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RESEARCH MEMORANDUM

WIND-TUNNEL INVESTIGATION OF THE EFFECTS OF A FENCE AND A LEADING-EDGE NOTCH ON THE AERODYNAMIC LOADING CHARACTERISTICS IN PITCH OF A 45° SWEPTBACK WING AT HIGH SUBSONIC SPEEDS

By Richard E. Kuhn, James W. Wiggins, and Andrew L. Byrnes, Jr.

Langley Aeronautical Laboratory
Langley Field, Va.

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RESEARCH MEMORANDUM

WIND-TUNNEL INVESTIGATION OF THE EFFECTS OF A FENCE AND A LEADING-EDGE NOTCH ON THE AERODYNAMIC LOADING CHARACTERISTICS IN PITCH OF A 45° SWEPTBACK WING AT HIGH SUBSONIC SPEEDS

By Richard E. Kuhn, James W. Wiggins,
and Andrew L. Byrnes, Jr.

SUMMARY

An investigation has been conducted in the Langley high-speed 7- by 10-foot tunnel to determine the aerodynamic loading characteristics of a 45° sweptback wing of aspect ratio 4, taper ratio 0.6, and an NACA 65A006 airfoil section. The investigation included the effects of fences and leading-edge notches over a Mach number range from 0.50 to 0.95 and an angle-of-attack range from 0° to about 25°.

The results indicate that, at Mach numbers below 0.93, the fences were effective in improving the lifting ability of the portion of the wing outboard of the fence in the angle-of-attack range from about 8° to about 17°. Above 20° at Mach numbers of 0.50 and 0.70, the fences were ineffective. The fences delayed the inward movement of the lateral center of pressure to a higher angle of attack but did not show any tendency to move the lateral center of pressure farther outboard than the most outward position shown for the clean wing. A limited investigation (at Mach numbers of 0.70 and 0.91) of the effect of a small leading-edge notch indicated that the effects were similar to, but smaller than, the effects shown by the fence.

INTRODUCTION

The aerodynamic loading characteristics of a 45° swept-wing—body combination in pitch, sideslip, and steady roll have been investigated in the Langley high-speed 7- by 10-foot tunnel. The general investigation included the effects of a full-chord wing fence located 65 percent of the semispan from the fuselage center line and covered a Mach number range

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from 0.50 to 0.95 which corresponds to a Reynolds number range from 2.1×10^6 to 3.0×10^6.

This paper presents only the load distributions in pitch for the clean wing and the fence-on condition. Some results on the effect of a notch in the leading edge of the wing are also presented. The lift, drag, and pitching-moment data for this wing-fuselage combination are presented in reference 1. The aerodynamic loading characteristics of this configuration, without the fence, at transonic speeds are presented in reference 2. Additional information concerning the effects of a similar fence on the force data of this configuration is presented in reference 3.

In order to expedite the publication of these results, they are presented here with only a brief discussion.

COEFFICIENTS AND SYMBOLS

The symbols used in the present paper are defined as follows:

M	Mach number
R	Reynolds number
P	pressure coefficient, $\frac{p_l - p_o}{q}$ or $\frac{p_u - p_o}{q}$
p	local static pressure, lb/sq ft
p_o	free-stream static pressure, lb/sq ft
q	dynamic pressure, $\frac{1}{2}\rho V^2$, lb/sq ft
ρ	air density, slugs/cu ft
V	free-stream velocity, ft/sec
c	local wing chord, ft
c_{av}	average wing chord, S/b, ft
b	wing span, ft
S	wing area, sq ft

x chordwise distance from leading edge of local chord, ft

y spanwise distance from fuselage center line, ft

$\Delta\frac{x}{c}$ increment of local chord over which the pressure at a particular orifice is assumed to act

c_n section normal-force coefficient, $\sum_{\frac{x}{c}=0}^{1} \left(P_l - P_u\right)\Delta\left(\frac{x}{c}\right)$

c_m section pitching-moment coefficient about 0.25 local chord,

$$\sum_{\frac{x}{c}=0}^{1} \left(P_u - P_l\right)\left(\frac{x}{c} - 0.25\right) \Delta\left(\frac{x}{c}\right)$$

C_{N_e} exposed-wing normal-force coefficient,

$$\int_0^{1.0} \left(c_n \frac{c}{c_{av}}\right) d\left(\frac{y}{b/2}\right)_e$$

C_{B_e} exposed-wing bending-moment coefficient,

$$\int_0^{1} \left(c_n \frac{c}{c_{av}}\right)\left(\frac{y}{b/2}\right)_e d\left(\frac{y}{b/2}\right)_e$$

α angle of attack, deg

X/c local longitudinal center of pressure, $\frac{c_m}{c_n} - 0.25$

$\left(\frac{Y}{b/2}\right)_e$ lateral center of pressure measured from fuselage surface, C_{B_e}/C_{N_e}

Subscripts:

e exposed wing

u upper surface

l lower surface

MODEL AND APPARATUS

A drawing of the wing-fuselage configuration tested is shown in figure 1. A tabulation of the fuselage ordinates is presented in reference 1. The wing had a quarter-chord sweep of 45°, aspect ratio of 4, taper ratio of 0.6, and an NACA 65A006 airfoil section and was of composite construction, consisting of a steel core with a bismuth-tin covering to give the desired contour. One hundred and fifteen static-pressure orifices were located in the upper and lower surfaces of the wing and were distributed along five spanwise stations parallel to the plane of symmetry (20, 60, and 95 percent semispan on the right wing and 40 and 80 percent semispan on the left wing). The wing was mounted to the fuselage in a midwing position with zero dihedral and zero incidence.

The model was tested on the sting-type support system shown in figure 2. With this support system the model can be remotely operated through a 28° angle range.

The details of the fences used in the investigation are shown in figure 1. The brass fences were mounted on the wing at the 65-percent-semispan station such that the mounting clips did not protrude above the wing surface.

The leading-edge notches were installed in the wing at the 65-percent-semispan station after the pressure tubes had been installed. It was not possible, therefore, to make them as deep (chordwise) as desired. The details of the notches tested are shown in figure 1. The location of the notch was determined from a study of unpublished low-speed data on a similar wing.

TESTS AND CORRECTIONS

The tests were conducted in the Langley high-speed 7- by 10-foot tunnel through a Mach number range from approximately 0.50 to 0.95 corresponding to a Reynolds number range from 2.1×10^6 to 3.0×10^6. The size of the model caused the tunnel to choke at a Mach number of 0.96 at zero angle of attack. The blocking corrections which were applied to the Mach number were determined by the method of reference 4.

The angle of attack, which was varied from 0° to about 25°, has been corrected for the deflection of the sting support system.

The aeroelastic deflection characteristics of this wing (as determined from static loadings) are presented in reference 1. No aeroelastic corrections have been applied to these data. The variation of Reynolds number with Mach number is presented in figure 3.

PRESENTATION OF RESULTS

All the detailed pressure coefficients are presented in tables I, II, and III and selected results from this investigation are presented in the following figures:

	Figure
Span load distributions	4
Local chordwise center of pressure	5
Effect of Mach number on local chordwise center of pressure at $\alpha = 4^{\circ}$	6
Local section normal-force curves	7
Wing normal-force curves	8
Root bending moments	9
Lateral center of pressure	10
Some representative chordwise loadings	11

A few of the chordwise load distributions have been plotted and are presented in figure 11. In the process of plotting and analyzing these data, it was found that less than 2 percent of the individual pressure coefficients were in error. Therefore, in order to expedite the publication of the data, it was decided that it would be unnecessary to plot all the chordwise load distributions, and the pressure coefficients (tables I, II, and III) have been presented in the form obtained from the tabulating machine. Tests at Mach numbers of 0.91 and 0.95 were repeated on the clean-wing configuration as an indication of the repeatability of the data.

At Mach numbers below 0.93, the fences were effective in improving the lifting characteristics of the outer portion of the wing in the angle-of-attack range from about 8° to about 17°, however, in general the fence reduced the load over the section inboard of the fence at these angles of attack. Above an angle of attack of about 20°, at Mach numbers of 0.50 and 0.70, the fences were ineffective (figs. 4 and 10).

The results in figure 10 show that the lateral center of pressure moved inward at angles of attack above 8° for the clean-wing configuration because of the partial stalling of the tip sections of the wing. The fence delayed this inward movement to a higher angle of attack but did not exhibit any definite tendency to move the lateral center of pressure farther outboard than the most outward position shown for the clean wing.

The effects of the leading-edge notch were similar to the effects of the fence but of somewhat smaller magnitude and the effectiveness of the notch was lost at lower angles of attack (figs. 4, 5, and 10). However, as mentioned previously, the notch was rather small. Unpublished data on the effect of notches and fences on the lift and pitching-moment

characteristics of similar wings indicate that deeper (chordwise) notches produce results more nearly approaching those for fences.

CONCLUSIONS

The results of the investigation of the aerodynamic loading characteristics, including the effects of wing fences and leading-edge notches, on a 45° sweptback wing of aspect ratio 4, taper ratio 0.6, and a NACA 65A006 airfoil section indicate the following conclusions:

1. The wing fences, at Mach numbers below 0.93, were effective in improving the lift-carrying ability of the outer portion of the wing in the angle-of-attack range from about 8° to 17°. Above 20°, at Mach numbers of 0.50 and 0.70 the fences were ineffective.

2. The fences delayed the inward movement of the lateral center of pressure to a higher angle of attack but did not show any tendency to move the lateral center of pressure farther outboard than the most outward position shown for the clean wing.

3. A limited investigation (at Mach numbers of 0.70 and 0.91) of the effects of a small leading-edge notch indicated that the effects were similar to, but smaller than, the effects shown by the fence.

Langley Aeronautical Laboratory,
National Advisory Committee for Aeronautics,
Langley Field, Va., August 17, 1953.

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TABLE I.- TABULATION OF PRESSURE COEFFICIENTS FOR THE CLEAN WING

(a) M = 0.50.

$\frac{y}{b/2}$		$\frac{x}{c}$	$\alpha=0°$	$\alpha=4.2°$	$\alpha=8.3°$	$\alpha=12.5°$	$\alpha=16.5°$	$\alpha=20.6°$	$\alpha=24.6°$
=.20	Upper	.000	.435	.006	-1.057	-2.443	-2.984	-1.132	-.990
		.025	-.107	-.628	-1.921	-2.634	-2.909	-1.124	-.994
		.075	-.147	-.425	-1.604	-3.237	-3.125	-1.122	-.997
		.150	-.136	-.340	-.448	-.666	-2.555	-1.126	-.998
		.250	-.168	-.323	-.489	-.632	-.991	-1.130	-1.008
		.350	-.176	-.306	-.453	-.547	-.738	-1.118	-1.010
		.450	-.166	-.265	-.381	-.439	-.690	-1.028	-.939
		.550	-.187	-.280	-.376	-.425	-.577	-1.064	-.998
		.650	-.174	-.247	-.329	-.366	-.514	-1.003	-.988
		.750	-.127	-.184	-.222	-.258	-.357	-.896	-.948
		.850	-.097	-.127	-.167	-.186	-.314	-.806	-.920
		.950	-.052	-.056	-.074	-.102	-.216	-.695	-.881
	Lower	.950	-.037	-.003	.002	.009	.004	-.142	-.177
		.850	-.082	-.028	-.003	.037	.062	-.013	-.006
		.750	-.129	-.051	-.003	.053	.102	.077	.114
		.650	-.166	-.069	-.003	.071	.137	.139	.198
		.550	-.195	-.084	.001	.094	.171	.199	.269
		.450	-.176	-.051	.047	.158	.245	.287	.362
		.350	-.156	-.011	.099	.220	.320	.366	.448
		.250	-.142	.024	.154	.285	.402	.452	.536
		.150	-.118	.092	.244	.387	.507	.550	.632
		.075	-.110	.162	.340	.487	.596	.636	.695
		.025	-.056	.317	.487	.542	.550	.585	.588
=.40	Upper	.000	.180	-.284	-1.161	-1.889	-1.439	-1.085	-1.060
		.025	-.156	-.834	-1.248	-1.578	-1.320	-1.072	-1.048
		.075	-.162	-.537	-1.084	-1.636	-1.330	-1.072	-1.052
		.150	-.169	-.417	-.926	-1.646	-1.301	-1.076	-1.058
		.250	-.176	-.359	-.763	-1.508	-1.885	-1.048	-1.052
		.350	-.192	-.330	-.612	-1.326	-1.270	-1.017	-1.039
		.450	-.199	-.304	-.490	-1.117	-1.217	-1.003	-1.028
		.550	-.181	-.262	-.392	-.875	-1.124	-.989	-1.008
		.650	-.144	-.196	-.300	-.603	-1.006	-.956	-.984
		.750	-.101	-.140	-.215	-.331	-.870	-.899	-.948
		.850	-.051	-.072	-.135	-.136	-.712	-.815	-.905
		.950	-.013	-.023	-.071	-.042	-.602	-.775	-.884
	Lower	.950	-.031	-.013	-.007	.013	-.142	-.314	-.338
		.850	-.054	-.021	.000	.037	-.047	-.159	-.155
		.750	-.083	-.036	.007	.053	.027	-.040	-.006
		.650	-.144	-.071	-.008	.052	.068	.031	.074
		.550	-.182	-.077	.003	.081	.119	.117	.162
		.450	-.181	-.063	.039	.131	.188	.200	.260
		.350	-.181	-.039	.077	.186	.256	.283	.349
		.250	-.165	.003	.141	.262	.341	.371	.437
		.150	-.148	.061	.224	.360	.434	.465	.522
		.075	-.141	.146	.322	.453	.522	.533	.568
		.025	-.107	.289	.442	.496	.504	.478	.449
=.60	Upper	.000	.441	-.084	-.635	-1.072	-.949	-.785	-.805
		.025	-.141	-.741	-.923	-.943	-.854	-.773	-.819
		.075	-.152	-.592	-.898	-.928	-.844	-.776	-.810
		.150	-.154	-.442	-.863	-.898	-.825	-.748	-.788
		.250	-.162	-.355	-.831	-.864	-.793	-.730	-.770
		.350	-.160	-.301	-.798	-.853	-.768	-.718	-.761
		.450	-.159	-.255	-.726	-.837	-.752	-.716	-.752
		.550	-.134	-.202	-.614	-.796	-.734	-.718	-.746
		.650	-.102	-.149	-.468	-.754	-.733	-.714	-.733
		.750	-.070	-.101	-.332	-.690	-.698	-.685	-.701
		.850	-.024	-.032	-.199	-.641	-.663	-.643	-.677
		.950	.014	.007	-.111	-.598	-.628	-.615	-.647
	Lower	.950	.011	.031	.007	-.206	-.271	-.290	-.295
		.850	-.019	.007	.002	-.098	-.167	-.180	-.172
		.750	-.072	-.020	-.008	-.039	-.077	-.092	-.072
		.650	-.113	-.039	-.003	.002	-.013	-.021	.019
		.550	-.156	-.046	.026	.044	.052	.051	.091
		.450	-.152	-.044	.092	.086	.119	.128	.174
		.350	-.150	-.001	.144	.162	.205	.225	.278
		.250	-.143	.031	.216	.229	.304	.307	.366
		.150	-.126	.099	.225	.333	.387	.407	.450
		.075	-.112	.191	.425	.370	.429	.433	.428
		.025	-.099	.295	.429	.498	.532	.496	.412

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TABLE I.- TABULATION OF PRESSURE COEFFICIENTS FOR THE

CLEAN WING - Continued

(a) M = 0.50. Concluded.

$\frac{y}{b/2}$		$\frac{x}{c}$	$\alpha=0°$	$\alpha=4.2°$	$\alpha=8.3°$	$\alpha=12.5°$	$\alpha=16.5°$	$\alpha=20.6°$	$\alpha=24.6°$
.80	Upper	.000	.434	-.455	-1.097	-.736	-.589	-.603	-.645
		.025	-.159	-.908	-1.169	-.596	-.537	-.592	-.644
		.075	-.169	-.570	-1.216	-.582	-.527	-.598	-.642
		.150	-.162	-.420	-.994	-.557	-.508	-.590	-.634
		.250	-.171	-.349	-.610	-.521	-.492	-.581	-.626
		.350	-.174	-.300	-.468	-.490	-.477	-.573	-.617
		.450	-.170	-.264	-.382	-.467	-.467	-.563	-.606
		.550	-.156	-.215	-.310	-.453	-.460	-.550	-.594
		.650	-.116	-.147	-.245	-.438	-.445	-.525	-.579
		.750	-.072	-.091	-.186	-.422	-.427	-.503	-.561
		.850	-.034	-.037	-.129	-.388	-.402	-.470	-.532
		.950	.010	.004	-.082	-.336	-.354	-.416	-.469
	Lower	.950	-.037	-.035	-.086	-.276	-.296	-.329	-.341
		.850	-.033	-.018	.034	-.160	-.170	-.174	-.171
		.750	-.079	-.052	.047	-.127	-.124	-.125	-.105
		.650	-.114	-.061	.033	-.078	-.066	-.058	-.023
		.550	-.147	-.071	.024	-.042	-.018	.003	.047
		.450	-.164	-.079	.009	.001	.037	.062	.116
		.350	-.174	.041	.052	.096	.131	.141	.136
		.250	-.154	.011	.119	.173	.229	.262	.326
		.150	-.149	.072	.209	.269	.327	.370	.424
		.075	-.129	.165	.311	.382	.431	.453	.487
		.025	-.101	.311	.433	.472	.483	.463	.445
.95	Upper	.000	.361	-.499	-.374	-.434	-.381	-.462	-.513
		.025	-.128	-.696	-.438	-.326	-.344	-.455	-.506
		.075	-.170	-.509	-.425	-.329	-.344	-.455	-.502
		.150	-.159	-.354	-.378	-.305	-.328	-.446	-.491
		.250	-.140	-.245	-.322	-.277	-.312	-.438	-.485
		.350	-.141	-.216	-.297	-.280	-.325	-.462	-.518
		.450	-.127	-.180	-.254	-.259	-.318	-.452	-.501
		.550	-.103	-.148	-.225	-.260	-.317	-.446	-.495
		.650	-.055	-.104	-.174	-.231	-.287	-.404	-.443
		.750	-.032	-.067	-.181	-.255	-.313	-.424	-.455
		.850	.003	-.037	-.171	-.250	-.301	-.398	-.424
		.950	.041	-.004	-.148	-.221	-.267	-.347	-.374
	Lower	.950	.020	-.011	-.119	-.176	-.202	-.237	-.240
		.850	.020	.001	-.064	-.109	-.122	-.149	-.143
		.750	-.027	-.042	-.084	-.127	-.136	-.157	-.151
		.650	-.075	-.069	-.092	-.122	-.124	[illegible]	-.120
		.550	-.108	-.091	-.094	-.106	-.099	-.106	-.090
		.450	-.118	-.094	-.081	-.079	-.061	-.065	-.042
		.350	-.133	-.092	-.064	-.035	-.013	.007	.023
		.250	-.148	-.077	.037	.015	.049	.067	.106
		.150	-.135	.003	.062	.136	.190	.205	.244
		.075	-.135	.094	.177	.250	.300	.324	.346
		.025	-.097	.246	.329	.376	.395	.377	.358

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TABLE I.- TABULATION OF PRESSURE COEFFICIENTS FOR THE

CLEAN WING - Continued

(b) M = 0.70.

		$\frac{x}{c}$	$\alpha=0°$	$\alpha=42°$	$\alpha=84°$	$\alpha=126°$	$\alpha=167°$	$\alpha=208°$	$\alpha=248°$
$\frac{y}{b/2}=.20$	Upper	.000	.453	.133	-.523	-1.265	-1.645	-.896	-.828
		.025	-.087	-.597	-1.647	-2.179	-1.608	-.899	-.833
		.075	-.101	-.406	-1.682	-2.320	-1.540	-.901	-.839
		.150	-.119	-.337	-.350	-1.917	-1.442	-.901	-.851
		.250	-.158	-.330	-.489	-.569	-1.246	-.904	-.857
		.350	-.166	-.313	-.462	-.549	-.939	-.903	-.867
		.450	-.160	-.274	-.394	-.485	-.812	-.830	-.814
		.550	-.188	-.292	-.396	-.508	-.693	-.889	-.879
		.650	-.178	-.260	-.339	-.431	-.579	-.872	-.877
		.750	-.138	-.189	-.231	-.286	-.479	-.834	-.867
		.850	-.090	-.128	-.157	-.218	-.377	-.796	-.859
		.950	-.031	-.049	-.059	-.185	-.377	-.753	-.843
	Lower	.950	-.032	-.010	.012	.007	-.043	-.147	-.118
		.850	-.083	-.035	.009	.039	.035	.008	.058
		.750	-.134	-.059	.009	.066	.091	.111	.171
		.650	-.170	-.078	.009	.088	.133	.171	.247
		.550	-.203	-.089	.018	.107	.174	.227	.315
		.450	-.177	-.052	.066	.174	.253	.315	.405
		.350	-.150	-.010	.123	.240	.326	.398	.491
		.250	-.131	.030	.177	.307	.407	.483	.584
		.150	-.095	.098	.266	.405	.510	.587	.678
		.075	-.090	.163	.359	.505	.608	.671	.744
		.025	-.037	.314	.503	.591	.634	.664	.674
$\frac{y}{b/2}=.40$	Upper	.000	.476	-.086	-.825	-1.294	-1.091	-.897	-.925
		.025	-.153	-.854	-1.413	-1.180	-1.044	-.891	-.927
		.075	-.159	-.553	-1.218	-1.188	-1.028	-.891	-.925
		.150	-.162	-.434	-.924	-1.246	-1.003	-.888	-.925
		.250	-.173	-.378	-.754	-1.272	-.972	-.880	-.928
		.350	-.191	-.343	-.609	-1.234	-.952	-.870	-.928
		.450	-.197	-.316	-.487	-1.147	-.937	-.857	-.923
		.550	-.180	-.266	-.383	-1.031	-.906	-.849	-.915
		.650	-.138	-.201	-.290	-.860	-.863	-.835	-.898
		.750	-.088	-.128	-.204	-.679	-.816	-.809	-.877
		.850	-.038	-.061	-.125	-.471	-.755	-.766	-.854
		.950	.009	-.009	-.063	-.293	-.705	-.787	-.833
	Lower	.950	-.007	.001	.005	-.032	-.248	-.893	-.291
		.850	-.034	-.009	.014	.008	-.111	-.149	-.127
		.750	-.061	-.021	.021	.041	-.005	-.025	.016
		.650	-.137	-.066	.001	.050	.040	.039	.099
		.550	-.171	-.072	.015	.080	.099	.119	.190
		.450	-.175	-.054	.049	.130	.175	.205	.280
		.350	-.177	-.028	.093	.192	.248	.293	.366
		.250	-.159	.018	.156	.266	.333	.380	.455
		.150	-.137	.081	.239	.360	.431	.476	.539
		.075	-.121	.163	.341	.457	.517	.546	.585
		.025	-.077	.300	.461	.525	.538	.527	.505
$\frac{y}{b/2}=.60$	Upper	.000	.448	-.001	-.383	-.813	-.852	-.741	-.784
		.025	-.148	-.813	-.764	-.786	-.736	-.724	-.814
		.075	-.155	-.655	-.731	-.777	-.752	-.739	-.810
		.150	-.158	-.495	-.677	-.751	-.715	-.714	-.772
		.250	-.164	-.388	-.605	-.712	-.693	-.698	-.758
		.350	-.166	-.328	-.560	-.696	-.674	-.689	-.751
		.450	-.160	-.275	-.540	-.684	-.653	-.681	-.741
		.550	-.138	-.216	-.510	-.674	-.643	-.680	-.733
		.650	-.099	-.156	-.468	-.671	-.630	-.678	-.719
		.750	-.063	-.099	-.405	-.649	-.601	-.662	-.697
		.850	-.013	-.034	-.334	-.612	-.565	-.640	-.678
		.950	.027	.011	-.275	-.576	-.532	-.606	-.657
	Lower	.950	.031	.024	-.077	-.839	-.253	-.288	-.279
		.850	-.011	.003	-.031	-.136	-.164	-.182	-.159
		.750	-.053	-.025	-.016	-.065	-.094	-.093	-.059
		.650	-.105	-.044	-.007	-.014	-.031	-.019	.028
		.550	-.116	-.052	.010	.034	.031	.057	.102
		.450	-.152	-.047	.033	.086	.102	.130	.196
		.350	-.147	-.007	.094	.160	.193	.229	.294
		.250	-.139	.030	.148	.236	.275	.315	.368
		.150	-.122	.098	.230	.329	.378	.412	.464
		.075	-.093	.176	.366	.409	.415	.447	.461
		.025	-.074	.296	.432	.495	.497	.461	.439

NACA

TABLE I.- TABULATION OF PRESSURE COEFFICIENTS FOR THE

CLEAN WING - Continued

(b) M = 0.70. Concluded.

