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RESEARCH MEMORANDUM

PRESSURE MEASUREMENTS AT TRANSONIC AND LOW SUPERSONIC SPEEDS ON A THIN CONICAL CAMBERED LOW-ASPECT-RATIO DELTA WING IN COMBINATION WITH BASIC AND INDENTED BODIES

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NATIONAL ADVISORY COMMITTEE
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PRESSURE MEASUREMENTS AT TRANSONIC AND LOW SUPERSONIC SPEEDS ON A THIN CONICAL CAMBERED LOW-ASPECT-RATIO DELTA WING IN COMBINATION WITH BASIC AND INDENTED BODIES

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SUMMARY

Pressure distributions on a thin conical cambered low-aspect-ratio delta wing have been obtained in the Langley 8-foot transonic tunnel over a Mach number range from 0.60 to 1.12 and at a Mach number of 1.43. Data were obtained for the wing in combination with a basic Sears-Haack body and a body indented symmetrically for a Mach number of 1.2. The angle-of-attack range was from -4° to 20° in most instances.

The results indicate that a leading-edge separation vortex forms on the wing at moderate angles of attack. This vortex causes separation over the outboard sections of the wing which extends inboard with increases in angle of attack. Shock waves at transonic speeds caused increased loading over the wing trailing-edge regions. The effects of body indentation were small, as would be expected for this wing configuration.

INTRODUCTION

Flight tests have shown that conical leading-edge camber on a thin low-aspect-ratio delta wing results in increased performance characteristics at transonic and low supersonic speeds (ref. 1). Therefore, a wing of this description has been included in a general program being conducted in the Langley 8-foot transonic tunnel to investigate the detailed pressure distributions and loads on a series of thin wings suitable for transonic and supersonic flight. References 2 and 3 present the results of two previous investigations of this general program. The present paper presents the detailed pressure distributions at transonic and low supersonic speeds on a thin conical cambered low-aspect-ratio delta wing very similar to the cambered wing of reference 1. In addition,

the effects on the pressure distributions of body contouring in accordance with the supersonic area rule (ref. 4) have been included. These data are being presented without analysis in order to expedite publication.

SYMBOLS AND COEFFICIENTS

b	wing span to rounded tips
$b'/2$	unsupported semispan, distance from outer face of wing mounting block to wing tip
c	airfoil section chord, measured parallel to plane of symmetry
$\bar{c}$	wing mean aerodynamic chord
l	body length
M	free-stream Mach number
p	local static pressure
p_o	free-stream static pressure
q_o	free-stream dynamic pressure
x	distance measured streamwise from section leading edge or body nose
y	distance measured laterally from plane of symmetry
y'	distance measured laterally from outer face of wing mounting block
α	angle of attack of body center line
$\Delta\alpha$	angle of attack of wing station minus angle of attack of body center line
$\frac{\partial\Delta\alpha}{\partial n}$	change in wing station twist angle due to normal force at wing section quarter chord
$\frac{\partial\Delta\alpha}{\partial m}$	change in wing station twist angle due to pitching moment about wing section quarter chord

C_p pressure coefficient, $\frac{p - p_o}{q_o}$

$C_{p,sonic}$ pressure coefficient corresponding to local Mach number of 1

D_{max} maximum diameter

R Reynolds number

APPARATUS

Tunnels

The investigation at subsonic and transonic speeds was conducted in the Langley 8-foot transonic tunnel. This facility is a single-return wind tunnel operated at approximately atmospheric stagnation pressures. The dodecagonal-shaped test section has been slotted longitudinally to allow testing through the speed of sound with negligible effects of choking and blockage. A description of the tunnel and its calibration is given in reference 5. Data at a Mach number of 1.43 were obtained in the Langley 8-foot transonic pressure tunnel by enclosing the longitudinal slots with specially designed channels which converted the slotted test section to a supersonic nozzle. Details of the resulting nozzle shape and the test-section Mach number distribution have been published in reference 6.

Models

The delta wing tested has 60° sweepback of the leading edge, a taper ratio of 0, and NACA 65A003 airfoil sections parallel to the model plane of symmetry over the uncambered portion of the wing. The leading-edge portion of the wing has conical camber over the outboard 15 percent of each semispan. This camber was designed for a lift coefficient of 0.15 near $M = 1.0$. Details of the camber distribution and ordinates for the cambered sections have been presented in reference 7. The actual wing plan form deviated from the theoretical delta wing plan form in that the wing tips were rounded. Rounding the tips reduced the wing area by a small amount (a reduction of 0.6 percent of the total wing area of 0.855 square foot) and produced negligible changes in mean aerodynamic chord length and location. The theoretical aspect ratio, which assumes pointed tips, is 2.31. The wing was constructed of steel and was tested as a midwing configuration.

The wing was tested in combination with basic and indented bodies. The basic body is a body of revolution designed to have minimum wave drag

for a given length and volume (Sears-Haack body). The other body was a body of revolution indented symmetrically for a Mach number of 1.2 in accordance with the supersonic area-rule concept (ref. 4). Dimensional details of both models tested are presented in figure 1(a). Ordinates for both the basic and indented bodies are presented in figure 1(b).

TESTS

Both the basic and the indented wing-body combinations were tested at Mach numbers from 0.60 to 1.12 and at a Mach number of 1.43. Generally, the angle-of-attack range extended from -4° to 20°. However, at $M = 1.43$, the angle-of-attack measuring device did not function properly and the resultant angle deviated from the values used at other Mach numbers. In addition, at $M = 1.43$, data for the indented body configuration were not obtained for the two highest angles of attack.

Transition strips were fixed on both configurations during all the tests. The strips were about 0.10 inch wide and were formed by sprinkling No. 120 carborundum grains on a plastic adhesive. The strips extended from the wing-body juncture to the wing tip at 10 percent of the local chord on the upper and lower wing surfaces and formed a ring around the body at 10 percent of the body length.

The Reynolds number based on the wing mean aerodynamic chord varied during the tests from about 2.8×10^{6} at a Mach number of 0.60 to 3.3×10^{6} at the highest test Mach number of 1.43 (see fig. 2). The free-stream dynamic pressure varied from about 400 to 900 pounds per square foot over the same Mach number range. On figure 2 the bands represent the variations in Reynolds number and dynamic pressure due to the changes in atmospheric pressure which occurred during the testing period.

MEASUREMENTS AND ACCURACY

Measurements of the local static pressures on the models were made by use of about 139 orifices distributed over the upper and lower wing surfaces at three wing semispan locations and along five body meridian rows. Orifice locations are given in tables in figure 3. Pressure coefficients determined from these measurements are estimated to be accurate within ±0.005.

The angle of attack of the model was measured by means of a strain-gage attitude transmitter mounted in the nose of the model and is estimated to be accurate within $\pm 0.1^{\circ}$. Calibrations of the Langley 8-foot

transonic tunnel with the test section empty indicate that local deviations from the average free-stream Mach number are of the order of ±0.003 at subsonic speeds. With increases in Mach number, these deviations increased but did not exceed ±0.010 at $M = 1.2$. (See ref. 5.) In the Langley 8-foot transonic pressure tunnel at $M = 1.43$, local deviations from the average free-stream Mach number did not exceed ±0.015 during these tests (ref. 6). The average stream Mach number was held to within ±0.003 of the nominal values shown on the figures.

The wing was calibrated statically for deflection due to load by using an arrangement of a hydraulic jack and a balance scale to apply loads in conjunction with an optical device to read vertical deflections at several stations on the wing. Influence coefficients were obtained from these measurements of wing deflection and are presented in table I. Wing-twist angles, computed by using the experimental wing section data in conjunction with the influence coefficients of table I, are estimated to be accurate to within one-quarter of a degree.

CORRECTIONS

No corrections have been applied to any of the data for boundary-interference effects. At subsonic speeds, the slotted test section minimized boundary-interference effects such as blockage and boundary-induced upwash. At $1.03 < M < 1.12$ boundary-reflected disturbances struck the model; therefore, no data were recorded in this Mach number range.

No corrections have been applied to any of the data for the effects of wing aeroelasticity. In order to provide an indication of the magnitude of these effects, some of the aeroelastic twist characteristics have been calculated and presented in figure 4. The most outboard station where the influence coefficients were measured is 0.80lb/2. In figure 4, the twist calculated at this outboard station plotted against Mach number reaches a maximum value of about -2° at a Mach number of 1.03. Therefore, at $M = 1.03$, the variation of aeroelastic twist across the span has been computed and presented also in figure 4 at several angles of attack. These variations have been faired only out to 0.80lb/2. The influence coefficients were determined and aeroelastic twist angles were computed by using the method found in reference 8.

RESULTS AND DISCUSSION

The wing chordwise pressure distributions in the presence of the basic and indented bodies are presented in figure 5. Longitudinal pressure

distributions along the basic and indented bodies in the presence of the wing are presented in figure 6. Tabulated pressure coefficients for all configurations tested are presented in tables II to V.

Wing Pressure Distributions

At a Mach number of 0.60 (fig. 5(a)) and at moderate angle of attack ($\alpha \approx 8^{\circ}$) the high negative pressure peaks inboard and the broadening and lowering of these peaks with increasing distance from the wing-body juncture indicate the presence of a leading-edge separation vortex. This vortex-flow phenomenon appears to be common to thin sweptback wings and has been noted previously in references 2, 3, and 8. In addition, reference 9, which presents the pressure distributions on a thin 60° delta wing without camber, shows ink-flow studies of this type of flow. This separation vortex appears to form above an angle of attack of 4° on this wing and is fully developed at 8° angle of attack. With increases in spanwise distance the conical vortex system increases in chordwise extent as is shown by the wider regions of negative pressures. With increases in angle of attack above 8°, the center of the vortex is swept progressively rearward. At $\alpha = 10^{\circ}$, the core has swept back until the vortex has moved off the trailing edge inboard of 0.80b/2 and has left the outboard regions separated. At $\alpha = 20^{\circ}$, it appears the vortex is leaving the wing trailing edge about 0.534b/2. This same type of flow phenomenon is apparent to some extent at all of the Mach numbers tested.

As the Mach number is increased, changes in the shapes of the chordwise pressure distributions become apparent at transonic speeds. At $M = 0.90$ (fig. 5(c)), the pressure distributions indicate the presence of a shock wave around 80 percent of the chord at the two inboard stations. Too few orifices were located near the trailing edge at the outboard station to precisely define the distributions in that region. This shock wave moved rearward to about 90 percent of the chord at $M = 0.94$ (fig. 5(d)) and then to the trailing edge at higher Mach numbers. This movement of the shock wave resulted in increased loading over the trailing-edge regions and corresponding rearward shifts in the center of pressure of the wing.

