1.738	1.766	1.795	
.6000	1.730	1.690	1.734	1.753	1.741	1.754	1.755	1.761	1.772	
.8000	1.687	1.672	1.708	1.714	1.699	1.699	1.700	1.707	1.729	

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	1.441	1.493	2.096	1.021	.944	1.171	1.481	1.615	1.889	
.0250	1.328	1.362	2.149	.994	.773	.855	.962	1.026	1.066	
.0500	1.227	1.246	1.905	.964	.690	.638	.637	.638	.643	
.0750	1.178	1.189	1.584	.901	.664	.591	.533	.516	.513	
.1000	1.159	1.152	1.381	.849	.655	.574	.496	.479	.450	
.1500	1.129	1.113	1.298	.825	.676	.580	.484	.444	.418	
.2000	1.098	1.088	1.143	.807	.681	.586	.484	.436	.409	
.2500	1.083	1.070	1.086	.795	.696	.597	.499	.462	.421	
.3000	1.073	1.049	1.019	.807	.708	.612	.516	.473	.435	
.3500	1.061	1.037	.978	.819	.729	.629	.527	.481	.458	
.4000	1.037	1.021	.927	.807	.729	.644	.536	.507	.464	
.4500	1.024	1.006	.883	.819	.743	.644	.548	.521	.487	
.5000	.975	.961	.873	.801	.729	.644	.559	.521	.484	
.5500	.932	.939	.845	.777	.714	.644	.553	.527	.493	
.6000	.880	.894	.787	.741	.679	.615	.539	.507	.493	
.6500	.846	.863	.740	.705	.643	.586	.516	.487	.473	
.7517	.639	.748	.670	.602	.561	.510	.455	.439	.415	
.7934	.604	.693	.635	.557	.507	.475	.424	.405	.392	
.8094	.604	.684	.606	.524	.484	.458	.409	.393	.383	
.8127	.552	.651	.632	.548	.493	.455	.406	.390	.389	
.8439	.586	.635	.565	.512	.463	.438	.386	.382	.363	
.8772	.558	.593	.505	.428	.389	.336	.294	.276	.268	
.9117	.528	.559	.483	.407	.354	.307	.268	.262	.254	
.9448	.503	.528	.445	.377	.310	.273	.233	.222	.219	
.9781	.472	.499	.397	.349	.286	.246	.213	.208	.193	
Vane										
.0250	1.640	1.810	1.378	1.437	1.159	.983	.888	.866	.801	
.0500	.935	.605	.470	.491	.319	.215	.161	.142	.118	
.1000	.574	.404	.171	.154	.065	.032	.000	.000	.000	
.1500	.485	.365	.149	.099	.044	.014	.000	.000	.000	
.2000	.432	.362	.137	.081	.041	.020	.009	.000	.000	
.3000	.399	.356	.133	.078	.047	.035	.014	.023	.035	
.4000	.374	.353	.133	.078	.056	.055	.040	.051	.040	
.5000	.353	.347	.171	.096	.080	.081	.058	.063	.052	
.6000	.365	.362	.194	.127	.097	.104	.092	.091	.089	
.7000	.408	.395	.229	.169	.158	.159	.147	.154	.144	
.8000	.506	.505	.337	.295	.283	.296	.282	.279	.271	
.9200	1.031	.982	.870	.840	.802	.806	.807	.795	.784	
Flap										
.0125	.859	.459	.381	.271	.180	.162	.150	.160	.147	
.0250	.466	.256	.124	.038	.009	.012	.000	.000	.000	
.0500	.294	.243	.029	.009	.009	.006	.000	.000	.000	
.0750	.285	.258	.025	.009	.009	.014	.000	.006	.000	
.1000	.251	.252	.102	.039	.027	.029	.006	.003	.000	
.1500	.227	.286	.137	.063	.062	.058	.040	.040	.029	
.2000	.221	.310	.159	.096	.089	.075	.061	.054	.049	
.4000	.301	.486	.197	.196	.186	.174	.141	.142	.124	
.6000	.456	.572	.352	.319	.307	.287	.259	.236	.281	
.8000	.607	.711	.533	.488	.487	.438	.418	.390	.398	
.9000	.764	.854	.718	.681	.670	.620	.588	.573	.571	

TABLE XV.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(d) Extended double slotted flap configuration; $\delta_f = 60^\circ$; $\delta_c = 20^\circ$; $\frac{y}{b/2} = 0.43$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	1.032	.702	1.412	2.264	2.815	3.224	3.546	3.393	3.267	
.0125	.789	1.058	1.331	2.339	2.891	3.265	3.528	3.445	3.332	
.0250	.891	1.110	1.378	2.413	3.009	3.376	3.617	3.558	3.381	
.0500	.895	1.101	1.393	2.277	3.042	3.326	3.522	3.419	3.298	
.0750	.959	1.123	1.458	2.165	3.400	3.341	3.467	3.378	3.298	
.1000	1.003	1.150	1.502	2.041	3.806	3.652	3.594	3.491	3.392	
.1500	1.016	1.150	1.511	1.873	3.263	4.431	4.493	4.063	3.725	
.2000	1.061	1.193	1.545	1.836	2.451	3.822	4.453	4.130	3.736	
.2500	1.080	1.187	1.548	1.798	2.015	2.944	3.876	3.832	3.546	
.3000	1.121	1.236	1.539	1.805	1.863	2.454	3.398	3.563	3.389	
.3500	1.141	1.227	1.508	1.758	1.736	2.055	2.870	3.139	3.082	
.4000	1.157	1.236	1.508	1.749	1.706	1.880	2.577	2.884	2.898	
.4500	1.201	1.273	1.486	1.752	1.685	1.778	2.372	2.670	2.739	
.5000	1.176	1.267	1.446	1.699	1.642	1.673	2.136	2.422	2.506	
.5500	1.227	1.297	1.449	1.705	1.654	1.667	2.038	2.303	2.392	
.6000	1.265	1.325	1.446	1.693	1.642	1.650	1.963	2.182	2.281	
.6993	1.128	1.319	1.412	1.640	1.609	1.629	1.807	1.983	2.065	
.7507	1.112	1.303	1.412	1.596	1.606	1.615	1.755	1.887	1.966	
.7704	1.227	1.368	1.455	1.646	1.636	1.662	1.795	1.922	1.992	
.8099	1.227	1.420	1.502	1.680	1.694	1.714	1.816	1.931	1.992	
.8512	1.345	1.491	1.554	1.718	1.712	1.761	1.844	1.922	1.977	
.8911	1.431	1.616	1.681	1.820	1.836	1.866	1.948	2.009	2.026	
.9322	1.693	1.840	1.867	1.997	2.036	2.073	2.136	2.165	2.176	
.9724	2.163	2.383	2.381	2.522	2.542	2.626	2.686	2.682	2.611	
Vane										
.0000	5.828	4.785	5.105	5.532	5.312	5.314	5.335	5.202	4.850	
.0250	7.118	6.744	6.610	7.072	7.084	7.869	7.536	7.317	6.693	
.0500	7.080	6.701	6.554	6.945	6.972	7.267	7.438	7.219	6.563	
.1000	7.109	6.671	6.533	6.861	6.878	7.174	7.340	7.069	6.330	
.1500	7.157	6.658	6.551	6.849	6.845	7.139	7.320	7.026	6.253	
.2000	7.291	6.876	6.762	7.088	7.017	7.311	7.508	7.239	6.452	
.3000	6.569	6.220	6.139	6.476	6.399	6.652	6.825	6.546	5.833	
.4000	5.850	5.524	5.517	5.845	5.778	6.031	6.228	5.962	5.270	
.5000	5.243	4.892	4.926	5.215	5.081	5.227	5.361	5.179	4.637	
.6000	4.479	4.214	4.276	4.560	4.433	4.542	4.663	4.500	4.065	
.7000	3.901	3.696	3.762	4.041	3.919	4.008	4.119	3.982	3.631	
.8000	3.377	3.144	3.248	3.482	3.363	3.428	3.510	3.419	3.165	
.9000	2.907	2.714	2.811	2.979	2.866	2.918	2.971	2.907	2.713	
Flap										
.0000	2.566	1.929	2.464	2.516	2.327	2.384	2.464	2.402	2.242	
.0125	2.671	2.616	2.759	2.938	2.806	2.842	2.908	2.847	2.671	
.0250	3.067	3.233	3.272	3.569	3.442	3.475	3.551	3.494	3.279	
.0500	3.051	3.263	3.266	3.609	3.472	3.489	3.571	3.514	3.232	
.0750	2.923	3.156	3.186	3.532	3.406	3.413	3.484	3.448	3.279	
.1000	3.016	3.039	3.074	3.382	3.227	3.247	3.288	3.263	3.111	
.1500	2.320	2.362	2.418	2.649	2.533	2.504	2.539	2.535	2.463	
.2000	2.173	2.012	2.155	2.270	2.176	2.160	2.193	2.170	2.088	
.4000	1.586	1.610	1.690	1.817	1.794	1.772	1.790	1.803	1.770	
.6000	1.399	1.491	1.542	1.699	1.712	1.702	1.697	1.714	1.668	
.8000	1.067	1.316	1.390	1.562	1.630	1.615	1.608	1.613	1.563	

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	1.377	1.031	1.440	1.103	1.230	1.443	1.692	1.731	1.719	
.0250	1.249	1.049	1.381	.888	.924	.997	1.110	1.136	1.134	
.0500	1.198	1.046	1.303	.780	.785	.770	.787	.795	.787	
.0750	1.150	1.046	1.229	.749	.742	.682	.677	.665	.656	
.1000	1.147	1.046	1.176	.742	.721	.650	.620	.604	.585	
.1500	1.105	1.043	1.068	.736	.721	.618	.568	.552	.514	
.2000	1.077	1.043	.972	.742	.721	.612	.553	.526	.491	
.2500	1.058	1.040	.923	.752	.724	.612	.542	.517	.480	
.3000	1.029	1.024	.867	.755	.718	.609	.545	.520	.480	
.3500	1.022	1.018	.845	.755	.727	.624	.553	.529	.491	
.4000	1.003	1.009	.811	.755	.724	.618	.553	.529	.500	
.4500	.959	.978	.786	.749	.712	.618	.553	.529	.500	
.5000	.923	.960	.771	.730	.697	.606	.545	.514	.486	
.5500	.872	.926	.740	.705	.679	.595	.539	.514	.486	
.6000	.812	.886	.672	.668	.636	.557	.507	.488	.466	
.6993	.623	.788	.697	.599	.591	.519	.481	.454	.438	
.7499	.569	.745	.647	.544	.539	.475	.455	.431	.412	
.7700	.527	.724	.635	.534	.509	.466	.435	.425	.401	
.7744	.543	.739	.635	.531	.521	.463	.447	.422	.406	
.8110	.540	.669	.573	.491	.488	.431	.403	.379	.366	
.8519	.498	.592	.529	.394	.385	.335	.303	.283	.278	
.8919	.463	.574	.495	.363	.342	.303	.291	.277	.264	
.9332	.441	.524	.455	.332	.306	.280	.268	.251	.241	
.9738	.396	.469	.446	.307	.288	.262	.248	.246	.227	
Vane										
.0250	1.355	.629	1.050	.932	.739	.656	.631	.621	.565	
.0500	.626	.255	.489	.298	.206	.160	.156	.142	.134	
.1000	.358	.181	.307	.093	.045	.029	.043	.061	.043	
.1500	.217	.181	.223	.022	.027	.026	.014	.043	.028	
.2000	.179	.184	.214	.025	.027	.035	.066	.049	.048	
.3000	.147	.230	.201	.037	.042	.061	.087	.055	.048	
.4000	.131	.251	.198	.050	.079	.082	.066	.078	.063	
.5000	.131	.261	.217	.084	.097	.108	.101	.107	.097	
.6000	.144	.285	.238	.109	.142	.157	.141	.136	.125	
.7000	.201	.334	.294	.174	.197	.198	.213	.211	.182	
.8000	.345	.463	.437	.339	.358	.367	.375	.358	.332	
.9200	1.048	1.061	1.093	1.037	1.021	1.044	1.052	1.032	.957	
Flap										
.0125	.498	.163	.601	.323	.273	.292	.320	.321	.293	
.0250	.103	.092	.291	.040	.027	.035	.009	.049	.048	
.0500	.016	.144	.220	.003	.018	.006	.014	.020	.017	
.0750	.006	.193	.207	.006	.018	.017	.012	.017	.023	
.1000	.000	.221	.207	.016	.042	.020	.032	.040	.043	
.1500	.051	.279	.226	.043	.055	.035	.032	.040	.045	
.2000	.064	.337	.232	.068	.064	.061	.049	.075	.063	
.4000	.192	.577	.180	.168	.170	.149	.135	.127	.131	
.6000	.300	.595	.372	.276	.267	.227	.225	.229	.219	
.8000	.470	.751	.430	.444	.439	.402	.380	.367	.352	
.9900	.569	.788	.607	.581	.579	.554	.516	.500	.497	

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TABLE XV.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Continued

(e) Extended double slotted flap configuration; $\delta_f = 60^\circ$; $\delta_a = 20^\circ$; $\frac{y}{b/2} = 0.55$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	.764	2.049	2.235	2.123	2.552	2.885	2.911	2.838	2.709	
.0125	1.018	1.788	2.254	2.214	2.694	2.937	2.934	2.882	2.810	
.0250	1.003	1.511	2.235	2.163	2.623	2.916	2.968	2.926	2.842	
.0500	1.046	1.350	2.216	2.196	2.761	3.093	3.095	2.991	2.853	
.0750	1.058	1.316	2.245	2.325	2.746	3.056	3.006	2.880	2.801	
.1000	1.089	1.307	2.242	2.437	2.752	2.971	2.928	2.849	2.807	
.1500	1.123	1.298	2.238	2.428	2.962	2.931	2.882	2.846	2.830	
.2000	1.144	1.289	2.153	2.223	3.348	3.131	3.009	2.937	2.896	
.2500	1.172	1.295	2.096	2.108	3.466	3.630	3.358	3.171	3.015	
.3000	1.187	1.298	2.022	2.003	3.186	3.763	3.551	3.288	3.064	
.3500	1.208	1.322	1.956	1.949	2.817	3.560	3.559	3.322	3.066	
.4000	1.245	1.332	1.880	1.883	2.421	3.174	3.424	3.236	2.997	
.4500	1.261	1.328	1.794	1.855	2.165	2.841	3.277	3.142	2.931	
.5000	1.303	1.362	1.734	1.831	2.012	2.574	3.104	3.031	2.870	
.6194	1.221	1.325	1.549	1.723	1.723	2.044	2.559	2.647	2.623	
.6644	1.251	1.335	1.505	1.684	1.688	1.867	2.314	2.419	2.441	
.7066	1.288	1.374	1.524	1.687	1.661	1.864	2.271	2.390	2.409	
.7594	1.343	1.408	1.505	1.678	1.620	1.809	2.136	2.219	2.236	
.8101	1.423	1.493	1.581	1.762	1.702	1.850	2.141	2.228	2.277	
.8622	1.552	1.623	1.686	1.843	1.791	1.919	2.156	2.211	2.254	
.9142	1.748	1.848	1.870	2.009	1.982	2.096	2.274	2.296	2.323	
.9652	2.291	2.399	2.381	2.488	2.499	2.635	2.738	2.698	2.651	
Vane										
.0000	4.686	3.903	4.178	4.440	4.366	4.459	4.352	4.120	3.807	
.0250	7.014	6.779	6.175	6.726	6.962	7.496	7.306	6.838	6.156	
.0500	6.977	6.743	6.156	6.681	6.941	7.421	7.283	6.840	6.139	
.1000	7.180	6.949	6.340	6.813	7.127	7.639	7.433	6.886	6.044	
.1500	7.232	6.977	6.391	6.846	7.177	7.688	7.459	6.875	5.972	
.2000	7.217	6.952	6.410	6.837	7.183	7.703	7.461	6.849	5.931	
.3000	6.796	6.557	6.134	6.452	6.753	7.285	7.084	6.513	5.637	
.4000	6.113	5.919	5.563	5.822	6.183	6.694	6.482	5.920	5.098	
.5000	5.505	5.305	5.115	5.259	5.514	5.972	5.862	5.382	4.652	
.6000	4.791	4.627	4.531	4.608	4.817	5.189	5.098	4.698	4.095	
.7000	4.183	4.040	3.988	4.030	4.180	4.511	4.453	4.123	3.631	
.8000	3.613	3.496	3.502	3.488	3.602	3.873	3.839	3.575	3.208	
.9000	3.085	2.973	3.023	2.988	3.050	3.256	3.234	3.054	2.790	
Flap										
.0000	1.849	1.651	2.242	2.129	1.968	2.076	2.110	1.996	1.839	
.0125	2.812	2.824	2.946	2.916	2.912	3.105	3.124	2.972	2.749	
.0250	3.297	3.380	3.397	3.392	3.431	3.664	3.689	3.507	3.251	
.0500	3.417	3.523	3.515	3.530	3.567	3.801	3.822	3.632	3.395	
.0750	3.411	3.554	3.556	3.554	3.564	3.801	3.824	3.638	3.401	
.1000	3.358	3.432	3.432	3.398	3.401	3.592	3.617	3.456	3.222	
.1500	2.776	2.751	2.746	2.696	2.670	2.690	2.723	2.664	2.577	
.2000	2.355	2.365	2.407	2.334	2.310	2.406	2.406	2.313	2.199	
.4000	1.920	1.906	1.908	1.825	1.782	1.818	1.816	1.769	1.755	
.6000	1.714	1.748	1.680	1.605	1.559	1.551	1.553	1.541	1.568	
.8000	1.359	1.505	1.416	1.346	1.295	1.281	1.288	1.291	1.360	