		$\frac{x}{c}$	$\alpha=0°$	$\alpha=42°$	$\alpha=84°$	$\alpha=126°$	$\alpha=167°$	$\alpha=208°$	$\alpha=248°$
$\frac{y}{b/2}=.80$	Upper	.000	-.454	-.223	-.825	-.607	-.569	-.604	-.674
		.025	-.155	-.919	-.993	-.492	-.485	-.593	-.669
		.075	-.163	-.648	-.986	-.482	-.479	-.600	-.680
		.150	-.156	-.454	-.955	-.466	-.467	-.596	-.671
		.250	-.167	-.358	-.813	-.447	-.453	-.577	-.654
		.350	-.170	-.298	-.648	-.484	-.440	-.566	-.642
		.450	-.167	-.251	-.515	-.411	-.434	-.558	-.630
		.550	-.148	-.198	-.402	-.404	-.429	-.545	-.615
		.650	-.103	-.132	-.307	-.395	-.419	-.527	-.593
		.750	-.054	-.072	-.230	-.379	-.410	-.504	-.566
		.850	-.013	-.020	-.153	-.353	-.396	-.474	-.534
		.950	.041	.032	-.093	-.298	-.343	-.407	-.461
	Lower	.950	-.008	-.020	-.078	-.277	-.314	-.347	-.364
		.850	-.005	-.002	-.023	-.157	-.178	-.191	-.182
		.750	-.060	-.041	-.040	-.132	-.142	-.140	-.121
		.650	-.089	-.052	-.028	-.084	-.079	-.067	-.031
		.550	-.123	-.066	-.017	-.044	-.029	-.005	.045
		.450	-.153	-.070	.001	.002	.030	.064	.118
		.350	-.149	-.028	.066	.088	.125	.168	.226
		.250	-.141	.020	.140	.181	.225	.269	.327
		.150	-.126	.090	.223	.281	.328	.370	.420
		.075	-.114	.179	.328	.387	.430	.455	.485
		.025	-.081	.320	.453	.485	.492	.475	.452
$\frac{y}{b/2}=.95$	Upper	.000	.383	-.288	-.403	-.413	-.423	-.455	-.509
		.025	-.120	-.645	-.474	-.322	-.381	-.445	-.504
		.075	-.157	-.514	-.458	-.320	-.384	-.441	-.499
		.150	-.145	-.366	-.412	-.302	-.374	-.429	-.485
		.250	-.123	-.244	-.355	-.270	-.359	-.418	-.473
		.350	-.126	-.208	-.341	-.281	-.380	-.449	-.505
		.450	-.104	-.166	-.306	-.273	-.373	-.445	-.497
		.550	-.086	-.131	-.283	-.273	-.368	-.443	-.494
		.650	.008	-.078	-.221	-.262	-.339	-.406	-.452
		.750	-.008	-.058	-.252	-.270	-.349	-.420	-.460
		.850	.033	-.028	-.234	-.263	-.325	-.391	-.433
		.950	.073	.004	-.198	-.230	-.279	-.334	-.377
	Lower	.950	.050	.011	-.106	-.188	-.215	-.252	-.274
		.850	.058	.032	-.029	-.105	-.129	-.151	-.163
		.750	.001	-.021	-.060	-.141	-.162	-.177	-.184
		.650	-.040	-.053	-.073	-.136	-.154	-.160	-.157
		.550	-.080	-.074	-.080	-.119	-.134	-.131	-.118
		.450	-.094	-.080	-.073	-.088	-.095	-.084	-.062
		.350	-.115	-.084	-.058	-.044	-.040	.020	.008
		.250	-.127	-.073	-.027	.009	.025	.064	.101
		.150	-.121	.007	.087	.144	.176	.219	.250
		.075	-.117	.103	.211	.266	.299	.334	.357
		.025	-.055	.262	.357	.389	.395	.390	.364

NACA

TABLE I.- TABULATION OF PRESSURE COEFFICIENTS FOR THE CLEAN WING - Continued

(c) M = 0.85.

$\frac{y}{b/2}$		$\frac{x}{c}$	$\alpha=0°$	$\alpha=4.3°$	$\alpha=8.6°$	$\alpha=12.8°$	$\alpha=16.8°$
.20	Upper	.000	.473	.269	-.119	-.577	-1.011
		.025	-.061	-.545	-1.266	-1.506	-1.353
		.075	-.096	-.391	-1.200	-1.509	-1.325
		.150	-.110	-.353	-.844	-1.509	-1.273
		.250	-.162	-.360	-.507	-.659	-1.197
		.350	-.177	-.367	-.533	-.663	-1.080
		.450	-.173	-.339	-.542	-.523	-.888
		.550	-.218	-.375	-.583	-.415	-.860
		.650	-.215	-.351	-.601	-.639	-.774
		.750	-.160	-.245	-.346	-.517	-.692
		.850	-.115	-.165	-.177	-.436	-.631
		.950	-.046	-.068	-.072	-.267	-.537
	Lower	.950	-.040	-.023	.001	-.056	-.085
		.850	-.108	-.053	.004	.000	.021
		.750	-.170	-.080	.005	.043	.088
		.650	-.210	-.097	.010	.067	.131
		.550	-.238	-.103	.018	.095	.177
		.460	-.200	-.054	.074	.166	.260
		.350	-.162	-.008	.133	.237	.336
		.250	-.136	.034	.189	.306	.414
		.150	-.090	.107	.281	.407	.528
		.075	-.082	.173	.371	.509	.623
		.025	-.023	.319	.519	.621	.685
.40	Upper	.000	.489	.090	-.396	-.915	-1.049
		.025	-.138	-1.315	-1.513	-1.082	-.988
		.075	-.160	-.561	-1.403	-1.042	-.973
		.150	-.174	-.480	-.890	-1.023	-.941
		.250	-.194	-.454	-.788	-.986	-.906
		.350	-.222	-.431	-.724	-.954	-.879
		.450	-.231	-.400	-.625	-.943	-.857
		.550	-.209	-.311	-.467	-.898	-.834
		.650	-.158	-.227	-.326	-.847	-.810
		.750	-.094	-.139	-.220	-.787	-.785
		.850	-.039	-.061	-.134	-.709	-.755
		.950	.008	-.004	-.075	-.625	-.723
	Lower	.950	.000	.001	-.013	-.182	-.262
		.850	-.034	-.012	.001	-.091	-.134
		.750	-.070	-.026	.014	-.018	-.022
		.650	-.155	-.079	-.011	-.006	.021
		.550	-.201	-.090	-.001	.037	.090
		.450	-.210	-.072	.036	.099	.165
		.350	-.209	-.046	.077	.161	.241
		.250	-.184	.006	.142	.241	.327
		.150	-.155	.073	.227	.334	.427
		.075	-.133	.153	.325	.435	.516
		.025	-.089	.289	.448	.521	.554
.60	Upper	.000	.450	.099	-.258	-.619	-.857
		.025	-.150	-1.101	-.943	-.804	-.749
		.075	-.171	-.792	-.911	-.801	-.793
		.150	-.178	-.592	-.874	-.779	-.736
		.250	-.186	-.441	-.838	-.752	-.714
		.350	-.190	-.378	-.821	-.734	-.704
		.450	-.182	-.313	-.796	-.709	-.687
		.550	-.155	-.242	-.745	-.679	-.678
		.650	-.109	-.171	-.682	-.650	-.659
		.750	-.068	-.106	-.604	-.616	-.641
		.850	-.014	-.033	-.500	-.565	-.561
		.950	.031	.014	-.392	-.522	-.596
	Lower	.950	.031	.022	-.046	-.250	-.302
		.850	-.011	-.003	-.006	-.176	-.204
		.750	-.068	-.033	.003	-.115	-.127
		.650	-.120	-.056	.007	-.059	-.060
		.550	-.134	-.066	.018	-.006	.010
		.450	-.182	-.062	.042	.048	.081
		.350	-.175	-.020	.100	.129	.176
		.250	-.164	.022	.127	.206	.260
		.150	-.149	.085	.158	.305	.360
		.075	-.120	.190	.241	.411	.416
		.025	-.099	.284	.317	.479	.485

NACA

TABLE I.- TABULATION OF PRESSURE COEFFICIENTS FOR THE CLEAN WING - Continued

(c) M = 0.85. Concluded.

$\frac{y}{b/2}$		$\frac{x}{c}$	$\alpha=0°$	$\alpha=4.3°$	$\alpha=8.6°$	$\alpha=12.8°$	$\alpha=16.8°$
.80	Upper	.000	.469	-.044	-.492	-.596	-.640
		.025	-.142	-1.505	-.737	-.534	-.579
		.075	-.162	-.786	-.712	-.524	-.576
		.150	-.162	-.502	-.682	-.502	-.568
		.250	-.160	-.390	-.652	-.480	-.551
		.350	-.181	-.330	-.621	-.464	-.535
		.450	-.179	-.267	-.594	-.454	-.525
		.550	-.152	-.200	-.561	-.453	-.517
		.650	-.101	-.128	-.520	-.447	-.506
		.750	-.045	-.067	-.471	-.430	-.495
		.850	.006	-.013	-.402	-.408	-.479
		.950	.064	.041	-.327	-.342	-.409
	Lower	.950	.006	-.014	-.212	-.344	-.406
		.850	.011	-.002	-.101	-.220	-.257
		.750	-.051	-.047	-.087	-.193	-.214
		.650	-.096	-.064	-.061	-.140	-.137
		.550	-.141	-.082	-.043	-.092	-.070
		.450	-.168	-.081	-.019	-.039	-.005
		.350	-.168	-.035	.050	.056	.102
		.250	-.163	.026	.130	.155	.202
		.150	-.146	.088	.211	.255	.310
		.075	-.133	.175	.313	.359	.406
		.025	-.100	.313	.439	.462	.475
.95	Upper	.000	.413	-.199	-.250	-.366	-.489
		.025	-.115	-.763	-.353	-.318	-.456
		.075	-.173	-.605	-.344	-.318	-.447
		.150	-.179	-.426	-.312	-.305	-.433
		.250	-.136	-.272	-.260	-.280	-.414
		.350	-.136	-.230	-.256	-.310	-.448
		.450	-.109	-.176	-.234	-.308	-.440
		.550	-.086	-.138	-.230	-.316	-.443
		.650	-.031	-.087	-.197	-.278	-.380
		.750	.005	-.062	-.230	-.313	-.410
		.850	.050	-.033	-.224	-.305	-.382
		.950	.094	.005	-.187	-.261	-.320
	Lower	.950	.069	.018	-.172	-.236	-.288
		.850	.085	.050	-.084	-.139	-.183
		.750	.015	-.015	-.128	-.196	-.243
		.650	-.031	-.051	-.131	-.195	-.240
		.550	-.080	-.082	-.129	-.184	-.221
		.450	-.097	-.096	-.109	-.145	-.171
		.350	-.127	-.110	-.082	-.099	-.107
		.250	-.146	-.110	-.047	-.033	-.014
		.150	-.157	-.015	.082	.123	.157
		.075	-.144	.108	.213	.258	.294
		.025	-.070	.267	.362	.386	.387

NACA

TABLE I.- TABULATION OF PRESSURE COEFFICIENTS FOR THE

CLEAN WING - Continued

(d) M = 0.91.

$\frac{y}{b/2}$		$\frac{x}{c}$	$\alpha=0°$	$\alpha=4.4°$	$\alpha=8.6°$	$\alpha=12.8°$
.20	Upper	.000	.485	.330	.019	-.378
		.025	-.052	-.550	-1.152	-1.342
		.075	-.076	-.387	-1.004	-1.342
		.150	-.102	-.317	-.735	-1.321
		.250	-.162	-.354	-.482	-.621
		.350	-.187	-.363	-.493	-.602
		.450	-.193	-.362	-.500	-.600
		.550	-.249	-.402	-.564	-.682
		.650	-.269	-.436	-.598	-.664
		.750	-.193	-.405	-.560	-.569
		.850	-.157	-.388	-.591	-.489
		.950	-.061	-.159	-.370	-.342
	Lower	.950	-.061	-.069	-.114	-.087
		.850	-.146	-.095	-.075	-.011
		.750	-.245	-.115	-.054	.030
		.650	-.267	-.123	-.038	.063
		.550	-.322	-.127	-.022	.091
		.450	-.210	-.068	.046	.166
		.350	-.169	-.014	.109	.241
		.250	-.138	.033	.173	.310
		.150	-.084	.110	.265	.410
		.075	-.077	.175	.354	.514
		.025	-.018	.322	.510	.637
.40	Upper	.000	.485	.159	-.208	-.687
		.025	-.153	-1.153	-1.316	-1.317
		.075	-.174	-.587	-1.289	-1.249
		.150	-.195	-.486	-1.219	-1.185
		.250	-.227	-.466	-.704	-1.104
		.350	-.271	-.483	-.657	-1.046
		.450	-.304	-.517	-.677	-.988
		.550	-.268	-.543	-.703	-.930
		.650	-.181	-.498	-.698	-.877
		.750	-.114	-.187	-.477	-.808
		.850	-.053	-.056	-.203	-.730
		.950	-.004	.005	-.092	-.637
	Lower	.950	-.014	-.009	-.064	-.177
		.850	-.046	-.028	-.063	-.092
		.750	-.085	-.045	-.050	-.025
		.650	-.185	-.109	-.083	-.018
		.550	-.259	-.127	-.061	.022
		.450	-.268	-.103	-.012	.087
		.350	-.265	-.069	.040	.153
		.250	-.215	-.013	.111	.232
		.150	-.172	.059	.197	.325
		.075	-.143	.140	.296	.424
		.025	-.091	.273	.424	.520
.60	Upper	.000	.429	.166	-.127	-.491
		.025	-.190	-1.212	-1.314	-.826
		.075	-.213	-1.076	-1.200	-.818
		.150	-.229	-.563	-1.085	-.805
		.250	-.237	-.553	-1.001	-.787
		.350	-.230	-.559	-.902	-.771
		.450	-.216	-.581	-.776	-.742
		.550	-.176	-.257	-.649	-.726
		.650	-.129	-.122	-.548	-.704
		.750	-.087	-.062	-.472	-.680
		.850	-.028	-.008	-.393	-.641
		.950	.022	.031	-.339	-.614
	Lower	.950	.027	.020	-.140	-.316
		.850	-.025	-.011	-.095	-.219
		.750	-.084	-.045	-.075	-.148
		.650	-.139	-.072	-.066	-.089
		.550	-.170	-.087	-.051	-.035
		.450	-.216	-.088	-.026	.022
		.350	-.201	-.045	.035	.107
		.250	-.197	-.003	.091	.185
		.150	-.186	.062	.176	.283
		.075	-.145	.182	.258	.376
		.025	-.122	.254	.379	.464

NACA

TABLE I.- TABULATION OF PRESSURE COEFFICIENTS FOR THE

CLEAN WING - Continued

(d) M = 0.91. Concluded.

$\frac{y}{b/2}$		$\frac{x}{c}$	$\alpha=0°$	$\alpha=4.4°$	$\alpha=8.6°$	$\alpha=12.8°$
.80	Upper	.000	.463	.028	-.378	-.660
		.025	-.168	-1.297	-1.006	-.664
		.075	-.192	-1.297	-1.001	-.641
		.150	-.187	-1.031	-.935	-.615
		.250	-.208	-.618	-.851	-.605
		.350	-.215	-.354	-.755	-.589
		.450	-.211	-.174	-.695	-.571
		.550	-.182	-.143	-.630	-.558
		.650	-.112	-.090	-.562	-.544
		.750	-.049	-.033	-.503	-.527
		.850	.001	.014	-.426	-.505
		.950	.064	.067	-.324	-.427
	Lower	.950	.002	.002	-.249	-.462
		.850	.005	.001	-.151	-.317
		.750	-.059	-.056	-.144	-.272
		.650	-.108	-.088	-.118	-.194
		.550	-.170	-.104	-.097	-.130
		.450	-.201	-.103	-.067	-.067
		.350	-.196	-.052	.000	.032
		.250	-.186	.010	.079	.133
		.150	-.171	.068	.157	.230
		.075	-.156	.154	.258	.332
		.025	-.120	.289	.383	.439
.95	Upper	.000	.410	-.204	-.376	-.529
		.025	-.146	-1.006	-.526	-.483
		.075	-.220	-1.006	-.531	-.478
		.150	-.271	-.599	-.455	-.460
		.250	-.168	-.411	-.377	-.425
		.350	-.156	-.191	-.369	-.454
		.450	-.132	-.118	-.332	-.439
		.550	-.090	-.109	-.326	-.442
		.650	-.101	-.087	-.243	-.377
		.750	.003	-.059	-.295	-.421
		.850	.044	-.030	-.276	-.398
		.950	.097	.011	-.229	-.332
	Lower	.950	.072	.043	-.196	-.325
		.850	.096	.073	-.097	-.230
		.750	.011	-.009	-.151	-.326
		.650	-.038	-.049	-.165	-.327
		.550	-.087	-.085	-.177	-.305
		.450	-.109	-.103	-.173	-.253
		.350	-.145	-.144	-.161	-.196
		.250	-.174	-.190	-.115	-.100
		.150	-.217	-.029	.050	.091
		.075	-.160	.112	.194	.238
		.025	-.073	.271	.339	.363

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TABLE I.- TABULATION OF PRESSURE COEFFICIENTS FOR THE CLEAN WING - Continued

(e) M = 0.93 and 0.95.

M=.93

$\frac{y}{b/2}$		$\frac{x}{c}$	$\alpha=0°$	$\alpha=4.4°$	$\alpha=8.6°$
.20	Upper	.000	.496	.342	.061
		.025	-.048	-.551	-1.092
		.075	-.076	-.381	-.923
		.150	-.104	-.295	-.676
		.250	-.165	-.334	-.475
		.350	-.197	-.346	-.473
		.450	-.208	-.354	-.480
		.550	-.253	-.397	-.540
		.650	-.289	-.432	-.569
		.750	-.301	-.422	-.555
		.850	-.254	-.419	-.579
		.950	-.124	-.255	-.420
	Lower	.950	-.116	-.116	-.141
		.850	-.244	-.130	-.091
		.750	-.312	-.146	-.068
		.650	-.295	-.144	-.048
		.550	-.323	-.141	-.029
		.450	-.209	-.072	.046
		.350	-.171	-.018	.111
		.250	-.139	.032	.175
		.150	-.084	.112	.273
		.075	-.078	.177	.364
		.025	-.015	.323	.517
.40	Upper	.000	.483	.193	-.148
		.025	-.157	-1.080	-1.262
		.075	-.178	-.555	-1.202
		.150	-.206	-.474	-1.147
		.250	-.237	-.447	-.688
		.350	-.291	-.468	-.615
		.450	-.334	-.508	-.643
		.550	-.371	-.535	-.674
		.650	-.290	-.538	-.681
		.750	-.130	-.371	-.616
		.850	-.050	-.135	-.361
		.950	-.003	-.039	-.220
	Lower	.950	-.016	-.041	-.146
		.850	-.049	-.062	-.129
		.750	-.101	-.080	-.099
		.650	-.281	-.150	-.122
		.550	-.351	-.165	-.094
		.450	-.335	-.129	-.036
		.350	-.280	-.093	-.020
		.250	-.226	-.028	.094
		.150	-.183	.044	.183
		.075	-.153	.129	.285
		.025	-.101	.263	.414
.60	Upper	.000	.412	.201	-.066
		.025	-.210	-1.186	-1.242
		.075	-.238	-1.005	-1.179
		.150	-.268	-.541	-1.011
		.250	-.303	-.537	-.918
		.350	-.313	-.559	-.858
		.450	-.264	-.595	-.765
		.550	-.170	-.572	-.642
		.650	-.133	-.235	-.518
		.750	-.094	-.083	-.430
		.850	-.037	-.032	-.363
		.950	.010	.035	-.314
	Lower	.950	.020	.011	-.189
		.850	-.035	-.021	-.145
		.750	-.097	-.062	-.125
		.650	-.147	-.096	-.113
		.550	-.190	-.117	-.099
		.450	-.257	-.118	-.072
		.350	-.270	-.077	-.002
		.250	-.261	-.030	.063
		.150	-.230	.036	.123
		.075	-.191	.193	.155
		.025	-.159	.232	.361

M=.95

$\frac{y}{b/2}$		$\frac{x}{c}$	$\alpha=0°$	$\alpha=4.4°$	$\alpha=6.5°$
.20	Upper	.000	.505	.368	.235
		.025	-.035	-.530	-.803
		.075	-.059	-.362	-.586
		.150	-.090	-.275	-.426
		.250	-.155	-.314	-.411
		.350	-.192	-.327	-.405
		.450	-.204	-.347	-.412
		.550	-.243	-.387	-.465
		.650	-.277	-.425	-.499
		.750	-.290	-.415	-.483
		.850	-.284	-.424	-.499
		.950	-.219	-.312	-.376
	Lower	.950	-.211	-.176	-.167
		.850	-.286	-.180	-.137
		.750	-.299	-.188	-.131
		.650	-.277	-.167	-.112
		.550	-.295	-.172	-.098
		.450	-.187	-.076	-.016
		.350	-.151	-.022	.048
		.250	-.122	.031	.108
		.150	-.063	.113	.194
		.075	-.055	.174	.277
		.025	.007	.320	.427
.40	Upper	.000	-.170	.223	.055
		.025	-.145	-1.035	-1.266
		.075	-.168	-.542	-1.096
		.150	-.189	-.454	-.704
		.250	-.225	-.432	-.506
		.350	-.283	-.461	-.532
		.450	-.326	-.498	-.561
		.550	-.369	-.527	-.597
		.650	-.375	-.536	-.604
		.750	-.271	-.497	-.569
		.850	-.105	-.268	-.346
		.950	-.024	-.151	-.217
	Lower	.950	-.040	-.128	-.170
		.850	-.100	-.145	-.162
		.750	-.242	-.155	
		.650	-.366	-.231	-.175
		.550	-.360	-.219	-.162
		.450	-.320	-.163	-.099
		.350	-.263	-.114	-.049
		.250	-.211	-.046	.022
		.150	-.172	.032	.114
		.075	-.141	.120	.213
		.025	-.085	.252	.341
.60	Upper	.000	-.422	.219	.095
		.025	-.199	-1.074	-1.248
		.075	-.233	-.949	-1.139
		.150	-.260	-.518	-1.029
		.250	-.308	-.523	-.758
		.350	-.348	-.559	-.624
		.450	-.375	-.585	-.650
		.550	-.300	-.617	-.664
		.650	-.182	-.568	-.661
		.750	-.079	-.280	-.442
		.850	-.021	-.135	-.317
		.950	.031	-.061	-.196
	Lower	.950	.036	-.069	-.151
		.850	-.015	-.099	-.157
		.750	-.083	-.142	-.159
		.650	-.196	-.181	-.179
		.550	-.297	-.202	-.167
		.450	-.351	-.189	-.149
		.350	-.311	-.133	-.076
		.250	-.270	-.079	-.013
		.150	-.236	-.005	.071
		.075	-.186	.117	.204
		.025	-.146	.197	.284

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TABLE I.- TABULATION OF PRESSURE COEFFICIENTS FOR THE

CLEAN WING - Continued

(e) M = 0.93 and 0.95. Concluded.