At $\alpha = 0^{\circ}$, the pressure distributions of figure 5 show almost no loading inboard and a small negative load over the outboard sections. This would be expected since with this type of camber the outboard sections are operating at a lower angle of attack than the wing-body center line (see fig. 1(c) of ref. 7).

Effects of Body Indentation on Wing Pressures

Figure 5 shows that the effects of body indentation on the wing pressures are generally small. In several instances, some rather large differences were noted over the outboard sections (for example, fig. 5(d), $\alpha = 8^{\circ}$ and 16°); however these differences occurred only over a limited Mach number and angle-of-attack range. Reasons for these differences are not known. The consistent differences in the outboard distributions at $\alpha = 8^{\circ}$ through the Mach number range indicate that body indentation had some effect on the separation characteristics of the wing.

Body Pressure Distributions

Examination of the longitudinal pressure distributions for the body in the presence of the wing (fig. 6(a)) indicates that the presence of the wing causes pressure peaks to occur on the body in the region of the wing. The location and magnitude of these pressure peaks, particularly those on the upper surface of the body, are closely associated with the pressures on the wing near the wing-body juncture. With increases in angle of attack, the magnitude of the minimum pressure peak on the upper surfaces of the body (stations A and B, fig. 6) generally increased up to the maximum angle of attack of 20°. At the higher angles of attack ($\alpha = 16^{\circ}$ and 20°), there are indications that separation may have occurred over the upper rear portion of the body. With increases in Mach number, the minimum pressure peak on the upper surfaces of the body showed a pronounced rearward shift while the positive peak on the lower surfaces of the body remained relatively unchanged.

Effects of Body Indentation on Body Pressures

Figure 6 shows that the effects of body indentation on the body pressures are also small. Generally, these effects are confined to the regions of local acceleration and deceleration of the flow corresponding to the changes in the curvature of the indentation.

CONCLUDING REMARKS

The results of tests of a thin conical cambered low-aspect-ratio delta wing in combination with both basic and indented bodies have led to the following concluding remarks:

1. A leading-edge separation vortex has the predominant effect on the flow over the upper surface of the wing. This vortex causes separation over the outboard sections of the wing at moderate angles of attack which extends inboard with an increase in angle of attack.

2. Shock waves at transonic speeds caused increased loading over the wing trailing-edge regions and corresponding rearward shifts in the center of pressure of the wing.

3. The effects of body indentation on both the wing and body pressures are small.

Langley Aeronautical Laboratory,
National Advisory Committee for Aeronautics,
Langley Field, Va., July 1, 1957.

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TABLE I.- WING DEFLECTION CHARACTERISTICS

Twist-measurement station, $\frac{y}{b/2}$	Rate of change in twist angle due to a load at section quarter chord, $\frac{\partial \Delta\alpha}{\partial n}$, deg/lb, at -		
	$\frac{y'}{b'/2} = 0.142$	$\frac{y'}{b'/2} = 0.428$	$\frac{y'}{b'/2} = 0.715$
0.267	0.00015	-0.00041	-0.00190
0.534	0.00006	0.00109	-0.00563
0.801	0.00024	0.00172	-0.00570

Twist-measurement station, $\frac{y}{b/2}$	Rate of change in twist angle due to a pitching moment about section quarter chord, $\frac{\partial \Delta\alpha}{\partial m}$, deg/in-lb, at -		
	$\frac{y'}{b'/2} = 0.142$	$\frac{y'}{b'/2} = 0.428$	$\frac{y'}{b'/2} = 0.715$
0.267	0.00040	0.00040	0.00064
0.534	0.00076	0.00202	0.00118
0.801	0.00152	0.00452	0.00811

TABLE II.- PRESSURE COEFFICIENTS FOR WING IN PRESENCE OF BASIC BODY

(a) M = 0.60

	x/c	α = −4°	α = 0°	α = 4°	α = 8°	α = 10°	α = 12°	α = 16°	α = 20°	x/c
	26.7– percent-semispan station									
Upper surface	.010	.260	.093	−.301	−1.227	−1.430	−1.707	−2.215	−2.676	.010
	.030	.184	−.002	−.277	−1.285	−1.652	−1.823	−2.195	−2.829	.030
	.060	.117	−.037	−.216	−.341	−.939	−1.768	−2.437	−2.670	.060
	.100	.088	−.041	−.176	−.289	−.293	−.486	−1.598	−2.408	.100
	.150	.068	−.052	−.173	−.293	−.333	−.371	−.709	−1.381	.150
	.200	.040	−.065	−.170	−.284	−.329	−.387	−.566	−.990	.200
	.300	.026	−.057	−.150	−.237	−.280	−.328	−.432	−.633	.300
	.400	.004	−.072	−.152	−.222	−.269	−.318	−.381	−.526	.400
	.500	−.008	−.074	−.145	−.205	−.250	−.277	−.313	−.462	.500
	.600	−.015	−.069	−.128	−.184	−.210	−.225	−.248	−.402	.600
	.700	−.018	−.062	−.107	−.150	−.166	−.174	−.220	−.332	.700
	.800	−.010	−.047	−.079	−.110	−.118	−.121	−.199	−.249	.800
	.900	.007	−.013	−.034	−.043	−.050	−.054	−.122	−.122	.900
	.950	.009	−.004	−.018	−.026	−.028	−.031	−.084	−.077	.950
Lower surface	.025	−.647	−.047	.120	.272	.317	.361	.438	.495	.025
	.060	−.137	−.009	.107	.247	.297	.345	.443	.527	.060
	.100	−.166	−.028	.077	.220	.269	.318	.420	.515	.100
	.150	−.168	−.041	.055	.184	.231	.285	.384	.486	.150
	.200	−.171	−.055	.039	.158	.203	.250	.353	.454	.200
	.300	−.156	−.059	.024	.127	.168	.212	.304	.399	.300
	.400	−.159	−.069	.002	.095	.131	.172	.258	.347	.400
	.500	−.158	−.084	−.018	.063	.097	.135	.209	.287	.500
	.600	−.140	−.080	−.026	.046	.054	.107	.172	.239	.600
	.700	−.116	−.068	−.021	.040	.064	.089	.141	.197	.700
	.800	−.082	−.044	−.013	.036	.053	.074	.110	.154	.800
	.850	−.067	−.041	−.013	.028	.039	.058	.086	.122	.850
	.925	−.033	−.015	.004	.034	.045	.053	.069	.092	.925
	53.4– percent-semispan station									
Upper surface	.010	.252	.144	−.372	−.871	−.975	−1.078	−1.293	−1.365	.010
	.030	.211	.034	−.347	−.874	−.981	−1.082	−1.289	−1.346	.030
	.060	.152	−.035	−.342	−.867	−.976	−1.081	−1.268	−1.311	.060
	.100	.106	−.065	−.312		−1.151	−1.102	−1.293	−1.298	.100
	.200	.054	−.079	−.238	−.916	−1.293	−1.179	−1.245	−1.231	.200
	.300	.026	−.079	−.201	−.639	−1.173	−1.560	−1.183	−1.164	.300
	.400	.007	−.081	−.181	−.210	−.537	−1.070	−1.555	−1.219	.400
	.500	−.010	−.084	−.165	−.133	−.147	−.477	−1.123	−1.118	.500
	.600	−.016	−.074	−.139	−.128		−.170	−.820	−.986	.600
	.700	−.007	−.054	−.102	−.102	−.063	−.057	−.633	−.844	.700
	.800	.006	−.030	−.064	−.059	−.034	−.009	−.469	−.723	.800
	.900	.021	−.004	−.028	−.020	−.004	.006	−.329	−.602	.900
Lower surface	.025	−.008	−.009	−.011	.258	.284	.300	.334	.346	.025
	.060	−.552	−.120	.124	.234	.269	.304	.359	.398	.060
	.100	−.626	−.066	.103	.212	.248	.284	.348	.404	.100
	.200	−.734	−.021	.077	.180	.211	.248	.318	.383	.200
	.300	−.274	−.051	.038	.137	.170	.207	.279	.347	.300
	.400	.012	−.053	.035	.110	.142	.179	.244	.310	.400
	.500	−.060	−.069	−.002	.077	.106	.136	.196	.257	.500
	.600	−.077	−.066	−.007	.061	.083	.107	.159	.210	.600
	.700	−.060	−.049	−.004	.050	.068	.087	.125	.165	.700
	.800	−.034	−.028	.004	.045	.059	.071	.084	.109	.800
	.850	−.017	−.010	.012	.044	.055	.059	.060	.068	.850
	80.1 – percent-semispan station									
Upper surface	.030	.229	.086	−.398	−.587	−.632	−.695	−.687	−.769	.030
	.060	.193	.026	−.378	−.581	−.627	−.689	−.684	−.768	.060
	.100	.154	−.009	−.361	−.570	−.620	−.677	−.677	−.768	.100
	.200	.103	−.054	−.292	−.551	−.602	−.651	−.662	−.766	.200
	.300	.048	−.081	−.270	−.536	−.595	−.632	−.656	−.747	.300
	.400	.022	−.093	−.248	−.518	−.587	−.617	−.641	−.721	.400
	.500	.002	−.091	−.217	−.512	−.562	−.581	−.629	−.688	.500
	.600	−.004	−.077	−.172	−.510	−.555	−.525	−.623	−.659	.600
	.700	.000	−.050	−.119	−.507	−.619	−.474	−.584	−.626	.700
	.850	−.003	−.017	−.059	−.460	−.669	−.565	−.517	−.592	.850
Lower surface	.045	−.491	−.294	.065	.205	.229	.246	.260	.275	.045
	.150	−.493	−.255	.074	.180	.210	.236	.274	.315	.150
	.250	−.506	−.228	.060	.148	.180	.206	.244	.286	.250
	.350	−.573	−.079	.047	.131	.152	.175	.212	.252	.350
	.450	−.623	.008	.040	.113	.134	.151	.183	.221	.450
	.550	−.639	−.007	.021	.080	.094	.109	.135	.168	.550
	.650	−.580	−.017	.013	.052	.064	.077	.089	.117	.650
	.750	−.441	−.009	.012	.029	.031	.036	.038	.055	.750