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	1.193	.763	1.556	1.160	1.457	1.763	1.945	1.974	1.989	
.0250	1.162	.827	1.276	.904	1.050	1.183	1.306	1.313	1.346	
.0500	1.126	.882	1.105	.768	.817	.858	.916	.903	.925	
.0750	1.084	.882	1.010	.735	.726	.733	.746	.741	.749	
.1000	1.077	.906	.965	.723	.693	.687	.677	.652	.674	
.1500	1.049	.906	.892	.714	.661	.623	.591	.567	.576	
.2000	1.021	.915	.857	.711	.664	.603	.571	.536	.539	
.2500	1.012	.924	.829	.714	.664	.603	.559	.524	.522	
.3000	.991	.909	.813	.714	.661	.594	.556	.527	.510	
.3500	.981	.918	.791	.723	.658	.597	.548	.513	.510	
.4000	.957	.921	.768	.705	.658	.594	.548	.507	.504	
.4500	.911	.894	.736	.702	.640	.591	.542	.501	.490	
.5000	.880	.888	.724	.681	.628	.577	.533	.487	.473	
.6195	.758	.809	.708	.633	.581	.539	.484	.453	.461	
.6834	.715	.778	.692	.599	.566	.522	.487	.447	.461	
.7080	.693	.781	.660	.583	.519	.487	.452	.410	.415	
.7137	.675	.778	.667	.566	.528	.481	.452	.410	.427	
.7603	.632	.699	.616	.527	.504	.473	.435	.410	.415	
.8119	.543	.596	.584	.467	.431	.403	.369	.342	.354	
.8643	.509	.578	.556	.443	.407	.371	.343	.316	.323	
.9162	.460	.502	.521	.404	.378	.333	.308	.274	.285	
.9669	.408	.435	.489	.380	.339	.319	.305	.276	.280	
Vane										
.0250	1.129	.614	1.359	1.114	.953	.916	.865	.809	.749	
.0500	.344	.128	.692	.380	.271	.258	.245	.222	.210	
.1000	.092	.052	.432	.142	.083	.081	.072	.060	.081	
.1500	.031	.085	.356	.096	.071	.075	.069	.057	.075	
.2000	.046	.119	.324	.084	.080	.084	.078	.069	.081	
.3000	.067	.146	.298	.111	.062	.099	.118	.077	.101	
.4000	.086	.176	.292	.114	.112	.122	.115	.105	.115	
.5000	.110	.192	.305	.123	.133	.142	.133	.123	.127	
.6000	.144	.225	.324	.175	.171	.183	.170	.154	.167	
.7000	.196	.292	.362	.235	.233	.249	.225	.219	.228	
.8000	.371	.426	.514	.392	.398	.426	.415	.382	.366	
.9200	1.132	1.131	1.200	1.108	1.142	1.215	1.196	1.111	1.032	
Flap										
.0125	.261	.134	.797	.494	.354	.383	.386	.350	.334	
.0250	.091	.043	.403	.163	.094	.101	.101	.085	.107	
.0500	.031	.085	.289	.096	.062	.072	.061	.060	.069	
.0750	.021	.097	.279	.133	.086	.084	.078	.063	.081	
.1000	.052	.152	.276	.108	.086	.090	.081	.068	.078	
.1500	.083	.219	.267	.120	.106	.107	.095	.088	.101	
.2000	.117	.258	.286	.143	.124	.119	.112	.097	.115	
.4000	.242	.565	.241	.229	.212	.203	.187	.160	.184	
.6000	.344	.666	.318	.307	.289	.270	.251	.236	.265	
.8000	.454	.663	.505	.437	.416	.386	.372	.353	.375	
.9000	.558	.733	.613	.548	.502	.496	.467	.453	.484	

TABLE XV.- PRESSURE COEFFICIENT C_p AT THE SIX SPANWISE STATIONS THROUGH THE ANGLE-OF-ATTACK RANGE - Concluded

(f) Extended double slotted flap configuration; $\delta_f = 60^\circ$; $\delta_c = 20^\circ$; $\frac{y}{b/2} = 0.72$

Upper surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0000	1.102	2.046	2.189	2.288	2.563	2.452	2.454	2.317	2.213	
.0125	1.255	2.049	2.169	2.187	2.474	2.393	2.399	2.288	2.202	
.0250	1.231	2.055	2.169	2.208	2.501	2.408	2.422	2.326	2.236	
.0500	1.225	2.107	2.189	2.248	2.590	2.461	2.501	2.403	2.283	
.0750	1.218	2.165	2.210	2.303	2.721	2.537	2.559	2.443	2.297	
.1000	1.182	2.159	2.189	2.263	2.762	2.555	2.539	2.394	2.283	
.1500	1.203	2.214	2.210	2.291	2.777	2.590	2.492	2.368	2.285	
.2000	1.231	2.125	2.251	2.337	2.736	2.517	2.452	2.374	2.311	
.2500	1.243	1.776	2.292	2.362	2.712	2.455	2.422	2.383	2.329	
.3000	1.277	1.408	2.292	2.395	2.718	2.440	2.414	2.388	2.332	
.3500	1.298	1.310	2.332	2.457	2.727	2.417	2.428	2.408	2.349	
.4000	1.305	1.316	2.353	2.527	2.801	2.414	2.440	2.414	2.355	
.4500	1.326	1.343	2.373	2.582	2.937	2.473	2.478	2.428	2.363	
.5000	1.348	1.395	2.414	2.687	3.207	2.593	2.556	2.477	2.395	
.5500	1.348	1.395	2.394	2.662	3.296	2.669	2.583	2.477	2.381	
.6000	1.338	1.420	2.373	2.665	3.391	2.745	2.624	2.491	2.392	
.6500	1.366	1.445	2.332	2.631	3.376	2.769	2.629	2.468	2.366	
.7000	1.403	1.451	2.353	2.613	3.332	2.751	2.644	2.457	2.346	
.7500	1.458	1.524	2.332	2.589	3.240	2.745	2.644	2.463	2.343	
.8000	1.471	1.509	2.292	2.543	3.133	2.710	2.624	2.437	2.323	
.8500	1.505	1.543	2.251	2.438	2.896	2.587	2.554	2.377	2.257	
.9000	1.545	1.601	2.210	2.349	2.688	2.508	2.478	2.320	2.219	

Lower surface

x/c	C_p for -									
	$\alpha = -4^\circ$	$\alpha = 0^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	$\alpha = 22^\circ$	$\alpha = 24^\circ$	$\alpha = 26^\circ$
Wing										
.0125	.886	.800	1.269	1.141	1.394	1.469	1.600	1.586	1.585	
.0250	.917	.776	1.125	.945	1.104	1.164	1.268	1.269	1.271	
.0500	.911	.791	.982	.791	.878	.897	.959	.966	.971	
.0750	.898	.816	.941	.754	.792	.795	.834	.840	.839	
.1000	.902	.825	.900	.742	.745	.742	.758	.763	.769	
.1500	.892	.846	.880	.724	.697	.683	.679	.677	.674	
.2000	.905	.853	.880	.733	.694	.675	.656	.649	.651	
.2500	.917	.868	.900	.745	.709	.678	.656	.643	.640	
.3000	.923	.889	.921	.776	.733	.707	.685	.669	.648	
.3500	.929	.905	.921	.797	.754	.722	.679	.651	.643	
.4000	.917	.905	.880	.785	.730	.680	.644	.620	.599	
.4500	.917	.917	.880	.779	.727	.678	.624	.597	.571	
.5000	.920	.923	.880	.797	.736	.686	.618	.586	.559	
.5500	.911	.932	.900	.810	.754	.704	.647	.611	.588	
.6000	.908	.945	.982	.819	.771	.727	.673	.640	.628	
.6500	.932	.972	1.064	.859	.810	.774	.726	.694	.692	
.7000	.960	.997	1.187	.896	.872	.836	.796	.771	.784	
.7500	1.028	1.037	1.350	.975	.958	.933	.909	.880	.896	
.8000	1.102	1.107	1.514	1.089	1.092	1.088	1.064	1.060	1.066	
.8500	1.206	1.215	1.760	1.273	1.308	1.302	1.283	1.286	1.306	
.9000	1.342	1.371	2.026	1.558	1.608	1.593	1.568	1.543	1.565	
.9500	1.517	1.543	2.251	1.810	1.881	1.898	1.892	1.854	1.888	

TABLE XVI.- INTEGRATED SECTION DATA FOR THE PLAIN WING CONFIGURATION

(a) $\beta = 0^{\circ}$

$\frac{y}{b/2}$	α, deg.	$c_{n,w}$	$c_{m,w}$
0.21	-4	-.1615	.0260
0.21	0	-.0373	.0176
0.21	4	.1100	.0114
0.21	8	.2964	.0019
0.21	12	.5335	.0114
0.21	16	.7997	.0174
0.21	20	1.1453	.0258
0.21	22	1.2985	.0245
0.21	24	1.4313	-.0060
0.21	26	1.3621	-.0918
0.30	-4	-.2150	.0433
0.30	0	-.0699	.0334
0.30	4	.1055	.0281
0.30	8	.3287	.0376
0.30	12	.5976	.0473
0.30	16	.9234	.0319
0.30	20	1.3054	-.0186
0.30	22	1.4273	-.0552
0.30	24	1.5479	-.0978
0.30	26	1.5209	-.1608
0.43	-4	-.2292	.0249
0.43	0	-.0564	.0188
0.43	4	.1400	.0229
0.43	8	.4073	.0309
0.43	12	.7397	.0186
0.43	16	1.0985	-.0628
0.43	20	1.3624	-.1490
0.43	22	1.4163	-.1795
0.43	24	1.5177	-.2207
0.43	26	1.5109	-.2525
0.55	-4	-.2949	.0358
0.55	0	-.0799	.0303
0.55	4	.1488	.0235
0.55	8	.5002	.0213
0.55	12	.8978	-.0814
0.55	16	1.0732	-.1595
0.55	20	1.1820	-.1886
0.55	22	1.2037	-.1973
0.55	24	1.2670	-.2201
0.55	26	1.2565	-.2296
0.72	-4	-.2218	-.0113
0.72	0	-.0173	-.0100
0.72	4	.2882	-.0122
0.72	8	.7557	-.1599
0.72	12	.7824	-.1557
0.72	16	.8557	-.1619
0.72	20	.9071	-.1728
0.72	22	.9301	-.1763
0.72	24	.9602	-.1842
0.72	26	.9620	-.1898

(b) $\beta = -10^{\circ}$

$\frac{y}{b/2}$	α, deg	$c_{n,w}$	$c_{m,w}$
0.21	-4	-.1354	.0249
0.21	0	-.0460	.0190
0.21	4	.0715	.0149
0.21	8	.2030	.0144
0.21	12	.3875	.0254
0.21	16	.6301	.0396
0.21	20	.8463	.0519
0.21	22	1.0114	.0579
0.21	24	1.1798	.0632
0.21	26	1.2370	.0070
0.30	-4	-.2028	.0392
0.30	0	-.0657	.0306
0.30	4	.0795	.0275
0.30	8	.2709	.0233
0.30	12	.5379	.0222
0.30	16	.8409	.0246
0.30	20	1.1189	.0236
0.30	22	1.2632	.0087
0.30	24	1.3809	-.0196
0.30	26	1.4455	-.0731
0.43	-4	-.2099	.0223
0.43	0	-.0551	.0203
0.43	4	.1074	.0178
0.43	8	.2822	.0176
0.43	12	.5744	.0051
0.43	16	.9208	-.0223
0.43	20	1.1870	-.0582
0.43	22	1.2972	-.0922
0.43	24	1.3769	-.1351
0.43	26	1.4146	-.1894
0.55	-4	-.2604	.0288
0.55	0	-.0850	.0291
0.55	4	.0970	.0302
0.55	8	.3197	.0255
0.55	12	.6385	-.0152
0.55	16	1.0013	-.0973
0.55	20	1.2002	-.1623
0.55	22	1.2245	-.1913
0.55	24	1.2413	-.2088
0.55	26	1.2319	-.2280
0.72	-4	-.1926	-.0114
0.72	0	.0065	-.0087
0.72	4	.2754	-.0240
0.72	8	.6005	-.1118
0.72	12	.8387	-.1929
0.72	16	.8909	-.2039
0.72	20	.9473	-.2230
0.72	22	.9894	-.2325
0.72	24	.9912	-.2266
0.72	26	.9520	-.2119

(c) $\beta = 10^{\circ}$

$\frac{y}{b/2}$	α, deg	$c_{n,w}$	$c_{m,w}$
0.21	-4	-.1389	.0240
0.21	0	-.0282	.0209
0.21	4	.0863	.0377
0.21	8	.3069	.0103
0.21	12	.5503	.0123
0.21	16	.8762	.0078
0.21	20	1.3248	-.0068
0.21	22	1.1784	-.1583
0.21	24	1.2191	-.1902
0.21	26	1.2796	-.2113
0.30	-4	-.1973	.0326
0.30	0	-.0542	.0340
0.30	4	.0660	.0489
0.30	8	.3918	.0486
0.30	12	.6957	.0601
0.30	16	1.1452	.0163
0.30	20	1.5066	-.0887
0.30	22	1.1753	-.1733
0.30	24	1.1728	-.1850
0.30	26	1.2178	-.2019
0.43	-4	-.2468	.0264
0.43	0	-.0367	.0213
0.43	4	.1909	.0169
0.43	8	.5186	.0435
0.43	12	1.0032	-.0294
0.43	16	1.3182	-.1495
0.43	20	1.3923	-.1924
0.43	22	1.1377	-.1916
0.43	24	1.1573	-.2030
0.43	26	1.2057	-.2205
0.55	-4	-.3045	.0351
0.55	0	-.0605	.0307
0.55	4	.1591	.0275
0.55	8	.7274	-.0535
0.55	12	1.0047	-.1368
0.55	16	1.0739	-.1539
0.55	20	1.1024	-.1641
0.55	22	.9619	-.1510
0.55	24	.9983	-.1637
0.55	26	1.0509	-.1794
0.72	-4	-.2045	-.0131
0.72	0	.0167	-.0088
0.72	4	.2925	-.0065
0.72	8	.7011	-.1094
0.72	12	.7525	-.1146
0.72	16	.8090	-.1240
0.72	20	.8378	-.1354
0.72	22	.8327	-.1414
0.72	24	.8528	-.1495
0.72	26	.9126	-.1641

TABLE XVII.- INTEGRATED SECTION DATA FOR THE DOUBLE SLOTTED FLAP CONFIGURATION

(a) $\delta_f = 50^\circ$

$\frac{y}{b/2}$	α, deg	$c_{n,w}$	$c_{m,w}$	$c_{n,v}$	$c_{m,v}$	$c_{n,f}$	$c_{m,f}$
0.21	-4	.5283	-.2886	2.3262	-1.0399	1.3064	-.5333
0.21	0	.6670	-.2926	2.2595	-1.0007	1.2938	-.5275
0.21	4	.8027	-.2903	2.2341	-.9841	1.2886	-.5169
0.21	8	.9659	-.2853	2.2875	-1.0065	1.2836	-.5344
0.21	12	1.1959	-.2768	2.4558	-1.0791	1.3374	-.5615
0.21	16	1.4912	-.2808	2.7908	-1.2107	1.5014	-.6337
0.21	20	1.8175	-.2809	2.9921	-1.3257	1.6596	-.7075
0.21	22	1.9061	-.2890	3.0405	-1.3684	1.7265	-.7376
0.21	24	1.7585	-.3507	2.9366	-1.3427	1.7194	-.7259
0.21	26	1.7351	-.4031	2.6233	-1.2043	1.6555	-.7072
0.30	-4	.7114	-.3770	3.7309	-1.6381	1.4264	-.5506
0.30	0	.8514	-.3728	3.5806	-1.5387	1.4011	-.5523
0.30	4	1.0060	-.3522	3.4027	-1.4508	1.3507	-.5378
0.30	8	1.2011	-.3432	3.5318	-1.4932	1.3831	-.5569
0.30	12	1.4871	-.3313	3.5902	-1.5104	1.4281	-.5769
0.30	16	1.8277	-.3578	3.8968	-1.6686	1.6020	-.6433
0.30	20	2.1753	-.4172	4.2600	-1.8349	1.7625	-.7018
0.30	22	2.2027	-.4373	4.0663	-1.7582	1.7887	-.7147
0.30	24	2.0384	-.4415	3.2293	-1.4285	1.7146	-.6946
0.30	26	1.7987	-.4428	2.5729	-1.1526	1.6413	-.6853
0.43	-4	.8897	-.4350	4.2497	-1.8414	1.4765	-.5024
0.43	0	1.0626	-.4282	4.1722	-1.8096	1.4761	-.5106
0.43	4	1.2459	-.4099	4.1438	-1.7976	1.4750	-.5143
0.43	8	1.4953	-.3926	4.1874	-1.8208	1.5182	-.5361
0.43	12	1.8065	-.4057	4.1980	-1.8327	1.5698	-.5627
0.43	16	2.0646	-.4653	3.9248	-1.7234	1.6584	-.6165
0.43	20	2.0760	-.4829	3.5415	-1.5611	1.6924	-.6511
0.43	22	1.9704	-.4664	3.1327	-1.3821	1.6398	-.6437
0.43	24	1.8177	-.4426	2.3658	-1.0547	1.5610	-.6499
0.43	26	1.7345	-.4469	2.1425	-.9649	1.5782	-.6687
0.55	-4	1.0926	-.5233	4.1122	-1.7923	1.5983	-.5799
0.55	0	1.2774	-.5113	4.0804	-1.7706	1.5993	-.5808
0.55	4	1.5406	-.4821	4.0396	-1.7672	1.5681	-.5674
0.55	8	1.8511	-.5091	4.0037	-1.7775	1.6572	-.6038
0.55	12	1.9882	-.5279	3.3677	-1.4868	1.5391	-.5594
0.55	16	1.9284	-.5246	2.8419	-1.2508	1.6107	-.6472
0.55	20	1.9127	-.5213	2.6721	-1.1750	1.6193	-.6613
0.55	22	1.8479	-.5107	2.5182	-1.1142	1.6083	-.6651
0.55	24	1.7071	-.4824	2.1459	-.9564	1.5498	-.6547
0.55	26	1.6357	-.4693	2.0263	-.9090	1.5112	-.6451
0.72	-4	.1353	.0825				
0.72	0	.3270	.1156				
0.72	4	1.0548	-.0744				
0.72	8	.9258	-.0661				
0.72	12	.8739	-.0793				
0.72	16	.8802	-.0893				
0.72	20	.8385	-.0854				
0.72	22	.7871	-.0764				
0.72	24	.7563	-.0740				
0.72	26	.7606	-.0764				