$\frac{y}{b/2}$		$\frac{x}{c}$	$\alpha=0°$	$\alpha=4.4°$	$\alpha=8.6°$	$\frac{x}{c}$	$\alpha=0°$	$\alpha=4.4°$	$\alpha=6.5°$
=.80	Upper	.000	.443	.081	-.276	.000	.416	.135	-.016
		.025	-.204	-1.258	-1.277	.025	-.244	-1.091	-1.257
		.075	-.224	-1.260	-1.270	.075	-.278	-.984	-1.142
		.150	-.211	-.956	-1.098	.150	-.284	-.883	-1.065
		.250	-.222	-.630	-.997	.250	-.299	-.556	-1.014
		.350	-.235	-.618	-.883	.350	-.290	-.619	-.981
		.450	-.232	-.449	-.776	.450	-.264	-.669	-.838
		.550	-.216	-.135	-.694	.550	-.223	-.632	-.745
		.650	-.124	-.039	-.616	.650	-.127	-.321	-.703
		.750	-.050	.007	-.569	.750	-.037	-.115	-.408
		.850	.002	.042	-.497	.850	.021	-.011	-.216
		.950	.067	.086	-.367	.950	.087	.056	-.080
	Lower	.950	.003	.008	-.286	.950	.025	-.056	-.219
		.850	.000	.001	-.197	.850	.023	-.104	-.239
		.750	-.064	-.080	-.200	.750	-.045	-.204	-.263
		.650	-.131	-.123	-.166	.650	-.127	-.204	-.229
		.550	-.210	-.129	-.137	.550	-.208	-.197	-.205
		.450	-.231	-.125	-.111	.450	-.258	-.196	-.188
		.350	-.216	-.077	-.043	.350	-.266	-.145	-.120
		.250	-.207	-.016	.038	.250	-.272	-.088	-.049
		.150	-.196	.042	.117	.150	-.261	-.027	.026
		.075	-.188	.124	.215	.075	-.236	.054	.117
		.025	-.155	.257	.347	.025	-.195	.190	.259
=.95	Upper	.000	.409	-.155	-.489	.000	.393	-.095	-.239
		.025	-.164	-.724	-.767	.025	-.209	-.759	-1.168
		.075	-.241	-.724	-.757	.075	-.289	-.752	-1.039
		.150	-.309	-.720	-.740	.150	-.331	-.739	-.956
		.250	-.223	-.505	-.679	.250	-.278	-.739	-.874
		.350	-.162	-.246	-.665	.350	-.175	-.522	-.914
		.450	-.149	-.070	-.550	.450	-.122	-.312	-.831
		.550	-.099	-.062	-.524	.550	-.075	-.136	-.751
		.650	-.129	-.070	-.332	.650	-.083	-.055	-.376
		.750	.000	-.037	-.409	.750	.022	-.017	-.487
		.850	.042	-.011	-.361	.850	.064	.007	-.286
		.950	.095	.030	-.295	.950	.115	.046	-.121
	Lower	.950	.071	.053	-.191	.950	.092	.056	-.083
		.850	.094	.079	-.090	.850	.113	.083	-.073
		.750	.007	.003	-.161	.750	.028	-.015	-.244
		.650	-.051	-.037	-.219	.650	-.024	-.111	-.360
		.550	-.100	-.072	-.295	.550	-.074	-.263	-.401
		.450	-.116	-.122	-.301	.450	-.085	-.348	-.369
		.350	-.164	-.254	-.274	.350	-.195	-.349	-.330
		.250	-.236	-.238	-.199	.250	-.288	-.289	-.246
		.150	-.243	-.045	.016	.150	-.244	-.079	-.029
		.075	-.173	.093	.162	.075	-.187	.051	.111
		.025	-.099	.250	.310	.025	-.121	.198	.265

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[redacted]

TABLE I.- TABULATION OF PRESSURE COEFFICIENTS FOR THE CLEAN WING - Continued

(f) M = 0.91 (repeat tests).

		$\frac{x}{c}$	$\alpha=0°$	$\alpha=4.4°$	$\alpha=8.6°$	$\alpha=12.8°$
$\frac{y}{b/2}=.20$	Upper	.000	.488	.320	-.038	-.439
		.025	-.077	-.691	-1.238	-1.430
		.075	-.101	-.402	-1.005	-1.344
		.150	-.119	-.352	-.719	-1.283
		.250	-.179	-.360	-.518	-.713
		.350	-.216	-.400	-.554	-.626
		.450	-.243	-.421	-.572	-.669
		.550	-.264	-.447	-.606	-.696
		.650	-.298	-.481	-.622	-.655
		.750	-.258	-.475	-.621	-.550
		.850	-.174	-.447	-.608	-.498
		.950	-.078	-.245	-.452	-.380
	Lower	.950	-.071	-.112	-.135	-.087
		.850	-.145	-.118	-.079	-.001
		.750	-.239	-.140	-.063	.034
		.650	-.289	-.147	-.046	.061
		.550	-.322	-.150	-.033	.095
		.450	-.211	-.081	.041	.174
		.350	-.171	-.028	.106	.246
		.250	-.140	.019	.171	.317
		.150	-.086	.100	.264	.420
		.075	-.076	.170	.362	.524
		.025	-.006	.319	.513	.640
$\frac{y}{b/2}=.40$	Upper	.000	.495	.186	-.180	-.711
		.025	-.179	-1.184	-1.411	-1.206
		.075	-.194	-.637	-1.322	-1.163
		.150	-.213	-.513	-1.244	-1.117
		.250	-.246	-.506	-.753	-1.045
		.350	-.290	-.520	-.681	-.992
		.450	-.318	-.547	-.703	-.940
		.550	-.285	-.584	-.730	-.879
		.650	-.191	-.560	-.728	-.838
		.750	-.122	-.274	-.552	-.779
		.850	-.058	-.101	-.235	-.729
		.950	-.004	-.035	-.116	-.657
	Lower	.950	-.019	-.042	-.094	-.202
		.850	-.051	-.056	-.088	-.102
		.750	-.095	-.074	-.076	-.030
		.650	-.190	-.138	-.100	-.016
		.550	-.226	-.133	-.059	.042
		.450	-.276	-.129	-.028	.090
		.350	-.260	-.092	.028	.158
		.250	-.221	-.034	.097	.237
		.150	-.177	.040	.189	.330
		.075	-.145	.126	.290	.430
		.025	-.086	.264	.416	.520
$\frac{y}{b/2}=.60$	Upper	.000	.443	.153	-.159	-.532
		.025	-.824	-1.833	-1.400	-.842
		.075	-.243	-1.088	-1.246	-.855
		.150	-.255	-.631	-1.098	-.798
		.250	-.251	-.587	-1.004	-.778
		.350	-.241	-.614	-.917	-.762
		.450	-.229	-.645	-.788	-.739
		.550	-.192	-.421	-.660	-.725
		.650	-.145	-.165	-.550	-.696
		.750	-.094	-.080	-.459	-.672
		.850	-.036	-.022	-.386	-.641
		.950	.016	.018	-.325	-.592
	Lower	.950	.012	-.009	-.162	-.316
		.850	-.037	-.038	-.122	-.244
		.750	-.098	-.075	-.100	-.148
		.650	-.151	-.100	-.090	-.091
		.550	-.195	-.119	-.074	-.040
		.450	-.230	-.119	-.053	.019
		.350	-.212	-.072	.015	.113
		.250	-.207	-.027	.076	.189
		.150	-.182	.037	.166	.285
		.075	-.169	.112	.248	.367
		.025	-.131	.234	.367	.461

TABLE I.- TABULATION OF PRESSURE COEFFICIENTS FOR THE CLEAN WING - Continued

(f) M = 0.91 (repeat tests). Concluded.

$\frac{y}{b/2}$		$\frac{x}{c}$	$\alpha=0°$	$\alpha=4.4°$	$\alpha=8.6°$	$\alpha=12.8°$
.80	Upper	.000	.461	.003	-.402	-.650
		.025	-.218	-1.277	-.997	-.632
		.075	-.221	-1.147	-.981	-.612
		.150	-.208	-1.047	-.931	-.597
		.250	-.227	-.665	-.841	-.577
		.350	-.232	-.585	-.773	-.561
		.450	-.228	-.218	-.711	-.545
		.550	-.195	-.112	-.642	-.533
		.650	-.119	-.066	-.573	-.517
		.750	-.053	-.024	-.515	-.508
		.850	.003	.015	-.443	-.481
		.950	.051	.051	-.361	-.422
	Lower	.950	.026	-.002	-.254	-.406
		.850	.004	-.016	-.178	-.301
		.750	-.061	-.073	-.171	-.257
		.650	-.120	-.118	-.152	-.194
		.550	-.175	-.129	-.118	-.122
		.450	-.219	-.122	-.085	-.060
		.350	-.206	-.072	-.013	.035
		.250	-.182	-.012	.064	.135
		.150	-.163	.054	.140	.229
		.075	-.135	.152	.169	.331
		.025	-.094	.277	.372	.444
.95	Upper	.000	.410	-.264	-.430	-.526
		.025	-.173	-1.242	-.517	-.438
		.075	-.244	-1.131	-.534	-.445
		.150	-.252	-.412	-.424	-.371
		.250	-.141	-.211	-.367	-.356
		.350	-.169	-.115	-.399	-.429
		.450	-.142	-.121	-.356	-.418
		.550	-.105	-.092	-.343	-.423
		.650	-.055	-.076	-.331	-.415
		.750	-.005	-.050	-.317	-.400
		.850	.044	-.005	-.302	-.378
		.950	.054	.016	-.267	-.334
	Lower	.950	.063	.016	-.283	-.302
		.850	.055	-.034	-.163	-.260
		.750	.001	-.076	-.177	-.289
		.650	-.052	-.114	-.188	-.296
		.550	-.107	-.131	-.205	-.292
		.450	-.132	-.202	-.201	-.247
		.350	-.167	-.243	-.191	-.185
		.250	-.202	-.060	-.155	-.097
		.150	-.241	.084	.021	.100
		.075	-.175	.250	.175	.240
		.025	-.095	.250	.329	.370

TABLE I.- TABULATION OF PRESSURE COEFFICIENTS FOR THE CLEAN WING - Continued

(g) M = 0.95 (repeat tests).

$\frac{y}{b/2}$		$\frac{x}{c}$	$\alpha=0°$	$\alpha=4.4°$	$\alpha=6.5°$
.20	Upper	.000	.493	.365	.218
		.025	-.068	-.654	-1.094
		.075	-.092	-.351	-.519
		.150	-.110	-.323	-.442
		.250	-.174	-.334	-.413
		.350	-.212	-.371	-.446
		.450	-.254	-.396	-.464
		.550	-.278	-.424	-.498
		.650	-.319	-.456	-.510
		.750	-.331	-.456	-.510
		.850	-.321	-.447	-.501
		.950	-.285	-.365	-.403
	Lower	.950	-.264	-.197	-.163
		.850	-.314	-.195	-.121
		.750	-.317	-.208	-.121
		.650	-.314	-.198	-.108
		.550	-.306	-.194	-.093
		.450	-.208	-.088	-.009
		.350	-.171	-.032	.054
		.250	-.134	.019	.112
		.150	-.076	.103	.202
		.075	-.069	.169	.288
		.025	.001	.319	.440
.40	Upper	.000	.484	.247	.081
		.025	-.183	-1.062	-1.300
		.075	-.199	-.652	-1.105
		.150	-.224	-.447	-.810
		.250	-.249	-.462	-.514
		.350	-.307	-.490	-.548
		.450	-.356	-.519	-.575
		.550	-.404	-.562	-.610
		.650	-.408	-.563	-.615
		.750	-.362	-.529	-.571
		.850	-.193	-.323	-.349
		.950	-.098	-.206	-.217
	Lower	.950	-.109	-.181	-.169
		.850	-.177	-.192	-.160
		.750	-.319	-.201	-.130
		.650	-.398	-.259	-.174
		.550	-.364	-.218	-.130
		.450	-.345	-.183	-.095
		.350	-.276	-.131	-.041
		.250	-.287	-.062	.030
		.150	-.183	.020	.117
		.075	-.149	.108	.215
		.025	-.088	.244	.352
.60	Upper	.000	.417	.210	.086
		.025	-.244	-1.073	-2.248
		.075	-.272	-.954	-1.149
		.150	-.296	-.554	-1.055
		.250	-.346	-.559	-.836
		.350	-.394	-.581	-.640
		.450	-.442	-.621	-.654
		.550	-.414	-.639	-.684
		.650	-.310	-.649	-.698
		.750	-.145	-.400	-.431
		.850	-.058	-.242	-.253
		.950	-.003	-.161	-.160
	Lower	.950	-.003	-.150	-.128
		.850	-.062	-.175	-.148
		.750	-.152	-.215	-.160
		.650	-.302	-.251	-.182
		.550	-.393	-.258	-.173
		.450	-.398	-.231	-.146
		.350	-.332	-.164	-.070
		.250	-.288	-.104	-.010
		.150	-.250	-.024	.202
		.075	-.211	.055	.161
		.025	-.126	.172	.287

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TABLE I.- TABULATION OF PRESSURE COEFFICIENTS FOR THE CLEAN WING - Concluded

(g) M = 0.95 (repeat tests). Concluded.

		$\frac{x}{c}$	$\alpha=0°$	$\alpha=4.4°$	$\alpha=6.5°$
$\frac{y}{b/2}=.80$	Upper	.000	.405	.110	-.048
		.025	-.288	-1.076	-1.241
		.075	-.325	-.980	-1.141
		.150	-.336	-.895	-1.065
		.250	-.374	-.600	-1.017
		.350	-.382	-.631	-.993
		.450	-.360	-.683	-.890
		.550	-.291	-.740	-.720
		.650	-.149	-.634	-.510
		.750	-.057	-.300	-.306
		.850	.005	-.159	-.163
		.950	.052	-.072	-.044
	Lower	.950	.027	-.185	-.115
		.850	-.001	-.249	-.175
		.750	-.079	-.293	-.230
		.650	-.163	-.271	-.213
		.550	-.263	-.243	-.187
		.450	-.343	-.242	-.170
		.350	-.341	-.187	-.104
		.250	-.326	-.128	-.030
		.150	-.259	-.068	.039
		.075	-.232	.061	.186
		.025	-.204	.161	.281
$\frac{y}{b/2}=.95$	Upper	.000		-.115	-.280
		.025		-1.036	-1.194
		.075		-.948	-1.125
		.150		-.792	-.917
		.250		-.721	-.881
		.350		-.617	-.882
		.450		-.580	-.620
		.550		-.484	-.502
		.650		-.341	-.437
		.750		-.135	-.425
		.850		-.073	-.302
		.950		-.043	-.148
	Lower	.950		-.013	-.009
		.850		-.039	-.012
		.750		-.187	-.108
		.650		-.354	-.250
		.550		-.445	-.377
		.450		-.420	-.369
		.350		-.386	-.334
		.250		-.327	-.253
		.150		-.116	-.034
		.075		.010	.107
		.025		.164	.266

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TABLE II.- TABULATION OF PRESSURE COEFFICIENTS FOR THE

FENCE-ON CONDITION

(a) M = 0.50.

		$\frac{x}{c}$	$\alpha=0°$	$\alpha=4.2°$	$\alpha=8.3°$	$\alpha=12.5°$	$\alpha=16.5°$	$\alpha=20.6°$	$\alpha=24.6°$
$\frac{y}{b/2}=.20$	Upper	.000	.433	.006	-1.788	-2.399	-2.504	-1.109	-1.017
		.025	-.107	-.639	-1.490	-2.581	-2.443	-1.109	-1.017
		.075	-.124	-.434	-.443	-3.178	-2.541	-1.109	-1.020
		.150	-.134	-.348	-.475	-.686	-2.814	-1.108	-1.023
		.250	-.169	-.331	-.435	-.579	-1.337	-1.108	-1.032
		.350	-.172	-.310	-.378	-.496	-.836	-1.102	-1.030
		.450	-.170	-.278	-.366	-.409	-.652	-1.032	-.971
		.550	-.188	-.283	-.317	-.409	-.607	-1.063	-1.014
		.650	-.180	-.249	-.237	-.376	-.550	-1.003	-1.005
		.750	-.129	-.193	-.157	-.287	-.425	-.922	-.960
		.850	-.098	-.132	-.076	-.211	-.368	-.810	-.931
		.950	-.038	-.065	-.076	-.120	-.259	-.702	-.887
	Lower	.950	-.043	-.022	-.008	-.004	-.021	-.173	-.184
		.850	-.091	-.049	-.007	.028	.049	-.038	-.008
		.750	-.134	-.074	-.010	.048	.093	.062	.108
		.650	-.170	-.087	-.010	.064	.134	.122	.187
		.550	-.201	-.104	-.003	.087	.173	.180	.258
		.450	-.183	-.068	.050	.146	.238	.266	.358
		.350	-.167	-.032	.099	.215	.319	.348	.437
		.250	-.153	.005	.153	.281	.393	.446	.524
		.150	-.121	.072	.240	.381	.498	.534	.625
		.075	-.117	.136	.301	.491	.603	.620	.689
		.025	-.065	.291	.487	.550	.571	.564	.570
$\frac{y}{b/2}=.40$	Upper	.000	.459	-.285	-1.245	-1.548	-1.165	-1.101	-1.063
		.025	-.147	-.836	-1.069	-1.234	-1.115	-1.081	-1.054
		.075	-.153	-.536	-.901	-1.253	-1.096	-1.096	-1.050
		.150	-.160	-.422	-.765	-1.251	-1.054	-1.096	-1.050
		.250	-.170	-.361	-.621	-1.237	-1.037	-1.056	-1.050
		.350	-.185	-.334	-.494	-1.124	-.999	-1.030	-1.040
		.450	-.189	-.306	-.382	-.962	-.962	-1.020	-1.027
		.550	-.172	-.262	-.290	-.804	-.902	-1.004	-1.010
		.650	-.140	-.208	-.209	-.654	-.821	-.974	-.982
		.750	-.095	-.143	-.131	-.527	-.751	-.915	-.948
		.850	-.797	-.073	-.072	-.404	-.664	-.826	-.913
		.950	-.006	-.020	-.012	-.303	-.619	-.784	-.892
	Lower	.950	-.027	-.014	-.006	-.059	-.204	-.330	-.363
		.850	-.053	-.025	.001	-.014	-.083	-.172	-.167
		.750	-.081	-.041	-.016	.023	.002	-.046	-.020
		.650	-.146	-.073	-.005	.030	.050	.024	.071
		.550	-.170	-.083	.033	.059	.096	.107	.160
		.450	-.173	-.065	.076	.108	.167	.185	.252
		.350	-.175	-.043	.133	.163	.236	.272	.336
		.250	-.167	-.002	.214	.241	.323	.356	.427
		.150	-.152	.060	.319	.339	.417	.454	.515
		.075	-.140	.133	.434	.443	.505	.529	.561
		.025	-.107	.286		.503	.504	.469	.442
$\frac{y}{b/2}=.60$	Upper	.000	.438	-.092	-.430	-.752	-.859	-.783	-.810
		.025	-.141	-.758	-.694	-.694	-.698	-.764	-.797
		.075	-.149	-.607	-.661	-.693	-.698	-.764	-.809
		.150	-.163	-.478	-.616	-.667	-.682	-.758	-.841
		.250	-.177	-.386	-.565	-.616	-.649	-.729	-.801
		.350	-.170	-.320	-.550	-.570	-.627	-.717	-.780
		.450	-.162	-.269	-.483	-.514	-.586	-.709	-.762
		.550	-.144	-.223	-.394	-.462	-.532	-.709	-.742
		.650	-.116	-.170	-.321	-.423	-.469	-.697	-.789
		.750	-.082	-.114	-.297	-.388	-.435	-.672	-.693
		.850	-.033	-.055	-.237	-.320	-.394	-.648	-.670
		.950	.000	-.004	-.192	-.293	-.333	-.612	-.640
	Lower	.950	.010	.005	-.055	-.140	-.172	-.296	-.304
		.850	-.987	-.014	-.046	-.121	-.105	-.189	-.182
		.750	-.076	-.040	-.022	-.046	-.051	-.101	-.083
		.650	-.119	-.063	-.017	-.010	-.002	.031	.010
		.550	-.153	-.066	-.008	.021	.054	.047	.091
		.450	-.163	-.066	.015	.062	.111	.113	.172
		.350	-.165	-.028	.066	.130	.197	.216	.275
		.250	-.157	.009	.180	.202	.262	.312	.366
		.150	-.144	.079	.212	.291	.362	.409	.459
		.075	-.139	.186	.310	-.257	.122	-.219	.027
		.025	-.137	.271	.398	-.866	-.289	-.889	-.336

NACA

TABLE II.- TABULATION OF PRESSURE COEFFICIENTS FOR THE FENCE-ON CONDITION - Continued

(a) M = 0.50. Concluded.

		$\frac{x}{c}$	$\alpha=0°$	$\alpha=4.2°$	$\alpha=8.3°$	$\alpha=12.5°$	$\alpha=16.5°$	$\alpha=20.6°$	$\alpha=24.6°$
$\frac{y}{b/2}=.80$	Upper	.000	.441	-.420	-1.292	-1.681	-1.577	-.639	-.688
		.025	-.143	-.886	-1.345	-1.462	-1.428	-.639	-.689
		.075	-.156	-.558	-1.345	-1.528	-1.541	-.639	-.687
		.150	-.153	-.415	-1.194	-1.751	-1.714	-.634	-.677
		.250	-.167	-.341	-.790	-1.617	-1.610	-.626	-.669
		.350	-.166	-.294	-.509	-.730	-.991	-.617	-.658
		.450	-.161	-.256	-.377	-.272	-.642	-.600	-.641
		.550	-.144	-.214	-.289	-.206	-.444	-.583	-.625
		.650	-.107	-.147	-.213	-.183	-.341	-.552	-.602
		.750	-.065	-.091	-.154	-.172	-.314	-.527	-.580
		.850	-.028	-.032	-.094	-.142	-.288	-.496	-.550
		.950	.014	.014	-.041	-.084	-.245	-.435	-.485
	Lower	.950	-.055	-.055	-.069	-.107	-.192	-.363	-.378
		.850	-.027	-.013	-.013	-.038	-.079	-.195	-.188
		.750	-.072	-.050	-.038	-.046	-.059	-.142	-.116
		.650	-.127	-.079	-.047	-.043	-.034	-.091	-.058
		.550	-.141	-.074	-.025	.001	.030	-.004	.033
		.450	-.160	-.079	-.008	.038	.076	.048	.106
		.350	-.676	-.050	.120	-.682	-.691	-.340	-.739
		.250	-.144	.006	.204	.194	.253	.252	.309
		.150	-.134	.067	.316	.292	.350	.318	.406
		.075	-.127	.160	.441	-1.141	.442	.433	.466
		.025	-.008	.309	-1.154	.468	.452	.445	.425
$\frac{y}{b/2}=.95$	Upper	.000	.397	-.597	-.659	-.994	-.824	-.462	-.519
		.025	-.120	-.694	-.632	-.720	-.756	-.453	-.504
		.075	-.166	-.500	-.603	-.752	-.793	-.460	-.511
		.150	-.115	-.293	-.463	-.620	-.674	-.378	-.434
		.250	-.123	-.225	-.441	-.619	-.633	-.412	-.462
		.350	-.136	-.212	-.452	-.677	-.679	-.460	-.519
		.450	-.124	-.181	-.434	-.672	-.670	-.457	-.506
		.550	-.108	-.153	-.456	-.653	-.637	-.452	-.502
		.650	-.031	-.096	-.373	-.565	-.583	-.452	-.458
		.750	-.034	-.072	-.410	-.543	-.508	-.420	-.450
		.850	.005	-.039	-.359	-.471	-.439	-.394	-.419
		.950	.038	-.008	-.292	-.382	-.358	-.351	-.371
	Lower	.950	.020	-.011	-.090	-.129	-.153	-.233	-.243
		.850	.020	.002	-.032	-.054	-.069	-.147	-.142
		.750	-.026	-.041	-.057	-.061	-.079	-.153	-.144
		.650	-.074	-.072	-.069	-.068	-.075	-.129	-.107
		.550	-.108	-.094	-.083	-.074	-.064	-.106	-.076
		.450	-.117	-.094	-.083	-.060	-.036	-.067	-.030
		.350	-.137	-.092	-.064	-.022	.008	.001	.033
		.250	-.149	-.081	-.035	.025	.074	.069	.112
		.150	-.139	.009	.067	.156	.216	.206	.259
		.075	-.139	.093	.196	.280	.337	.325	.359
		.025	-.109	.241	.343	.392	.406	.382	.358

NACA

TABLE II.- TABULATION OF PRESSURE COEFFICIENTS FOR THE

FENCE-ON CONDITION - Continued

(b) M = 0.70.