TABLE II.- PRESSURE COEFFICIENTS FOR WING IN PRESENCE OF BASIC BODY - Continued

(b) M = 0.80

	x/c	α = -4°	α = 0°	α = 4°	α = 8°	α = 10°	α = 12°	α = 16°	α = 20°	x/c
	26.7- percent -semispan station									
Upper surface	.010	.266	.101	-.279	-1.047	-1.235	-1.331	-1.533	-1.887	.010
	.030	.180	.002	-.267	-.907	-1.315	-1.455	-1.601	-1.884	.030
	.060	.115	-.030	-.214	-.601	-1.159	-1.618	-1.777	-1.949	.060
	.100	.090	-.036	-.172	-.390	-.287	-.544	-1.669	-1.944	.100
	.150	.065	-.054	-.179	-.343	-.353	-.402	-1.039	-1.586	.150
	.200	.039	-.067	-.180	-.320	-.360	-.424	-.732	-1.148	.200
	.300	.027	-.066	-.165	-.276	-.317	-.368	-.535	-.710	.300
	.400	.001	-.082	-.176	-.273	-.324	-.386	-.491	-.617	.400
	.500	-.016	-.087	-.174	-.263	-.317	-.357	-.391	-.641	.500
	.600	-.022	-.084	-.157	-.228	-.268	-.290	-.280	-.489	.600
	.700	-.025	-.076	-.131	-.185	-.211	-.217	-.266	-.409	.700
	.800	-.018	-.059	-.098	-.128	-.144	-.147	-.303	-.365	.800
	.900	.004	-.017	-.041	-.047	-.057	-.060	-.217	-.261	.900
	.950	.009	-.007	-.023	-.024	-.029	-.032	-.163	-.207	.950
Lower surface	.025	-.783	-.045	.119	.273	.326	.378	.470	.549	.025
	.060	-.127	-.009	.110	.250	.302	.354	.456	.551	.060
	.100	-.169	-.028	.079	.222	.274	.325	.427	.530	.100
	.150	-.170	-.042	.059	.192	.237	.288	.392	.493	.150
	.200	-.181	-.060	.042	.163	.208	.257	.360	.461	.200
	.300	-.171	-.065	.026	.134	.176	.220	.312	.404	.300
	.400	-.180	-.080	-.000	.101	.137	.179	.266	.351	.400
	.500	-.186	-.098	-.024	.066	.101	.139	.216	.293	.500
	.600	-.172	-.096	-.031	.047	.070	.114	.177	.246	.600
	.700	-.144	-.082	-.029	.041	.066	.093	.147	.203	.700
	.800	-.102	-.056	-.020	.038	.055	.079	.112	.152	.800
	.850	-.081	-.047	-.017	.029	.044	.062	.083	.116	.850
	.925	-.037	-.016	.002	.037	.047	.060	.058	.072	.925
	53.4- percent - semispan station									
Upper surface	.010	.251	.143	-.372	-.863	-.950	-1.029	-1.276	-1.188	.010
	.030	.207	.032	-.359	-.861	-.954	-1.033	-1.262	-1.192	.030
	.060	.145	-.041	-.363	-.857	-.954	-1.032	-1.256	-1.190	.060
	.100	.098	-.067	-.332			-1.052	-1.234	-1.198	.100
	.200	.046	-.090	-.263	-.951	-1.149	-1.089	-1.277	-1.156	.200
	.300	.020	-.093	-.230	-.803	-1.207	-1.350	-1.093	-1.113	.300
	.400	.001	-.096	-.210	-.277	-.941	-1.280	-.997	-1.067	.400
	.500	-.017	-.101	-.196	-.155	-.351	-.820	-1.100	-.988	.500
	.600	-.021	-.092	-.172	-.165	-.220	-.254	-.977	-.911	.600
	.700	-.013	-.068	-.126	-.124	-.082	-.019	-.823	-.832	.700
	.800	-.001	-.038	-.077	-.068	-.034	-.041	-.665	-.758	.800
	.900	.017	-.005	-.028	-.016	.008	-.141	-.516	-.686	.900
Lower surface	.025				.256	.281	.303	.345	.373	.025
	.060	-.553	-.138	.121	.232	.268	.307	.363	.407	.060
	.100	-.613	-.084	.098	.209	.244	.284	.349	.404	.100
	.200	-.720		.071	.179	.212	.250	.321	.384	.200
	.300	-.384	-.064	.034	.137	.172	.211	.283	.350	.300
	.400	-.005	-.066	.033	.110	.144	.185	.249	.311	.400
	.500	-.092	-.087	-.010	.076	.110	.138	.200	.262	.500
	.600	-.107	-.081	-.017	.060	.087	.109	.161	.215	.600
	.700	-.084	-.061	-.010	.049	.073	.085	.127	.169	.700
	.800	-.048	-.034	.002	.046	.064	.060	.086	.113	.800
	.850	-.023	-.016	.011	.046	.059	.045	.060	.074	.850
	80.1 - percent -semispan station									
Upper surface	.030	.226	.081	-.441	-.664	-.714	-.797	-.708	-.744	.030
	.060	.186	.019	-.425	-.656	-.707	-.787	-.709	-.744	.060
	.100	.149	-.019	-.410	-.644	-.702	-.776	-.713	-.736	.100
	.200	.100	-.070	-.339	-.629	-.675	-.746	-.703	-.733	.200
	.300	.042	-.098	-.313	-.610	-.665	-.716	-.688	-.726	.300
	.400	.017	-.111	-.293	-.591	-.659	-.702	-.673	-.712	.400
	.500	-.002	-.110	-.254	-.574	-.641	-.693	-.659	-.700	.500
	.600	-.008	-.094	-.203	-.563	-.645	-.630	-.642	-.685	.600
	.700	-.001	-.061	-.136	-.548	-.670	-.564	-.604	-.666	.700
	.850	-.004	-.022	-.063	-.503	-.692	-.522	-.546	-.643	.850
Lower surface	.045	-.533	-.305	.055	.200	.225	.238	.254	.277	.045
	.150	-.529	-.266	.071	.178	.209	.237	.272	.305	.150
	.250	-.543	-.266	.055	.152	.184	.210	.244	.282	.250
	.350	-.592	-.126	.042	.138	.158	.185	.215	.251	.350
	.450	-.634	-.010	.041	.117	.141	.168	.191	.221	.450
	.550	-.659	-.018	.017	.083	.101	.123	.143	.173	.550
	.650	-.627	-.020	.011	.059	.077	.093	.100	.125	.650
	.750	-.518	-.011	.011	.037	.045	.056	.051	.071	.750

TABLE II.- PRESSURE COEFFICIENTS FOR WING IN PRESENCE OF BASIC BODY - Continued

(c) M = 0.90

	x/c	α = −4°	α = 0°	α = 4°	α = 8°	α = 10°	α = 12°	α = 16°	α = 20°	x/c
	26.7– percent-semispan station									
Upper surface	.010	.271	.107	-.246	-.908	-1.088	-1.170	-1.267	-1.501	.010
	.030	.182	.006	-.251	-.706	-1.080	-1.202	-1.301	-1.494	.030
	.060	.116	-.028	-.205	-.469	-1.015	-1.297	-1.499	-1.554	.060
	.100	.089	-.033	-.158	-.379	-.270	-.607	-1.431	-1.570	.100
	.150	.063	-.058	-.176	-.354	-.307	-.366	-.976	-1.386	.150
	.200	.038	-.071	-.183	-.351	-.363	-.417	-.662	-1.099	.200
	.300	.026	-.072	-.172	-.298	-.325	-.391	-.535	-.741	.300
	.400	-.003		-.196	-.313	-.344	-.424	-.540	-.620	.400
	.500	-.022	-.108	-.208	-.337	-.387	-.453	-.576	-.660	.500
	.600	-.032	-.108	-.193	-.324	-.392	-.439	-.539	-.610	.600
	.700	-.039	-.101	-.171	-.278	-.360	-.419	-.352	-.560	.700
	.800	-.026	-.076	-.122	-.168	-.201	-.257	-.197	-.466	.800
	.900	.002	-.027	-.046	-.055	-.055	-.081	-.186	-.396	.900
	.950	.007	-.012	-.021	-.026	-.021	-.047	-.192	-.359	.950
Lower surface	.025	-.621	-.044	.117	.271	.332	.389	.494	.590	.025
	.060	-.105	-.007	.109	.248	.310	.362	.474	.582	.060
	.100	-.164	-.029	.077	.220	.279	.330	.444	.556	.100
	.150	-.161	-.045	.059	.191	.244	.293	.403	.516	.150
	.200	-.179	-.065	.039	.160	.212	.261	.370	.482	.200
	.300	-.176	-.074	.022	.129	.178	.223	.321	.426	.300
	.400	-.194	-.094	-.007	.095	.138	.180	.274	.372	.400
	.500	-.220	-.119	-.036	.059	.099	.139	.223	.314	.500
	.600	-.218	-.123	-.045	.039	.075	.112	.186	.267	.600
	.700	-.196	-.110	-.043	.030	.064	.094	.155	.223	.700
	.800	-.131	-.078	-.031	.029	.056	.078	.120	.175	.800
	.850	-.098	-.065	-.027	.020	.044	.064	.087	.137	.850
	.925	-.040	-.023	.000	.030	.049	.060	.055	.088	.925
	53.4– percent-semispan station									
Upper surface	.010	.248		-.337	-.859	-.914	-.972	-1.116	-1.024	.010
	.030	.200	.026	-.356	-.858	-.916	-.971	-1.111	-1.028	.030
	.060	.137	-.050	-.379	-.857	-.915	-.974	-1.129	-1.035	.060
	.100	.091	-.077	-.342			-1.005	-1.121	-1.039	.100
	.200	.040	-.103	-.274	-.886	-1.036	-1.065	-1.274	-1.024	.200
	.300	.011	-.112	-.260	-.827	-1.107	-1.195	-1.262	-1.004	.300
	.400	-.010	-.118	-.242	-.405	-1.027	-1.224	-1.238	-.967	.400
	.500	-.029	-.127	-.247	-.275	-.661	-1.129	-1.169	-.902	.500
	.600	-.029	-.118	-.221	-.292	-.366	-.879	-1.126	-.850	.600
	.700	-.024	-.090	-.159	-.157	-.237	-.474	-.891	-.802	.700
	.800	-.009	-.054	-.087	-.076	-.011	-.056	-.745	-.754	.800
	.900	.015	-.012	-.023	-.017	.032	.056	-.593	-.711	.900
Lower surface	.025				.246	.277	.299	.362	.407	.025
	.060	-.547	-.151	.112	.225	.265	.302	.370	.430	.060
	.100	-.599	-.099	.091	.203	.242	.282	.355	.424	.100
	.200	-.690	-.050	.063	.171	.210	.242	.321	.398	.200
	.300	-.456	-.085	.022	.129	.167	.203	.282	.364	.300
	.400	-.046	-.091	.021	.100	.141	.180	.249	.327	.400
	.500	-.143	-.117	-.025	.067	.104	.136	.206	.280	.500
	.600	-.167	-.111	-.031	.051	.082	.112	.171	.237	.600
	.700	-.125	-.085	-.023	.042	.071	.096	.141	.194	.700
	.800	-.062	-.046	-.007	.041	.064	.081	.105	.146	.800
	.850	-.029	-.024	.005	.041	.062	.077	.080	.112	.850
	80.1 – percent-semispan station									
Upper surface	.030	.212	.064	-.458	-.794	-.891	-.985	-.696	-.764	.030
	.060	.171	.000	-.471	-.791	-.881	-.976	-.679	-.759	.060
	.100	.136	-.041	-.454	-.771	-.874	-.971	-.656	-.743	.100
	.200	.088	-.092	-.418	-.757	-.866	-.982	-.616	-.744	.200
	.300	.031	-.123	-.358	-.728	-.832	-.962	-.592	-.736	.300
	.400	.006	-.140	-.333	-.689	-.794	-.903	-.579	-.729	.400
	.500	-.011	-.136	-.331	-.654	-.763	-.848	-.580	-.714	.500
	.600	-.013	-.116	-.240	-.617	-.751	-.782	-.592	-.701	.600
	.700	-.003	-.078	-.144	-.588	-.733	-.725	-.618	-.680	.700
	.850	-.001	-.026	-.051	-.543	-.696	-.723	-.641	-.662	.850
Lower surface	.045	-.593	-.327	.036	.180	.205	.215	.265	.288	.045
	.150	-.585	-.289	.056	.164	.192	.219	.281	.317	.150
	.250	-.603	-.306	.042	.141	.173	.195	.257	.297	.250
	.350	-.641	-.176	.031	.128	.154	.174	.228	.268	.350
	.450	-.668	-.048	.027	.112	.135	.157	.202	.245	.450
	.550	-.681	-.044	.003	.080	.104	.122	.161	.200	.550
	.650	-.654	-.038	.002	.058	.083	.101	.125	.162	.650
	.750	-.562	-.023	.005	.041	.059	.068	.082	.117	.750