TABLE XVII.- INTEGRATED SECTION DATA FOR THE DOUBLE SLOTTED FLAP CONFIGURATION - Continued

(b) $\delta_f = 60^\circ$

$\frac{y}{b/2}$	α, deg	$c_{n,w}$	$c_{m,w}$	$c_{n,v}$	$c_{m,v}$	$c_{n,f}$	$c_{m,f}$
0.21	-4	.5592	-.2878	2.2861	-.9860	1.3752	-.5882
0.21	0	.7005	-.2973	2.3379	-1.0002	1.3849	-.5877
0.21	4	.8384	-.2992	2.3635	-1.0071	1.3693	-.5832
0.21	8	.9995	-.2960	2.4619	-1.0465	1.3961	-.5972
0.21	12	1.2309	-.2852	2.6913	-1.1351	1.4652	-.6278
0.21	16	1.4657	-.2793	2.9826	-1.2409	1.6011	-.6905
0.21	20	1.7969	-.2748	3.1983	-1.3474	1.7562	-.7667
0.21	22	1.8646	-.2890	3.2341	-1.3734	1.8107	-.7932
0.21	24	1.6948	-.3535	3.0224	-1.3099	1.7748	-.7720
0.21	26	1.6072	-.3772	2.4988	-1.1041	1.6618	-.7334
0.30	-4	.7523	-.3852	4.2879	-1.7158	1.5003	-.6245
0.30	0	.8806	-.3883	4.0988	-1.6342	1.5323	-.6442
0.30	4	1.0755	-.3702	3.8806	-1.5403	1.5061	-.6393
0.30	8	1.2900	-.3521	3.9351	-1.5447	1.5237	-.6512
0.30	12	1.5217	-.3438	3.9538	-1.5521	1.5908	-.6842
0.30	16	1.8283	-.3640	4.2541	-1.7097	1.7417	-.7447
0.30	20	2.0437	-.4000	4.3026	-1.7449	1.8230	-.7762
0.30	22	2.0193	-.4141	3.9441	-1.6357	1.8267	-.7731
0.30	24	1.8420	-.4164	3.0863	-1.3303	1.7442	-.7420
0.30	26	1.6443	-.4143	2.3871	-1.0630	1.7040	-.7452
0.43	-4	.9003	-.4209	5.0353	-2.0305	1.4403	-.4995
0.43	0	1.1529	-.4534	5.0694	-2.0489	1.6084	-.5848
0.43	4	1.3449	-.4364	5.0878	-2.0584	1.6220	-.5936
0.43	8	1.5808	-.4206	5.1293	-2.0753	1.6551	-.6109
0.43	12	1.8707	-.4230	4.9093	-1.9995	1.6824	-.6361
0.43	16	1.9631	-.4436	4.1694	-1.7294	1.6974	-.6636
0.43	20	1.9447	-.4541	3.6621	-1.5392	1.7403	-.7027
0.43	22	1.8539	-.4411	3.1679	-1.3431	1.7031	-.7047
0.43	24	1.7200	-.4234	2.3705	-1.0401	1.6590	-.7174
0.43	26	1.6060	-.4178	2.0856	-.9346	1.6461	-.7247
0.55	-4	1.1489	-.5338	5.1344	-2.1010	1.6992	-.6277
0.55	0	1.3494	-.5465	5.0551	-2.0645	1.7976	-.6665
0.55	4	1.6743	-.5214	5.1599	-2.1188	1.7835	-.6538
0.55	8	1.9876	-.5378	4.7785	-1.9921	1.8333	-.6909
0.55	12	1.9539	-.5284	3.5172	-1.4799	1.7767	-.7110
0.55	16	1.8162	-.4921	2.8654	-1.2200	1.7047	-.7097
0.55	20	1.7123	-.4668	2.5605	-1.1014	1.6268	-.6877
0.55	22	1.6152	-.4488	2.3504	-1.0209	1.5751	-.6758
0.55	24	1.5688	-.4385	2.0485	-.9058	1.5672	-.6865
0.55	26	1.5098	-.4384	1.9499	-.8746	1.5853	-.7024
0.72	-4	.1102	.1307				
0.72	0	.3844	.1672				
0.72	4	1.0508	-.0445				
0.72	8	.8999	-.0713				
0.72	12	.8586	-.0791				
0.72	16	.7583	-.0618				
0.72	20	.7420	-.0637				
0.72	22	.7137	-.0596				
0.72	24	.6887	-.0677				
0.72	26	.7315	-.0796				

TABLE XVII.- INTEGRATED SECTION DATA FOR THE DOUBLE SLOTTED FLAP CONFIGURATION - Concluded

(c) $\delta_f = 65^{\circ}$

$\frac{y}{b/2}$	α, deg	$c_{n,w}$	$c_{m,w}$	$c_{n,v}$	$c_{m,v}$	$c_{n,f}$	$c_{m,f}$
0.21	-4	.5482	-.2744	2.1699	-.9128	1.3677	-.5978
0.21	0	.6740	-.2750	2.0499	-.8769	1.3586	-.5899
0.21	4	.8228	-.2775	2.0500	-.8768	1.3615	-.5900
0.21	8	.9720	-.2772	2.1388	-.9033	1.4173	-.6191
0.21	12	1.1893	-.2585	2.3004	-.9591	1.4826	-.6470
0.21	16	1.4416	-.2504	2.5283	-1.0368	1.6273	-.7136
0.21	20	1.7213	-.2414	2.6481	-1.0910	1.7280	-.7642
0.21	22	1.6511	-.2773	2.6462	-1.0944	1.7406	-.7709
0.21	24	1.5850	-.3317	2.4379	-1.0310	1.7163	-.7654
0.21	26	1.5239	-.3533	2.1230	-.9379	1.6550	-.7407
0.30	-4	.7678	-.3793	4.4088	-1.7051	1.5059	-.6379
0.30	0	.9383	-.3883	4.2771	-1.6507	1.5727	-.6713
0.30	4	1.0915	-.3695	4.0947	-1.5764	1.5411	-.6642
0.30	8	1.3134	-.3574	4.1982	-1.6057	1.5966	-.6940
0.30	12	1.5166	-.3386	4.1201	-1.5796	1.6429	-.7153
0.30	16	1.8194	-.3590	4.4685	-1.7525	1.7902	-.7717
0.30	20	1.9011	-.3712	4.2014	-1.6648	1.7769	-.7666
0.30	22	1.9920	-.4052	4.1158	-1.6672	1.8854	-.8112
0.30	24	1.6864	-.3972	2.7760	-1.2002	1.7205	-.7490
0.30	26	1.5729	-.3908	2.2246	-1.0048	1.6819	-.7446
0.43	-4	.9566	-.4407	5.4388	-2.1238	1.5726	-.5725
0.43	0	1.1527	-.4426	5.2558	-2.0513	1.6274	-.6198
0.43	4	1.3587	-.4249	5.2766	-2.0573	1.6469	-.6328
0.43	8	1.5679	-.4114	5.3197	-2.0753	1.6822	-.6541
0.43	12	1.8107	-.3999	4.8071	-1.8917	1.6663	-.6671
0.43	16	1.9187	-.4316	4.1592	-1.6821	1.7424	-.7108
0.43	20	1.8619	-.4327	3.6092	-1.4870	1.7530	-.7282
0.43	22	1.7870	-.4233	3.1798	-1.3292	1.7155	-.7227
0.43	24	1.6204	-.3987	2.2449	-.9864	1.6440	-.7239
0.43	26	1.5205	-.3923	1.9070	-.8761	1.6358	-.7289
0.55	-4	1.1703	-.5362	5.5567	-2.2067	1.7644	-.6608
0.55	0	1.4606	-.5596	5.6724	-2.2460	1.9183	-.7201
0.55	4	1.7220	-.5365	5.6748	-2.2632	1.8871	-.7029
0.55	8	2.0078	-.5356	5.0780	-2.0487	1.8669	-.7125
0.55	12	1.8368	-.4893	3.3230	-1.3735	1.7237	-.7114
0.55	16	1.7146	-.4577	2.7470	-1.1562	1.6621	-.7085
0.55	20	1.5411	-.4181	2.3744	-1.0114	1.5412	-.6639
0.55	22	1.5790	-.4358	2.3690	-1.0175	1.6115	-.7003
0.55	24	1.4643	-.4110	1.8926	-.8463	1.5559	-.6949
0.55	26	1.4163	-.4052	1.7198	-.7976	1.5564	-.6991
0.72	-4	.1332	.1387				
0.72	0	.4074	.1766				
0.72	4	1.0525	-.0273				
0.72	8	.8742	-.0568				
0.72	12	.7884	-.0522				
0.72	16	.6812	-.0380				
0.72	20	.6548	-.0353				
0.72	22	.6264	-.0393				
0.72	24	.6582	-.0526				
0.72	26	.6878	-.0672				

TABLE XVIII.- INTEGRATED SECTION DATA FOR THE EXTENDED DOUBLE SLOTTED FLAP CONFIGURATION

(a) $\delta_f = 45^\circ$

$\frac{y}{b/2}$	α, deg	$c_{n,w}$	$c_{m,w}$	$c_{n,v}$	$c_{m,v}$	$c_{n,f}$	$c_{m,f}$
0.21	-4	.5547	-.3958	2.4745	-1.1256	1.3824	-.5658
0.21	0	.6791	-.3877	2.3255	-1.0433	1.2979	-.5320
0.21	4	.8367	-.3956	2.2709	-1.0114	1.2800	-.5259
0.21	8	1.0022	-.3957	2.2036	-.9792	1.2750	-.5311
0.21	12	1.2782	-.3972	2.3027	-1.0279	1.3294	-.5585
0.21	16	1.5903	-.3976	2.4327	-1.0828	1.4218	-.6052
0.21	20	1.9259	-.3965	2.5525	-1.1484	1.5064	-.6509
0.21	22	2.1630	-.3959	2.7011	-1.2154	1.5652	-.6764
0.21	24	2.2532	-.4444	2.8212	-1.2759	1.6463	-.7112
0.21	26	2.0354	-.5950	2.5588	-1.1901	1.6478	-.7034
0.30	-4	.6841	-.4881	3.7451	-1.6689	1.3854	-.4954
0.30	0	.8225	-.4914	3.6214	-1.5998	1.3717	-.5008
0.30	4	1.0315	-.5056	3.6812	-1.6176	1.3960	-.5152
0.30	8	1.2776	-.4997	3.6865	-1.6085	1.4146	-.5374
0.30	12	1.5486	-.4844	3.4935	-1.5332	1.4214	-.5494
0.30	16	1.9345	-.5264	3.9097	-1.7215	1.5669	-.5945
0.30	20	2.4081	-.5870	4.0466	-1.7830	1.6960	-.6469
0.30	22	2.5972	-.6238	4.0530	-1.8033	1.7232	-.6565
0.30	24	2.5302	-.6613	3.6600	-1.6410	1.6985	-.6570
0.30	26	2.0044	-.6535	2.4701	-1.1047	1.5434	-.6309
0.43	-4	.8245	-.5920	3.9883	-1.7467	1.3694	-.4510
0.43	0	1.0329	-.6006	3.8208	-1.6702	1.3897	-.4773
0.43	4	1.2513	-.5984	3.8959	-1.7040	1.4204	-.4922
0.43	8	1.5239	-.6020	3.9859	-1.7436	1.4630	-.5087
0.43	12	1.9447	-.6383	4.1711	-1.8272	1.5640	-.5542
0.43	16	2.4122	-.7150	4.1474	-1.8289	1.6400	-.5881
0.43	20	2.7282	-.8087	4.1826	-1.8438	1.6742	-.6064
0.43	22	2.8434	-.8703	3.9758	-1.7488	1.7258	-.6376
0.43	24	2.6067	-.8338	3.2795	-1.4583	1.6581	-.6336
0.43	26	2.1935	-.7760	2.3779	-1.0573	1.5487	-.6329
0.55	-4	.9076	-.6653	3.6208	-1.5912	1.3017	-.4593
0.55	0	1.1539	-.7033	3.5658	-1.5659	1.3940	-.4978
0.55	4	1.4129	-.6178	3.4616	-1.3839	.9917	-.2211
0.55	8	1.9306	-.7835	3.9912	-1.7736	1.5860	-.5657
0.55	12	2.3013	-.8490	3.8896	-1.7441	1.5662	-.5532
0.55	16	2.6515	-.9825	3.9775	-1.7809	1.6249	-.5725
0.55	20	2.8284	-1.0630	3.7850	-1.6816	1.6288	-.5837
0.55	22	2.7672	-1.0401	3.4361	-1.5189	1.5900	-.5882
0.55	24	2.4035	-.9266	2.5797	-1.1419	1.5096	-.6037
0.55	26	2.0768	-.8511	2.1033	-.9346	1.4465	-.6020
0.72	-4	.2594	-.0657				
0.72	0	.4846	-.0549				
0.72	4	1.0913	-.1794				
0.72	8	1.4263	-.3184				
0.72	12	1.4363	-.3114				
0.72	16	1.5411	-.3444				
0.72	20	1.5384	-.3429				
0.72	22	1.5282	-.3360				
0.72	24	1.3422	-.2831				
0.72	26	1.2252	-.2572				

TABLE XVIII.- INTEGRATED SECTION DATA FOR THE EXTENDED DOUBLE SLOTTED FLAP CONFIGURATION - Continued

(b) $\delta_f = 50^\circ$

$\frac{y}{b/2}$	α, deg	$c_{n,w}$	$c_{m,w}$	$c_{n,v}$	$c_{m,v}$	$c_{n,f}$	$c_{m,f}$
0.21	-4	.5627	-.3982	2.6004	-1.1280	1.3524	-.5473
0.21	0	.6996	-.3943	2.4262	-1.0332	1.2816	-.5381
0.21	4	.8778	-.4069	2.3766	-1.0046	1.2846	-.5385
0.21	8	1.0490	-.3943	2.2219	-.9479	1.2420	-.5226
0.21	12	1.2973	-.3884	2.2553	-.9590	1.2866	-.5475
0.21	16	1.6059	-.3896	2.3906	-1.0230	1.3722	-.5886
0.21	20	1.9421	-.3876	2.5989	-1.1089	1.4710	-.6360
0.21	22	2.1605	-.3924	2.7469	-1.1675	1.5411	-.6692
0.21	24	2.1880	-.5012	3.0707	-1.3412	1.7728	-.7662
0.21	26	1.9857	-.6041	2.5022	-1.1466	1.6717	-.7112
0.30	-4	.7126	-.4979	3.9936	-1.7174	1.4303	-.5430
0.30	0	.8725	-.5165	3.9069	-1.6661	1.4634	-.5663
0.30	4	1.0491	-.4985	3.6808	-1.5600	1.4085	-.5558
0.30	8	1.3184	-.4937	3.6342	-1.5237	1.4312	-.5787
0.30	12	1.5599	-.4696	3.3337	-1.4067	1.4284	-.5849
0.30	16	1.9393	-.4991	3.6655	-1.5523	1.5072	-.6110
0.30	20	2.3518	-.5541	3.8436	-1.6383	1.6181	-.6545
0.30	22	2.5662	-.6166	3.9974	-1.7219	1.7220	-.6958
0.30	24	2.3906	-.6711	3.3480	-1.4625	1.7520	-.7192
0.30	26	2.0422	-.6648	2.6816	-1.1840	1.6554	-.6920
0.43	-4	.8779	-.6219	4.3388	-1.8405	1.4614	-.5018
0.43	0	1.0616	-.6199	4.1967	-1.7786	1.4431	-.5012
0.43	4	1.3306	-.6316	4.3823	-1.8551	1.5081	-.5308
0.43	8	1.6100	-.6211	4.4096	-1.8693	1.5278	-.5430
0.43	12	1.9705	-.6426	4.4854	-1.9013	1.5678	-.5654
0.43	16	2.4234	-.7202	4.4793	-1.9129	1.6567	-.6071
0.43	20	2.7620	-.8315	4.5384	-1.9392	1.7194	-.6366
0.43	22	2.8149	-.8690	4.2429	-1.8145	1.7447	-.6526
0.43	24	2.4716	-.8122	3.0886	-1.3446	1.6816	-.6774
0.43	26	2.1099	-.7609	2.3367	-1.0250	1.5875	-.6684
0.55	-4	1.0112	-.7196	4.1168	-1.7558	1.4345	-.5144
0.55	0	1.2716	-.7744	4.1573	-1.7744	1.5677	-.5692
0.55	4	1.5932	-.7650	4.2360	-1.8127	1.5922	-.5794
0.55	8	2.0267	-.8178	4.4489	-1.9205	1.6956	-.6114
0.55	12	2.4043	-.8954	4.3613	-1.8962	1.6665	-.5933
0.55	16	2.7304	-1.0166	4.3259	-1.8811	1.6647	-.5919
0.55	20	2.7686	-1.0405	3.9346	-1.7038	1.6220	-.5907
0.55	22	2.6434	-.9990	3.3844	-1.4630	1.6062	-.6098
0.55	24	2.2874	-.9013	2.4479	-1.0707	1.5619	-.6421
0.55	26	2.0919	-.8491	2.1761	-.9571	1.5001	-.6334
0.72	-4	.2737	-.0631				
0.72	0	.5469	-.0479				
0.72	4	1.1085	-.1719				
0.72	8	1.5241	-.3394				
0.72	12	1.4736	-.3294				
0.72	16	1.5634	-.3493				
0.72	20	1.5547	-.3409				
0.72	22	1.4451	-.3067				
0.72	24	1.2534	-.2574				
0.72	26	1.1424	-.2330				