		$\frac{x}{c}$	$\alpha=0°$	$\alpha=42°$	$\alpha=84°$	$\alpha=126°$	$\alpha=167°$	$\alpha=208°$	$\alpha=248°$
$\frac{y}{b/2}=.20$	Upper	.000	.456	.189	-.515	-1.849	-1.663	-.892	-.867
		.025	-.088	-.600	-1.653	-2.144	-1.495	-.893	-.867
		.075	-.106	-.408	-1.651	-2.309	-1.476	-.898	-.872
		.150	-.118	-.342	-.351	-1.598	-1.407	-.898	-.880
		.250	-.160	-.334	-.498	-.784	-1.320	-.902	-.893
		.350	-.169	-.315	-.472	-.497	-1.168	-.901	-.899
		.450	-.188	-.287	-.414	-.393	-.960	-.849	-.858
		.550	-.180	-.295	-.398	-.428	-.847	-.885	-.908
		.650	-.180	-.263	-.348	-.411	-.735	-.866	-.905
		.750	-.134	-.198	-.233	-.326	-.612	-.819	-.883
		.850	-.097	-.130	-.164	-.252	-.538	-.811	-.879
		.950	-.047	-.050	-.069	-.156	-.484	-.740	-.856
	Lower	.950	-.046	-.015	-.005	-.018	-.072	-.147	-.130
		.850	-.100	-.046	-.008	.011	.008	.001	.040
		.750	-.145	-.071	-.008	.037	.066	.107	.157
		.650	-.185	-.088	-.008	.060	.111	.168	.233
		.550	-.219	-.099	-.003	.085	.158	.227	.299
		.450	-.190	-.057	.050	.153	.230	.315	.395
		.350	-.167	.015	.108	.218	.307	.403	.484
		.250	-.149	.023	.162	.288	.388	.480	.571
		.150	-.108	.093	.248	.392	.493	.586	.668
		.075	-.123	.164	.350	.493	.590	.670	.724
		.025	-.051	.307	.493	.582	.624	.659	.662
$\frac{y}{b/2}=.40$	Upper	.000	.462	-.084	-.824	-1.190	-1.047	-.890	-.948
		.025	-.146	-.852	-1.469	-1.080	-.983	-.887	-.938
		.075	-.157	-.557	-1.250	-1.055	-.959	-.892	-.941
		.150	-.164	-.436	-.897	-1.017	-.939	-.893	-.944
		.250	-.176	-.372	-.739	-.994	-.905	-.875	-.943
		.350	-.194	-.347	-.602	-.970	-.884	-.866	-.942
		.450	-.200	-.319	-.487	-.909	-.861	-.857	-.938
		.550	-.183	-.269	-.388	-.830	-.837	-.846	-.928
		.650	-.145	-.204	-.302	-.739	-.800	-.832	-.916
		.750	-.093	-.134	-.218	-.658	-.762	-.806	-.889
		.850	-.048	-.063	-.140	-.553	-.711	-.760	-.862
		.950	-.003	-.009	-.074	-.466	-.656	-.725	-.839
	Lower	.950	-.016	-.001	-.013	-.126	-.255	-.296	-.296
		.850	-.047	-.012	-.002	-.046	-.127	-.142	-.133
		.750	-.079	-.023	.004	.001	-.028	-.020	.007
		.650	-.150	-.069	-.014	.009	.013	.038	.084
		.550	-.182	-.075	-.007	.048	.078	.126	.182
		.450	-.185	-.059	.030	.106	.158	.209	.275
		.350	-.185	-.034	.078	.160	.224	.286	.359
		.250	-.172	.014	.141	.242	.312	.357	.449
		.150	-.150	.074	.224	.339	.409	.462	.535
		.075	-.137	.158	.331	.439	.498	.552	.584
		.025	-.101	.291	.447	.525	.527	.528	.502
$\frac{y}{b/2}=.60$	Upper	.000	.444	.000	-.302	-.614	-.840	-.719	-.783
		.025	-.133	-.818	-.680	-.655	-.702	-.704	-.756
		.075	-.146	-.659	-.655	-.650	-.702	-.706	-.813
		.150	-.172	-.523	-.612	-.643	-.687	-.735	-.901
		.250	-.185	-.415	-.497	-.614	-.671	-.699	-.807
		.350	-.174	-.326	-.445	-.586	-.658	-.684	-.778
		.450	-.171	-.271	-.413	-.543	-.621	-.673	-.766
		.550	-.144	-.213	-.380	-.504	-.588	-.671	-.756
		.650	-.112	-.171	-.339	-.451	-.544	-.666	-.735
		.750	-.070	-.100	-.292	-.395	-.498	-.652	-.711
		.850	-.015	-.040	-.249	-.346	-.455	-.629	-.692
		.950	.017	.011	-.107	-.302	-.417	-.602	-.675
	Lower	.950	.012	.016	-.063	-.168	-.216	-.285	-.305
		.850	-.086	-.004	-.040	-.115	-.146	-.173	-.170
		.750	-.072	-.041	-.030	-.072	-.083	-.089	-.065
		.650	-.120	-.051	-.009	-.047	-.042	-.041	.019
		.550	-.151	-.057	.012	.007	.027	.058	.101
		.450	-.171	-.055	.073	.053	.090	.129	.188
		.350	-.164	-.019	.129	.122	.165	.231	.281
		.250	-.158	.023	.198	.197	.251	.314	.370
		.150	-.148	.087	.343	.283	.345	.419	.461
		.075	-.115	.204	.422	.340	.375	.415	.435
		.025	-.084	.289	.415	.437	.413	.407	.413

NACA

TABLE II.- TABULATION OF PRESSURE COEFFICIENTS FOR THE FENCE-ON CONDITION - Continued

(b) M = 0.70. Concluded.

		$\frac{x}{c}$	$\alpha=0°$	$\alpha=4.2°$	$\alpha=8.4°$	$\alpha=12.6°$	$\alpha=16.7°$	$\alpha=20.8°$	$\alpha=24.8°$
$\frac{y}{b/2}=.80$	Upper	.000	.454	-.203	-.907	-1.429	-1.569	-.611	-.688
		.025	-.144	-.919	-1.098	-1.292	-1.426	-.604	-.687
		.075	-.162	-.637	-1.085	-1.358	-1.437	-.613	-.700
		.150	-.160	-.447	-1.073	-1.518	-1.450	-.610	-.698
		.250	-.170	-.358	-.985	-1.592	-1.584	-.591	-.678
		.350	-.177	-.304	-.796	-.982	-1.186	-.585	-.662
		.450	-.174	-.254	-.617	-.347	-.649	-.566	-.649
		.550	-.155	-.207	-.416	-.258	-.490	-.552	-.634
		.650	-.109	-.136	-.238	-.268	-.433	-.528	-.612
		.750	-.058	-.077	-.136	-.247	-.401	-.502	-.585
		.850	-.022	-.024	-.066	-.189	-.371	-.469	-.555
		.950	.033	.028	-.018	-.127	-.313	-.405	-.484
	Lower	.950	-.048	-.044	-.065	-.150	-.255	-.356	-.395
		.850	-.023	-.004	-.045	-.074	-.132	-.192	-.201
		.750	-.065	-.043	-.047	-.083	-.101	-.137	-.131
		.650	-.129	-.074	-.060	-.077	-.076	-.089	-.064
		.550	-.144	-.065	-.033	-.023	-.009	-.002	.031
		.450	-.166	-.072	-.028	.007	.045	.061	.105
		.350	-.166	-.032	.046	.081	.134	.166	.205
		.250	-.160	.018	.107	.169	.233	.266	.320
		.150	-.144	.082	.202	.265	.268	.365	.413
		.075	-.137	.162	.311	.373	.426	.451	.475
		.025	-.110	.312	.443	.457	.454	.472	.441
$\frac{y}{b/2}=.95$	Upper	.000	.431	-.387	-.584	-.922	-.644	-.446	-.511
		.025	-.108	-.687	-.614	-.702	-.576	-.431	-.499
		.075	-.161	-.530	-.582	-.723	-.631	-.446	-.504
		.150	-.117	-.299	-.451	-.605	-.561	-.365	-.417
		.250	-.117	-.215	-.427	-.572	-.561	-.384	-.443
		.350	-.133	-.203	-.448	-.614	-.601	-.436	-.509
		.450	-.120	-.168	-.419	-.606	-.571	-.442	-.500
		.550	-.096	-.131	-.339	-.605	-.536	-.433	-.496
		.650	-.049	-.085	-.369	-.542	-.501	-.441	-.462
		.750	-.016	-.055	-.327	-.524	-.441	-.404	-.455
		.850	.026	-.022	-.264	-.467	-.389	-.382	-.430
		.950	.062	.010	-.085	-.393	-.329	-.332	-.381
	Lower	.950	.042	.010	-.014	-.129	-.179	-.239	-.268
		.850	.050	.033	-.047	-.035	-.095	-.148	-.170
		.750	-.009	-.021	-.071	-.066	-.114	-.167	-.182
		.650	-.045	-.053	-.085	-.078	-.114	-.154	-.158
		.550	-.096	-.075	-.088	-.084	-.101	-.121	-.117
		.450	-.109	-.083	-.071	-.070	-.068	-.077	-.065
		.350	-.130	-.089	-.045	-.038	-.019	-.013	.008
		.250	-.147	-.073	.073	.017	.055	.070	.100
		.150	-.143	.004	.206	.154	.205	.223	.252
		.075	-.143	.110	.364	.285	.331	.341	.357
		.025	-.095	.261	.364	.404	.409	.393	.368

NACA

TABLE II.- TABULATION OF PRESSURE COEFFICIENTS FOR THE FENCE-ON CONDITION - Continued

(c) M = 0.85.

$\frac{y}{b/2}$		$\frac{x}{c}$	$\alpha=0°$	$\alpha=4.3°$	$\alpha=8.6°$	$\alpha=12.8°$	$\alpha=16.8°$
=.20	Upper	.000	.478	.258	-.120	-.573	-1.016
		.025	-.071	-.565	-1.282	-1.615	-1.391
		.075	-.096	-.405	-1.213	-1.612	-1.364
		.150	-.120	-.360	-.848	-1.613	-1.310
		.250	-.168	-.369	-.512	-.658	-1.238
		.350	-.185	-.373	-.538	-.440	-1.114
		.450	-.191	-.360	-.568	-.476	-.936
		.550	-.231	-.383	-.585	-.661	-.879
		.650	-.226	-.361	-.503	-.599	-.790
		.750	-.168	-.274	-.284	-.599	-.712
		.850	-.131	-.173	-.200	-.446	-.649
		.950	-.058	-.074	-.100	-.276	-.550
	Lower	.950	-.059	-.032	-.021	-.081	-.098
		.850	-.126	-.067	-.021	-.024	.012
		.750	-.189	-.089	-.020	.015	.075
		.650	-.228	-.103	-.012	.044	.122
		.550	-.263	-.114	-.007	.071	.164
		.450	-.219	-.064	.054	.146	.249
		.350	-.179	-.016	.114	.218	.328
		.250	-.149	.029	.171	.289	.408
		.150	-.105	.104	.260	.392	.514
		.075	-.099	.168	.352	.493	.615
		.025	-.037	.317	.503	.612	.677
=.40	Upper	.000	.482	.079	-.386	-.847	-1.079
		.025	-.152	-1.341	-1.596	-.914	-1.016
		.075	-.170	-.569	-1.428	-.917	-1.000
		.150	-.187	-.487	-.855	-.922	-.972
		.250	-.207	-.470	-.759	-.894	-.931
		.350	-.236	-.433	-.671	-.879	-.905
		.450	-.249	-.407	-.574	-.865	-.886
		.550	-.225	-.334	-.454	-.815	-.861
		.650	-.177	-.236	-.349	-.763	-.838
		.750	-.111	-.147	-.251	-.709	-.812
		.850	-.054	-.069	-.168	-.650	-.779
		.950	-.005	-.014	-.112	-.593	-.754
	Lower	.950	-.013	-.007	-.037	-.222	-.279
		.850	-.048	-.022	-.021	-.122	-.149
		.750	-.084	-.035	-.005	-.045	-.035
		.650	-.176	-.089	-.032	-.030	.008
		.550	-.224	-.099	-.021	.011	.077
		.450	-.226	-.080	.017	.072	.156
		.350	-.226	-.050	.068	.135	.228
		.250	-.197	.000	.130	.219	.316
		.150	-.167	.064	.211	.318	.415
		.075	-.147	.144	.310	.414	.506
		.025	-.104	.283	.433	.507	.547
=.60	Upper	.000	.446	.102	-.184	-.523	-.870
		.025	-.144	-1.096	-.740	-.719	-.746
		.075	-.160	-.774	-.712	-.704	-.747
		.150	-.205	-.618	-.666	-.694	-.760
		.250	-.224	-.553	-.622	-.672	-.754
		.350	-.208	-.362	-.591	-.654	-.717
		.450	-.200	-.305	-.539	-.622	-.700
		.550	-.166	-.243	-.478	-.586	-.682
		.650	-.134	-.186	-.425	-.543	-.661
		.750	-.082	-.104	-.368	-.497	-.637
		.850	-.023	-.036	-.314	-.449	-.457
		.950	.020	.012	-.278	-.410	-.598
	Lower	.950	.017	.017	-.149	-.256	-.316
		.850	-.026	-.007	-.104	-.202	-.216
		.750	-.082	-.040	-.070	-.140	-.142
		.650	-.133	-.062	-.053	-.093	-.073
		.550	-.159	-.074	-.030	-.047	-.005
		.450	-.204	-.070	-.003	.008	.063
		.350	-.193	-.026	.057	.092	.164
		.250	-.186	.020	.117	.167	.248
		.150	-.173	.075	.199	.263	.351
		.075	-.119	.178	.307	.373	.423
		.025	-.074	.289	.412	.454	.488

NACA

TABLE II.- TABULATION OF PRESSURE COEFFICIENTS FOR THE FENCE-ON CONDITION - Continued

(c) M = 0.85. Concluded.

		$\frac{x}{c}$	$\alpha=0°$	$\alpha=4.3°$	$\alpha=8.6°$	$\alpha=12.8°$	$\alpha=16.8°$
$\frac{y}{b/2}=.80$	Upper	.000	.459	-.056	-.596	-.977	-.703
		.025	-.156	-1.212	-1.309	-1.272	-.636
		.075	-.179	-.756	-1.263	-1.292	-.640
		.150	-.179	-.569	-1.216	-1.405	-.635
		.250	-.199	-.422	-1.102	-1.405	-.612
		.350	-.203	-.349	-.913	-1.271	-.606
		.450	-.197	-.283	-.711	-.577	-.576
		.550	-.174	-.214	-.478	-.418	-.557
		.650	-.116	-.137	-.296	-.403	-.541
		.750	-.063	-.070	-.194	-.369	-.527
		.850	-.013	-.014	-.112	-.319	-.507
		.950	.046	.041	-.035	-.242	-.436
	Lower	.950	-.031	-.036	-.092	-.256	-.441
		.850	-.006	-.008	-.047	-.164	-.280
		.750	-.065	-.052	-.081	-.172	-.233
		.650	-.136	-.091	-.097	-.161	-.182
		.550	-.161	-.089	-.069	-.102	-.090
		.450	-.193	-.092	-.045	-.056	-.022
		.350	-.192	-.045	.030	.037	.085
		.250	-.185	.018	.105	.127	.189
		.150	-.167	.080	.191	.224	.293
		.075	-.155	.169	.290	.327	.391
		.025	-.127	.308	.416	.427	.459
$\frac{y}{b/2}=.95$	Upper	.000	.445	-.312	-.455	-.701	-.567
		.025	-.119	-.871	-.547	-.578	-.475
		.075	-.191	-.674	-.571	-.642	-.490
		.150	-.145	-.372	-.468	-.580	-.392
		.250	-.138	-.223	-.470	-.606	-.406
		.350	-.151	-.213	-.519	-.676	-.475
		.450	-.127	-.172	-.461	-.624	-.473
		.550	-.099	-.139	-.425	-.596	-.457
		.650	-.059	-.092	-.336	-.541	-.433
		.750	-.008	-.056	-.385	-.518	-.423
		.850	-.035	-.024	-.360	-.467	-.400
		.950	.080	.011	-.305	-.396	-.339
	Lower	.950	.060	.022	-.110	-.184	-.298
		.850	.074	.045	-.012	-.085	-.203
		.750	.003	-.021	-.064	-.140	-.250
		.650	-.047	-.058	-.096	-.170	-.250
		.550	-.090	-.090	-.125	-.189	-.232
		.450	-.114	-.102	-.135	-.174	-.186
		.350	-.140	-.111	-.121	-.136	-.119
		.250	-.167	-.115	-.093	-.070	-.020
		.150	-.178	-.020	.054	.100	.158
		.075	-.164	.109	.204	.244	.288
		.025	-.094	.270	.358	.369	.383

NACA

TABLE II.- TABULATION OF PRESSURE COEFFICIENTS FOR THE FENCE-ON CONDITION - Continued

(d) M = 0.91.

		$\frac{x}{c}$	$\alpha=0°$	$\alpha=4.4°$	$\alpha=8.6°$	$\alpha=12.8°$
$\frac{y}{b/2}=.20$	Upper	.000	.490	.320	.015	-.387
		.025	-.058	-.564	-1.163	-1.401
		.075	-.082	-.400	-1.019	-1.401
		.150	-.108	-.323	-.729	-1.342
		.250	-.170	-.356	-.481	-.621
		.350	-.194	-.373	-.495	-.605
		.450	-.208	-.389	-.524	-.630
		.550	-.255	-.406	-.567	-.669
		.650	-.281	-.447	-.597	-.615
		.750	-.256	-.439	-.581	-.531
		.850	-.163	-.407	-.582	-.494
		.950	-.062	-.184	-.290	-.346
	Lower	.950	-.065	-.088	-.095	-.098
		.850	-.152	-.110	-.060	-.026
		.750	-.246	-.129	-.045	.015
		.650	-.269	-.137	-.030	.047
		.550	-.328	-.141	-.016	.077
		.450	-.217	-.077	.052	.156
		.350	-.173	-.023	.113	.228
		.250	-.139	.022	.177	.299
		.150	-.087	.103	.270	.403
		.075	-.075	.166	.363	.507
		.025	-.016	.316	.512	.625
$\frac{y}{b/2}=.40$	Upper	.000	.485	.159	-.217	-.694
		.025	-.166	-1.183	-1.397	-1.240
		.075	-.187	-.597	-1.395	-1.194
		.150	-.203	-.498	-1.202	-1.136
		.250	-.231	-.476	-.726	-1.075
		.350	-.276	-.494	-.672	-.981
		.450	-.309	-.528	-.693	-.928
		.550	-.281	-.554	-.700	-.898
		.650	-.197	-.509	-.510	-.855
		.750	-.117	-.236	-.357	-.799
		.850	-.056	-.074	-.264	-.726
		.950	-.011	-.003	-.173	-.633
	Lower	.950	-.015	-.020	-.084	-.208
		.850	-.051	-.041	-.062	-.116
		.750	-.091	-.059	-.044	-.044
		.650	-.187	-.125	-.071	-.038
		.550	-.272	-.140	-.058	.011
		.450	-.272	-.115	-.009	.079
		.350	-.264	-.079	.043	.138
		.250	-.218	-.025	.113	.221
		.150	-.173	.047	.203	.311
		.075	-.141	.133	.300	.416
		.025	-.085	.267	.427	.511
$\frac{y}{b/2}=.60$	Upper	.000	.436	.168	-.115	-1.022
		.025	-.173	-1.251	-.953	-1.438
		.075	-.176	-1.079	-.951	-1.438
		.150	-.266	-.564	-.904	-1.438
		.250	-.356	-.746	-.861	-1.420
		.350	-.229	-.574	-.815	-1.408
		.450	-.216	-.598	-.684	-1.380
		.550	-.164	-.882	-.574	-1.366
		.650	-.148	-.153	-.485	-1.329
		.750	-.084	-.071	-.407	-1.281
		.850	-.027	-.011	-.326	-.858
		.950	.022	.031	-.280	-1.143
	Lower	.950	.019	.012	-.162	-.852
		.850	-.029	-.016	-.117	-.730
		.750	-.087	-.054	-.092	-.634
		.650	-.142	-.088	-.073	-.561
		.550	-.174	-.099	-.054	-.487
		.450	-.221	-.102	-.030	-.410
		.350	-.208	-.057	.035	-.296
		.250	-.202	-.017	.093	-.195
		.150	-.201	.041	.173	-.064
		.075	-.139	.138	.289	.073
		.025	-.067	.252	.389	.192

NACA

TABLE II.- TABULATION OF PRESSURE COEFFICIENTS FOR THE FENCE-ON CONDITION - Continued

(d) M = 0.91. Concluded.

		$\frac{x}{c}$	$\alpha=0°$	$\alpha=4.4°$	$\alpha=8.6°$	$\alpha=12.8°$
$\frac{y}{b/2}=.80$	Upper	.000	.460	.033	-.390	-.780
		.025	-.178	-1.302	-1.113	-1.144
		.075	-.202	-1.402	-1.099	-1.112
		.150	-.193	-1.031	-1.016	-1.073
		.250	-.219	-.640	-.921	-.975
		.350	-.224	-.462	-.834	-.878
		.450	-.216	-.195	-.738	-.818
		.550	-.191	-.144	-.632	-.756
		.650	-.116	-.090	-.527	-.688
		.750	-.054	-.038	-.420	-.625
		.850	.001	.009	-.299	-.553
		.950	.063	.062	-.192	-.458
	Lower	.950	-.020	-.028	-.178	-.436
		.850	.002	-.009	-.109	-.324
		.750	-.063	-.066	-.138	-.296
		.650	-.136	-.120	-.149	-.238
		.550	-.179	-.120	-.106	-.151
		.450	-.207	-.117	-.078	-.089
		.350	-.198	-.066	-.003	.012
		.250	-.194	-.005	.072	.109
		.150	-.175	.054	.156	.211
		.075	-.161	.139	.255	.312
		.025	-.122	.276	.381	.413
$\frac{y}{b/2}=.95$	Upper	.000	.441	-.251	-.586	-1.400
		.025	-.145	-1.228	-.813	-1.285
		.075	-.229	-1.144	-.804	-1.329
		.150	-.212	-.595	-.643	-1.182
		.250	-.146	-.358	-.554	-1.182
		.350	-.165	-.166	-.588	-1.308
		.450	-.142	-.112	-.513	-1.279
		.550	-.096	-.112	-.468	-1.246
		.650	-.123	-.089	-.429	-1.174
		.750	.002	-.059	-.398	-1.101
		.850	.040	-.032	-.364	-1.023
		.950	.092	.005	-.296	-.904
	Lower	.950	.070	.038	-.129	-.801
		.850	.092	.062	-.027	-.690
		.750	.013	-.013	-.083	-.820
		.650	-.041	-.055	-.117	-.866
		.550	-.091	-.089	-.158	-.891
		.450	-.117	-.111	-.186	-.815
		.350	-.151	-.151	-.198	-.726
		.250	-.178	-.209	-.145	-.582
		.150	-.226	-.042	.043	-.300
		.075	-.166	.101	.195	-.110
		.025	-.077	.262	.341	-.069

NACA

TABLE II.- TABULATION OF PRESSURE COEFFICIENTS FOR THE FENCE-ON CONDITION - Continued.

(e) M = 0.93 and 0.95.

M=.93

$\frac{y}{b/2}$		$\frac{x}{c}$	α=0°	α=4.4°	α=8.6°
.20	Upper	.000	.496	.343	.058
		.025	-.048	-.547	-1.123
		.075	-.073	-.392	-.952
		.150	-.167	-.303	-.678
		.250	-.197	-.343	-.486
		.350	-.218	-.349	-.488
		.450	-.252	-.377	-.508
		.550	-.291	-.407	-.553
		.650	-.269	-.440	-.584
		.750	-.247	-.433	-.571
		.850	-.119	-.426	-.594
		.950	-.120	-.273	-.435
	Lower	.950	-.119	-.136	-.162
		.850	-.246	-.148	-.114
		.750	-.313	-.163	-.089
		.650	-.301	-.160	-.066
		.550	-.334	-.160	-.046
		.450	-.225	-.081	.028
		.350	-.182	-.027	.097
		.250	-.149	.022	.162
		.150	-.091	.102	.259
		.075	-.085	.167	.352
		.025	-.023	.312	.501
.40	Upper	.000	.487	.186	-.152
		.025	-.152	-1.125	-1.327
		.075	-.179	-.568	-1.284
		.150	-.204	-.487	-1.207
		.250	-.240	-.458	-.700
		.350	-.290	-.481	-.613
		.450	-.331	-.517	-.653
		.550	-.360	-.547	-.683
		.650	-.304	-.531	-.689
		.750	-.136	-.447	-.615
		.850	-.051	-.179	-.434
		.950	-.003	-.060	-.252
	Lower	.950	-.017	-.061	-.186
		.850	-.048	-.082	-.157
		.750	-.106	-.098	-.119
		.650	-.301	-.171	-.143
		.550	-.352	-.183	-.112
		.450	-.337	-.144	-.051
		.350	-.283	-.102	.005
		.250	-.229	-.036	.080
		.150	-.191	.039	.165
		.075	-.160	.119	.272
		.025	-.110	.254	.409
.60	Upper	.000	.423	.193	-.070
		.025	-.170	-1.179	-1.314
		.075	-.171	-1.013	-1.205
		.150	-.306	-.538	-1.017
		.250	-.414	-.706	-1.025
		.350	-.311	-.570	-.946
		.450	-.244	-.597	-.830
		.550	-.153	-.624	-.719
		.650	-.156	-.329	-.603
		.750	-.087	-.111	-.502
		.850	-.097	-.019	-.217
		.950	.021	.030	-.365
	Lower	.950	.006	.003	-.283
		.850	-.037	-.034	-.178
		.750	-.099	-.075	-.154
		.650	-.151	-.112	-.144
		.550	-.198	-.132	-.128
		.450	-.261	-.136	-.098
		.350	-.279	-.090	-.028
		.250	-.267	-.044	.040
		.150	-.285	.011	.121
		.075	-.207	.110	.247
		.025	-.139	.227	.352

M=.95

$\frac{y}{b/2}$		$\frac{x}{c}$	α=0°	α=4.4°	α=6.5°
.20	Upper	.000	.504		.236
		.025	-.039		-.798
		.075	-.063		-.550
		.150	-.094		-.425
		.250	-.156		-.406
		.350	-.196		-.405
		.450	-.220		-.427
		.550	-.248		-.465
		.650	-.287		-.498
		.750	-.301		-.481
		.850	-.292		-.500
		.950	-.230		-.378
	Lower	.950	-.224	-.197	-.173
		.850	-.303	-.195	-.142
		.750	-.303	-.200	-.137
		.650	-.283	-.178	-.116
		.550	-.306	-.193	-.104
		.450	-.210	-.085	-.020
		.350	-.164	-.035	.041
		.250	-.130	.016	.102
		.150	-.071	.102	.189
		.075	-.070	.166	.272
		.025	-.003	.309	.419
.40	Upper	.000	.486	.224	.063
		.025	-.146	-1.058	-1.276
		.075	-.173	-.530	-1.091
		.150	-.196	-.456	-.617
		.250	-.231	-.438	-.494
		.350	-.288	-.466	-.532
		.450	-.327	-.502	-.562
		.550	-.360	-.534	-.597
		.650	-.381	-.527	-.595
		.750	-.308	-.522	-.563
		.850	-.158	-.321	-.396
		.950	-.043	-.177	-.208
	Lower	.950	-.061	-.155	-.173
		.850	-.134	-.169	-.163
		.750	-.282	-.182	-.133
		.650	-.393	-.255	-.181
		.550	-.366	-.236	-.165
		.450	-.330	-.173	-.103
		.350	-.272	-.123	-.051
		.250	-.219	-.056	.021
		.150	-.178	.020	.110
		.075	-.152	.110	.204
		.025	-.095	.238	.343
.60	Upper	.000	.436	.219	.092
		.025	-.168	-1.091	-1.288
		.075	-.157	-.938	-1.143
		.150	-.304	-.501	-1.019
		.250	-.399	-.655	-.853
		.350	-.366	-.568	-.644
		.450	-.392	-.589	-.659
		.550	-.337	-.631	-.679
		.650	-.250	-.623	-.652
		.750	-.088	-.330	-.402
		.850	.169	-.162	-.181
		.950	.031	-.098	-.201
	Lower	.950	.020	-.103	-.152
		.850	-.031	-.129	-.159
		.750	-.104	-.176	-.162
		.650	-.240	-.218	-.193
		.550	-.356	-.221	-.186
		.450	-.389	-.153	-.161
		.350	-.326	-.097	-.084
		.250	-.285	-.035	-.028
		.150	-.328	.102	.051
		.075	-.238	.199	.172
		.025	-.137	.199	.285

NACA

TABLE II.- TABULATION OF PRESSURE COEFFICIENTS FOR THE

FENCE-ON CONDITION - Concluded

(e) M = 0.93 and 0.95. Concluded.