TABLE II.- PRESSURE COEFFICIENTS FOR WING IN PRESENCE OF BASIC BODY - Continued

(d) M = 0.94

	x/c	α = −4°	α = 0°	α = 4°	α = 8°	α = 10°	α = 12°	α = 16°	α = 20°	x/c
	26.7– percent -semispan station									
Upper surface	.010	.276	.112	-.221	-.845	-1.036	-1.095	-1.173	-1.379	.010
	.030	.185	.008	-.235	-.642	-1.003	-1.114	-1.201	-1.372	.030
	.060	.118	-.023	-.193	-.438	-.918	-1.184	-1.383	-1.435	.060
	.100	.093	-.028	-.141	-.338	-.268	-.592	-1.330	-1.455	.100
	.150	.066	-.057	-.169	-.321	-.281	-.336	-.904	-1.297	.150
	.200	.039	-.070	-.181	-.335	-.336	-.388	-.606	-1.030	.200
	.300	.025	-.074	-.173	-.296	-.306	-.366	-.499	-.728	.300
	.400	-.007	-.101	-.203	-.302	-.339	-.406	-.509	-.645	.400
	.500	-.031	-.122	-.227	-.329	-.385	-.443	-.544	-.585	.500
	.600	-.041	-.125	-.224	-.337	-.389	-.434	-.535	-.536	.600
	.700	-.051	-.127	-.232	-.353	-.403	-.439	-.558	-.623	.700
	.800	-.036	-.093	-.172	-.332	-.391	-.417	-.548	-.578	.800
	.900	-.005	-.030	-.048	-.117	-.219	-.232	-.435	-.486	.900
	.950	.005	-.012	-.014	-.044	-.109	-.169	-.353	-.473	.950
Lower surface	.025	-.540	-.043	.119	.275	.340	.397	.510	.609	.025
	.060	-.093	-.005	.111	.251	.315	.370	.488	.594	.060
	.100	-.169	-.026	.078	.222	.285	.338	.456	.570	.100
	.150	-.152	-.043	.063	.194	.248	.301	.417	.531	.150
	.200	-.168	-.066	.039	.164	.216	.267	.382	.498	.200
	.300	-.171	-.075	.019	.131	.181	.229	.332	.438	.300
	.400	-.200	-.100	-.012	.095	.139	.183	.282	.386	.400
	.500	-.231	-.132	-.044	.056	.098	.138	.231	.328	.500
	.600	-.240	-.142	-.058	.033	.071	.111	.192	.283	.600
	.700	-.263	-.139	-.057	.023	.058	.090	.163	.241	.700
	.800	-.216	-.096	-.042	.022	.049	.071	.131	.196	.800
	.850	-.153	-.074	-.036	.014	.035	.053	.099	.161	.850
	.925	-.042	-.025	-.005	.026	.038	.044	.067	.113	.925
	53.4– percent - semispan station									
Upper surface	.010	.245	.134	-.313	-.842	-.886	-.914	-1.021	-1.083	.010
	.030	.196	.025	-.353	-.842	-.887	-.910	-1.019	-1.075	.030
	.060	.131	-.053	-.369	-.837	-.867	-.910	-1.034	-1.080	.060
	.100	.085	-.079	-.338			-.939	-1.038	-1.058	.100
	.200	.033	-.112	-.269	-.837	-.977	-1.008	-1.211	-1.066	.200
	.300	.001	-.122	-.273	-.760	-1.038	-1.109	-1.197	-1.035	.300
	.400	-.021	-.132	-.276	-.389	-.983	-1.134	-1.199	-1.021	.400
	.500	-.043	-.153	-.273	-.292	-.683	-1.067	-1.192	-.972	.500
	.600	-.047	-.146	-.290	-.346	-.363	-.887	-1.286	-.927	.600
	.700	-.037	-.111	-.243	-.364	-.371	-.526	-1.170	-.878	.700
	.800	-.018	-.063	-.114	-.336	-.373	-.436	-1.016	-.828	.800
	.900	.010	-.011	-.006	-.010	-.205	-.304	-.782	-.776	.900
Lower surface	.025				.244	.280	.303	.374	.421	.025
	.060	-.536	-.156	.107	.224	.265	.303	.380	.446	.060
	.100	-.586	-.105	.088	.200	.244	.284	.364	.437	.100
	.200	-.668	-.057	.056	.167	.208	.244	.328	.412	.200
	.300	-.468	-.095	.013	.122	.166	.203	.289	.378	.300
	.400	-.063	-.106	.010	.095	.137	.179	.257	.341	.400
	.500	-.181	-.146	-.040	.060	.099	.133	.211	.295	.500
	.600	-.235	-.142	-.048	.041	.077	.106	.176	.255	.600
	.700	-.241	-.104	-.038	.033	.063	.088	.146	.216	.700
	.800	-.118	-.055	-.015	.035	.057	.074	.114	.172	.800
	.850	-.021	-.023	.000	.036	.055	.066	.092	.141	.850
	80.1 – percent -semispan station									
Upper surface	.030	.200	.050	-.457	-.798	-.866	-.937	-1.076	-.879	.030
	.060	.158	-.016	-.482	-.796	-.857	-.930	-1.069	-.875	.060
	.100	.119	-.059	-.473	-.775	-.850	-.925	-1.055	-.853	.100
	.200	.073	-.115	-.460	-.766	-.854	-.941	-1.060	-.837	.200
	.300	.014	-.142	-.414	-.763	-.851	-.934	-1.055	-.821	.300
	.400	-.011	-.161	-.393	-.756	-.860	-.940	-1.052	-.813	.400
	.500	-.026	-.165	-.412	-.736	-.879	-.956	-1.054	-.798	.500
	.600	-.024	-.128	-.353	-.672	-.871	-.964	-1.055	-.780	.600
	.700	-.010	-.085	-.187	-.588	-.784	-.857	-.978	-.760	.700
	.850	-.005	-.026	-.019	-.487	-.689	-.783	-.875	-.735	.850
Lower surface	.045	-.625	-.347	.014	.167	.194	.209	.245	.295	.045
	.150	-.625	-.307	.037	.152	.180	.211	.266	.322	.150
	.250	-.651	-.341	.023	.126	.160	.187	.245	.306	.250
	.350	-.683	-.216	.014	.115	.140	.165	.218	.282	.350
	.450	-.709	-.079	.010	.098	.124	.148	.199	.261	.450
	.550	-.718	-.067	-.013	.066	.090	.113	.160	.220	.550
	.650	-.667	-.044	-.009	.050	.074	.090	.132	.186	.650
	.750	-.573	-.024	-.003	.036	.052	.063	.098	.147	.750