TABLE XVIII.- INTEGRATED SECTION DATA FOR THE EXTENDED DOUBLE SLOTTED FLAP CONFIGURATION - Continued

(c) $\delta_f = 55^{\circ}$

$\frac{y}{b/2}$	α, deg	$c_{n,w}$	$c_{m,w}$	$c_{n,v}$	$c_{m,v}$	$c_{n,f}$	$c_{m,f}$
0.21	-4	.5379	-.3644	2.0713	-.8895	1.3473	-.5790
0.21	0	.6809	-.3727	2.0862	-.8912	1.3157	-.5565
0.21	4	.8449	-.3796	2.1015	-.8904	1.2909	-.5465
0.21	8	1.0081	-.3743	1.9836	-.8617	1.3116	-.5640
0.21	12	1.2684	-.3638	2.0688	-.8919	1.3277	-.5736
0.21	16	1.6123	-.3790	2.3475	-.9817	1.4147	-.6089
0.21	20	1.9124	-.3617	2.3482	-.9851	1.4940	-.6516
0.21	22	2.0802	-.3623	2.4464	-1.0266	1.5726	-.6869
0.21	24	1.9687	-.4836	2.7258	-1.1733	1.7141	-.7491
0.21	26	1.9469	-.5836	2.5547	-1.1308	1.7235	-.7531
0.30	-4	.6937	-.4818	3.8271	-1.5655	1.4608	-.6061
0.30	0	.8496	-.4917	3.7057	-1.5127	1.4646	-.6106
0.30	4	1.0351	-.4838	3.5447	-1.4437	1.4329	-.6022
0.30	8	1.2792	-.4804	3.6467	-1.4722	1.4408	-.6047
0.30	12	1.6160	-.4924	3.7222	-1.5150	1.4868	-.6157
0.30	16	1.9581	-.5104	4.0704	-1.6614	1.5300	-.6253
0.30	20	2.3376	-.5558	4.1079	-1.6881	1.6264	-.6726
0.30	22	2.4771	-.6044	4.1185	-1.7141	1.7189	-.7151
0.30	24	2.3381	-.6556	3.6239	-1.5384	1.7869	-.7441
0.30	26	1.9935	-.6582	2.7231	-1.1897	1.6926	-.7168
0.43	-4	.9320	-.6440	4.8172	-1.9678	1.5270	-.5323
0.43	0	1.1268	-.6485	4.6221	-1.8906	1.5333	-.5542
0.43	4	1.3465	-.6325	4.6678	-1.9079	1.5350	-.5574
0.43	8	1.6216	-.6351	4.7041	-1.9164	1.5818	-.5901
0.43	12	1.9912	-.6465	4.7260	-1.9245	1.6120	-.6167
0.43	16	2.4831	-.7417	4.8522	-1.9946	1.7365	-.6672
0.43	20	2.6960	-.8209	4.5955	-1.8895	1.7354	-.6821
0.43	22	2.6450	-.8325	4.1924	-1.7405	1.7585	-.6938
0.43	24	2.2352	-.7483	2.7989	-1.1990	1.6317	-.6772
0.43	26	2.0940	-.7510	2.4316	-1.0537	1.6425	-.7017
0.55	-4	1.0612	-.7596	4.6185	-1.9132	1.5785	-.5780
0.55	0	1.3308	-.8012	4.5659	-1.8952	1.6783	-.6208
0.55	4	1.6129	-.7790	4.6631	-1.9339	1.6613	-.6107
0.55	8	2.0236	-.8255	4.8793	-2.0411	1.7342	-.6302
0.55	12	2.4875	-.9302	5.0343	-2.1227	1.7579	-.6258
0.55	16	2.7521	-1.0255	4.9106	-2.0759	1.7439	-.6218
0.55	20	2.7076	-1.0210	4.1500	-1.7533	1.6815	-.6172
0.55	22	2.5076	-.9521	3.4203	-1.4514	1.6124	-.6176
0.55	24	2.2093	-.8652	2.4825	-1.0711	1.5680	-.6480
0.55	26	2.0424	-.8319	2.1691	-.9472	1.5515	-.6675
0.72	-4	.3312	-.0672				
0.72	0	.5947	-.0465				
0.72	4	1.0911	-.1526				
0.72	8	1.5148	-.3272				
0.72	12	1.5447	-.3445				
0.72	16	1.5670	-.3446				
0.72	20	1.4994	-.3194				
0.72	22	1.3431	-.2722				
0.72	24	1.2364	-.2464				
0.72	26	1.1325	-.2243				

TABLE XVIII.- INTEGRATED SECTION DATA FOR THE EXTENDED DOUBLE SLOTTED FLAP CONFIGURATION - Continued

(d) $\delta_f = 60^\circ$

$\frac{y}{b/2}$	α, deg	$c_{n,w}$	$c_{m,w}$	$c_{n,v}$	$c_{m,v}$	$c_{n,f}$	$c_{m,f}$
0.21	-4	.5396	-.3571	2.0821	-.9281	1.4032	-.6059
0.21	0	.6640	-.3611	1.9389	-.8615	1.3718	-.5919
0.21	4	.8218	-.3693	1.9641	-.8643	1.3571	-.5862
0.21	8	1.0187	-.3659	1.8558	-.8227	1.3458	-.5848
0.21	12	1.2602	-.3493	1.9105	-.8385	1.3387	-.5821
0.21	16	1.5516	-.3452	2.0325	-.8886	1.4278	-.6256
0.21	20	1.8866	-.3506	2.2892	-.9937	1.5629	-.6871
0.21	22	2.0620	-.3724	2.5664	-1.1161	1.6832	-.7386
0.21	24	1.9186	-.4889	2.7427	-1.2010	1.7433	-.7672
0.21	26	1.9130	-.5739	2.6399	-1.1709	1.7555	-.7728
0.30	-4	.7078	-.4787	4.0762	-1.6248	1.5172	-.6389
0.30	0	.8758	-.4975	3.9271	-1.5674	1.5355	-.6477
0.30	4	1.0779	-.4943	3.7807	-1.4977	1.5230	-.6489
0.30	8	1.3085	-.4723	3.5757	-1.3977	1.4848	-.6396
0.30	12	1.5744	-.4641	3.4792	-1.3778	1.4943	-.6428
0.30	16	1.9555	-.4853	3.8026	-1.4964	1.5704	-.6760
0.30	20	2.3447	-.5592	4.1541	-1.6619	1.7249	-.7395
0.30	22	2.4665	-.6199	4.3786	-1.7782	1.8619	-.7957
0.30	24	2.2919	-.6460	3.7536	-1.5656	1.8473	-.7843
0.30	26	2.0233	-.6648	2.9747	-1.2796	1.7980	-.7708
0.43	-4	.9474	-.6492	5.1440	-2.0336	1.5771	-.5662
0.43	0	1.1346	-.6525	4.9179	-1.9483	1.5831	-.5893
0.43	4	1.3584	-.6442	5.0017	-1.9752	1.5985	-.6037
0.43	8	1.6618	-.6350	4.9925	-1.9651	1.6378	-.6360
0.43	12	1.9942	-.6375	4.8859	-1.9169	1.6406	-.6567
0.43	16	2.4103	-.7150	4.8077	-1.9102	1.7280	-.6960
0.43	20	2.5564	-.7917	4.5858	-1.8314	1.7654	-.7185
0.43	22	2.4417	-.7844	3.9967	-1.6290	1.7594	-.7118
0.43	24	2.1645	-.7296	2.7651	-1.1691	1.6697	-.7059
0.43	26	2.0445	-.7305	2.4515	-1.0525	1.6707	-.7218
0.55	-4	1.0952	-.7674	5.0463	-2.0308	1.6514	-.6141
0.55	0	1.3542	-.8138	5.0285	-2.0345	1.7660	-.6593
0.55	4	1.7025	-.8167	5.3002	-2.1401	1.8071	-.6717
0.55	8	2.0764	-.8359	5.4107	-2.1975	1.8040	-.6576
0.55	12	2.4846	-.9241	5.4132	-2.2134	1.7823	-.6437
0.55	16	2.7341	-1.0164	5.0279	-2.0616	1.7357	-.6300
0.55	20	2.5835	-.9742	4.0746	-1.6822	1.6928	-.6446
0.55	22	2.3649	-.8971	3.2769	-1.3668	1.6279	-.6474
0.55	24	2.1275	-.8272	2.4634	-1.0503	1.5815	-.6686
0.55	26	2.0329	-.8180	2.2335	-.9681	1.5947	-.6882
0.72	-4	.3463	-.0714				
0.72	0	.6174	-.0452				
0.72	4	1.1873	-.1789				
0.72	8	1.5850	-.3427				
0.72	12	1.5880	-.3531				
0.72	16	1.5493	-.3342				
0.72	20	1.4094	-.2882				
0.72	22	1.3262	-.2617				
0.72	24	1.1714	-.2245				
0.72	26	1.1143	-.2108				

TABLE XVIII.- INTEGRATED SECTION DATA FOR THE EXTENDED DOUBLE SLOTTED FLAP CONFIGURATION - Continued

(e) $\delta_f = 65^{\circ}$

$\frac{y}{b/2}$	α, deg	$c_{n,w}$	$c_{m,w}$	$c_{n,v}$	$c_{m,v}$	$c_{n,f}$	$c_{m,f}$
0.21	-4	.5606	-.3673	2.2847	-.9546	1.4256	-.6324
0.21	0	.6914	-.3742	2.1315	-.8982	1.3991	-.6166
0.21	4	.8755	-.3845	2.1187	-.8928	1.3722	-.6009
0.21	8	1.0456	-.3705	1.8996	-.8230	1.3587	-.5951
0.21	12	1.2730	-.3532	1.9903	-.8441	1.3404	-.5898
0.21	16	1.5572	-.3532	2.1684	-.9120	1.4443	-.6378
0.21	20	1.8842	-.3537	2.4634	-1.0168	1.5663	-.6963
0.21	22	2.0289	-.3728	2.6863	-1.1067	1.6793	-.7528
0.21	24	1.9205	-.4830	2.9826	-1.2436	1.7818	-.7905
0.21	26	1.9322	-.5847	2.7441	-1.1908	1.8114	-.8042
0.30	-4	.7276	-.4853	4.4070	-1.6917	1.6022	-.6846
0.30	0	.8933	-.5080	4.3539	-1.6724	1.6519	-.7108
0.30	4	1.0985	-.4915	3.9995	-1.5344	1.5883	-.6866
0.30	8	1.3017	-.4783	3.8191	-1.4472	1.5471	-.6750
0.30	12	1.6089	-.4687	3.7012	-1.4224	1.5614	-.6792
0.30	16	1.9669	-.4896	4.0645	-1.5555	1.6437	-.7128
0.30	20	2.3183	-.5690	4.4058	-1.7195	1.8290	-.7936
0.30	22	2.3519	-.6026	4.4271	-1.7525	1.8798	-.8130
0.30	24	2.1416	-.6276	3.6371	-1.5050	1.8392	-.7844
0.30	26	1.9837	-.6402	2.9788	-1.2678	1.8020	-.7764
0.43	-4	.9646	-.6494	5.7214	-2.1863	1.6188	-.5790
0.43	0	1.1737	-.6702	5.4161	-2.0719	1.6817	-.6334
0.43	4	1.3941	-.6458	5.2582	-1.9994	1.6735	-.6462
0.43	8	1.6661	-.6287	5.2187	-1.9738	1.6753	-.6606
0.43	12	1.9760	-.6244	5.0883	-1.9142	1.6586	-.6661
0.43	16	2.3920	-.7105	5.1513	-1.9734	1.7511	-.7068
0.43	20	2.5151	-.7794	4.8187	-1.8697	1.7895	-.7330
0.43	22	2.3429	-.7499	4.0089	-1.6030	1.7560	-.7180
0.43	24	2.1353	-.7086	2.8901	-1.2019	1.7068	-.7269
0.43	26	2.0239	-.7231	2.5046	-1.0690	1.7413	-.7639
0.55	-4	1.1464	-.7938	5.6371	-2.2056	1.7823	-.6664
0.55	0	1.3940	-.8370	5.6560	-2.2207	1.8908	-.7112
0.55	4	1.7200	-.8241	5.7855	-2.2680	1.8847	-.7027
0.55	8	2.0738	-.8382	5.8965	-2.3202	1.8533	-.6806
0.55	12	2.5284	-.9280	5.9219	-2.3474	1.8103	-.6555
0.55	16	2.6616	-.9848	5.1616	-2.0591	1.7496	-.6445
0.55	20	2.4029	-.8991	3.8630	-1.5649	1.6590	-.6542
0.55	22	2.2064	-.8347	3.1633	-1.3008	1.6107	-.6571
0.55	24	2.0282	-.7862	2.3542	-1.0010	1.5974	-.6943
0.55	26	1.9704	-.7886	2.2281	-.9621	1.6286	-.7197
0.72	-4	.3735	-.0718				
0.72	0	.6697	-.0430				
0.72	4	1.2787	-.1945				
0.72	8	1.6046	-.3386				
0.72	12	1.5887	-.3466				
0.72	16	1.5088	-.3135				
0.72	20	1.3100	-.2529				
0.72	22	1.2113	-.2219				
0.72	24	1.0603	-.1878				
0.72	26	1.0430	-.1846				

TABLE XVIII.- INTEGRATED SECTION DATA FOR THE EXTENDED DOUBLE SLOTTED FLAP CONFIGURATION - Concluded

(f) $\delta_f = 60^\circ$; $\delta_c = 20^\circ$

$\frac{y}{b/2}$	α, deg	$c_{n,w}$	$c_{m,w}$	$c_{n,v}$	$c_{m,v}$	$c_{n,f}$	$c_{m,f}$
0.21	-4	.4455	-.3520	1.9676	-.8590	1.3033	-.5533
0.21	0	.4212	-.3200	1.7545	-.7553	1.1226	-.4819
0.21	4	.5505	-.3900	1.9126	-.8390	1.3679	-.5907
0.21	8	.9016	-.4163	2.0754	-.9173	1.4380	-.6159
0.21	12	1.1469	-.4139	2.0062	-.8809	1.4193	-.6182
0.21	16	1.4125	-.4023	2.0116	-.8797	1.4354	-.6301
0.21	20	1.7417	-.3897	2.0764	-.9039	1.4692	-.6453
0.21	22	1.9046	-.3829	2.1475	-.9317	1.5111	-.6692
0.21	24	2.0533	-.3872	2.2459	-.9781	1.5406	-.6813
0.30	-4	.6031	-.4657	3.6330	-1.4372	1.4197	-.6067
0.30	0	.6110	-.4490	3.4691	-1.3444	1.2794	-.5307
0.30	4	.7181	-.5431	3.9218	-1.5551	1.5244	-.6458
0.30	8	1.1037	-.5544	4.0022	-1.6048	1.5918	-.6720
0.30	12	1.4062	-.5283	3.8219	-1.5213	1.5658	-.6658
0.30	16	1.7709	-.5317	3.9306	-1.5488	1.5978	-.6859
0.30	20	2.1832	-.5386	4.0376	-1.5844	1.6302	-.7005
0.30	22	2.3982	-.5568	3.9908	-1.5765	1.6600	-.7152
0.30	24	2.5608	-.5900	3.8273	-1.5381	1.6981	-.7288
0.43	-4	.7614	-.5719	4.7940	-1.9196	1.3940	-.4832
0.43	0	.8317	-.5423	4.4618	-1.7478	1.2259	-.4172
0.43	4	1.1335	-.6311	4.4474	-1.7799	1.5019	-.5778
0.43	8	1.6346	-.7226	4.8698	-1.9622	1.7543	-.6807
0.43	12	1.8834	-.6900	4.7874	-1.9032	1.7398	-.6947
0.43	16	2.2222	-.7144	4.9561	-1.9569	1.7517	-.7018
0.43	20	2.6185	-.7859	5.0914	-2.0104	1.7795	-.7119
0.43	22	2.7057	-.8273	4.9073	-1.9416	1.7800	-.7190
0.43	24	2.6555	-.8230	4.4219	-1.7604	1.7188	-.6974
0.55	-4	1.0551	-.7716	5.0249	-2.0293	1.6822	-.6276
0.55	0	1.1832	-.7284	4.8134	-1.9247	1.5251	-.5499
0.55	4	1.6356	-.7663	4.3934	-1.8171	1.6291	-.6204
0.55	8	1.8825	-.8387	4.7643	-1.9304	1.6380	-.6102
0.55	12	2.2805	-.8577	5.0033	-2.0149	1.6288	-.6001
0.55	16	2.6426	-.9615	5.3880	-2.1705	1.6814	-.6110
0.55	20	2.8519	-1.0571	5.2745	-2.1343	1.7039	-.6231
0.55	22	2.8080	-1.0515	4.8794	-1.9774	1.6749	-.6228
0.55	24	2.6816	-1.0083	4.2487	-1.7259	1.6265	-.6234
0.72	-4	.3390	-.0764				
0.72	0	.6228	-.0543				
0.72	4	1.1003	-.1807				
0.72	8	1.5015	-.3264				
0.72	12	1.9745	-.4584				
0.72	16	1.6332	-.3566				
0.72	20	1.6065	-.3501				
0.72	22	1.5127	-.3189				
0.72	24	1.4226	-.2909				

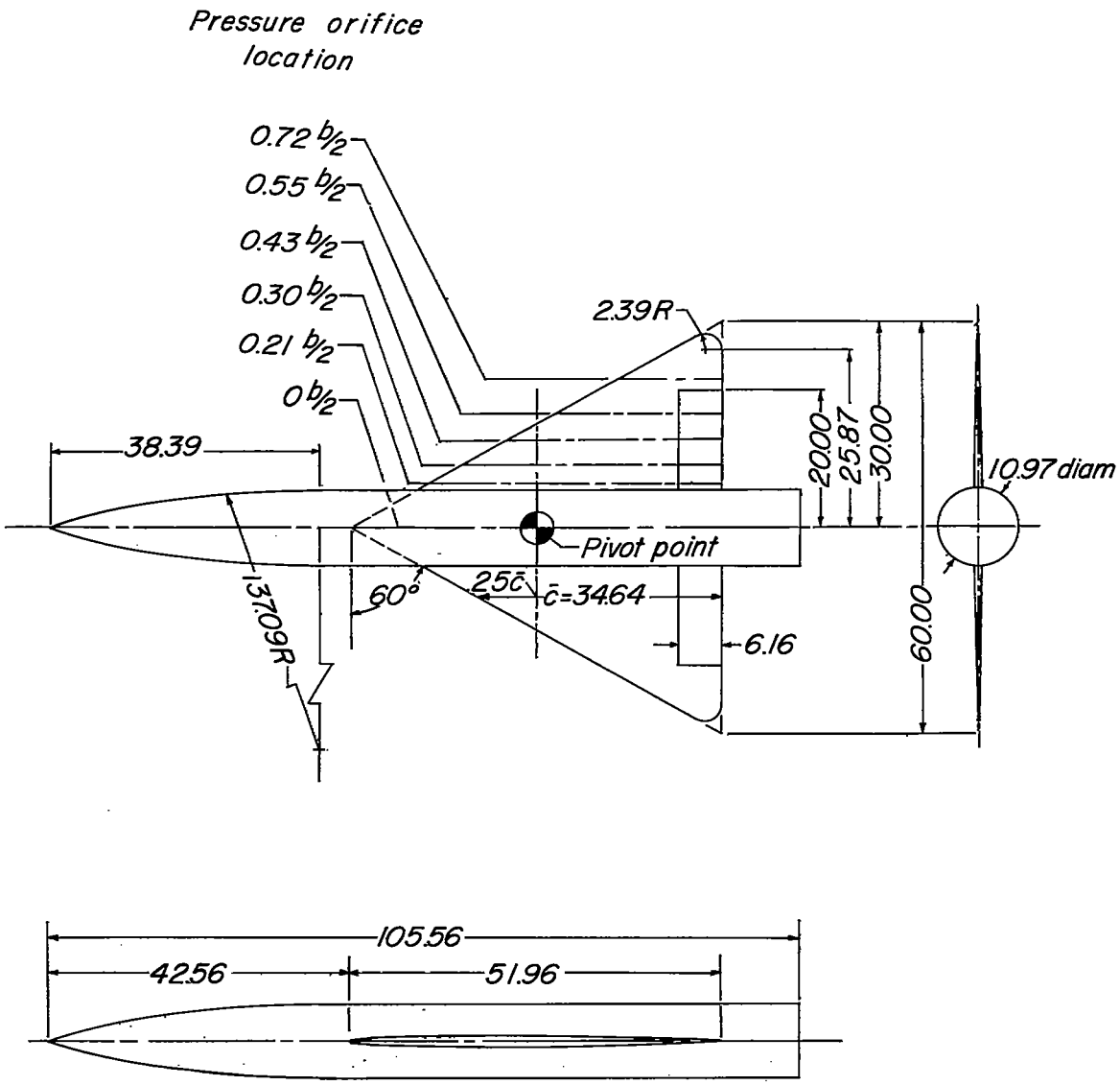


Figure 1.- General arrangement of 60° delta-wing model. (All dimensions are in inches unless otherwise noted.)