$\frac{y}{b/2}$		$\frac{x}{c}$	$\alpha=0°$	$\alpha=4.4°$	$\alpha=8.6°$	$\frac{x}{c}$	$\alpha=0°$	$\alpha=4.4°$	$\alpha=6.5°$
.80	Upper	.000	.446	.084	-.261	.000	.407	.140	-.005
		.025	-.199	-1.326	-1.328	.025	-.249	-1.095	-1.246
		.075	-.221	-1.326	-1.320	.075	-.288	-.985	-1.249
		.150	-.209	-.969	-1.182	.150	-.298	-.863	-1.074
		.250	-.236	-.623	-1.112	.250	-.324	-.581	-1.021
		.350	-.231	-.664	-1.025	.350	-.320	-.642	-.922
		.450	-.231	-.567	-.916	.450	-.296	-.673	-.826
		.550	-.224	-.193	-.769	.550	-.246	-.713	-.751
		.650	-.129	-.058	-.617	.650	-.139	-.477	-.682
		.750	-.054	-.003	-.596	.750	-.045	-.203	-.387
		.850	-.001	.034	-.520	.850	.016	-.070	-.225
		.950	.060	.074	-.351	.950	.078	.012	-.100
	Lower	.950	-.021	-.024	-.279	.950	.001	-.133	-.262
		.850	-.002	-.011	-.213	.850	.016	-.164	-.250
		.750	-.068	-.093	-.240	.750	-.058	-.243	-.277
		.650	-.157	-.157	-.225	.650	-.160	-.255	-.257
		.550	-.220	-.148	-.171	.550	-.229	-.223	-.212
		.450	-.236	-.144	-.144	.450	-.292	-.227	-.196
		.350	-.224	-.092	-.069	.350	-.305	-.171	-.123
		.250	-.215	-.034	.008	.250	-.304	-.114	-.049
		.150	-.207	.025	.087	.150	-.288	-.057	.024
		.075	-.202	.105	.191	.075	-.259	.022	.114
		.025	-.172	.241	.326	.025	-.218	.160	.255
.95	Upper	.000	.462	-.195	-.501	.000	.434	-.104	-.258
		.025	-.134	-1.165	-1.100	.025	-.178	-1.053	-1.164
		.075	-.226	-1.092	-1.083	.075	-.265	-.977	-1.111
		.150	-.238	-.850	-.827	.150	-.254	-.783	-.889
		.250	-.179	-.507	-.715	.250	-.245	-.716	-.831
		.350	-.158	-.255	-.820	.350	-.195	-.549	-.962
		.450	-.142	-.047	-.744	.450	-.122	-.435	-.898
		.550	-.097	-.056	-.718	.550	-.074	-.271	-.766
		.650	-.138	-.048	-.619	.650	-.013	-.077	-.360
		.750	.004	-.041	-.547	.750	.025	-.036	-.444
		.850	.050	-.011	-.470	.850	.071	.005	-.217
		.950	.099	.019	-.379	.950	.115	.048	-.083
	Lower	.950	.077	.050	-.192	.950	.091	.056	-.080
		.850	.095	.073	-.109	.850	.108	.065	-.094
		.750	.013	-.003	-.216	.750	.031	-.053	-.274
		.650	-.043	-.043	-.307	.650	-.021	-.189	-.380
		.550	-.097	-.075	-.367	.550	-.074	-.341	-.414
		.450	-.118	-.144	-.348	.450	-.097	-.374	-.379
		.350	-.159	-.280	-.300	.350	-.209	-.359	-.340
		.250	-.236	-.252	-.214	.250	-.312	-.300	-.253
		.150	-.254	-.055	.004	.150	-.265	-.088	-.030
		.075	-.182	.084	.152	.075	-.221	.038	.106
		.025	-.121	.243	.304	.025	-.160	.185	.257

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TABLE III.- TABULATION OF PRESSURE COEFFICIENTS FOR THE LEADING-EDGE-NOTCH CONDITION

(a) M = 0.70.

			α=0°	α=42°	α=84°	α=126°	α=167°	α=208°	α=248°
$\frac{y}{b/2}$=.20	Upper	.000	.454	.107	-.602	-1.324	-1.757	-.893	-.822
		.025	-.109	-.640	-1.725	-2.245	-1.605	-.901	-.825
		.075	-.123	-.444	-1.800	-2.295	-1.584	-.904	-.829
		.150	-.134	-.368	-.381	-1.708	-1.516	-.904	-.834
		.250	-.173	-.358	-.504	-.750	-1.425	-.907	-.844
		.350	-.188	-.346	-.804	-.606	-1.251	-.908	-.852
		.450	-.202	-.333	-.455	-.458	-1.041	-.897	-.852
		.550	-.206	-.320	-.417	-.448	-.858	-.895	-.858
		.650	-.195	-.287	-.357	-.428	-.734	-.876	-.855
		.750	-.148	-.218	-.263	-.345	-.612	-.842	-.839
		.850	-.108	-.152	-.173	-.260	-.518	-.801	-.825
		.950	-.048	-.075	-.079	-.166	-.407	-.756	-.812
	Lower	.950	-.048	-.023	.000	-.020	-.059	-.146	-.107
		.850	-.092	-.047	.008	.022	.033	.021	.074
		.750	-.140	-.077	.001	.044	.091	.113	.180
		.650	-.095	-.093	.005	.057	.133	.188	.247
		.550	-.218	-.066	.010	.090	.170	.240	.326
		.450	-.183	-.017	.061	.159	.251	.324	.414
		.350	-.159	.020	.117	.228	.324	.406	.501
		.250	-.139	.088	.177	.297	.403	.488	.584
		.150	-.104	.159	.266	.403	.508	.596	.684
		.075	-.097	.311	.360	.502	.605	.657	.753
		.025	-.031	.015	.502	.585	.689	.666	.674
$\frac{y}{b/2}$=.40	Upper	.000	.460	-.057	-.842	-1.253	-1.071	-.904	-.938
		.025	-.165	-.927	-1.404	-1.141	-1.036	-.908	-.951
		.075	-.171	-.582	-1.231	-1.113	-1.021	-.900	-.949
		.150	-.175	-.457	-.986	-1.062	-.997	-.889	-.944
		.250	-.182	-.391	-.821	-1.027	-.963	-.882	-.941
		.350	-.200	-.363	-.666	-.990	-.945	-.872	-.937
		.450	-.206	-.330	-.526	-.930	-.929	-.862	-.937
		.550	-.190	-.282	-.404	-.845	-.895	-.853	-.923
		.650	-.148	-.213	-.305	-.758	-.856	-.841	-.909
		.750	-.095	-.142	-.217	-.670	-.811	-.811	-.878
		.850	-.046	-.074	-.138	-.572	-.753	-.764	-.849
		.950	-.001	-.018	-.073	-.480	-.703	-.719	-.824
	Lower	.950	-.019	-.011	-.007	-.130	-.259	-.289	-.276
		.850	-.043	-.018	.006	-.045	-.116	-.134	-.102
		.750	-.074	-.031	.013	.006	-.014	-.011	.037
		.650	-.141	-.070	-.001	.018	.035	.054	.116
		.550	.099	.090	.015	.039	.099	.129	.186
		.450	-.188	-.067	.041	.106	.167	.215	.292
		.350	-.184	-.042	.087	.167	.242	.299	.377
		.250	-.165	.008	.153	.247	.328	.388	.461
		.150	-.146	.071	.235	.344	.425	.483	.547
		.075	-.130	.155	.336	.442	.513	.555	.592
		.025	-.083	.295	.455	.517	.532	.524	.507
$\frac{y}{b/2}$=.60	Upper	.000	.435	-.798	-.397	-.666	-.829	-.806	-.892
		.025	-.169	-.653	-.753	-.678	-.744	-.787	-.868
		.075	-.171	-.549	-.724	-.672	-.785	-.858	-1.009
		.150	-.171	-.436	-.680	-.667	-.783	-.829	-.903
		.250	-.191	-.363	-.613	-.645	-.696	-.675	-.710
		.350	-.183	-.304	-.566	-.619	-.684	-.669	-.699
		.450	-.180	-.242	-.516	-.588	-.667	-.669	-.708
		.550	-.154	-.174	-.463	-.559	-.653	-.673	-.706
		.650	-.117	-.130	-.421	-.525	-.640	-.675	-.694
		.750	-.078	-.062	-.371	-.490	-.611	-.665	-.680
		.850	-.031	-.010	-.311	-.446	-.579	-.638	-.658
		.950	.014	.002	-.267	-.398	-.544	-.607	-.632
	Lower	.950	.014	-.084	-.100	-.192	-.266	-.289	-.270
		.850	-.017	-.047	-.057	-.129	-.171	-.174	-.140
		.750	-.046	-.065	-.033	-.087	-.100	-.083	-.047
		.650	-.116	-.073	-.019	-.039	-.039	-.009	.042
		.550	-.157	-.066	-.001	.010	.026	.063	.126
		.450	-.179	-.019	.023	.049	.096	.139	.205
		.350	-.162	.020	.086	.139	.192	.244	.307
		.250	-.151	.109	.148	.209	.288	.335	.396
		.150	-.185	.096	.236	.296	.368	.409	.471
		.075	-.115	.137	.277	.351	.414	.449	.472
		.025	-.083	.255	.405	.417	.401	.417	.472

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TABLE III.- TABULATION OF PRESSURE COEFFICIENTS FOR THE LEADING-EDGE-NOTCH CONDITION - Continued

(a) M = 0.70. Concluded.

$\frac{y}{b/2}$		$\frac{x}{c}$	$\alpha=0°$	$\alpha=4.2°$	$\alpha=8.4°$	$\alpha=12.6°$	$\alpha=16.7°$	$\alpha=20.8°$	$\alpha=24.8°$
.80	Upper	.000	.459	-.249	-.919	-1.393	-.561	-.600	-.655
		.025	-.164	-.967	-1.087	-1.228	-.483	-.607	-.649
		.075	-.170	-.679	-1.127	-1.261	-.478	-.591	-.656
		.150	-.158	-.468	-1.148	-1.375	-.469	-.583	-.647
		.250	-.174	-.370	-1.090	-1.577	-.459	-.571	-.638
		.350	-.174	-.308	-.816	-.676	-.447	-.563	-.622
		.450	-.173	-.260	-.443	-.292	-.440	-.554	-.610
		.550	-.152	-.205	-.242	-.212	-.434	-.540	-.595
		.650	-.103	-.136	-.134	-.215	-.425	-.518	-.574
		.750	-.054	-.078	-.079	-.236	-.418	-.492	-.552
		.850	-.009	-.024	-.044	-.248	-.402	-.461	-.519
		.950	.030	.017	-.019	-.221	-.361	-.410	-.461
	Lower	.950	.006	-.006	-.057	-.147	-.283	-.297	-.304
		.850	-.008	-.005	-.037	-.086	-.175	-.174	-.162
		.750	-.057	-.038	-.051	-.082	-.136	-.124	-.093
		.650	-.100	-.062	-.046	-.056	-.086	-.058	-.016
		.550	-.136	-.069	-.031	-.025	-.030	.011	.063
		.450	-.167	-.073	-.008	.013	.030	.081	.137
		.350	-.159	-.030	.058	.082	.116	.161	.243
		.250	-.140	.019	.130	.181	.223	.277	.339
		.150	-.072	.063	.176	.251	.302	.329	.337
		.075	.369	.295	.306	.328	.354	.369	.372
		.025	-.072	.326	.450	.460	.489	.480	.459
.95	Upper	.000	.385	-.447	-.636	-.883	-.461	-.454	-.492
		.025	-.145	-.690	-.681	-.691	-.396	-.438	-.479
		.075	-.186	-.543	-.631	-.681	-.405	-.444	-.485
		.150	-.146	-.348	-.510	-.610	-.350	-.385	-.421
		.250	-.129	-.251	-.462	-.588	-.350	-.388	-.425
		.350	-.147	-.244	-.539	-.652	-.398	-.447	-.490
		.450	-.130	-.199	-.546	-.658	-.394	-.447	-.486
		.550	-.109	-.161	-.524	-.613	-.387	-.442	-.486
		.650	.017	-.099	-.474	-.543	-.385	-.423	-.444
		.750	-.036	-.084	-.408	-.455	-.361	-.418	-.450
		.850	.012	-.052	-.342	-.381	-.340	-.396	-.419
		.950	.022	-.031	-.202	-.255	-.282	-.320	-.337
	Lower	.950	.032	-.010	-.063	-.126	-.225	-.254	-.261
		.850	.021	-.007	-.025	-.077	-.167	-.183	-.186
		.750	-.022	-.043	-.050	-.093	-.167	-.174	-.169
		.650	-.066	-.076	-.073	-.105	-.157	-.157	-.142
		.550	-.105	-.102	-.090	-.111	-.143	-.131	-.109
		.450	-.124	-.110	-.089	-.093	-.104	-.084	-.053
		.350	-.140	-.109	-.071	-.054	-.048	-.017	.021
		.250	-.161	-.100	-.039	.002	.023	.064	.108
		.150	-.149	-.019	.074	.139	.169	.216	.260
		.075	-.145	.085	.209	.277	.293	.332	.359
		.025	-.088	.246	.368	.395	.390	.390	.373

TABLE III.- TABULATION OF PRESSURE COEFFICIENTS FOR THE LEADING-EDGE-NOTCH CONDITION - Continued

(b) M = 0.91.

		$\frac{x}{c}$	$\alpha=0°$	$\alpha=4.4°$	$\alpha=8.6°$	$\alpha=12.8°$
$\frac{y}{b/2}=.20$	Upper	.000	.489	.308	-.044	-.450
		.025	-.069	-.703	-1.263	-1.418
		.075	-.094	-.401	-1.038	-1.340
		.150	-.113	-.341	-.739	-1.285
		.250	-.172	-.353	-.513	-.710
		.350	-.207	-.393	-.550	-.615
		.450	-.238	-.417	-.574	-.589
		.550	-.258	-.438	-.607	-.514
		.650	-.289	-.472	-.622	-.567
		.750	-.236	-.464	-.619	-.610
		.850	-.163	-.412	-.600	-.547
		.950	-.071	-.183	-.410	-.332
	Lower	.950	-.067	-.081	-.129	-.070
		.850	-.138	-.093	-.074	.010
		.750	-.232	-.118	-.060	.047
		.650	-.263	-.124	-.050	.085
		.550	-.307	-.133	-.029	.110
		.450	-.205	-.068	.044	.184
		.350	-.164	-.015	.108	.258
		.250	-.134	.033	.173	.328
		.150	-.083	.115	.266	.432
		.075	-.009	.182	.363	.535
		.025	-.005	.330	.515	.649
$\frac{y}{b/2}=.40$	Upper	.000	.499	.166	-.198	-.724
		.025	-.163	-1.181	-1.415	-1.149
		.075	-.181	-.638	-1.316	-1.088
		.150	-.208	-.515	-1.216	-1.019
		.250	-.230	-.496	-.758	-.939
		.350	-.272	-.513	-.700	-.888
		.450	-.301	-.540	-.714	-.866
		.550	-.257	-.573	-.730	-.830
		.650	-.188	-.465	-.707	-.789
		.750	-.109	-.231	-.446	-.737
		.850	-.048	-.084	-.230	-.688
		.950	.008	-.014	-.119	-.613
	Lower	.950	-.001	-.024	-.077	-.193
		.850	-.039	-.036	-.070	-.093
		.750	-.079	-.053	-.058	-.015
		.650	-.179	-.112	-.085	.000
		.550	-.213	.118	-.058	.050
		.450	-.259	-.110	-.021	.107
		.350	-.252	-.077	.031	.173
		.250	-.210	-.019	.103	.254
		.150	-.166	.054	.192	.347
		.075	-.132	.139	.293	.443
		.025	-.078	.277	.421	.535
$\frac{y}{b/2}=.60$	Upper	.000	.426	.167	-.143	-.481
		.025	-.208	-1.235	-1.158	-.741
		.075	-.223	-1.113	-1.122	-.745
		.150	-.193	-.483	-.968	-.738
		.250	-.254	-.654	-.930	-.716
		.350	-.228	-.616	-.805	-.691
		.450	-.217	-.617	-.675	-.662
		.550	-.179	-.285	-.569	-.641
		.650	-.132	-.138	-.487	-.608
		.750	-.086	-.072	-.414	-.574
		.850	-.034	-.020	-.346	-.541
		.950	.022	.016	-.286	-.491
	Lower	.950	.021	.001	-.174	-.290
		.850	-.025	-.026	-.126	-.194
		.750	-.080	-.059	-.104	-.137
		.650	-.134	-.083	-.085	-.087
		.550	-.182	-.098	-.066	-.033
		.450	-.212	-.095	-.044	.024
		.350	-.197	-.048	.025	.117
		.250	-.186	.004	.097	.204
		.150	-.161	.086	.177	.287
		.075	-.154	.098	.221	.346
		.025	-.059	.235	.362	.446

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TABLE III.- TABULATION OF PRESSURE COEFFICIENTS FOR THE LEADING-EDGE-NOTCH CONDITION - Concluded

(b) M = 0.91. Concluded.

$\frac{y}{b/2}$		$\frac{x}{c}$	$\alpha=0°$	$\alpha=4.4°$	$\alpha=8.6°$	$\alpha=12.8°$
.80	Upper	.000	.468	-.003	-.405	-.589
		.025	-.187	-1.269	-.969	-.571
		.075	-.197	-1.149	-.969	-.554
		.150	-.189	-1.050	-.920	-.536
		.250	-.209	-.675	-.832	-.517
		.350	-.209	-.415	-.756	-.502
		.450	-.209	-.171	-.694	-.484
		.550	-.180	-.125	-.625	-.478
		.650	-.108	-.077	-.558	-.467
		.750	-.049	-.029	-.502	-.456
		.850	.008	.017	-.430	-.434
		.950	.059	.055	-.346	-.375
	Lower	.950	.038	.002	-.231	-.343
		.850	.017	-.009	-.154	-.248
		.750	-.042	-.066	-.150	-.212
		.650	-.100	-.100	-.135	-.161
		.550	-.159	-.111	-.104	-.097
		.450	-.204	-.106	-.072	-.038
		.350	-.191	-.058	-.002	.058
		.250	-.179	.004	.072	.159
		.150	-.135	.067	.164	.257
		.075	-.352	.274	.278	.283
		.025	-.095	.294	.382	.463
.95	Upper	.000	.417	-.263	-.650	-.744
		.025	-.141	-1.209	-.872	-.635
		.075	-.200	-1.101	-.806	-.655
		.150	-.192	-.623	-.672	-.544
		.250	-.124	-.382	-.560	-.517
		.350	-.151	-.205	-.588	-.621
		.450	-.115	-.132	-.524	-.611
		.550	-.100	-.126	-.501	-.594
		.650	-.042	-.107	-.452	-.530
		.750	.007	-.076	-.420	-.469
		.850	.058	-.049	-.371	-.405
		.950	.067	-.004	-.255	-.280
	Lower	.950	.077	.022	-.151	-.201
		.850	.065	.013	-.094	-.150
		.750	.018	-.024	-.115	-.185
		.650	-.035	-.066	-.146	-.219
		.550	-.084	-.106	-.186	-.243
		.450	-.113	-.123	-.203	-.217
		.350	-.140	-.162	-.210	-.161
		.250	-.166	-.212	-.171	-.080
		.150	-.213	-.046	.021	.116
		.075	-.155	.097	.172	.256
		.025	-.060	.262	.326	.382

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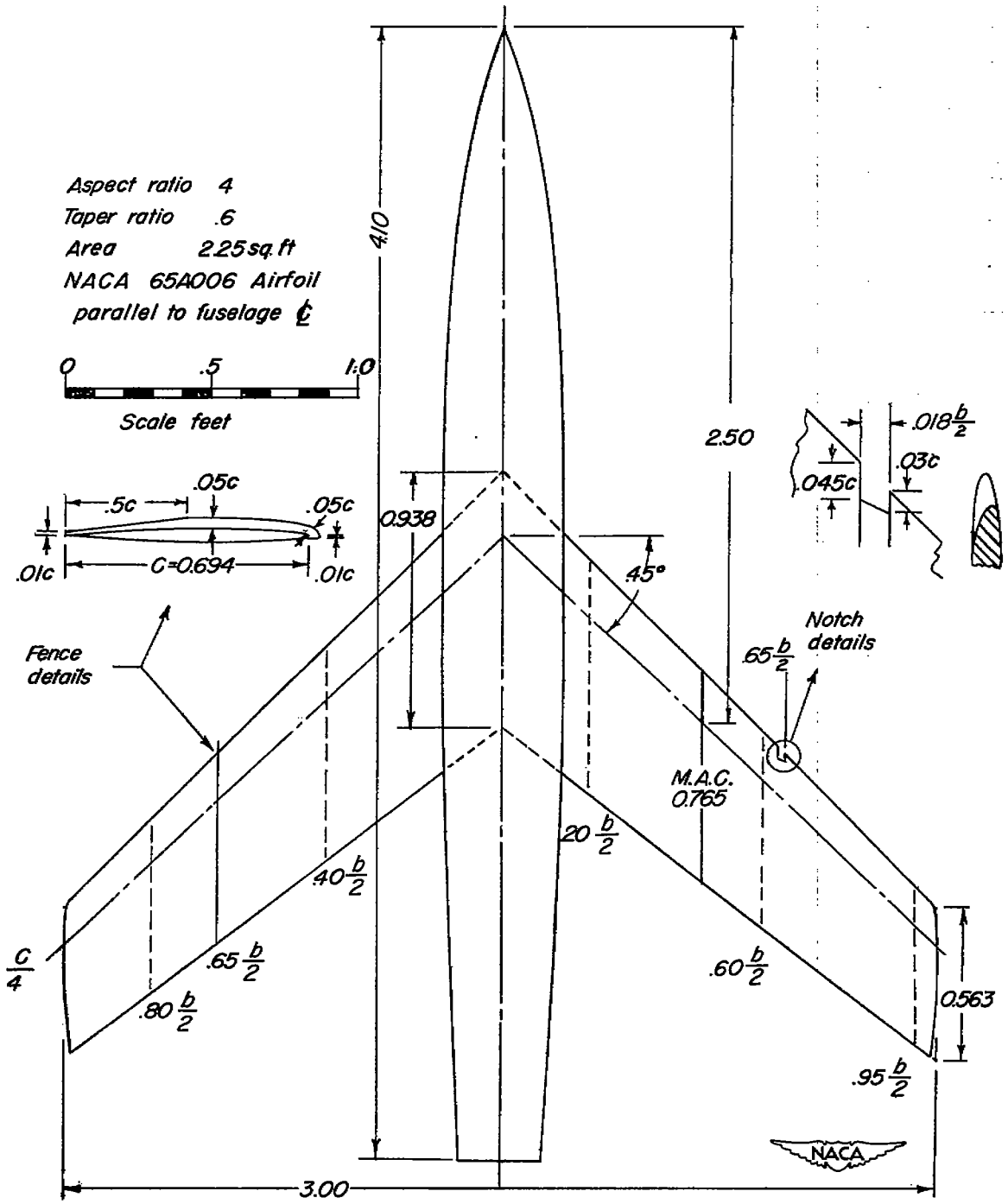

Figure 1.- Drawing of the model tested. All dimensions in feet.