TABLE II.- PRESSURE COEFFICIENTS FOR WING IN PRESENCE OF BASIC BODY - Continued

(e) M = 0.98

	x/c	α = -4°	α = 0°	α = 4°	α = 8°	α = 10°	α = 12°	α = 16°	α = 20°	x/c
	26.7- percent-semispan station									
Upper surface	.010	.287	.126	-.193	-.826	-.979	-1.019	-1.098	-1.271	.010
	.030	.197	.021	-.209	-.576	-.924	-1.031	-1.109	-1.262	.030
	.060	.131	-.013	-.180	-.401	-.820	-1.083	-1.271	-1.324	.060
	.100	.105	-.016	-.124	-.317	-.251	-.579	-1.233	-1.344	.100
	.150	.078	-.052	-.147	-.288	-.252	-.288	-.821	-1.198	.150
	.200	.049	-.066	-.176	-.301	-.309	-.352	-.539	-.959	.200
	.300	.033	-.066	-.160	-.269	-.278	-.336	-.443	-.670	.300
	.400	-.003	-.101	-.191	-.283	-.309	-.372	-.462	-.608	.400
	.500	-.034	-.131	-.219	-.318	-.356	-.411	-.497	-.625	.500
	.600	-.049	-.136	-.220	-.324	-.368	-.407	-.489	-.605	.600
	.700	-.081	-.170	-.247	-.342	-.382	-.417	-.513	-.612	.700
	.800	-.078	-.166	-.250	-.340	-.379	-.425	-.512	-.611	.800
	.900	-.069	-.134	-.237	-.327	-.366	-.404	-.496	-.593	.900
	.950	-.067	-.094	-.197	-.314	-.352	-.379	-.475	-.588	.950
Lower surface	.025	-.439	-.035	.127	.287	.355	.413	.529	.636	.025
	.060	-.106	.003	.120	.265	.329	.386	.507	.622	.060
	.100	-.149	-.018	.085	.232	.299	.354	.474	.595	.100
	.150	-.132	-.036	.071	.206	.262	.316	.433	.556	.150
	.200	-.140	-.060	.044	.174	.229	.284	.400	.521	.200
	.300	-.163	-.069	.024	.140	.194	.242	.350	.462	.300
	.400	-.195	-.097	-.009	.102	.149	.196	.301	.410	.400
	.500	-.220	-.136	-.048	.059	.106	.150	.248	.352	.500
	.600	-.233	-.151	-.068	.032	.077	.120	.210	.308	.600
	.700	-.259	-.183	-.093	.016	.061	.098	.183	.269	.700
	.800	-.257	-.172	-.089	.008	.046	.078	.150	.230	.800
	.850	-.263	-.187	-.099	-.012	.024	.054	.121	.194	.850
	.925	-.266	-.165	-.096	-.015	.013	.034	.089	.150	.925
	53.4- percent-semispan station									
Upper surface	.010	.252	.140	-.281	-.820	-.839	-.855	-.942	-1.077	.010
	.030	.199	.032	-.337	-.823	-.839	-.852	-.938	-1.073	.030
	.060	.136	-.048	-.366	-.807	-.839	-.852	-.950	-1.092	.060
	.100	.088	-.072	-.326			-.873	-.963	-1.091	.100
	.200	.034	-.105	-.265	-.761	-.913	-.948	-1.112	-1.143	.200
	.300	-.000	-.134	-.259	-.665	-.970	-1.011	-1.114	-1.163	.300
	.400	-.028	-.144	-.264	-.415	-.931	-1.038	-1.129	-1.161	.400
	.500	-.067	-.163	-.272	-.301	-.654	-.998	-1.131	-1.130	.500
	.600	-.082	-.192	-.292	-.338	-.340	-.859	-1.225	-1.146	.600
	.700	-.080	-.196	-.297	-.366	-.353	-.518	-1.111	-1.162	.700
	.800	-.071	-.186	-.282	-.361	-.360	-.421	-.972	-1.146	.800
	.900	-.066	-.151	-.275	-.357	-.369	-.394	-.818	-1.103	.900
Lower surface	.025				.248	.290	.317	.393	.443	.025
	.060	-.512	-.162	.108	.231	.274	.313	.399	.465	.060
	.100	-.553	-.108	.086	.206	.252	.294	.381	.455	.100
	.200	-.633	-.054	.054	.172	.217	.256	.345	.434	.200
	.300	-.461	-.100	.008	.124	.172	.213	.306	.398	.300
	.400	-.054	-.109	.004	.092	.141	.187	.274	.363	.400
	.500	-.179	-.161	-.061	.054	.101	.139	.228	.318	.500
	.600	-.232	-.183	-.085	.031	.076	.112	.196	.280	.600
	.700	-.255	-.196	-.095	.017	.059	.092	.167	.242	.700
	.800	-.274	-.195	-.087	.012	.046	.073	.137	.203	.800
	.850	-.266	-.184	-.075	.005	.036	.060	.117	.175	.850
	80.1 - percent-semispan station									
Upper surface	.030	.190	.042	-.424	-.760	-.815	-.877	-1.008	-1.064	.030
	.060	.141	-.031	-.466	-.755	-.810	-.870	-1.002	-1.063	.060
	.100	.100	-.072	-.459	-.731	-.806	-.867	-.996	-1.031	.100
	.200	.052	-.148	-.455	-.715	-.807	-.881	-1.007	-1.057	.200
	.300	-.018	-.178	-.424	-.718	-.807	-.874	-1.014	-1.052	.300
	.400	-.051	-.205	-.409	-.713	-.816	-.878	-1.022	-1.084	.400
	.500	-.084	-.257	-.427	-.716	-.832	-.893	-1.026	-1.067	.500
	.600	-.080	-.251	-.418	-.707	-.846	-.908	-1.025	-1.087	.600
	.700	-.064	-.226	-.399	-.694	-.866	-.931	-1.037	-1.082	.700
	.850	-.060	-.179	-.351	-.618	-.790	-.911	-.934	-1.087	.850
Lower surface	.045	-.603	-.359	-.024	.155	.196	.217	.262	.308	.045
	.150	-.601	-.330	.005	.140	.181	.216	.282	.333	.150
	.250	-.625	-.382	-.014	.112	.156	.193	.262	.324	.250
	.350	-.657	-.288	-.025	.100	.139	.169	.238	.302	.350
	.450	-.684	-.150	-.040	.080	.117	.151	.217	.280	.450
	.550	-.720	-.156	-.080	.043	.083	.114	.182	.242	.550
	.650	-.734	-.181	-.088	.029	.066	.096	.155	.212	.650
	.750	-.714	-.213	-.099	.016	.048	.071	.126	.178	.750

TABLE II.- PRESSURE COEFFICIENTS FOR WING IN PRESENCE OF BASIC BODY - Continued

(f) M = 1.03

	x/c	α = −4°	α = 0°	α = 4°	α = 8°	α = 10°	α = 12°	α = 16°	α = 20°	x/c
	26.7- percent-semispan station									
Upper surface	.010	.275	.112	-.188	-.820	-.891	-.942	-.993	-1.141	.010
	.030	.183	.007	-.209	-.532	-.807	-.929	-1.003	-1.126	.030
	.060	.123	-.028	-.192	-.378	-.696	-.955	-1.145	-1.191	.060
	.100	.102	-.018	-.139	-.290	-.362	-.632	-1.127	-1.211	.100
	.150	.072	-.052	-.139	-.260	-.226	-.237	-.760	-1.060	.150
	.200	.043	-.075	-.164	-.271	-.280	-.303	-.472	-.824	.200
	.300	.035	-.065	-.147	-.236	-.253	-.287	-.372	-.569	.300
	.400	.006	-.093	-.176	-.251	-.282	-.326	-.390	-.529	.400
	.500	-.029	-.118	-.202	-.283	-.326	-.361	-.424	-.539	.500
	.600	-.035	-.124	-.205	-.290	-.337	-.355	-.414	-.524	.600
	.700	-.071	-.154	-.233	-.303	-.345	-.365	-.434	-.530	.700
	.800	-.068	-.152	-.237	-.298	-.341	-.369	-.434	-.531	.800
	.900	-.071	-.151	-.230	-.285	-.329	-.353	-.422	-.520	.900
	.950	-.076	-.152	-.224	-.280	-.321	-.333	-.412	-.516	.950
Lower surface	.025	-.428	-.043	.116	.290	.351	.428	.571	.670	.025
	.060	-.077	.002	.114	.272	.325	.402	.548	.657	.060
	.100	-.152	-.047	.080	.241	.298	.373	.516	.633	.100
	.150	-.128	-.029	.073	.221	.266	.337	.475	.592	.150
	.200	-.131	-.045	.047	.190	.237	.306	.441	.558	.200
	.300	-.148	-.061	.031	.163	.205	.269	.395	.503	.300
	.400	-.179	-.091	.002	.128	.164	.229	.347	.449	.400
	.500	-.208	-.126	-.041	.089	.127	.188	.296	.394	.500
	.600	-.216	-.135	-.051	.064	.104	.161	.261	.353	.600
	.700	-.241	-.162	-.081	.050	.092	.145	.235	.316	.700
	.800	-.242	-.157	-.075	.044	.084	.129	.209	.282	.800
	.850	-.243	-.164	-.085	.025	.068	.109	.181	.248	.850
	.925	-.254	-.176	-.098	.023	.061	.095	.152	.209	.925
	53.4- percent-semispan station									
Upper surface	.010	.257	.140	-.249	-.801	-.793	-.785	-.848	-.973	.010
	.030	.202	.039	-.310	-.804	-.792	-.780	-.843	-.969	.030
	.060	.140	-.043	-.344	-.772	-.789	-.778	-.855	-.987	.060
	.100	.093	-.072	-.317			-.793	-.863	-.989	.100
	.200	.044	-.090	-.245	-.682	-.831	-.870	-1.003	-1.048	.200
	.300	.003	-.111	-.237	-.569	-.880	-.920	-1.011	-1.105	.300
	.400	-.014	-.133	-.244	-.375	-.856	-.938	-1.018	-1.087	.400
	.500	-.047	-.145	-.256	-.283	-.630	-.925	-1.021	-1.077	.500
	.600	-.072	-.170	-.263	-.306	-.309	-.798	-1.113	-1.117	.600
	.700	-.078	-.179	-.276	-.329	-.315	-.475	-1.012	-1.139	.700
	.800	-.072	-.171	-.264	-.323	-.323	-.373	-.886	-1.099	.800
	.900	-.068	-.166	-.257	-.319	-.331	-.344	-.741	-1.039	.900
Lower surface	.025				.270	.302	.345	.438	.487	.025
	.060	-.484	-.152	.114	.253	.289	.342	.443	.507	.060
	.100	-.519	-.100	.093	.231	.265	.324	.426	.499	.100
	.200	-.588	-.049	.061	.200	.233	.291	.390	.473	.200
	.300	-.437	-.087	.014	.159	.192	.251	.354	.439	.300
	.400	-.044	-.094	.020	.124	.163	.229	.322	.406	.400
	.500	-.165	-.146	-.043	.087	.129	.183	.278	.364	.500
	.600	-.214	-.165	-.068	.066	.110	.161	.248	.329	.600
	.700	-.237	-.174	-.083	.055	.100	.144	.222	.295	.700
	.800	-.254	-.181	-.083	.050	.092	.130	.198	.258	.800
	.850	-.249	-.173	-.078	.046	.087	.121	.178	.233	.850
	80.1 - percent-semispan station									
Upper surface	.030	.202	.061	-.373	-.704	-.759	-.800	-.904	-.971	.030
	.060	.153	-.015	-.426	-.699	-.757	-.787	-.898	-.973	.060
	.100	.111	-.061	-.430	-.675	-.752	-.783	-.894	-.966	.100
	.200	.061	-.137	-.431	-.661	-.751	-.797	-.904	-.980	.200
	.300	-.003	-.159	-.389	-.663	-.745	-.787	-.912	-.980	.300
	.400	-.038	-.178	-.387	-.659	-.752	-.793	-.922	-.989	.400
	.500	-.085	-.225	-.403	-.663	-.762	-.808	-.921	-.978	.500
	.600	-.080	-.233	-.395	-.652	-.773	-.820	-.915	-1.016	.600
	.700	-.068	-.216	-.365	-.642	-.785	-.841	-.922	-1.050	.700
	.850	-.053	-.184	-.329	-.581	-.768	-.850	-.905	-1.082	.850
Lower surface	.045	-.567	-.340	-.007	.187	.224	.261	.312	.356	.045
	.150	-.562	-.307	.019	.172	.209	.256	.333	.381	.150
	.250	-.583	-.349	-.004	.147	.188	.238	.311	.370	.250
	.350	-.613	-.275	-.017	.135	.174	.218	.290	.350	.350
	.450	-.635	-.143	-.026	.112	.156	.201	.272	.331	.450
	.550	-.667	-.138	-.066	.078	.124	.167	.237	.294	.550
	.650	-.683	-.158	-.076	.069	.116	.152	.214	.267	.650
	.750	-.671	-.195	-.091	.059	.100	.133	.187	.236	.750