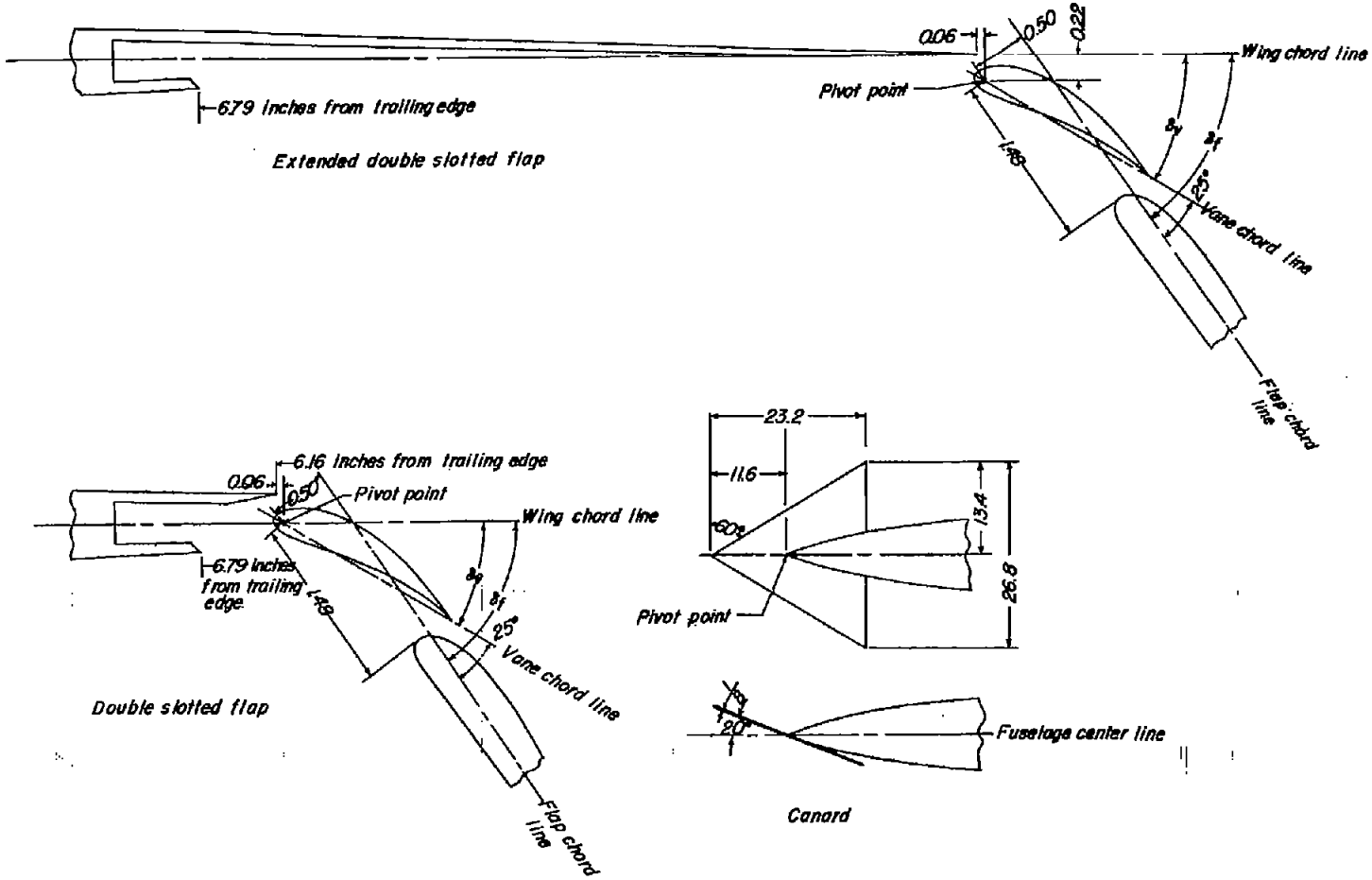


Figure 2.- Details of double slotted flaps, extended double slotted flaps, and canard. (All dimensions are in inches unless otherwise noted.)

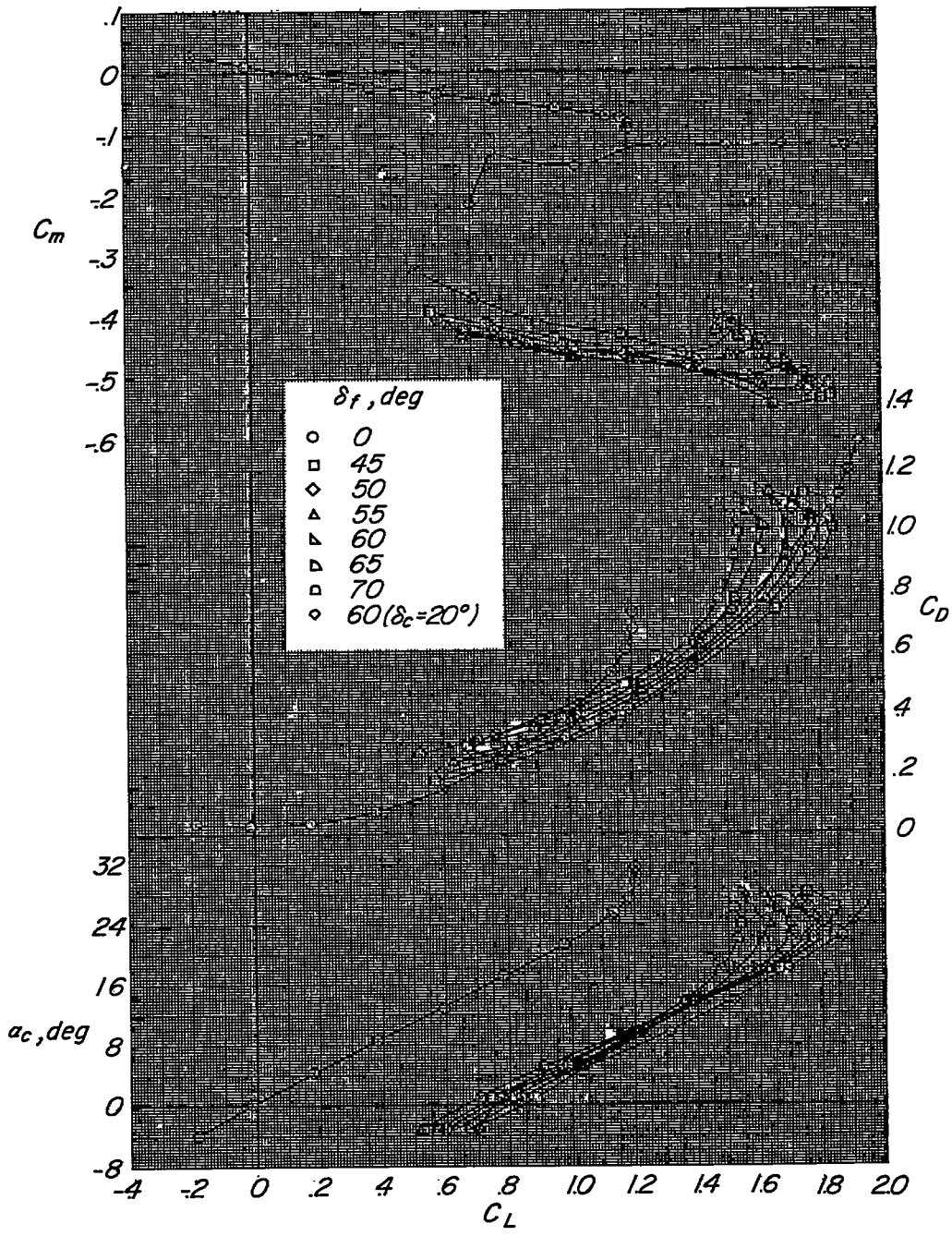


Figure 3.- The aerodynamic characteristics of plain-wing configuration, extended-double-slotted-flap configuration, and extended-double-slotted-flap configuration with a canard.

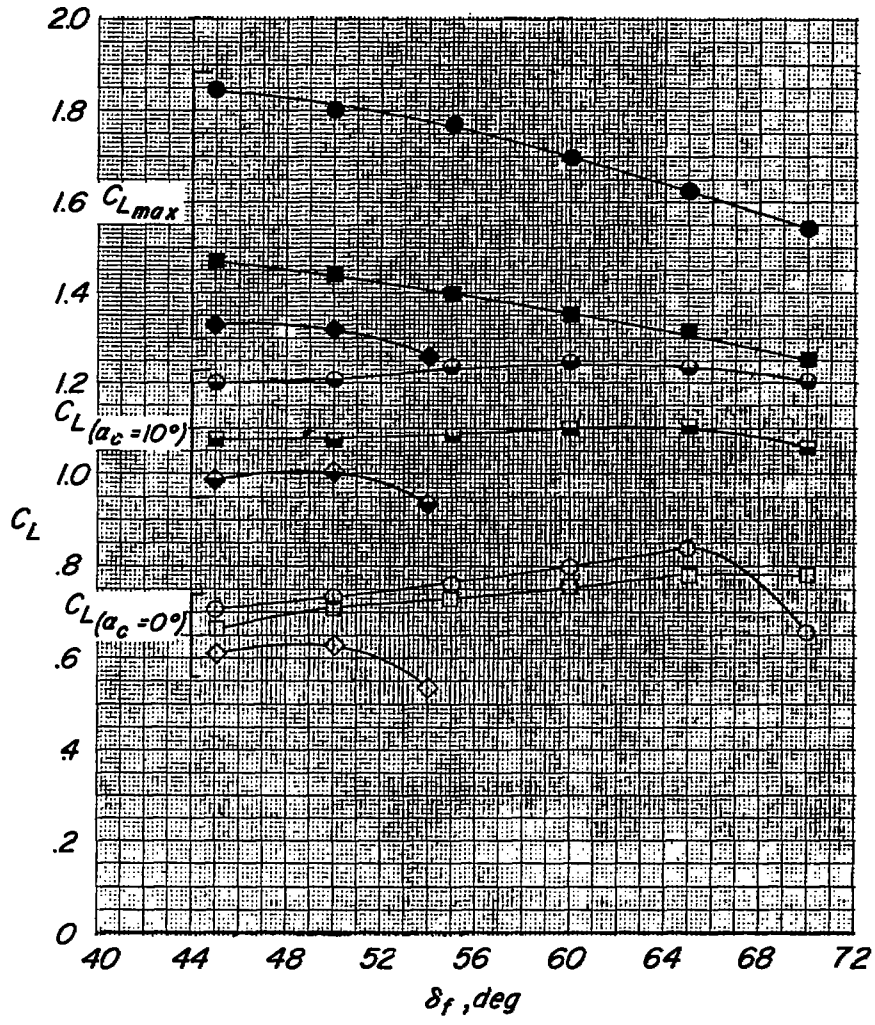
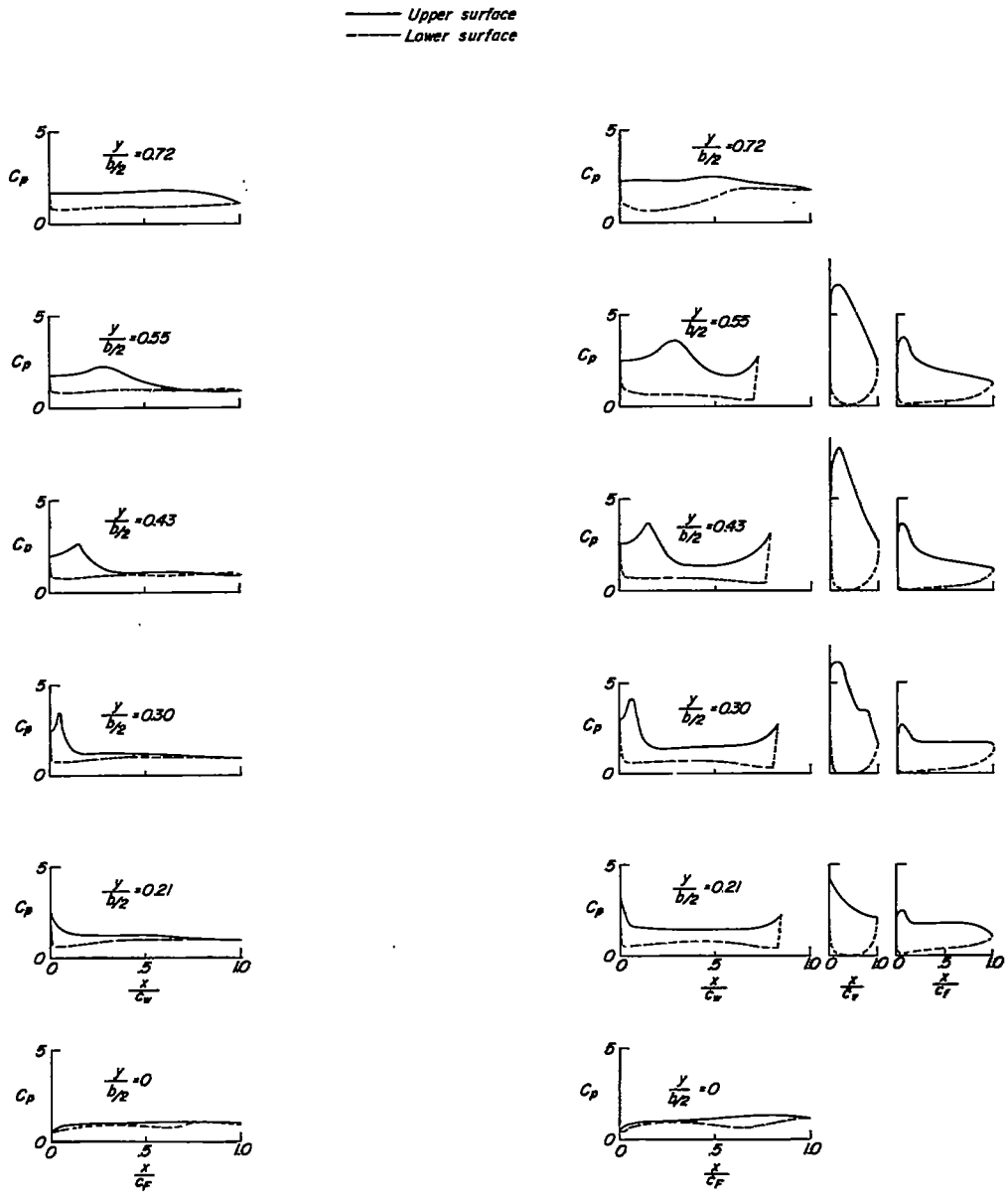


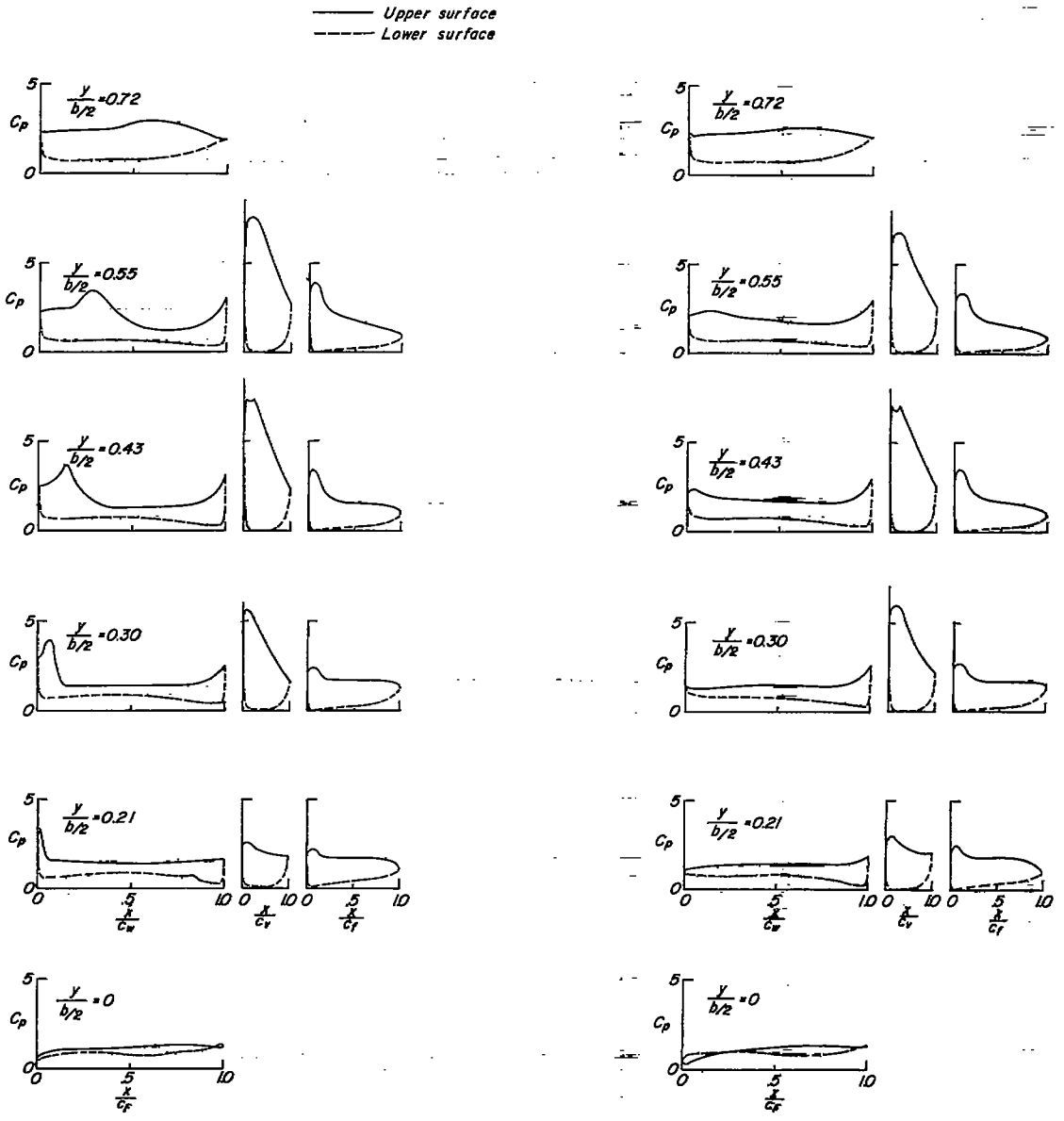
Figure 4.- The variation of lift coefficient with flap deflection for double-slotted-flap configuration and the extended-double-slotted-flap configuration at $\alpha = 0^{\circ}$, $\alpha = 10^{\circ}$, and at $C_{L_{max}}$.