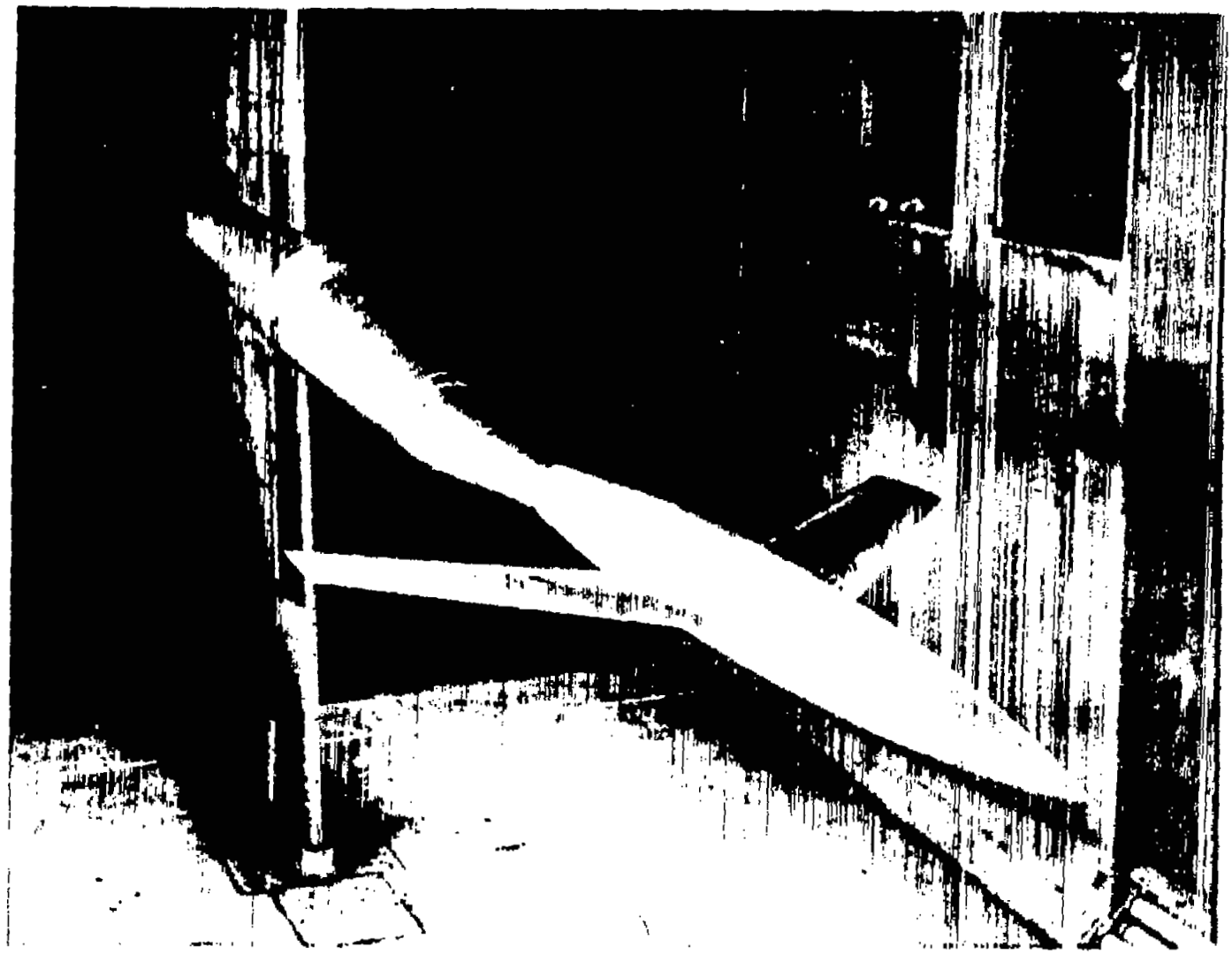

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Figure 2.- Photograph of the model installed on the variable-angle-of-attack support system.

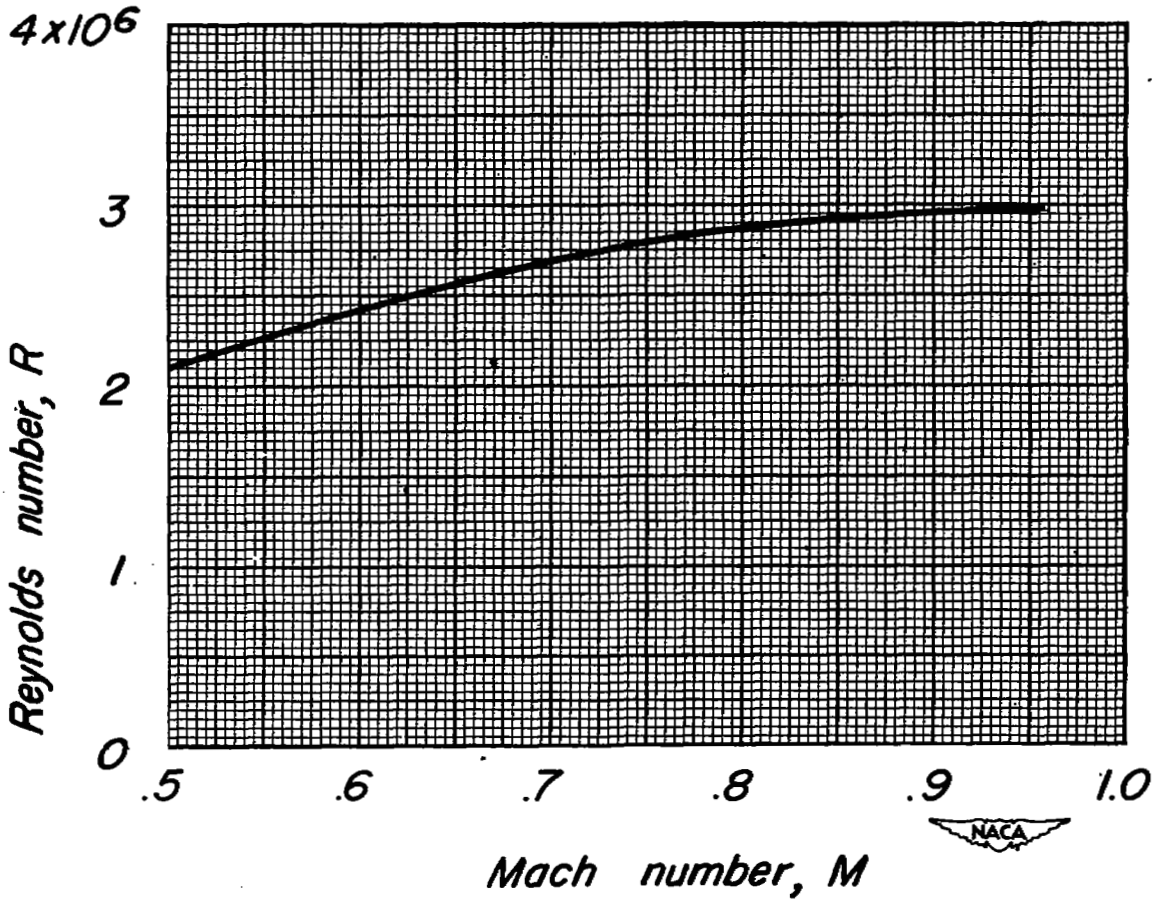
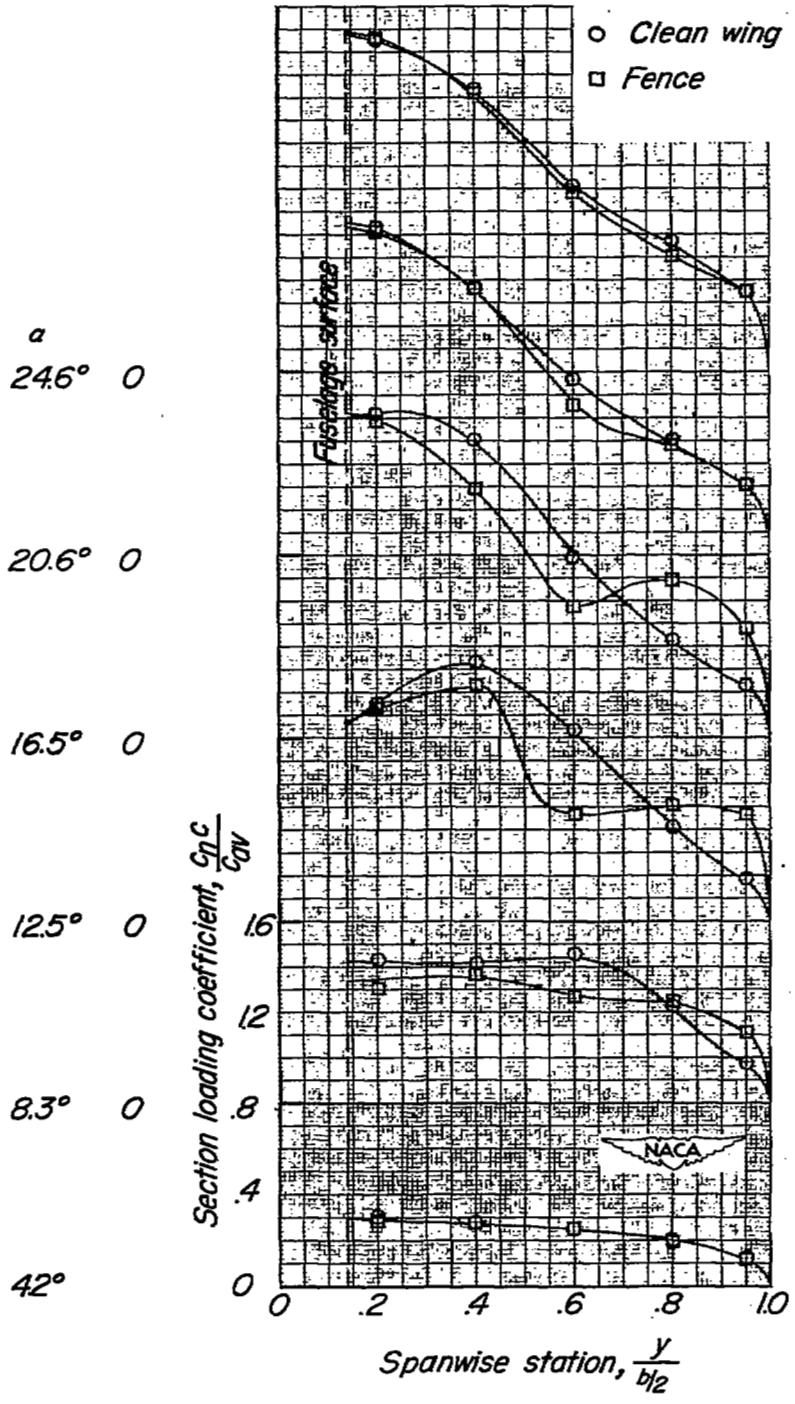
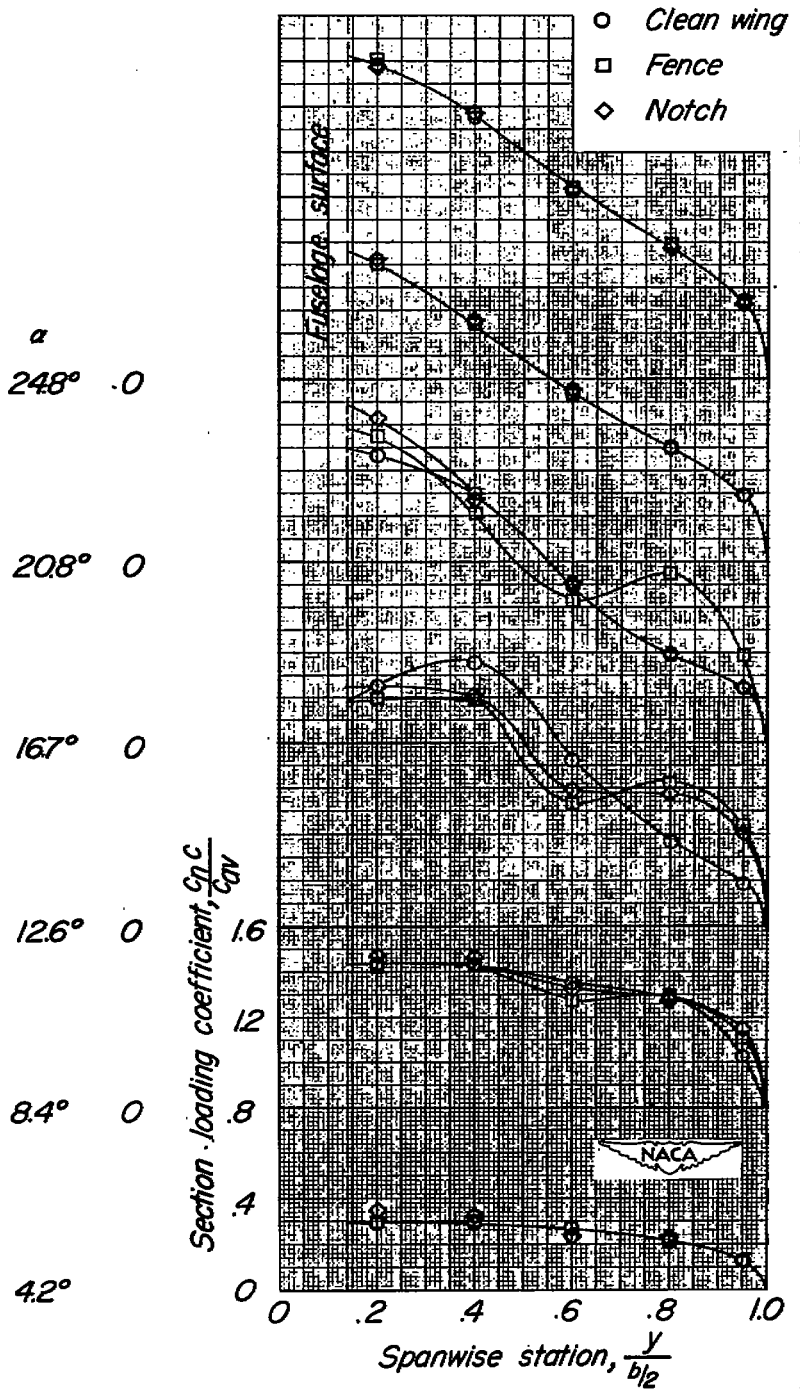


Figure 3.- Variation of mean test Reynolds number with Mach number based on the mean aerodynamic chord of the wing.



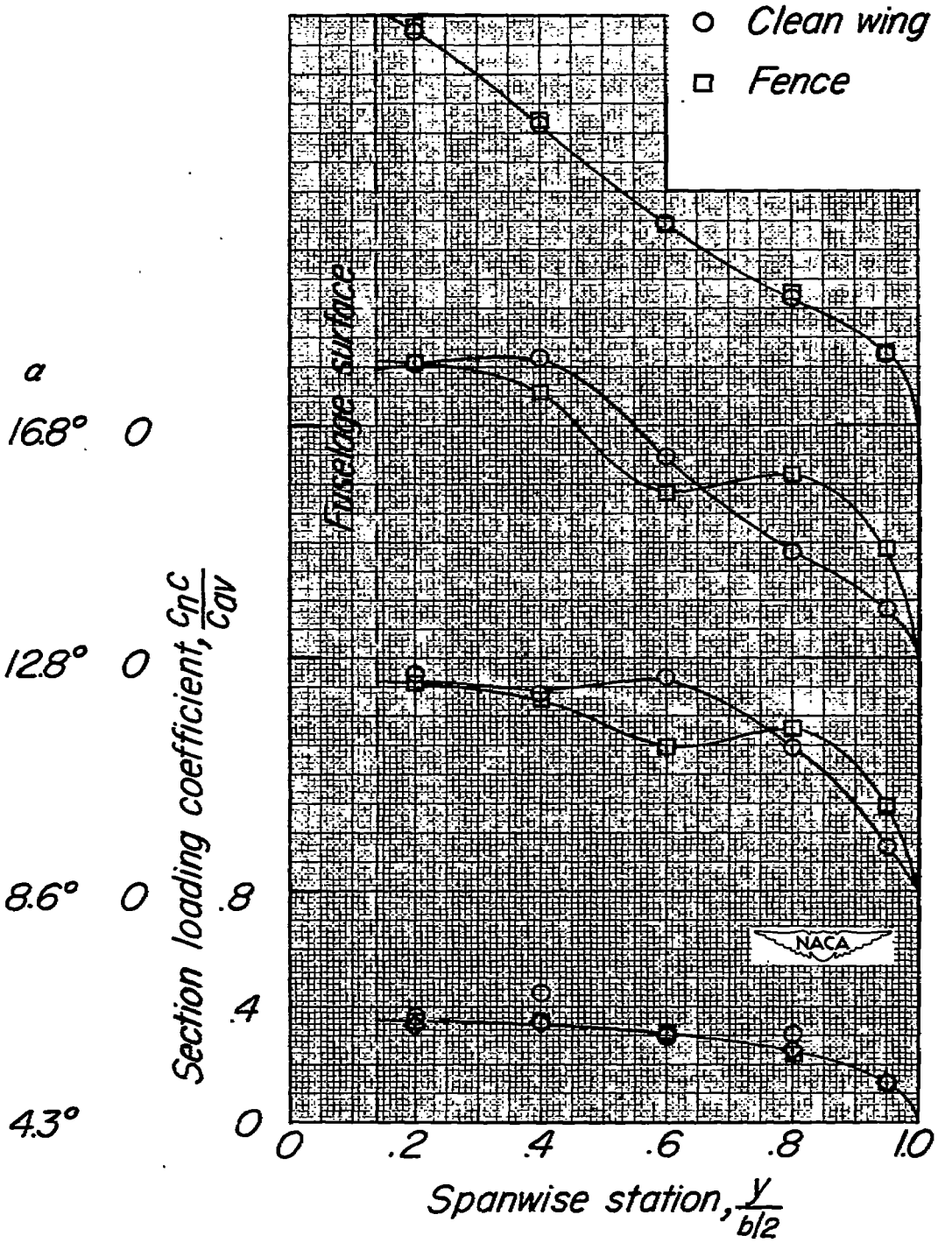
(a) M = 0.50.

Figure 4.- Effect of the fence and notch on the spanwise load distribution.



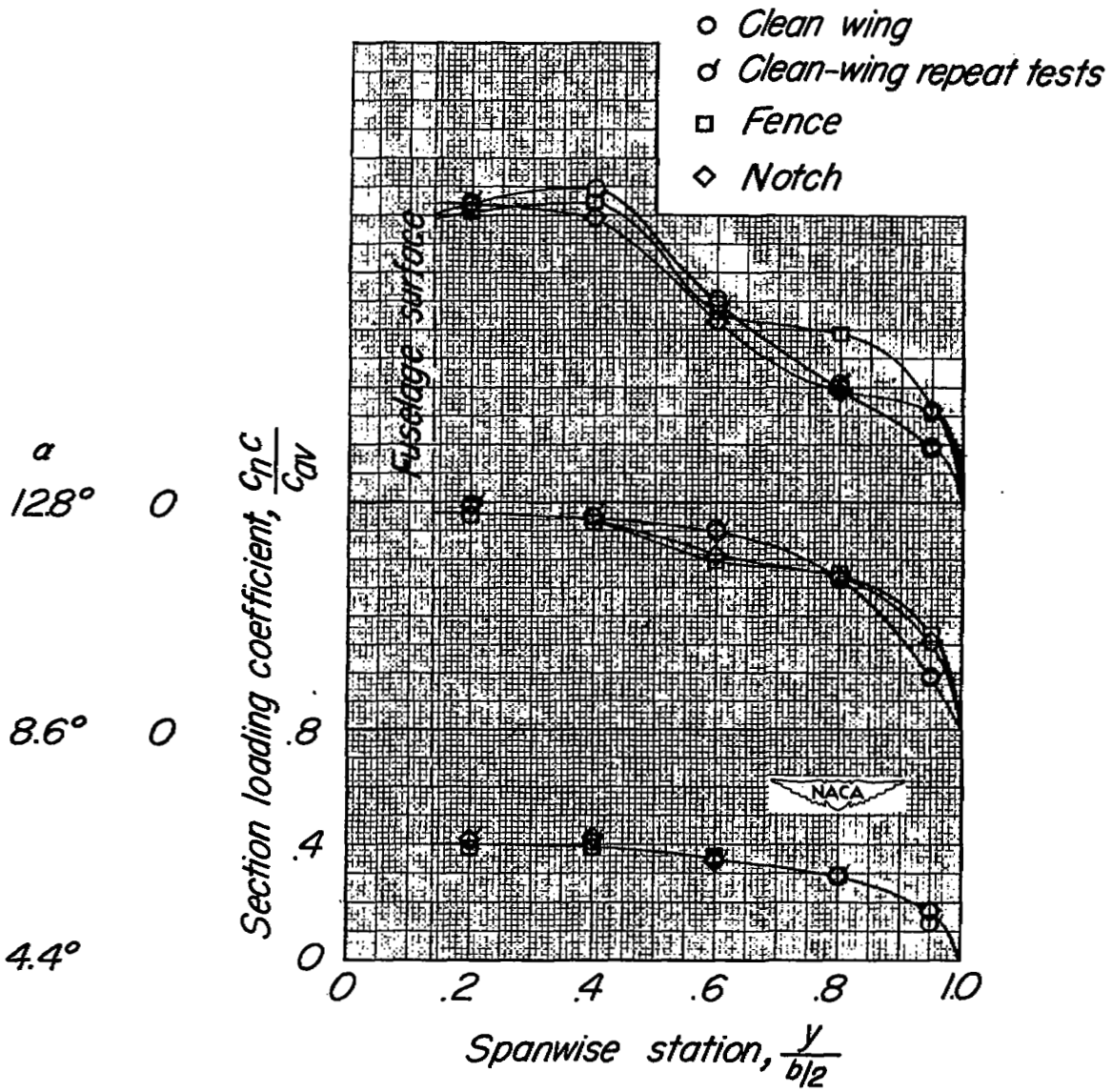
(b) M = 0.70.

Figure 4.- Continued.



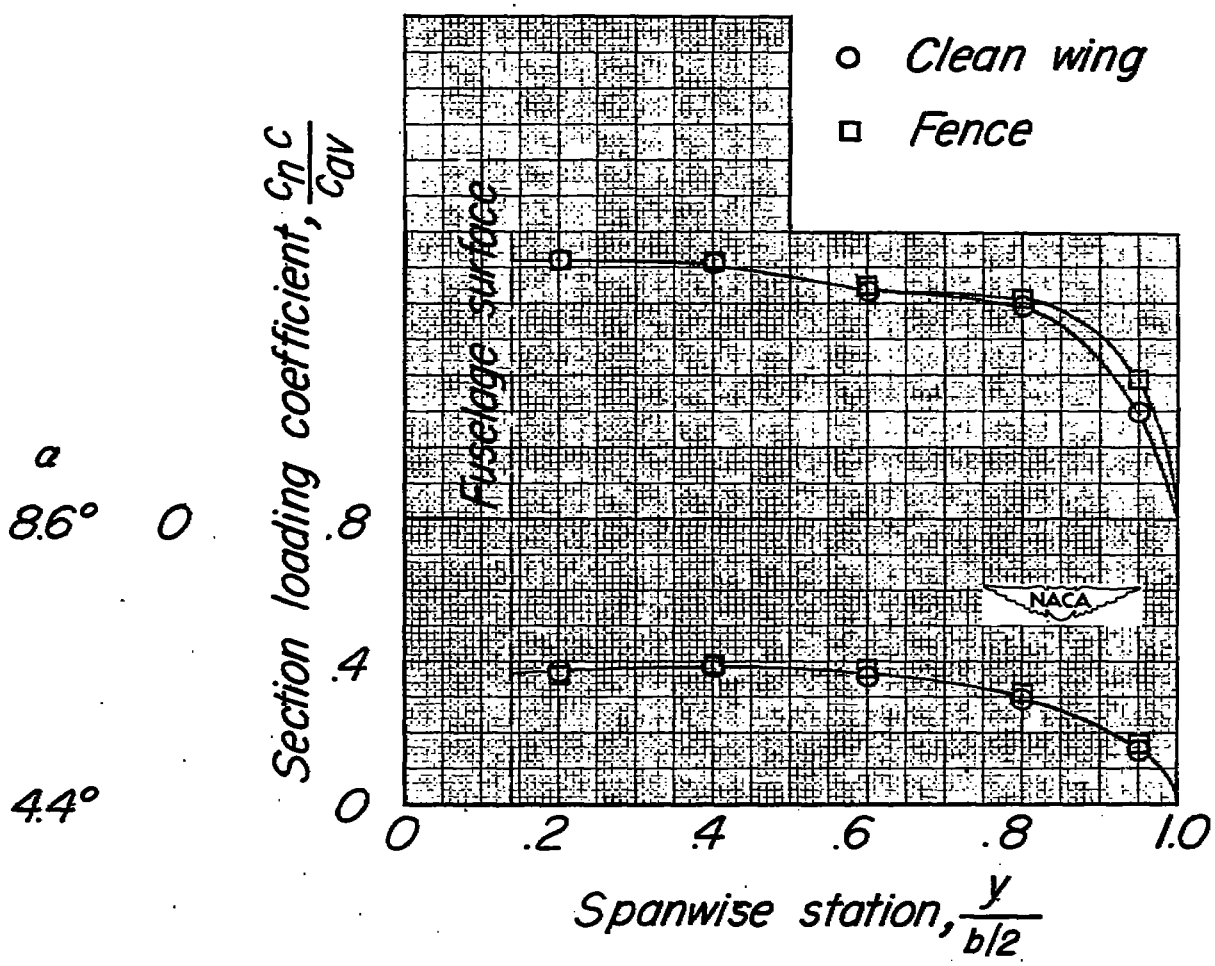
(c) M = 0.85.