TABLE II.- PRESSURE COEFFICIENTS FOR WING IN PRESENCE OF BASIC BODY - Continued

(g) M = 1.12

	x/c	α = −4°	α = 0°	α = 4°	α = 8°	α = 10°	α = 12°	α = 16°	α = 20°	x/c
	26.7– percent -semispan station									
Upper surface	.010	.275	.134	-.140	-.737	-.818	-.823	-.854	-.947	.010
	.030	.176	.030	-.159	-.459	-.633	-.770	-.855	-.931	.030
	.060	.111	-.012	-.151	-.317	-.482	-.759	-.930	-1.002	.060
	.100	.108	.002	-.115	-.242	-.403	-.652	-.969	-1.013	.100
	.150	.087	-.018	-.102	-.221	-.273	-.217	-.722	-.906	.150
	.200	.050	-.042	-.128	-.235	-.258	-.266	-.435	-.738	.200
	.300	.057	-.027	-.110	-.197	-.218	-.251	-.324	-.496	.300
	.400	.036	-.053	-.136	-.210	-.241	-.278	-.335	-.445	.400
	.500	.004	-.085	-.155	-.241	-.276	-.306	-.363	-.450	.500
	.600	-.005	-.091	-.156	-.244	-.274	-.293	-.351	-.431	.600
	.700	-.036	-.113	-.181	-.248	-.274	-.299	-.364	-.441	.700
	.800	-.026	-.105	-.185	-.237	-.266	-.300	-.365	-.432	.800
	.900	-.017	-.091	-.170	-.229	-.260	-.286	-.360	-.413	.900
	.950	-.021	-.092	-.163	-.223	-.252	-.269	-.354	-.403	.950
Lower surface	.025	-.278	-.007	.136	.266	.339	.410	.559	.711	.025
	.060	-.125	.030	.137	.248	.319	.386	.538	.687	.060
	.100	-.124	-.027	.081	.216	.291	.361	.511	.656	.100
	.150	-.104	-.005	.101	.213	.271	.333	.476	.620	.150
	.200	-.095	-.009	.090	.188	.244	.303	.445	.586	.200
	.300	-.113	-.028	.067	.173	.227	.281	.409	.534	.300
	.400	-.139	-.066	.030	.147	.198	.249	.368	.482	.400
	.500	-.162	-.092	-.004	.100	.161	.214	.326	.434	.500
	.600	-.164	-.096	-.019	.094	.147	.190	.298	.400	.600
	.700	-.179	-.115	-.045	.069	.122	.174	.281	.372	.700
	.800	-.185	-.106	-.039	.082	.129	.172	.266	.347	.800
	.850	-.185	-.104	-.030	.068	.115	.154	.243	.320	.850
	.925	-.187	-.112	-.032	.054	.102	.142	.228	.291	.925
	53.4– percent - semispan station									
Upper surface	.010	.284	.173	-.170	-.852	-.734	-.712	-.734	-.811	.010
	.030	.227	.066	-.240	-.822	-.729	-.707	-.729	-.807	.030
	.060	.162	-.009	-.281	-.770	-.725	-.706	-.736	-.819	.060
	.100	.120	-.032	-.275			-.711	-.755	-.823	.100
	.200	.077	-.063	-.198	-.388	-.693	-.763	-.790	-.867	.200
	.300	.043	-.075	-.180	-.337	-.695	-.805	-.854	-.935	.300
	.400	.010	-.091	-.181	-.312	-.685	-.823	-.906	-.916	.400
	.500	-.005	-.101	-.193	-.300	-.535	-.823	-.932	-.904	.500
	.600	-.024	-.105	-.189	-.295	-.279	-.703	-.961	-.921	.600
	.700	-.032	-.118	-.205	-.300	-.259	-.385	-.890	-.970	.700
	.800	-.024	-.115	-.201	-.293	-.267	-.305	-.802	-.938	.800
	.900	-.023	-.117	-.197	-.285	-.274	-.287	-.667	-.893	.900
Lower surface	.025			-.010	.270	.322	.361	.463	.534	.025
	.060	-.413	-.102	.138	.260	.315	.363	.469	.555	.060
	.100	-.439	-.062	.126	.239	.293	.344	.450	.544	.100
	.200	-.493	-.017	.094	.210	.264	.317	.418	.521	.200
	.300	-.359	-.049	.049	.165	.231	.277	.384	.491	.300
	.400	.008	-.000	.073	.147	.204	.254	.359	.464	.400
	.500	-.121	-.098	-.004	.112	.164	.210	.321	.424	.500
	.600	-.167	-.115	-.018	.094	.144	.192	.297	.396	.600
	.700	-.181	-.117	-.028	.080	.134	.183	.281	.370	.700
	.800	-.198	-.123	-.041	.078	.134	.177	.265	.340	.800
	.850	-.196	-.120	-.034	.078	.135	.175	.255	.321	.850
	80.1 – percent -semispan station									
Upper surface	.030	.248	.114	-.271	-.632	-.659	-.689	-.757	-.823	.030
	.060	.195	.048	-.326	-.651	-.661	-.679	-.751	-.823	.060
	.100	.155	-.007	-.335	-.604	-.646	-.676	-.748	-.816	.100
	.200	.098	-.076	-.356	-.551	-.635	-.681	-.756	-.827	.200
	.300	.045	-.107	-.317	-.535	-.626	-.673	-.762	-.831	.300
	.400	.013	-.129	-.313	-.524	-.632	-.682	-.769	-.845	.400
	.500	-.036	-.169	-.323	-.524	-.636	-.691	-.777	-.837	.500
	.600	-.042	-.170	-.318	-.517	-.641	-.702	-.789	-.837	.600
	.700	-.034	-.156	-.290	-.504	-.644	-.722	-.805	-.872	.700
	.850	-.015	-.135	-.249	-.456	-.643	-.748	-.795	-.899	.850
Lower surface	.045	-.478	-.271	.043	.204	.257	.292	.361	.418	.045
	.150	-.474	-.236	.069	.190	.243	.286	.375	.441	.150
	.250	-.492	-.266	.051	.166	.219	.270	.359	.433	.250
	.350	-.518	-.223	.035	.166	.214	.254	.342	.418	.350
	.450	-.535	-.112	.024	.150	.197	.235	.326	.400	.450
	.550	-.554	-.081	-.010	.106	.157	.203	.298	.371	.550
	.650	-.564	-.091	-.020	.098	.148	.196	.285	.349	.650
	.750	-.554	-.133	-.045	.080	.137	.187	.265	.323	.750

TABLE II.- PRESSURE COEFFICIENTS FOR WING IN PRESENCE OF BASIC BODY - Concluded

(h) M = 1.43

	x/c	α=-3.6°	α=-0.3°	α = 4.3°	α = 8.8°	α = 11.1°	α = 13.4°	α = 16.0°	α = 19.8°	x/c
	26.7 – percent – semispan station									
Upper surface	.010	.205	.050	-.177	-.393	-.500	-.553	-.579	-.602	.010
	.030	.137	-.001	-.186	-.382	-.489	-.543	-.576	-.599	.030
	.060	.113	.012	-.162	-.376	-.485	-.536	-.566	-.594	.060
	.100	.081	.006	-.089	-.281	-.447	-.524	-.562	-.595	.100
	.150	.075	-.002	-.107	-.204	-.345	-.489	-.551	-.588	.150
	.200	.034	-.043	-.132	-.209	-.207	-.416	-.530	-.569	.200
	.300	.038	-.028	-.100	-.164	-.193	-.211	-.270	-.398	.300
	.400	.023	-.044	-.110	-.168	-.192	-.217	-.253	-.343	.400
	.500	.008	-.045	-.103	-.173	-.196	-.220	-.256	-.328	.500
	.600	-.011	-.067	-.133	-.188	-.205	-.228	-.265	-.325	.600
	.700	-.007	-.062	-.131	-.184	-.201	-.233	-.268	-.318	.700
	.800	-.011	-.075	-.139	-.183	-.212	-.239	-.272	-.311	.800
	.900	-.013	-.071	-.128	-.184	-.207	-.233	-.264	-.300	.900
	.950	-.021	-.077	-.126	-.189	-.211	-.242	-.266	-.297	.950
Lower surface	.025	-.331	-.017	.137	.276	.341	.405	.478	.586	.025
	.060	-.069	.014	.123	.259	.311	.382	.467	.569	.060
	.100	-.065	.008	.097	.244	.302	.361	.428	.532	.100
	.150	-.089	-.005	.083	.200	.262	.318	.386	.499	.150
	.200	-.105	-.021	.065	.173	.240	.300	.372	.475	.200
	.250	-.092	-.037	.074	.151	.213	.269	.339	.432	.250
	.300	-.072	-.029	.058	.151	.195	.248	.322	.448	.300
	.400	-.096	-.039	.031	.136	.198	.246	.312	.425	.400
	.500	-.099	-.047	.033	.137	.183	.239	.314	.416	.500
	.600	-.116	-.068	.020	.110	.167	.228	.299	.400	.600
	.700	-.120	-.074	.003	.101	.146	.203	.271	.369	.700
	.800	-.126	-.077	-.006	.087	.139	.194	.264	.361	.800
	.850	-.109	-.063	.010	.099	.154	.205	.271	.366	.850
	.925	-.130	-.079	-.002	.085	.132	.181	.246	.339	.925
	53.4 – percent – semispan station									
Upper surface	.010	.286	.202	-.036	-.427	-.538	-.574	-.545	-.552	.010
	.030	.229	.110	-.122	-.379	-.465	-.502	-.477	-.487	.030
	.060	.138	.017	-.236	-.423	-.518	-.570	-.538	-.557	.060
	.100	.103	-.003	-.221	-.377	-.456	-.497	-.479	-.496	.100
	.200	.068	-.043	-.225	-.399	-.486	-.525	-.530	-.553	.200
	.300	.040	-.048	-.143	-.369	-.465	-.510	-.537	-.557	.300
	.400	.017	-.057	-.147	-.323	-.446	-.505	-.550	-.573	.400
	.500	.005	-.060	-.150	-.286	-.435	-.466	-.542	-.577	.500
	.600	-.005	-.073	-.156	-.262	-.388	-.422	-.520	-.570	.600
	.700	-.009	-.082	-.163	-.251	-.290	-.387	-.528	-.580	.700
	.800	-.015	-.083	-.158	-.236	-.230	-.366	-.534	-.588	.800
	.900	-.017	-.086	-.154	-.223	-.221	-.354	-.520	-.577	.900
Lower surface	.025	-.314	-.171	.122	.268	.326	.381	.433	.523	.025
	.060	-.316	-.117	.132	.259	.312	.367	.432	.534	.060
	.100	-.316	-.093	.122	.245	.300	.353	.416	.510	.100
	.200	-.297	-.023	.098	.220	.282	.334	.394	.491	.200
	.300	-.223	-.017	.073	.184	.242	.304	.371	.478	.300
	.400	-.132	-.048	.053	.162	.216	.274	.348	.456	.400
	.500	-.082	-.064	.033	.135	.187	.242	.310	.419	.500
	.600	-.117	-.085	.003	.112	.168	.217	.289	.396	.600
	.700	-.133	-.085	.001	.108	.154	.206	.281	.383	.700
	.800	-.135	-.090	-.006	.090	.143	.198	.267	.378	.800
	.850	-.131	-.090	-.009	.091	.142	.199	.270	.379	.850
	80.1 – percent – semispan station									
Upper surface	.030	.221	.123	-.175	-.490	-.575	-.533	-.528	-.547	.030
	.060	.210	.109	-.139	-.485	-.573	-.531	-.525	-.546	.060
	.100	.173	.073	-.105	-.447	-.548	-.532	-.530	-.549	.100
	.200	.105	-.005	-.215	-.449	-.537	-.531	-.530	-.547	.200
	.300	.050	-.073	-.276	-.467	-.536	-.526	-.531	-.552	.300
	.400	-.006	-.124	-.330	-.485	-.531	-.521	-.529	-.552	.400
	.500	-.028	-.132	-.341	-.487	-.525	-.516	-.529	-.552	.500
	.600	-.030	-.128	-.261	-.467	-.518	-.512	-.530	-.552	.600
	.700	-.019	-.113	-.234	-.451	-.504	-.508	-.531	-.553	.700
	.850	-.005	-.094	-.201	-.403	-.478	-.494	-.520	-.516	.850
Lower surface	.045	-.338	-.294	.085	.243	.295	.334	.386	.460	.045
	.150	-.344	-.282	.110	.234	.284	.338	.394	.474	.150
	.250	-.333	.011	.104	.208	.259	.312	.370	.457	.250
	.350	-.350	-.017	.085	.183	.239	.284	.342	.434	.350
	.450	-.357	-.037	.062	.162	.218	.269	.324	.424	.450
	.550	-.352	-.061	.034	.144	.198	.252	.317	.416	.550
	.650	-.341	-.072	.025	.140	.194	.245	.312	.417	.650
	.750	-.307	-.093	.005	.125	.183	.235	.300	.407	.750