(a) Plain-wing configuration; $\delta_f = 0^\circ$.

(b) Double-slotted-flap configuration; $\delta_f = 60^\circ$.

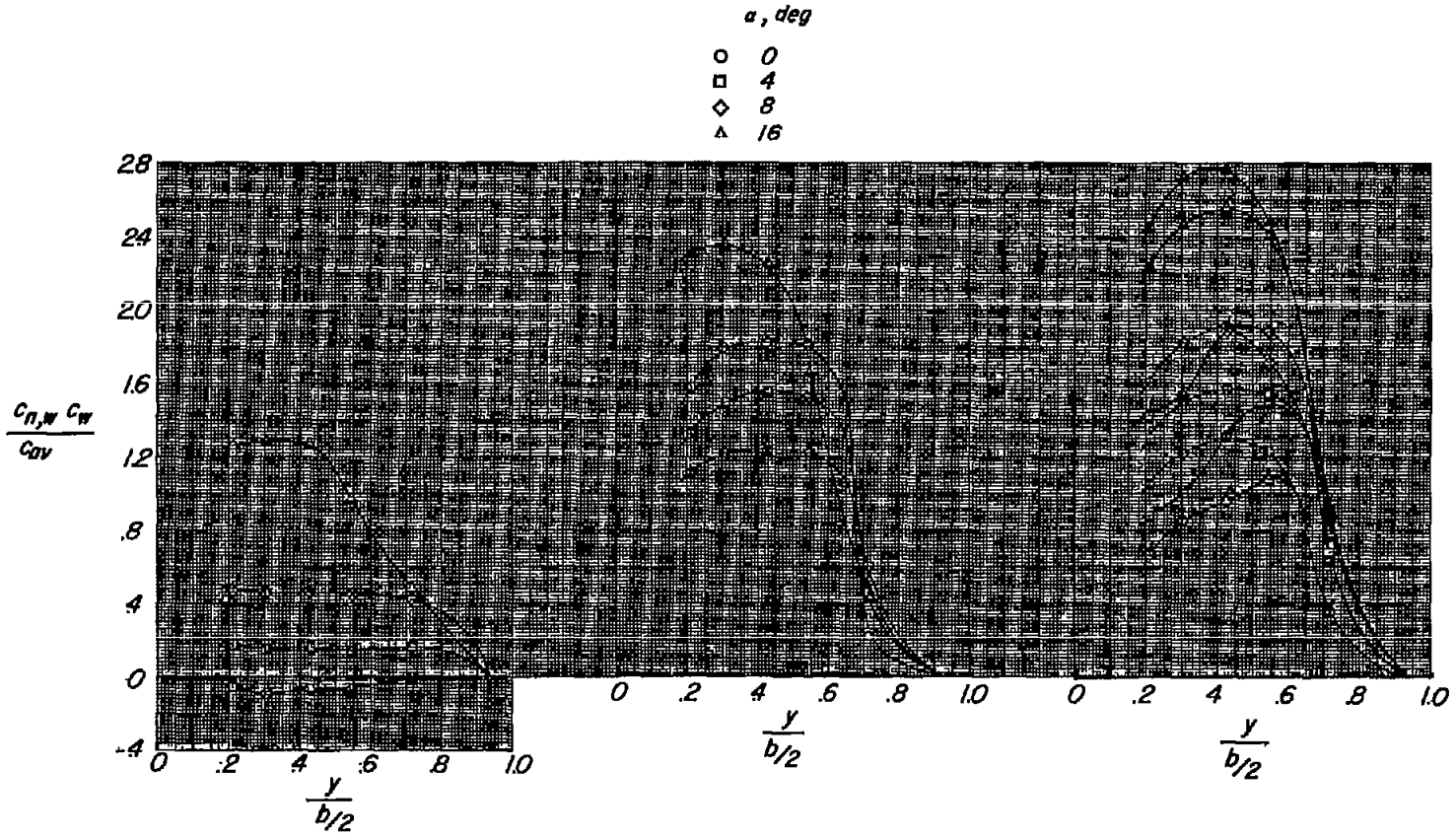
Figure 5.- Chordwise load distribution at several spanwise positions for plain-wing configuration and double-slotted-flap configuration at $\alpha = 8^\circ$. (Note x/c_F, x/c_w, x/c_v, and x/c_f are not to the same scale.)



(a) Extended-double-slotted-flap configuration; $\delta_f = 60^\circ$.

(b) Extended-double-slotted-flap configuration with canard; $\delta_f = 60^\circ$, $\delta_c = 20^\circ$.

Figure 6.- Chordwise load distribution at several spanwise positions for extended-double-slotted-flap configuration and the extended-double-slotted-flap configuration with a canard ($\delta_c = 20^\circ$) at $\alpha = 8^\circ$. (Note x/c_F, x/c_w, x/c_v, and x/c_f are not to the same scale.)

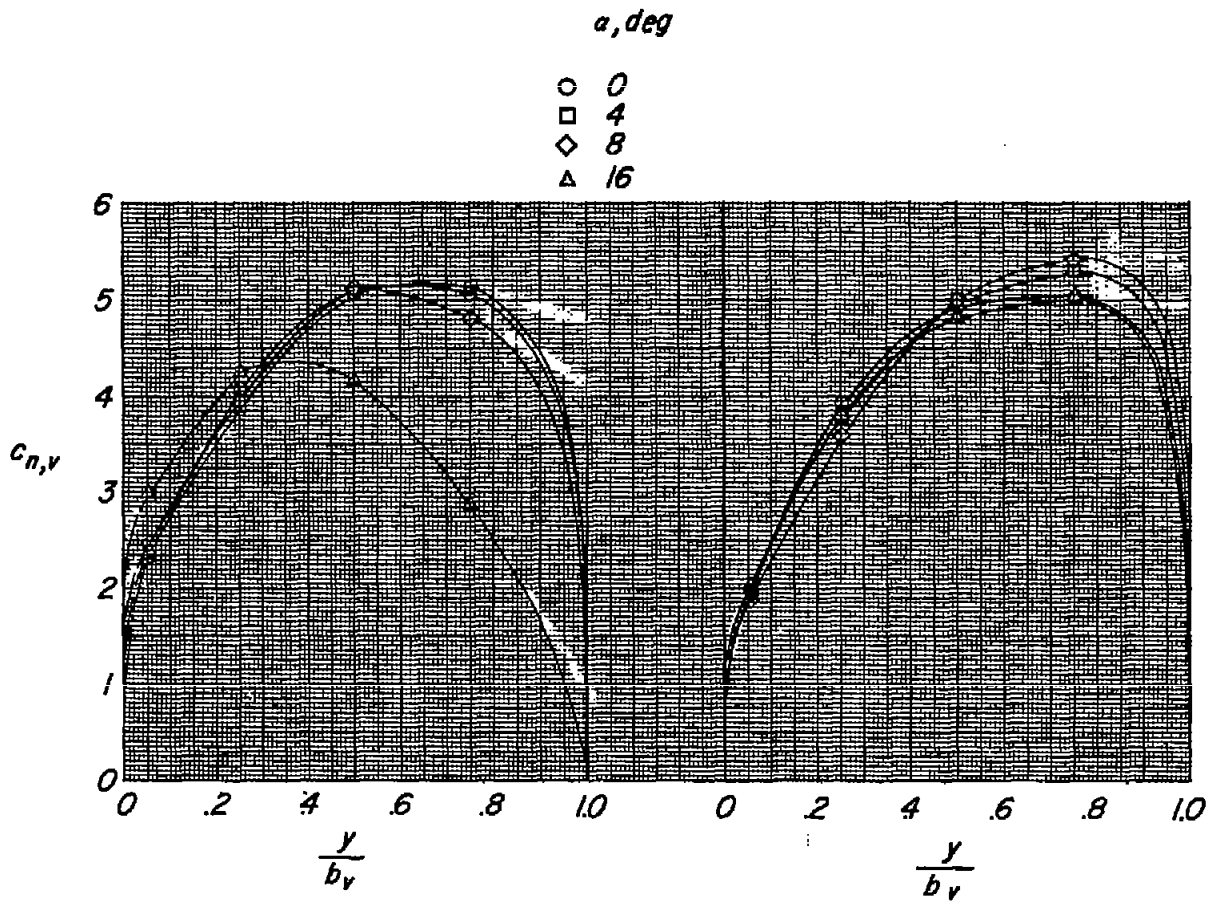


(a) Plain-wing configuration.

(b) Double-slotted-flap configuration; $\delta_f = 60^\circ$.

(c) Extended-double-slotted-flap configuration; $\delta_f = 60^\circ$. Flagged symbols are for canard configuration; $\delta_f = 60^\circ$; $\delta_c = 20^\circ$.

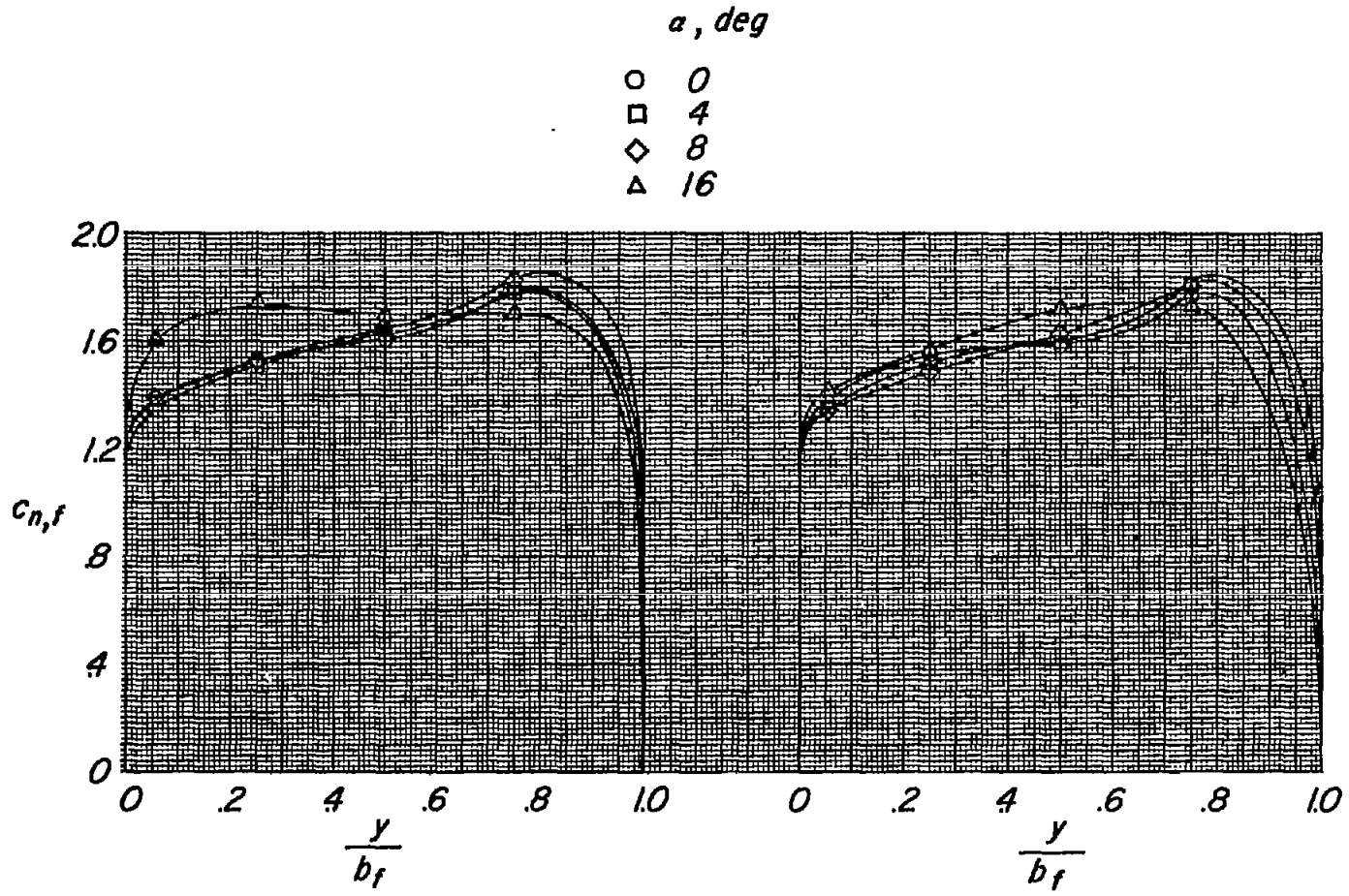
Figure 7.- Span load distribution of plain wing, double slotted flap, extended double slotted flap, and extended double slotted flap with canard at several angles of attack.



(a) Double-slotted-flap configuration; $\delta_f = 60^\circ$.

(b) Extended-double-slotted-flap configuration; $\delta_f = 60^\circ$.

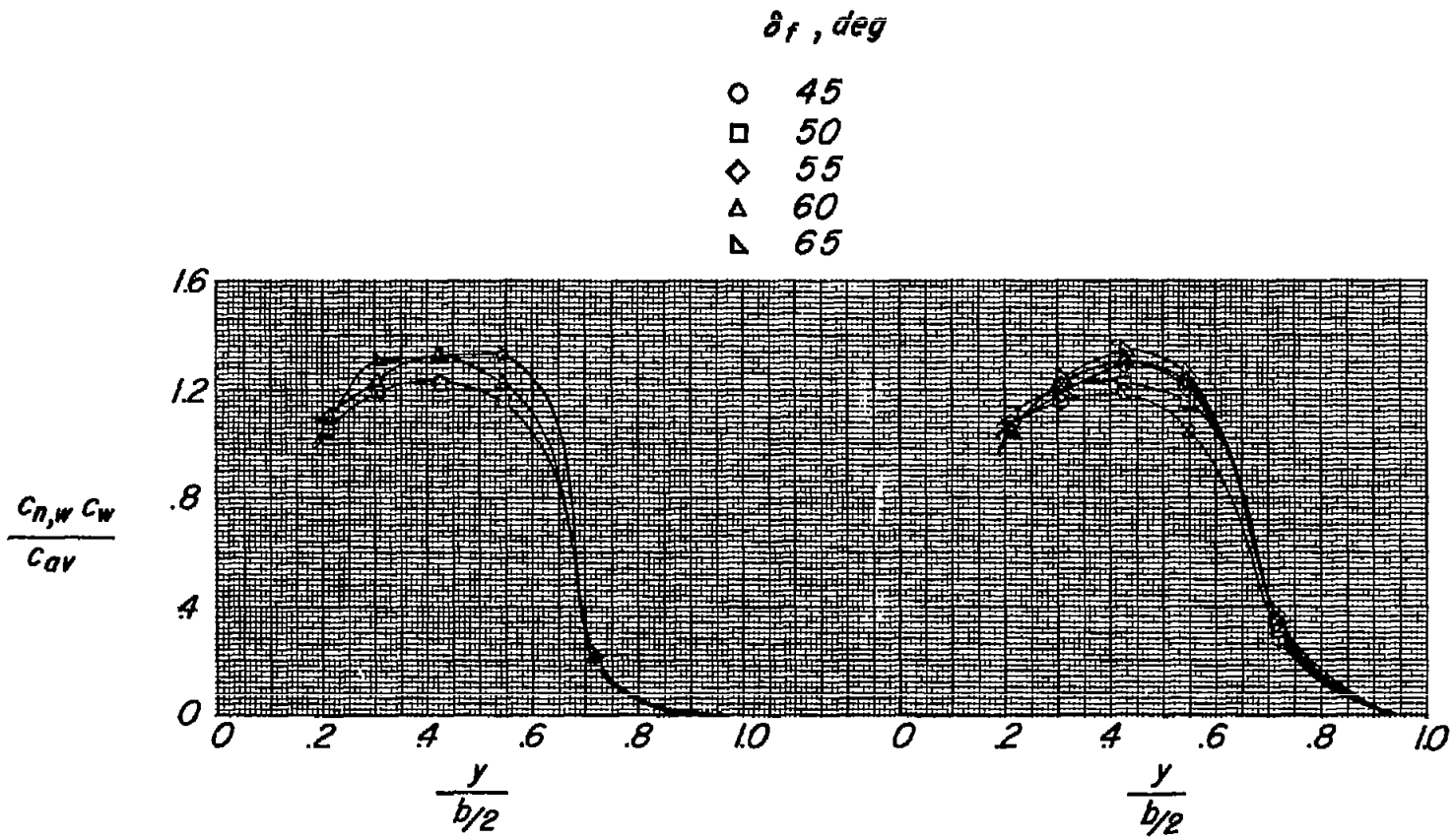
Figure 8.- Span load distribution over vane for double-slotted and extended-double-slotted-flap configurations at several angles of attack.



(a) Double-slotted-flap configuration; $\delta_f = 60^\circ$.

(b) Extended-double-slotted-flap configuration; $\delta_f = 60^\circ$.

Figure 9.- Span load distribution over flap for the double-slotted and extended-double-slotted-flap configurations at several angles of attack.



(a) Double-slotted-flap configuration; $\alpha = 0^{\circ}$.

(b) Extended-double-slotted-flap configuration; $\alpha = 0^{\circ}$.