Figure 4.- Continued.



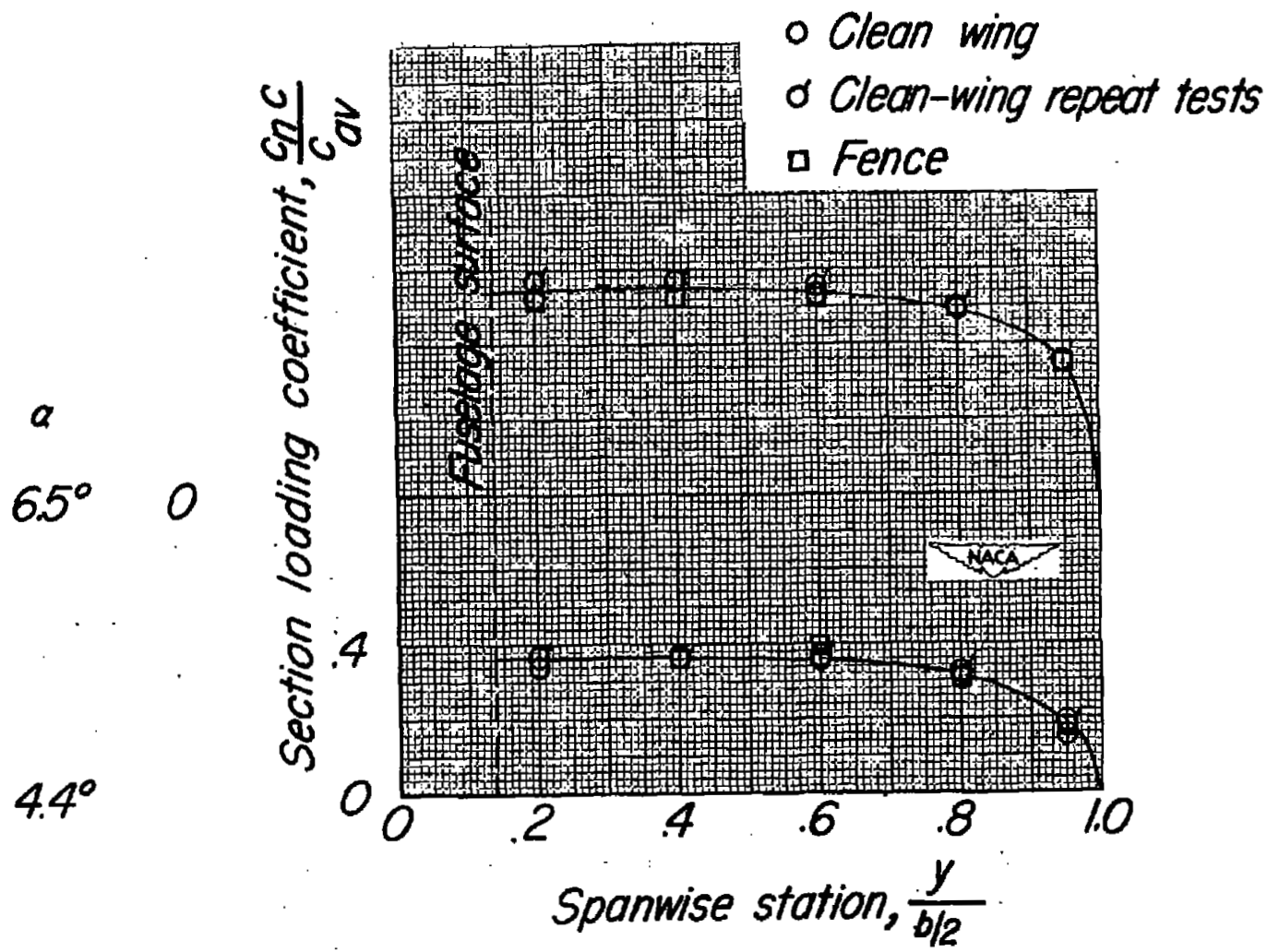
(d) M = 0.91.

Figure 4.- Continued.



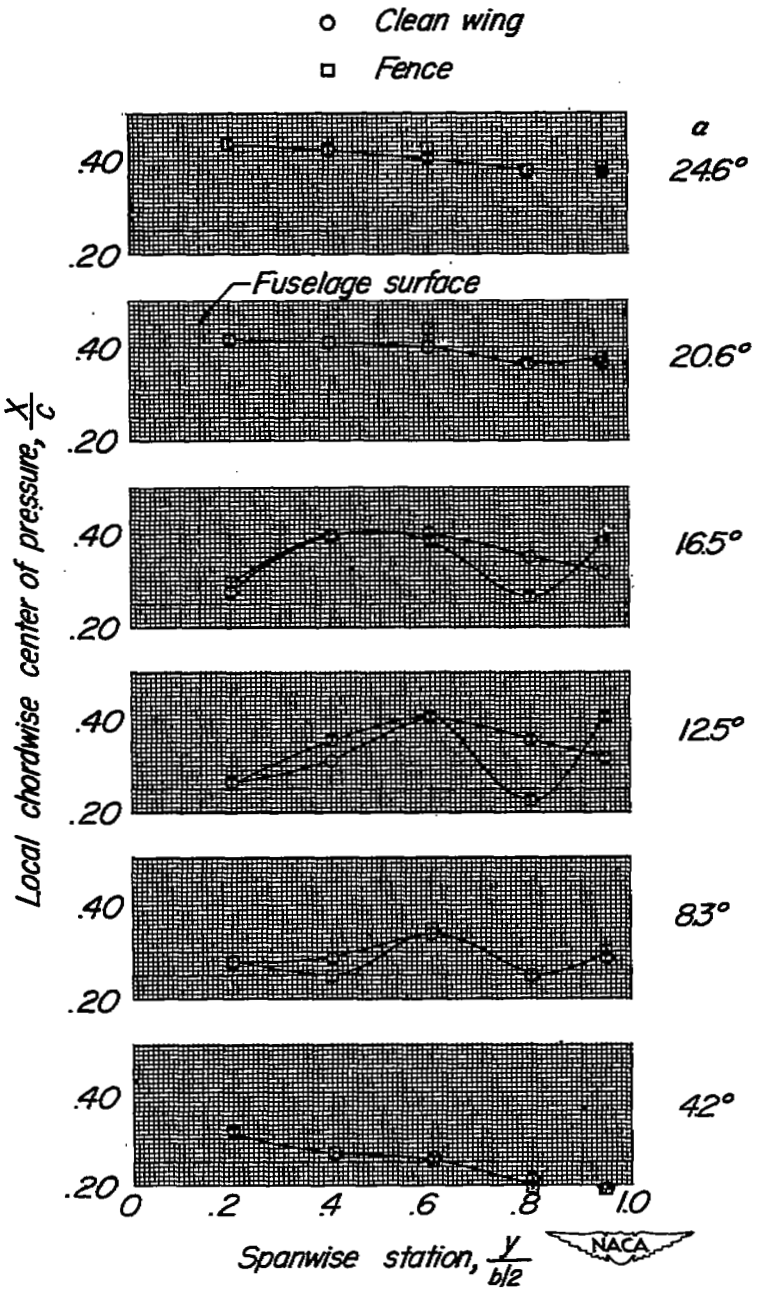
(e) M = 0.93.

Figure 4.- Continued.



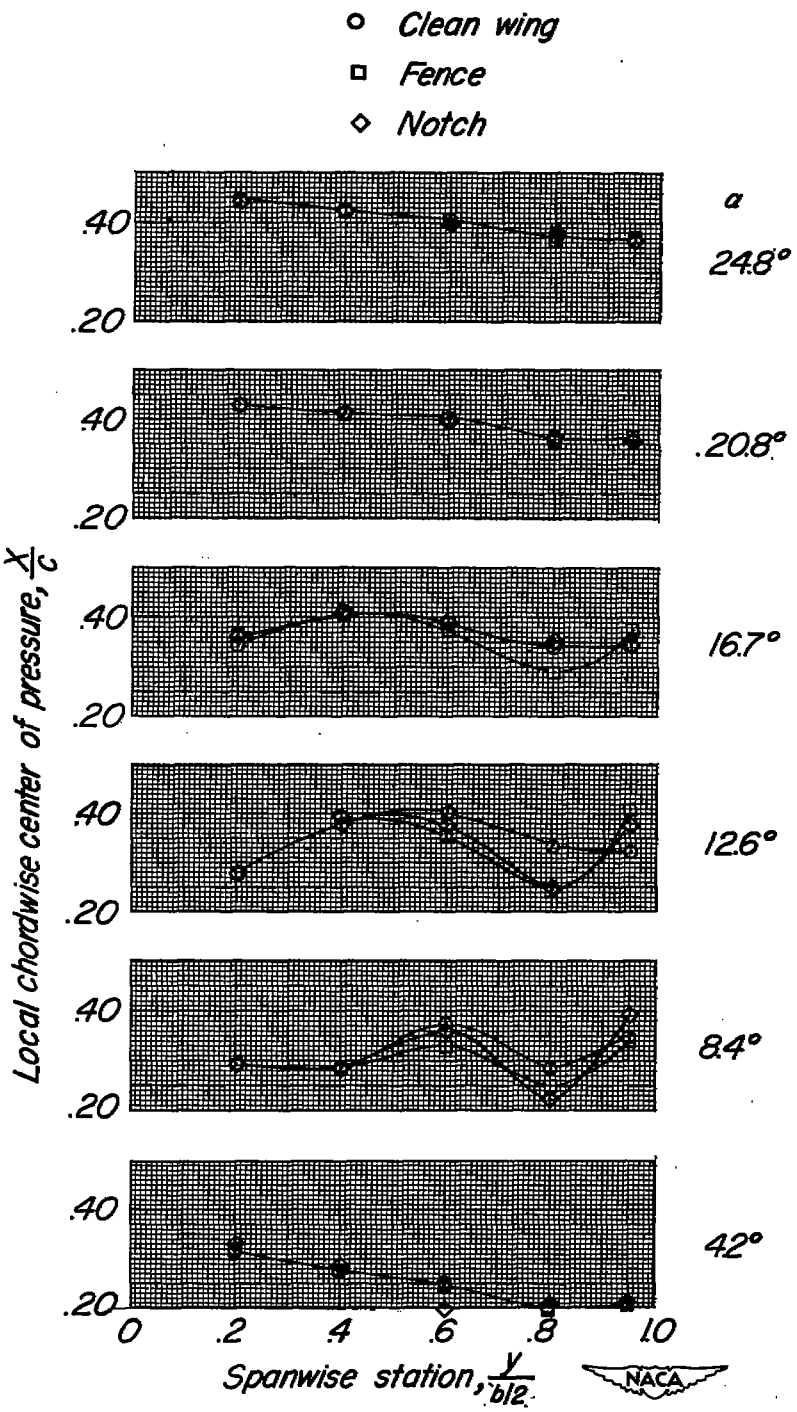
(f) M = 0.95.

Figure 4.- Concluded.



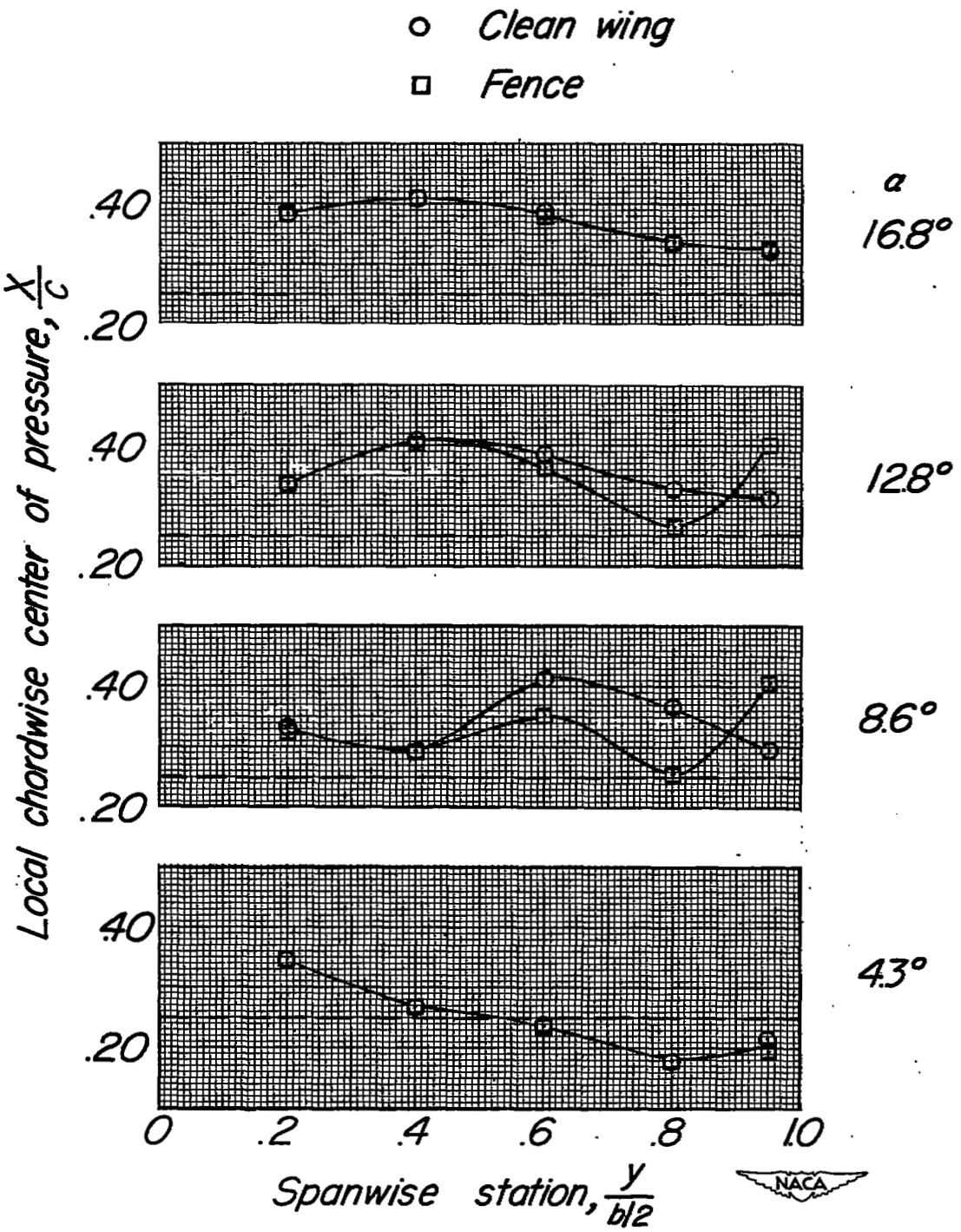
(a) M = 0.50.

Figure 5.- Effects of the fence and notch on the local chordwise centers of pressure.



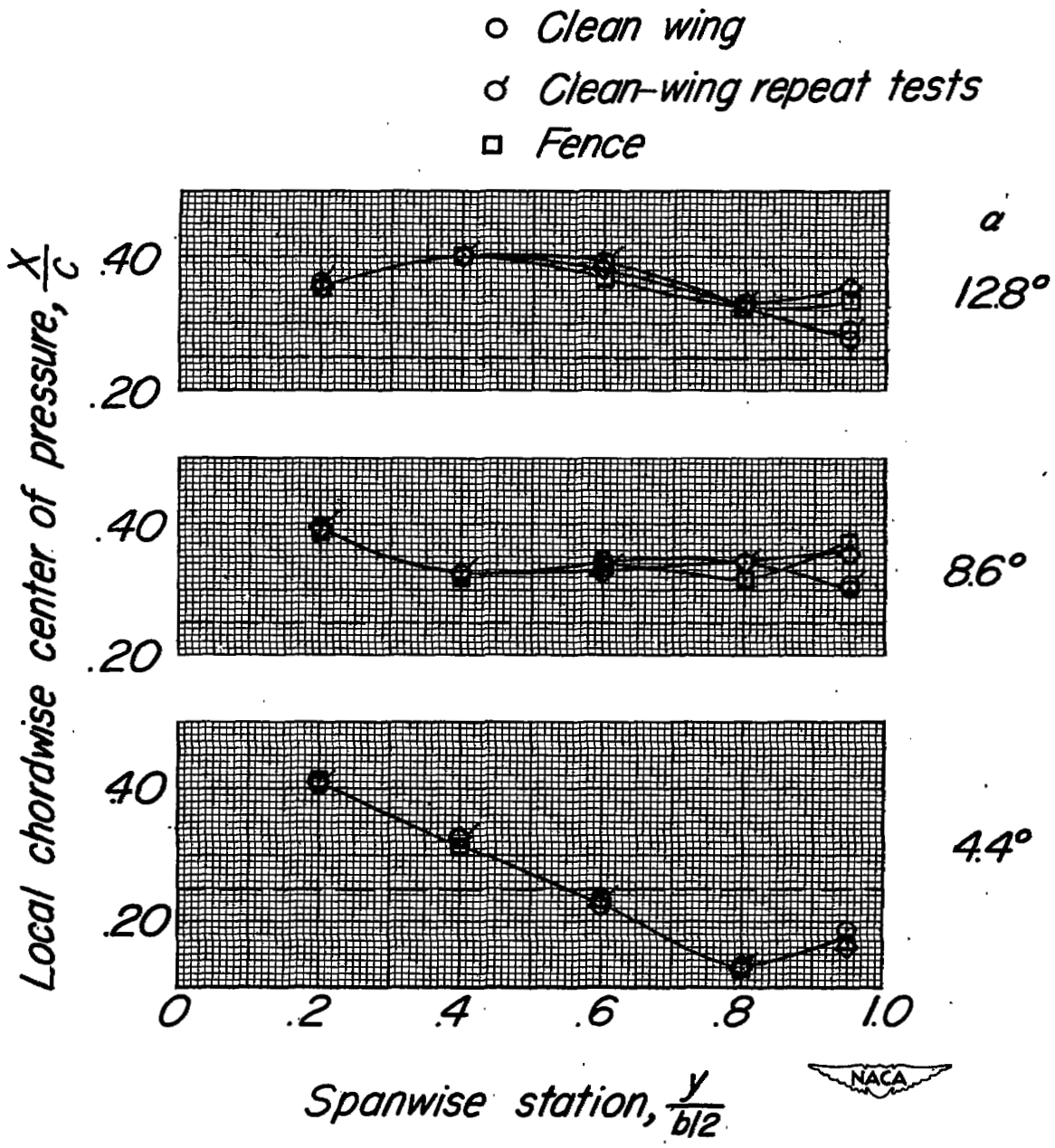
(b) M = 0.70.

Figure 5.- Continued.



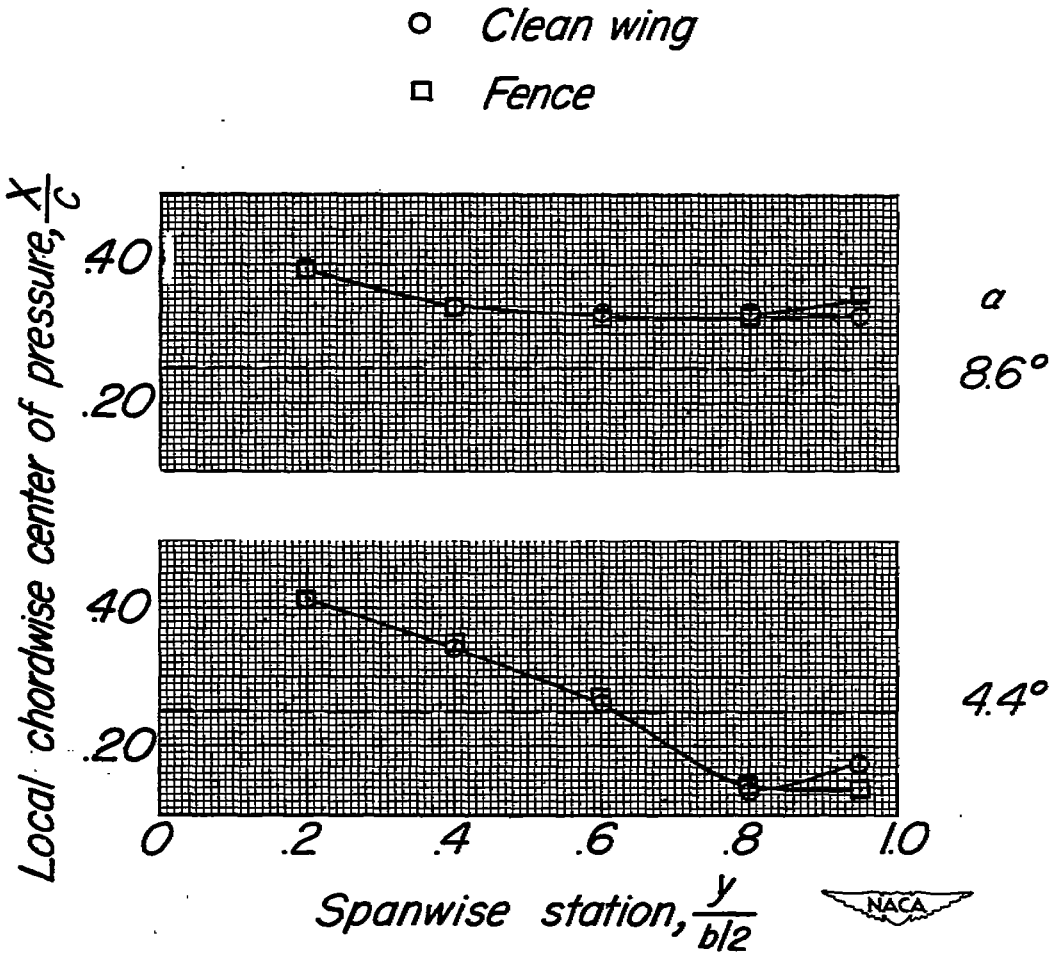
(c) M = 0.85.

Figure 5.- Continued.