TABLE III.- PRESSURE COEFFICIENTS FOR WING IN PRESENCE OF INDENTED BODY

(a) M = 0.60

	x/c	α = −4°	α = 0°	α = 4°	α = 8°	α = 10°	α = 12°	α = 16°	α = 20°	x/c
	26.7– percent -semispan station									
Upper surface	.010	.252	.081	-.323	-1.234	-1.462	-1.750	-2.088	-2.517	.010
	.030	.157	-.035	-.325	-1.266	-1.682	-1.772	-2.215	-2.686	.030
	.060	.104	-.052	-.237	-.417	-1.064	-1.869	-2.508	-2.739	.060
	.100	.076	-.045	-.188	-.301	-.310	-.545	-1.787	-2.602	.100
	.150	.063	-.054	-.175	-.304	-.330	-.361	-.820	-1.596	.150
	.200	.038	-.065	-.172	-.290	-.333	-.376	-.605	-1.116	.200
	.300	.032	-.049	-.141	-.237	-.274	-.324	-.430	-.660	.300
	.400	.018	-.054	-.127	-.208	-.239	-.282	-.349	-.498	.400
	.500	.020	-.038	-.105	-.170	-.197	-.224	-.271	-.450	.500
	.600	.016	-.035	-.087	-.141	-.159	-.182	-.224	-.377	.600
	.700	.006	-.035	-.077	-.118	-.130	-.149	-.208	-.329	.700
	.800	-.008	-.036	-.067	-.095	-.102	-.115	-.201	-.257	.800
	.900	.007	-.010	-.026	-.046	-.046	-.056	-.129	-.130	.900
	.950	.010	.001	-.012	-.025	-.024	-.034	-.090	-.081	.950
Lower surface	.025	-.929	-.075	.105	.265	.320	.362	.431	.490	.025
	.060	-.138	-.023	.096	.238	.294	.345	.434	.523	.060
	.100	-.168	-.035	.070	.207	.265	.313	.412	.506	.100
	.150	-.168	-.056	.064	.183	.239	.285	.383	.487	.150
	.200	-.171	-.059	.040	.156	.208	.254	.350	.455	.200
	.300	-.148	-.052	.030	.132	.179	.221	.308	.405	.300
	.400	-.140	-.057	.016	.109	.152	.189	.268	.357	.400
	.500	-.121	-.053	.010	.091	.127	.161	.232	.309	.500
	.600	-.100	-.044	.010	.079	.109	.143	.199	.266	.600
	.700	-.081	-.035	.008	.068	.096	.118	.164	.222	.700
	.800	-.066	-.032	.000	.049	.070	.088	.116	.164	.800
	.850	-.057	-.031	-.006	.033	.052	.063	.091	.126	.850
	.925	-.027	-.010	.006	.037	.051	.057	.070	.092	.925
	53.4– percent - semispan station									
Upper surface	.010	.254	.156	-.349	-.873	-.963	-1.076	-1.311	-1.321	.010
	.030	.215	.049	-.331	-.879	-.965	-1.087	-1.309	-1.306	.030
	.060	.155	-.020	-.325	-.879	-.965	-1.085	-1.285	-1.285	.060
	.100	.113	-.048	-.291	-.876	-1.008	-1.097	-1.268	-1.298	.100
	.200	.061	-.065	-.222	-.872	-1.257	-1.165	-1.221	-1.225	.200
	.300	.035	-.064	-.184	-.612	-1.148	-1.551	-1.164	-1.186	.300
	.400	.017	-.067	-.161	-.210	-.525	-1.076	-1.508	-1.222	.400
	.500	-.001	-.069	-.147	-.125	-.126	-.488	-1.141	-1.123	.500
	.600	-.007	-.062	-.124	-.121	-.070	-.172	-.876	-1.002	.600
	.700	.000	-.043	-.091	-.090	-.050	-.047	-.632	-.858	.700
	.800	.009	-.022	-.054	-.056	-.027	-.009	-.452	-.740	.800
	.900	.018	.001	-.019	-.020	.001	-.001	-.326	-.618	.900
Lower surface	.025	-.528	-.170	.134	.263	.295	.314	.332	.350	.025
	.060	-.547	-.121	.123	.241	.280	.312	.360	.399	.060
	.100	-.596	-.075	.108	.216	.262	.295	.354	.406	.100
	.200	-.719	-.017	.084	.183	.225	.255	.326	.384	.200
	.300	-.285	-.039	.048	.144	.184	.217	.285	.347	.300
	.400	.007	-.042	.033	.123	.159	.189	.251	.316	.400
	.500	-.048	-.060	.007	.086	.120	.145	.199	.257	.500
	.600	-.062	-.052	.001	.068	.095	.117	.161	.211	.600
	.700	-.054	-.041	.003	.056	.077	.093	.124	.163	.700
	.800	-.029	-.020	.010	.049	.066	.071	.084	.107	.800
	.850	-.013	-.007	.012	.045	.059	.059	.060	.064	.850
	80.1 – percent -semispan station									
Upper surface	.030	.233	.061	-.462	-.576	-.611	-.666	-.683	-.748	.030
	.060	.206	.043	-.356	-.568	-.605	-.659	-.676	-.753	.060
	.100	.175	.025	-.272	-.555	-.598	-.656	-.676	-.760	.100
	.200	.089	-.062	-.298	-.517	-.580	-.627	-.659	-.767	.200
	.300	.045	-.077	-.259	-.479	-.576	-.619	-.656	-.744	.300
	.400	.023	-.085	-.237	-.439	-.554	-.596	-.636	-.714	.400
	.500	.003	-.087	-.206	-.411	-.534	-.559	-.628	-.684	.500
	.600	-.006	-.073	-.167	-.377	-.540	-.506	-.620	-.655	.600
	.700	-.001	-.044	-.114	-.352	-.591	-.469	-.578	-.630	.700
	.850	-.005	-.014	-.055	-.316	-.611	-.584	-.513	-.596	.850
Lower surface	.045	-.491	-.284	.069	.183	.203	.223	.237	.222	.045
	.150	-.479	-.276	.087	.172	.212	.239	.272	.301	.150
	.250	-.498	-.273	.064	.146	.182	.207	.239	.277	.250
	.350	-.545	-.064	.054	.122	.159	.176	.210	.246	.350
	.450	-.602	.000	.032	.099	.126	.140	.168	.202	.450
	.550	-.603	-.004	.021	.069	.088	.102	.120	.153	.550
	.650	-.548	-.012	.015	.054	.065	.074	.088	.109	.650
	.750	-.430	-.005	.013	.035	.034	.033	.034	.049	.750