Figure 10.- Span load distribution for double-slotted and extended-double-slotted-flap configurations at several flap deflections at zero angle of attack.

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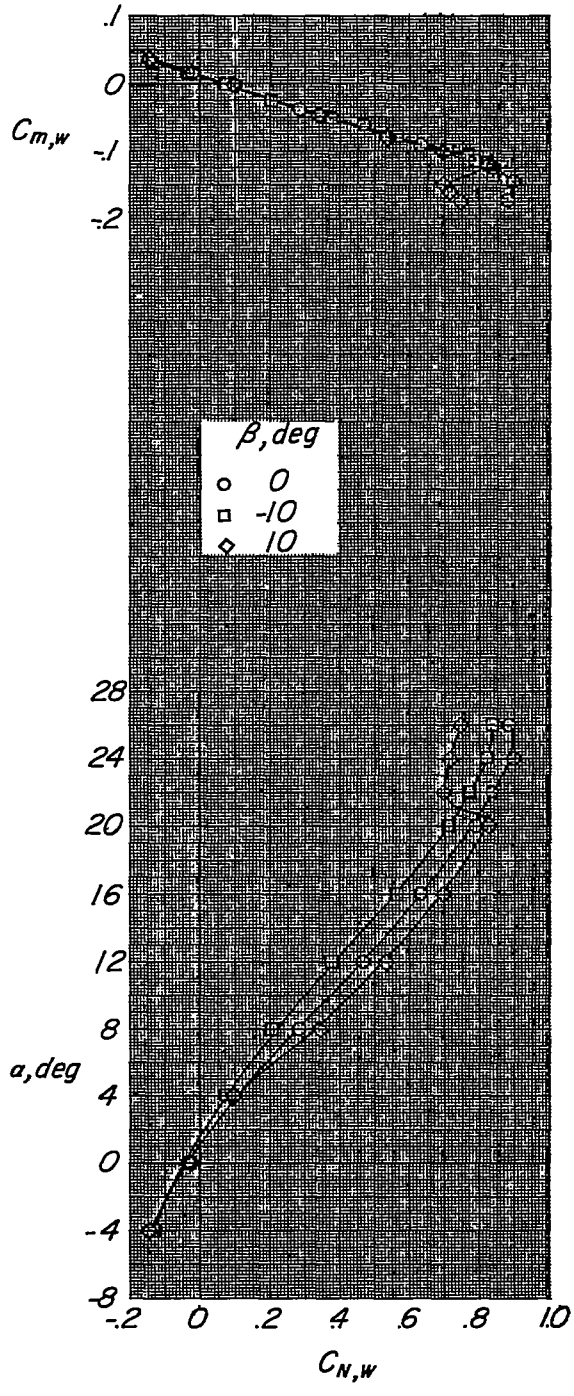

Figure 11.- Effect of sideslip angle on wing normal-force and wing pitching-moment coefficients of plain-wing configuration.

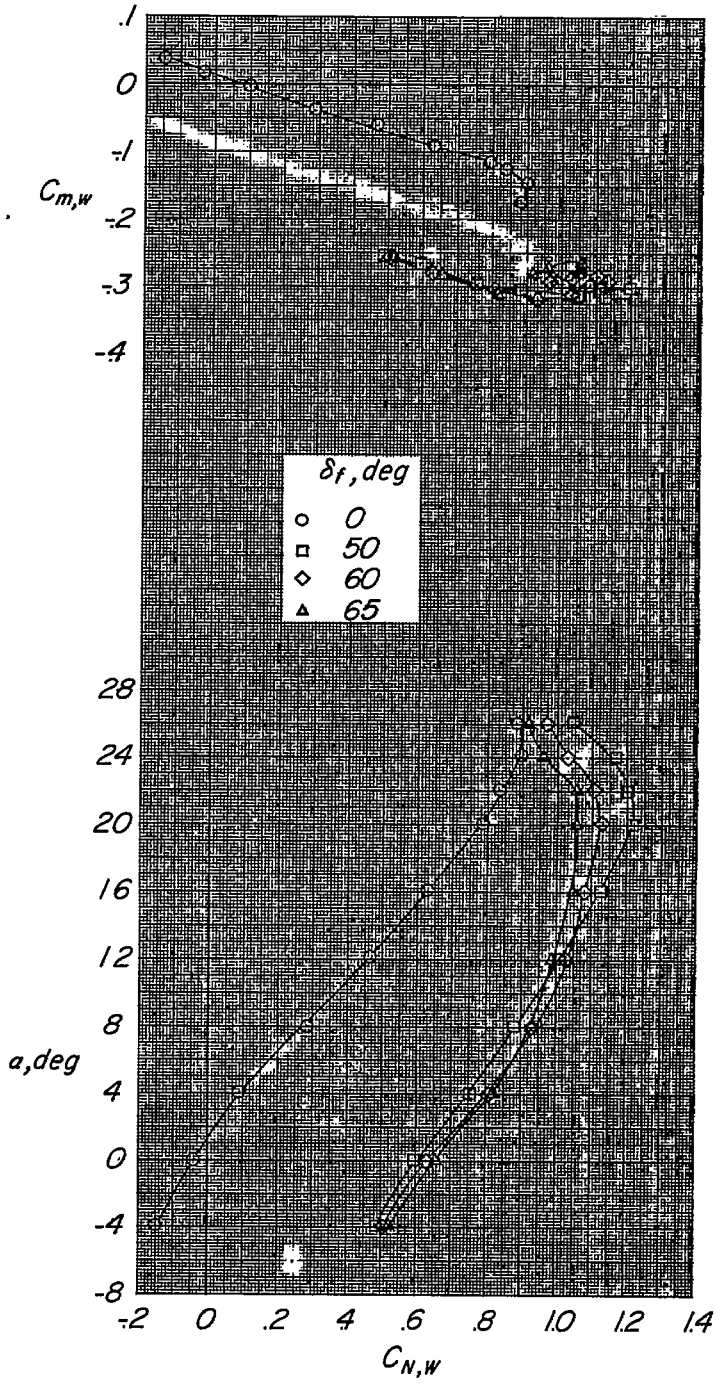

Figure 12.- Effect of flap deflection on wing normal-force and wing pitching-moment coefficients of double-slotted-flap configuration.

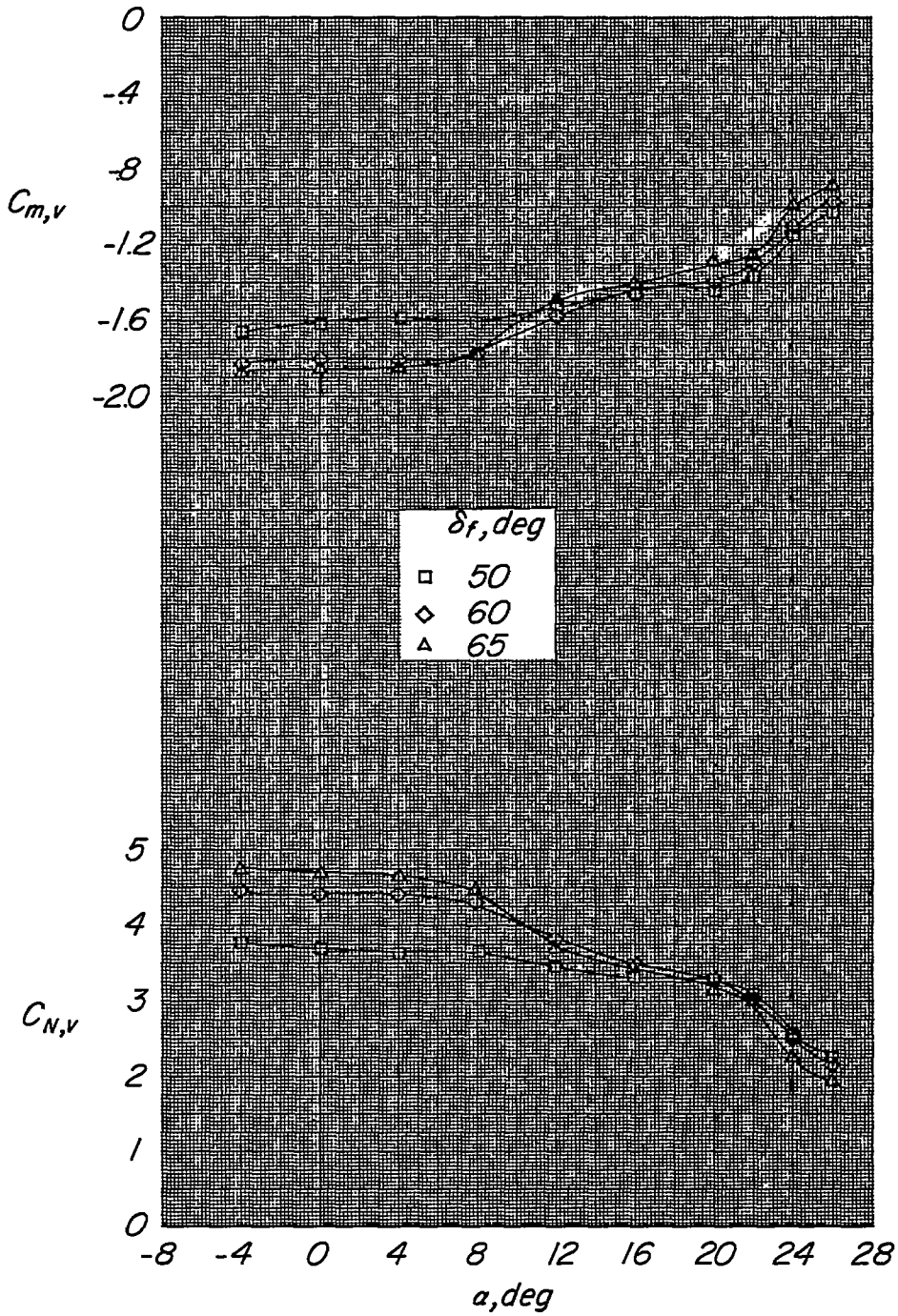

Figure 13.- Effect of flap deflection on vane normal-force and vane pitching-moment coefficients of double-slotted-flap configuration.

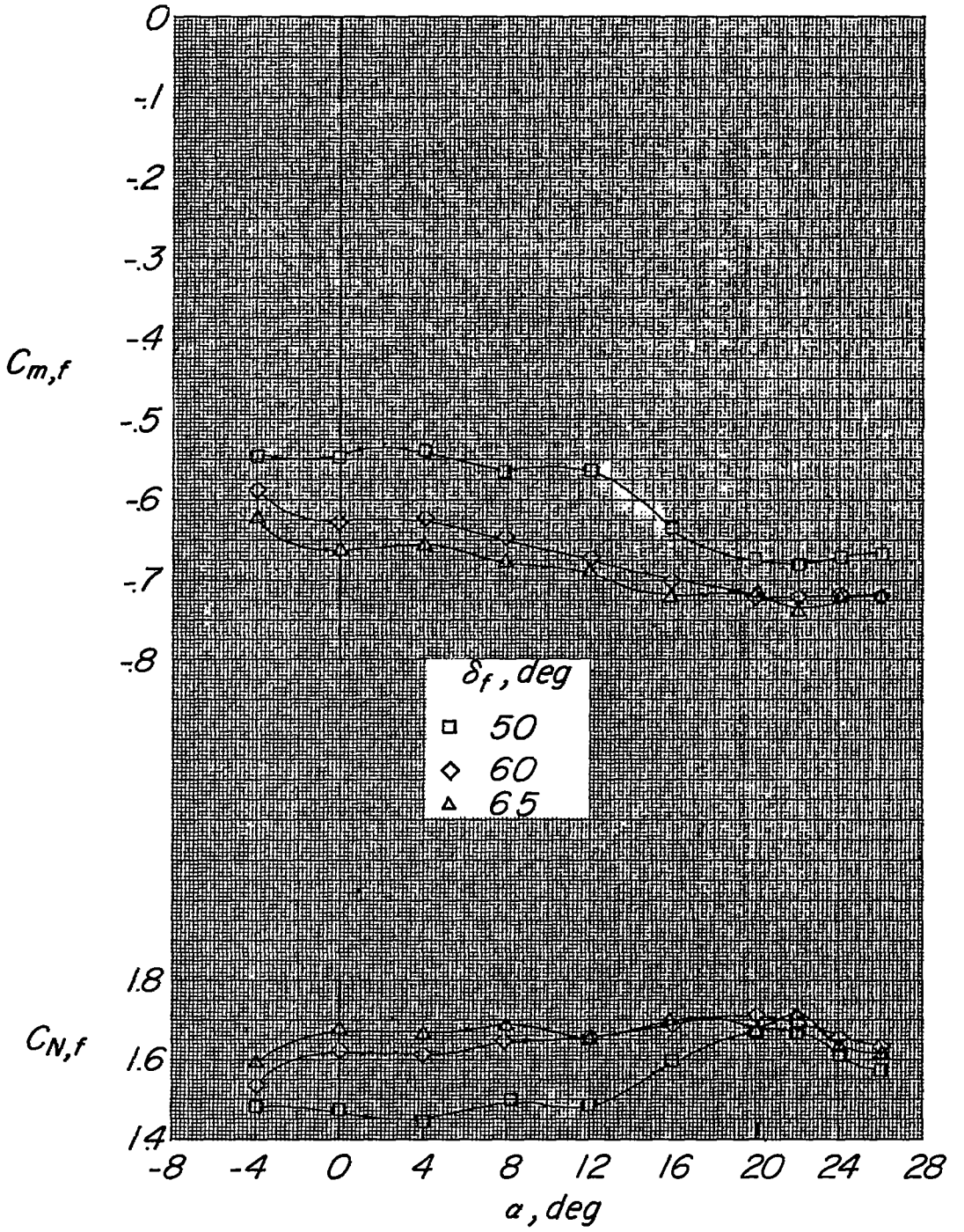

Figure 14.- Effect of flap deflection on flap normal-force and flap pitching-moment coefficients of double-slotted-flap configuration.

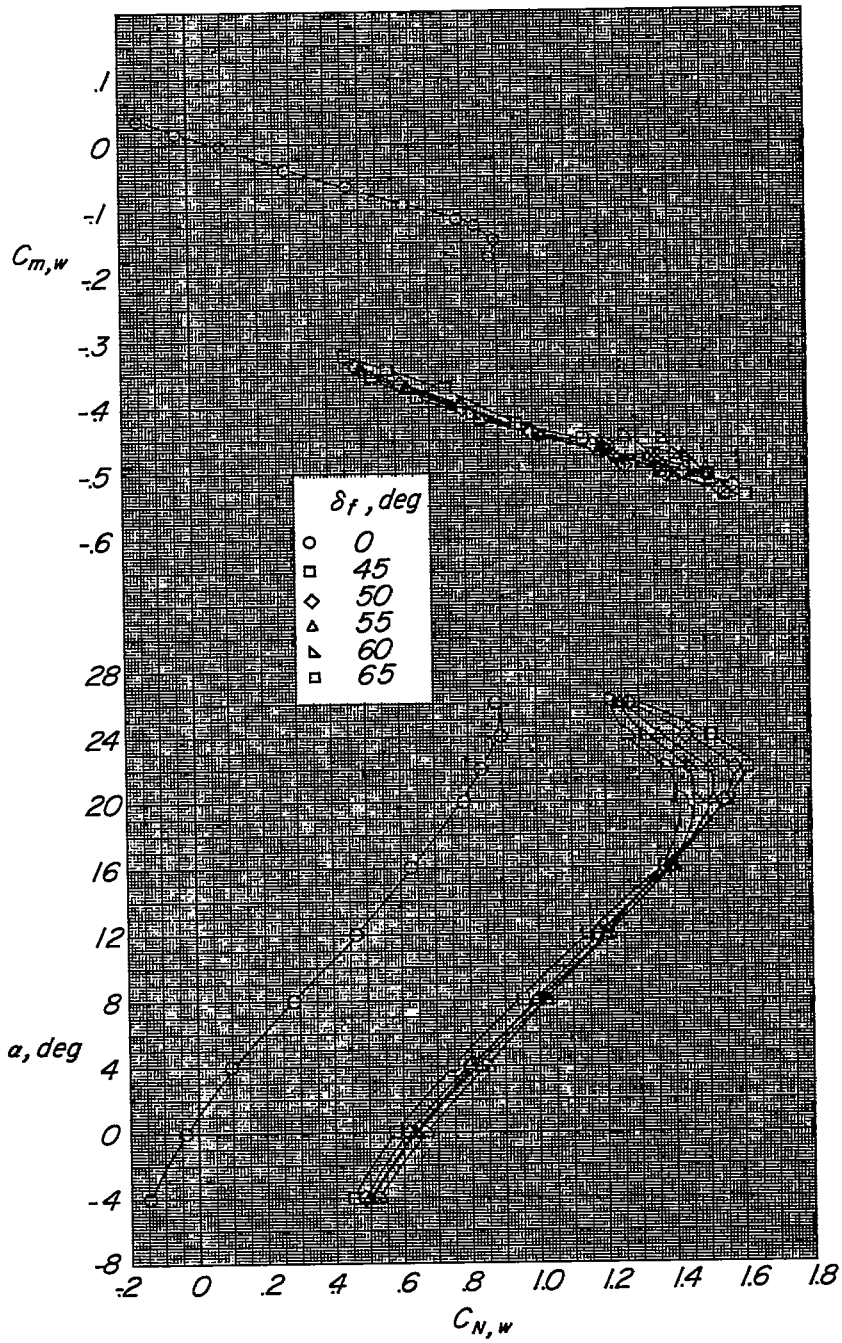

Figure 15.- Effect of flap deflection on wing normal-force and wing pitching-moment coefficients of extended-double-slotted-flap configuration.