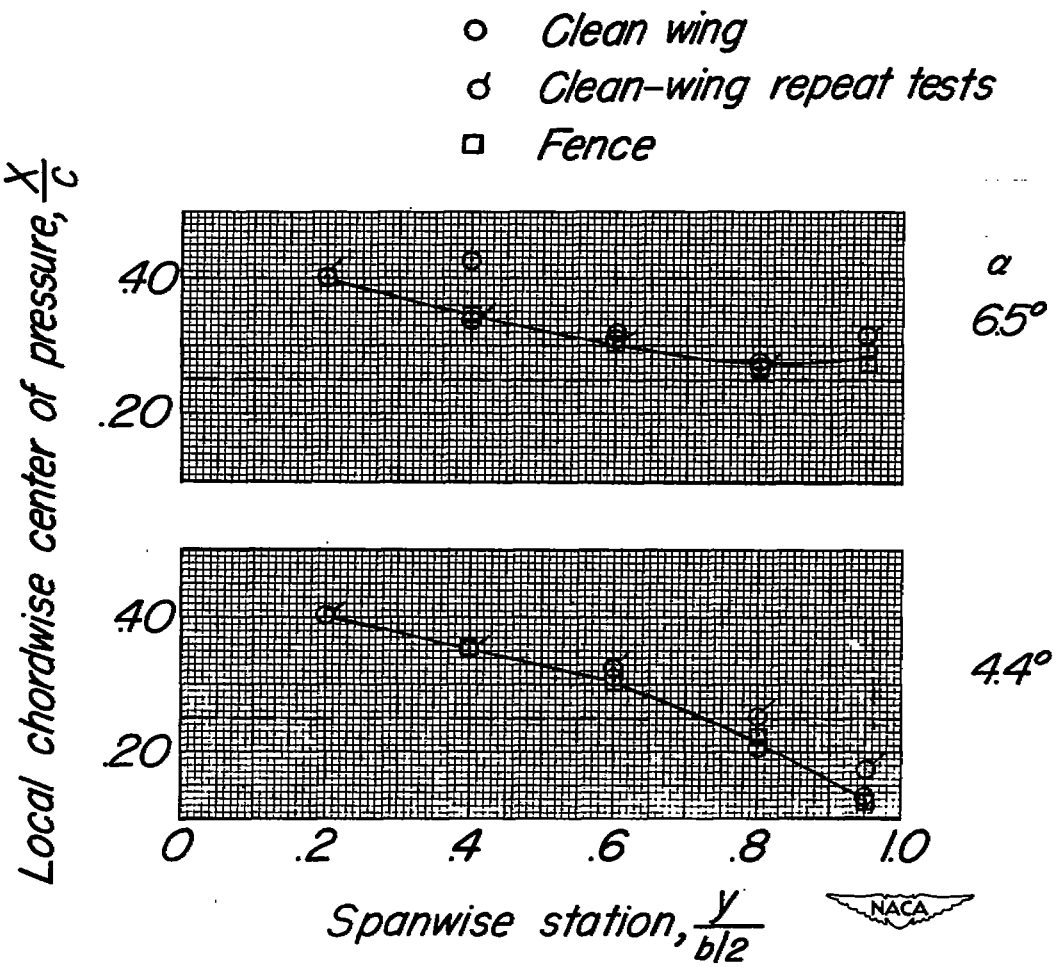
(d) M = 0.91.

Figure 5.- Continued.



(e) M = 0.93.

Figure 5.- Continued.



(f) M = 0.95.

Figure 5.- Concluded.

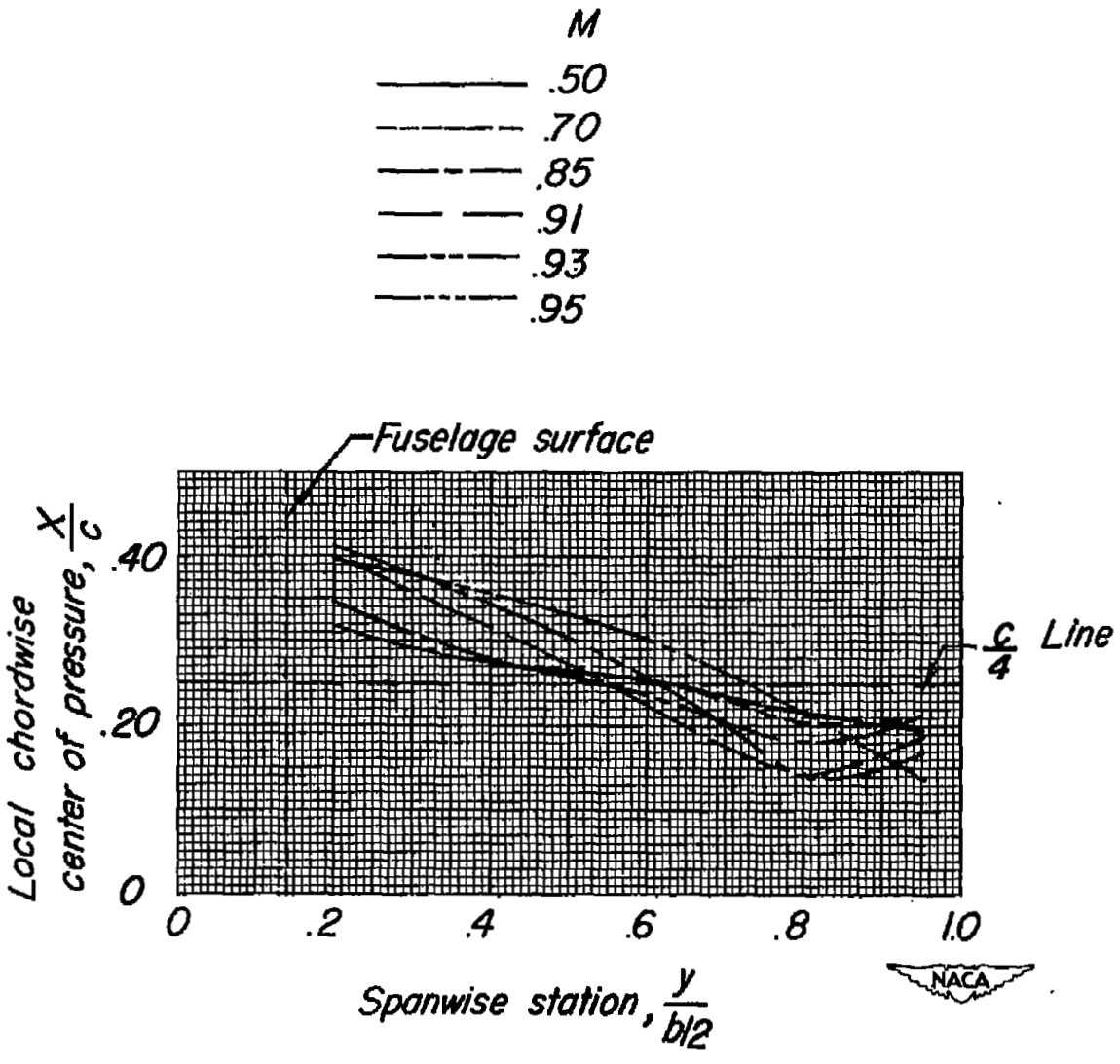


Figure 6.- Effect of Mach number on the local chordwise center of pressure. $\alpha \approx 4^{\circ}$; clean wing.

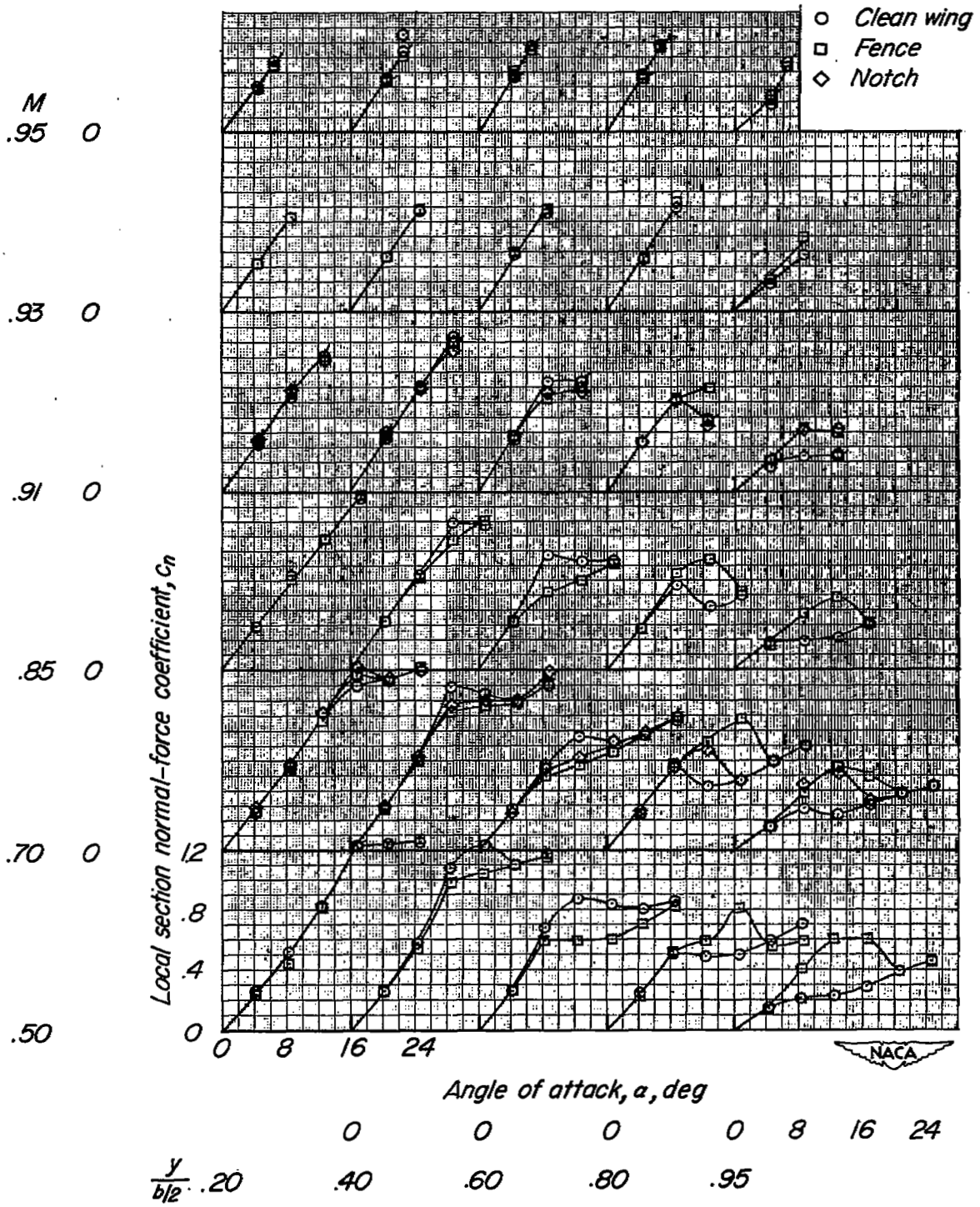


Figure 7.- Local normal-force curves for each of the spanwise stations.

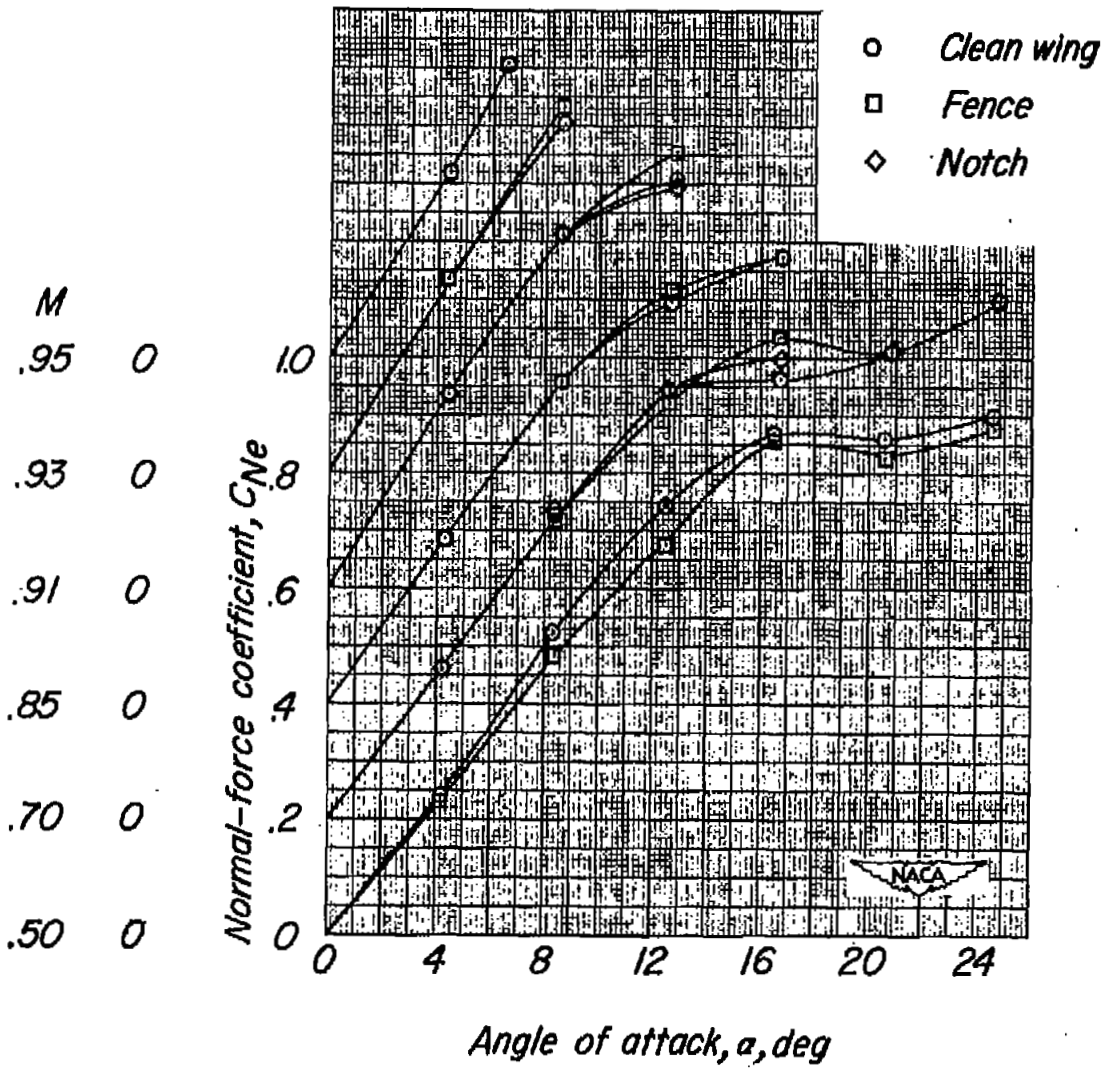


Figure 8.- Variation of the total normal-force coefficient of the exposed wing with angle of attack.

[redacted]

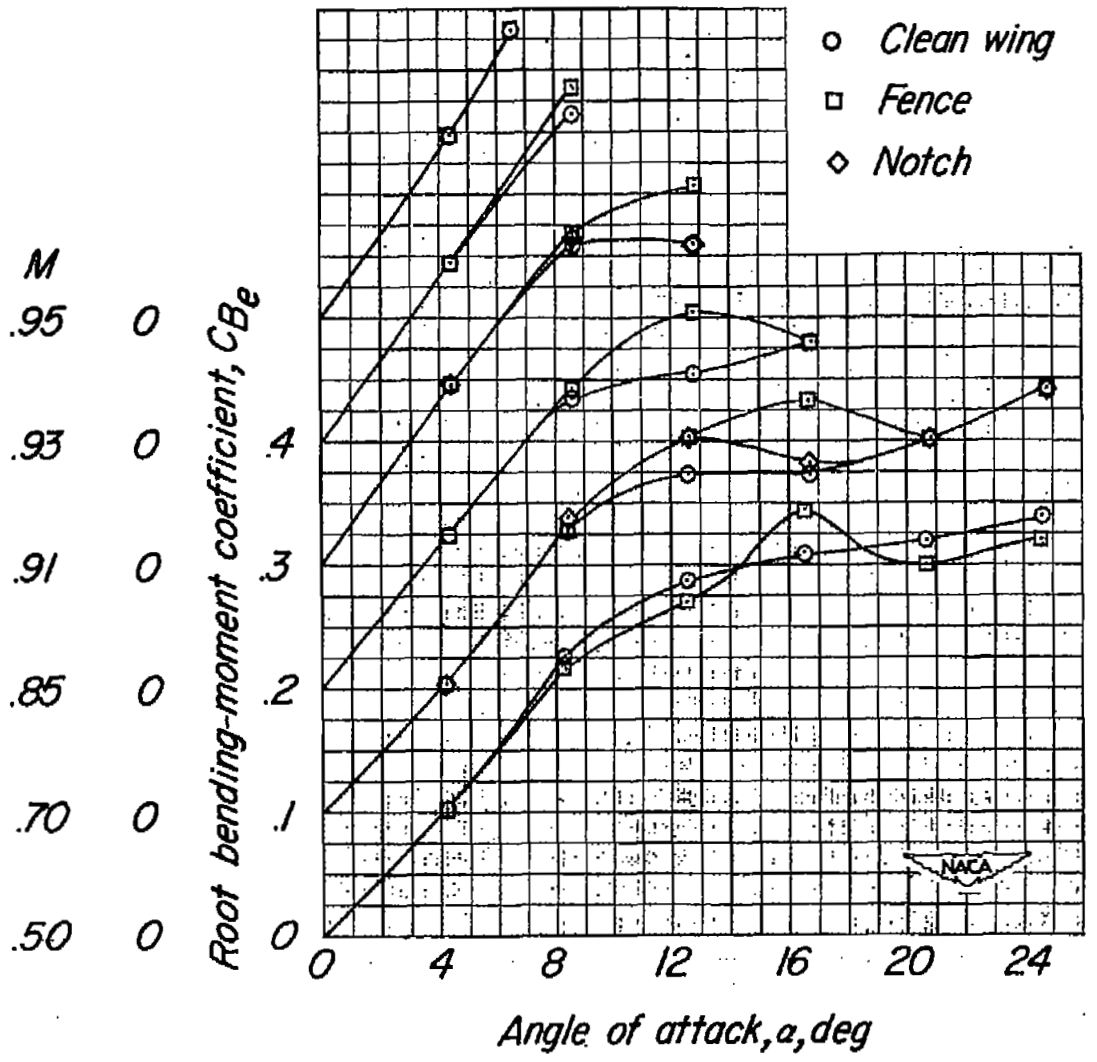


Figure 9.- Variation of the root bending-moment coefficient of the exposed wing with angle of attack.

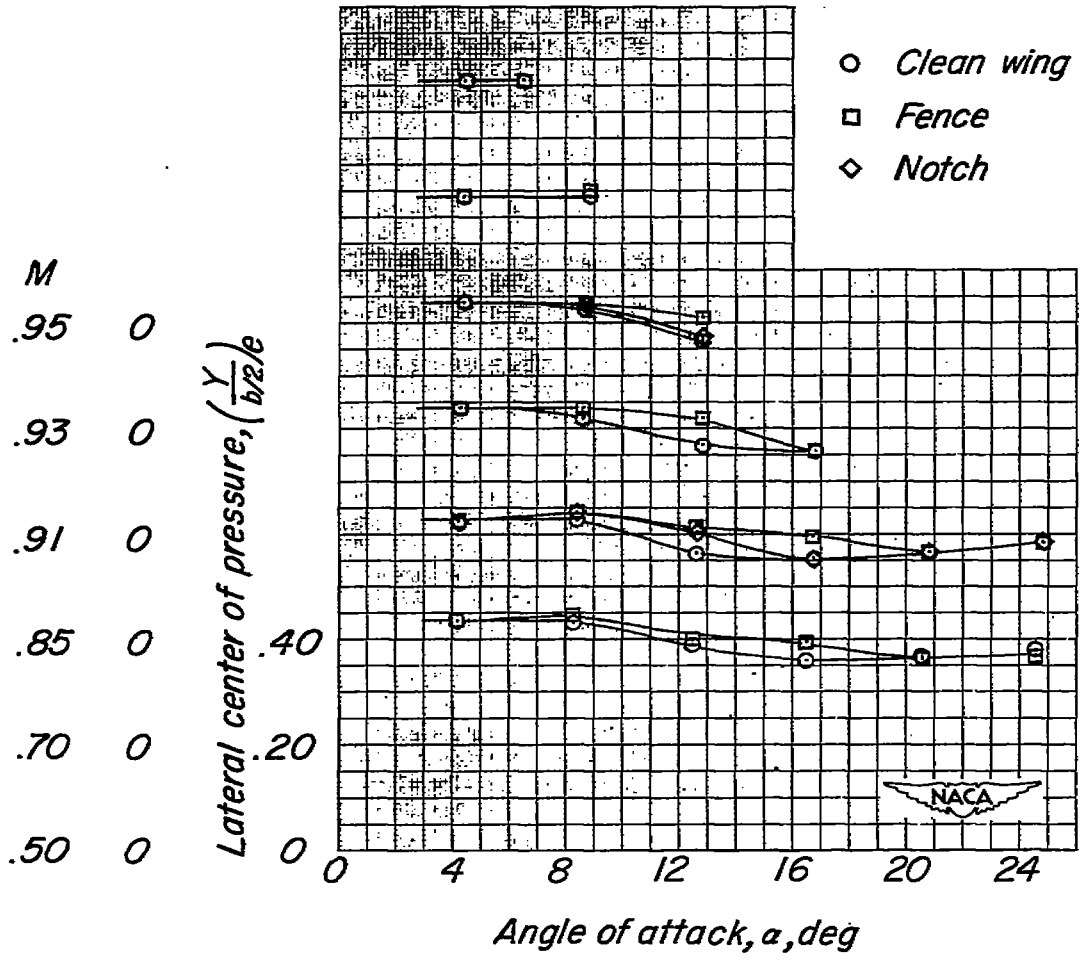


Figure 10.- Variation of the lateral center of pressure of the load on the exposed wing with angle of attack measured from the fuselage surface.

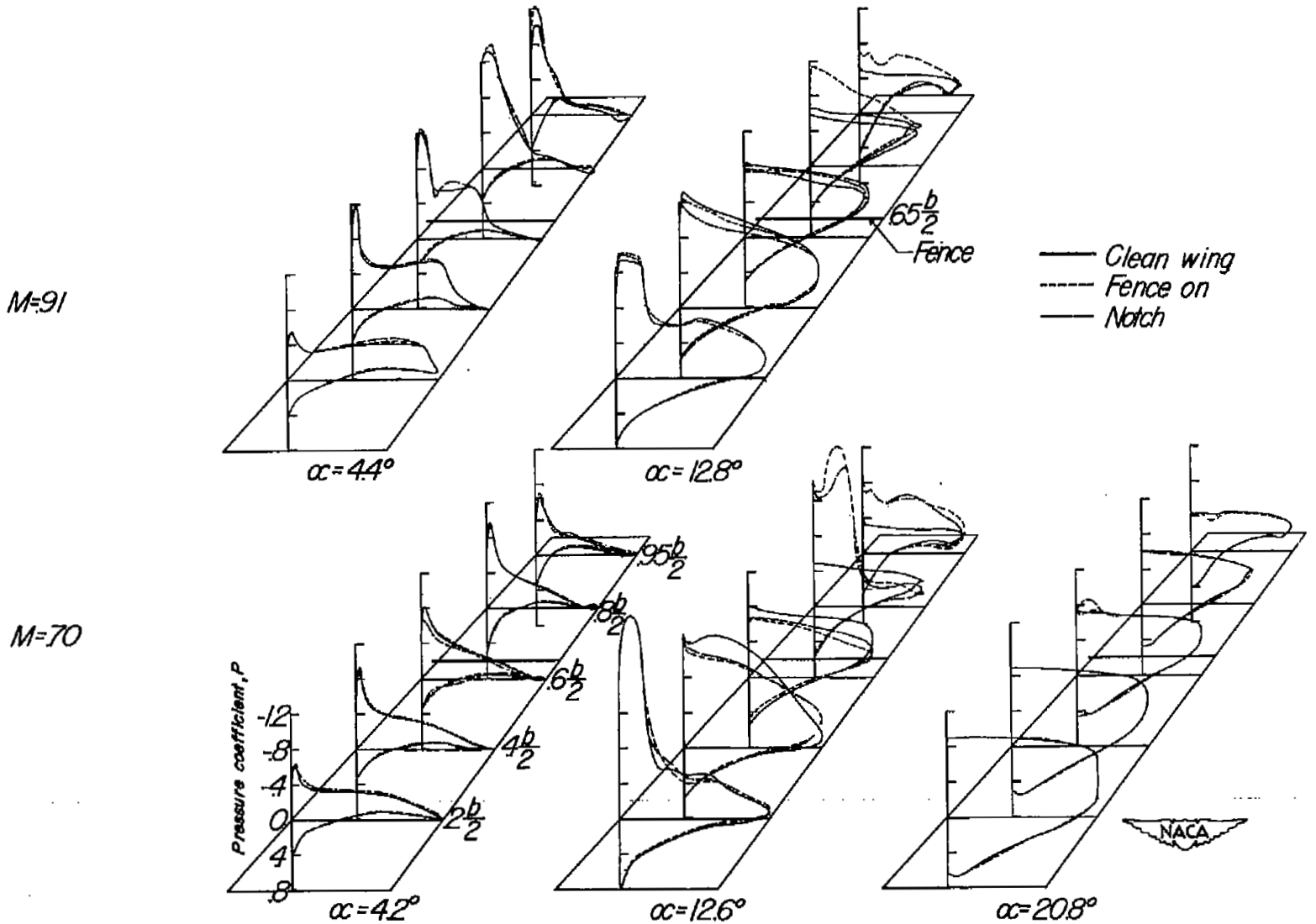


Figure 11.- Effect of fences on the chordwise load distribution.

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