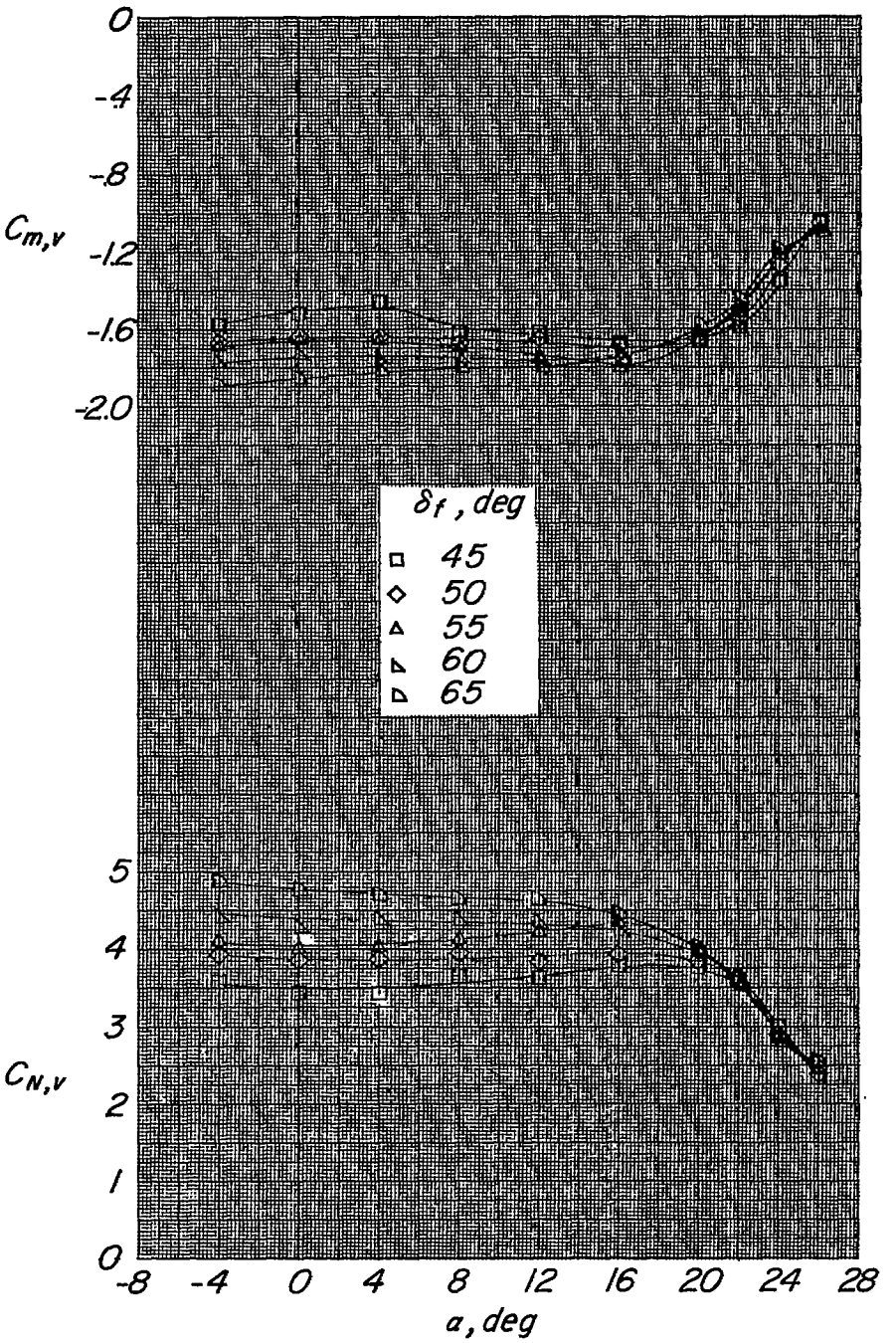

Figure 16.- Effect of flap deflection on vane normal-force and vane pitching-moment coefficients of extended-double-slotted-flap configuration.

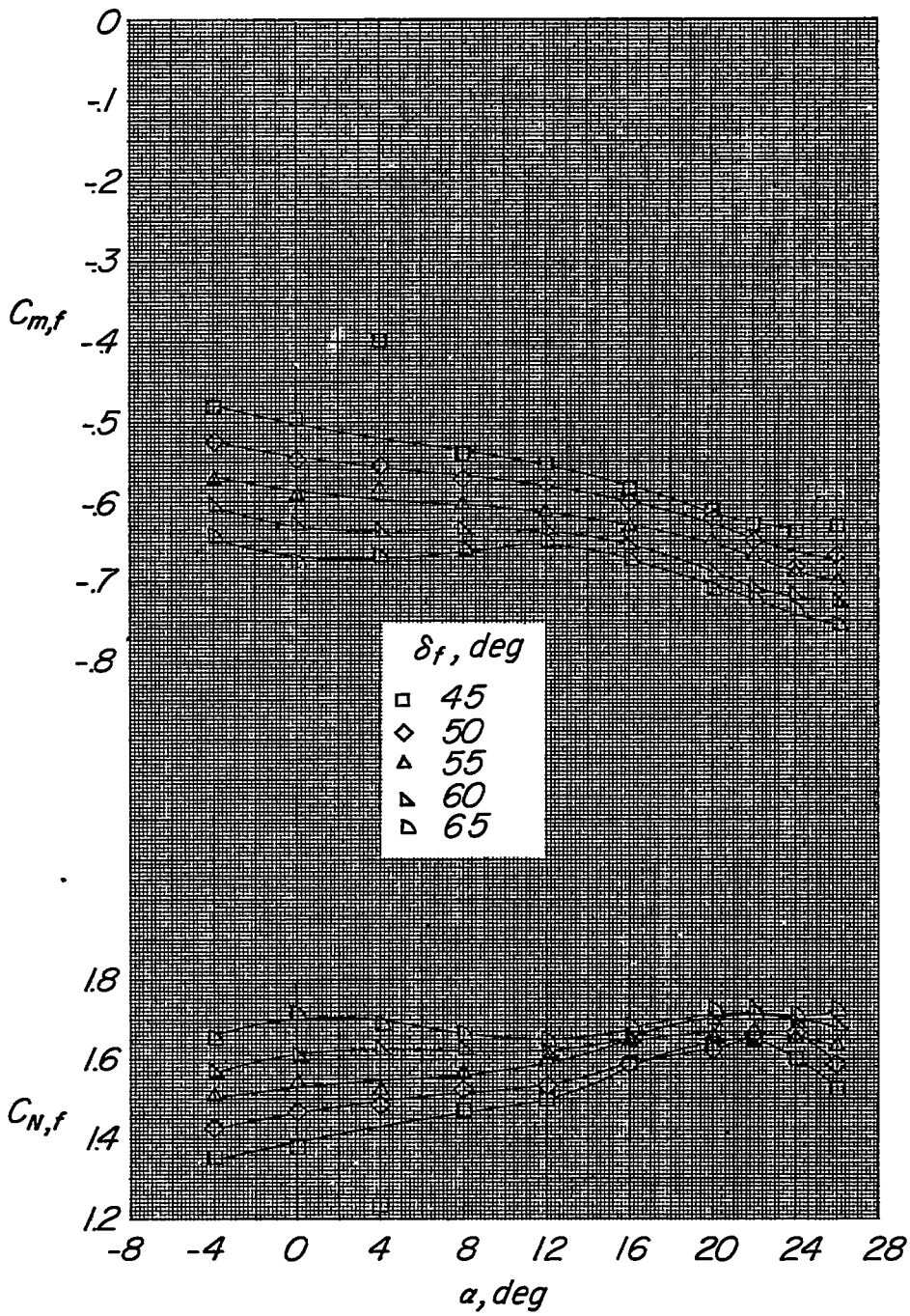

Figure 17.- Effect of flap deflection on flap normal-force and flap pitching-moment coefficients of extended-double-slotted-flap configuration.

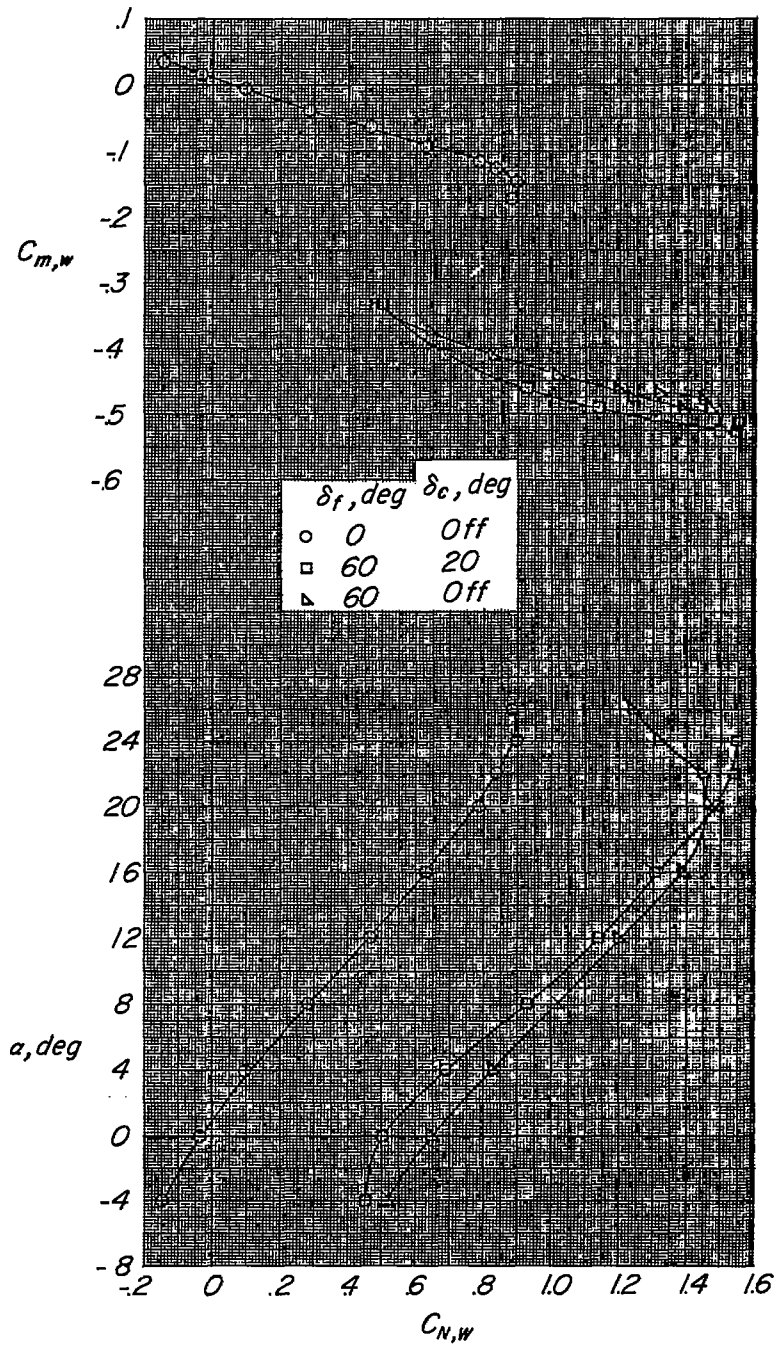

Figure 18.- Effect of canard on wing normal-force and wing pitching-moment coefficients of extended-double-slotted-flap configuration; $\delta_f = 60^\circ$.

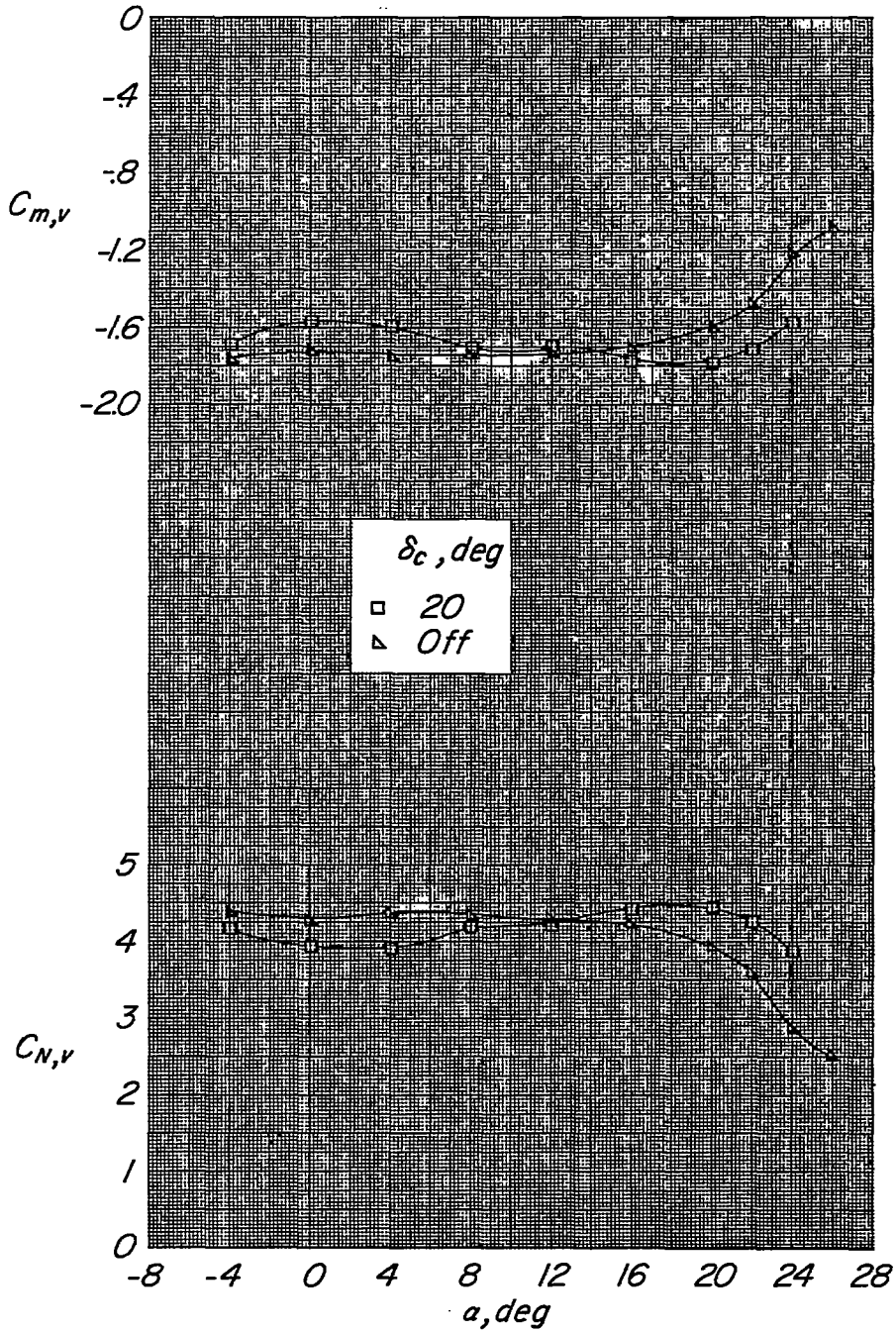

Figure 19.- Effect of canard on vane normal-force and vane pitching-moment coefficients of extended-double-slotted-flap configuration; $\delta_f = 60^{\circ}$.

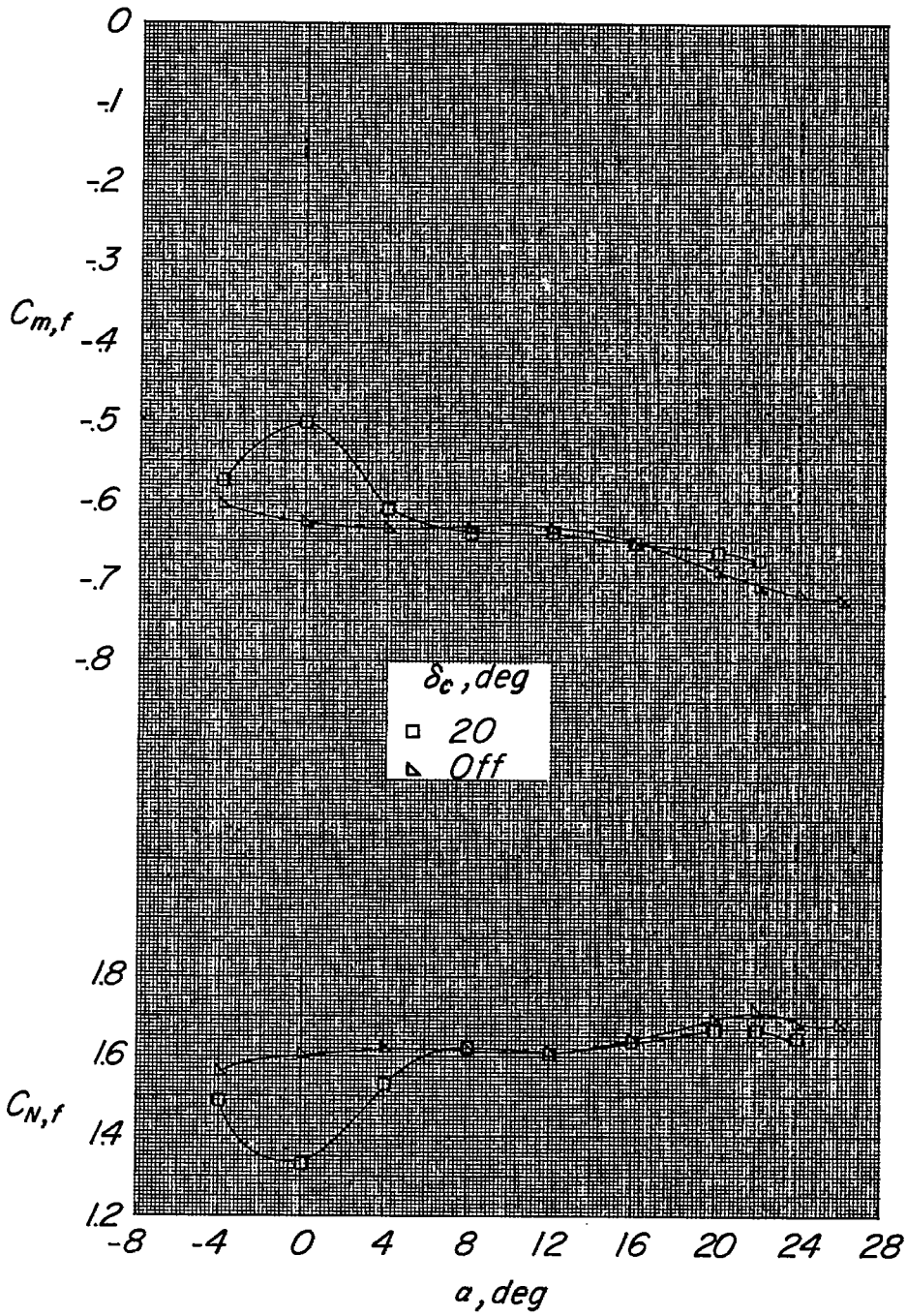

Figure 20.- Effect of canard on flap normal-force and flap pitching-moment coefficients of extended-double-slotted-flap configuration; $\delta_f = 60^\circ$.

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