TABLE III.- PRESSURE COEFFICIENTS FOR WING IN PRESENCE OF INDENTED BODY - Continued

(b) M = 0.80

	x/c	α = -4°	α = 0°	α = 4°	α = 8°	α = 10°	α = 12°	α = 16°	α = 20°	x/c
	26.7- percent-semispan station									
Upper surface	.010	.257	.079	-.302	-1.062	-1.219	-1.308	-1.539	-1.742	.010
	.030	.154	-.042	-.320	-.957	-1.292	-1.424	-1.580	-1.734	.030
	.060	.103	-.056	-.236	-.589	-1.238	-1.579	-1.730	-1.833	.060
	.100	.078	-.050	-.188	-.331	-.343	-.802	-1.799	-1.918	.100
	.150	.066	-.064	-.180	-.320	-.352	-.434	-1.216	-1.665	.150
	.200	.041	-.073	-.182	-.315	-.373	-.436	-.786	-1.261	.200
	.300	.037	-.057	-.152	-.267	-.313	-.365	-.566	-.786	.300
	.400	.025	-.062	-.142	-.244	-.289	-.331	-.444	-.582	.400
	.500	.028	-.046	-.117	-.204	-.243	-.271	-.315	-.587	.500
	.600	.023	-.039	-.099	-.169	-.201	-.222	-.236	-.528	.600
	.700	.010	-.043	-.092	-.145	-.169	-.185	-.236	-.372	.700
	.800	-.006	-.047	-.080	-.116	-.130	-.139	-.302	-.345	.800
	.900	.011	-.012	-.031	-.049	-.056	-.060	-.225	-.258	.900
	.950	.017	-.003	-.013	-.024	-.027	-.029	-.166	-.205	.950
Lower surface	.025	-.788	-.075	.105	.264	.320	.375	.462	.545	.025
	.060	-.112	-.024	.097	.237	.296	.349	.450	.548	.060
	.100	-.167	-.039	.075	.209	.267	.321	.422	.524	.100
	.150	-.173	-.062	.061	.188	.243	.291	.393	.499	.150
	.200	-.179	-.064	.041	.161	.213	.262	.359	.463	.200
	.300	-.161	-.057	.033	.141	.186	.232	.319	.416	.300
	.400	-.156	-.065	.018	.115	.160	.199	.280	.371	.400
	.500	-.138	-.058	.013	.099	.138	.176	.245	.324	.500
	.600	-.113	-.047	.014	.089	.121	.154	.215	.283	.600
	.700	-.096	-.042	.008	.075	.105	.130	.176	.238	.700
	.800	-.080	-.041	-.003	.050	.075	.096	.126	.173	.800
	.850	-.069	-.041	-.009	.034	.055	.074	.091	.130	.850
	.925	-.029	-.010	.007	.040	.055	.068	.066	.087	.925
	53.4- percent-semispan station									
Upper surface	.010	.263	.154	-.340	-.865	-.931	-.998	-1.223	-1.212	.010
	.030	.218	.044	-.340	-.864	-.933	-.998	-1.219	-1.201	.030
	.060	.160	-.029	-.340	-.861	-.939	-1.001	-1.210	-1.179	.060
	.100	.113	-.052	-.310	-.867	-.958	-1.039	-1.199	-1.156	.100
	.200	.064	-.074	-.234	-.933	-1.108	-1.045	-1.266	-1.108	.200
	.300	.042	-.073	-.199	-.737	-1.152	-1.320	-1.100	-1.073	.300
	.400	.021	-.078	-.180	-.211	-.905	-1.245	-.957	-1.029	.400
	.500	.003	-.082	-.166	-.145	-.321	-.782	-1.098	-.948	.500
	.600	-.004	-.076	-.144	-.154	-.118	-.237	-.953	-.883	.600
	.700	.000	-.056	-.106	-.116	-.068	-.003	-.820	-.812	.700
	.800	.009	-.033	-.064	-.064	-.029	-.088	-.664	-.754	.800
	.900	.024	-.003	-.018	-.016	.013	-.200	-.508	-.688	.900
Lower surface	.025	-.528	-.179	.129	.263	.295	.323	.353	.383	.025
	.060	-.543	-.130	.122	.242	.278	.319	.369	.417	.060
	.100	-.580	-.079	.110	.219	.260	.299	.360	.417	.100
	.200	-.695	-.020	.087	.187	.226	.263	.328	.394	.200
	.300	-.375	-.044	.049	.150	.190	.226	.292	.359	.300
	.400	-.011	-.048	.037	.131	.168	.201	.263	.328	.400
	.500	-.065	-.071	.006	.090	.123	.153	.210	.270	.500
	.600	-.082	-.065	-.001	.071	.101	.122	.173	.225	.600
	.700	-.070	-.050	.000	.057	.082	.093	.133	.179	.700
	.800	-.038	-.027	.009	.050	.072	.064	.092	.120	.800
	.850	-.018	-.012	.014	.047	.065	.047	.062	.081	.850
	80.1 - percent-semispan station									
Upper surface	.030	.236	.053	-.499	-.624	-.678	-.751	-.683	-.742	.030
	.060	.213	.038	-.414	-.618	-.668	-.741	-.679	-.744	.060
	.100	.181	.019	-.293	-.604	-.661	-.728	-.680	-.736	.100
	.200	.091	-.079	-.341	-.565	-.634	-.706	-.679	-.730	.200
	.300	.050	-.091	-.294	-.527	-.626	-.678	-.675	-.714	.300
	.400	.025	-.104	-.272	-.489	-.620	-.674	-.663	-.698	.400
	.500	.005	-.104	-.236	-.461	-.597	-.665	-.652	-.687	.500
	.600	-.001	-.088	-.191	-.431	-.592	-.595	-.634	-.672	.600
	.700	.004	-.057	-.126	-.406	-.620	-.517	-.597	-.656	.700
	.850	.000	-.020	-.054	-.371	-.662	-.445	-.536	-.634	.850
Lower surface	.045	-.516	-.288	.061	.183	.209	.218	.236	.227	.045
	.150	-.502	-.275	.086	.176	.215	.242	.276	.304	.150
	.250	-.520	-.307	.061	.152	.187	.212	.242	.281	.250
	.350	-.564	-.104	.053	.131	.167	.190	.215	.253	.350
	.450	-.610	-.018	.030	.107	.133	.157	.178	.214	.450
	.550	-.619	-.014	.019	.074	.099	.117	.136	.172	.550
	.650	-.589	-.019	.013	.060	.080	.095	.101	.130	.650
	.750	-.493	-.010	.014	.039	.049	.059	.054	.075	.750

TABLE III.- PRESSURE COEFFICIENTS FOR WING IN PRESENCE OF INDENTED BODY - Continued

(c) M = 0.90

	x/c	α = −4°	α = 0°	α = 4°	α = 8°	α = 10°	α = 12°	α = 16°	α = 20°	x/c
	26.7– percent -semispan station									
Upper surface	.010	.257	.078	-.276	-.909	-1.060	-1.134	-1.254	-1.394	.010
	.030	.147	-.049	-.310	-.726	-1.053	-1.163	-1.268	-1.391	.030
	.060	.099	-.058	-.238	-.505	-1.023	-1.240	-1.419	-1.471	.060
	.100	.077	-.056	-.185	-.391	-.357	-.961	-1.504	-1.521	.100
	.150	.064	-.070	-.187	-.362	-.342	-.398	-1.161	-1.469	.150
	.200	.035	-.082	-.193	-.353	-.391	-.440	-.719	-1.170	.200
	.300	.035	-.064	-.164	-.307	-.348	-.414	-.575	-.791	.300
	.400	.023	-.073	-.164	-.303	-.341	-.404	-.536	-.685	.400
	.500	.029	-.054	-.138	-.261	-.323	-.387	-.516	-.494	.500
	.600	.024	-.048	-.119	-.216	-.278	-.355	-.408	-.475	.600
	.700	.005	-.058	-.119	-.200	-.245	-.320	-.201	-.651	.700
	.800	-.012	-.065	-.105	-.158	-.185	-.224	-.211	-.519	.800
	.900	.010	-.021	-.039	-.057	-.064	-.071	-.349	-.329	.900
	.950	.016	-.007	-.016	-.028	-.029	-.033	-.290	-.307	.950
Lower surface	.025	-.651		.091	.259	.324	.381	.487	.581	.025
	.060	-.089	-.029	.090	.234	.295	.355	.468	.573	.060
	.100	-.179	-.047	.069	.203	.266	.325	.436	.546	.100
	.150	-.175	-.070	.055	.189	.242	.297	.406	.516	.150
	.200	-.184		.034	.159	.212	.266	.372	.482	.200
	.300	-.171	-.066	.026	.140	.187	.235	.330	.433	.300
	.400	-.177	-.076	.010	.114	.159	.204	.290	.386	.400
	.500	-.160	-.068	.009	.101	.142	.179	.258	.343	.500
	.600	-.135	-.056	.012	.092	.128	.163	.228	.303	.600
	.700	-.132	-.057	.001	.074	.106	.138	.188	.258	.700
	.800	-.111	-.059	-.012	.048	.076	.100	.137	.194	.800
	.850	-.091	-.057	-.020	.029	.055	.077	.098	.150	.850
	.925	-.034	-.017	.003	.039	.056	.071	.067	.104	.925
	53.4– percent - semispan station									
Upper surface	.010	.262	.149	-.317	-.879	-.926	-.973	-1.115	-1.066	.010
	.030	.215	.041	-.342	-.882	-.927	-.972	-1.113	-1.054	.030
	.060	.156	-.034	-.363	-.882	-.930	-.982	-1.119	-1.060	.060
	.100	.111	-.057	-.321	-.885	-.948	-1.017	-1.144	-1.053	.100
	.200	.062	-.080	-.253	-.891	-1.045	-1.098	-1.186	-1.021	.200
	.300	.039	-.083	-.220	-.771	-1.103	-1.193	-1.209	-.995	.300
	.400	.018	-.088	-.199	-.281	-.984	-1.237	-1.148	-.963	.400
	.500	-.001	-.096	-.197	-.203	-.494	-1.098	-1.049	-.885	.500
	.600	-.009	-.097	-.184	-.231	-.250	-.759	-1.054	-.820	.600
	.700	-.006	-.075	-.138	-.162	-.183	-.349	-.763	-.764	.700
	.800	.002	-.046	-.079	-.080	-.066	-.076	-.579	-.726	.800
	.900	.020	-.007	-.020	-.018	.009	.010	-.565	-.685	.900
Lower surface	.025	-.532	-.192	.118	.258	.291	.322	.368	.410	.025
	.060	-.546	-.143	.115	.240	.276	.314	.378	.436	.060
	.100	-.578	-.094	.106	.218	.256	.298	.368	.432	.100
	.200	-.675	-.028	.082	.188	.225	.261	.336	.407	.200
	.300	-.437	-.053	.046	.151	.189	.224	.298	.370	.300
	.400	-.037	-.059	.030	.130	.147	.203	.273	.342	.400
	.500	-.097	-.087	.000	.088	.123	.159	.220	.288	.500
	.600	-.126	-.084	-.011	.069	.102	.132	.185	.244	.600
	.700	-.107	-.069	-.011	.055	.085	.111	.148	.200	.700
	.800	-.058	-.040	.001	.049	.074	.094	.107	.152	.800
	.850	-.027	-.019	.009	.048	.069	.085	.082	.115	.850
	80.1 – percent -semispan station									
Upper surface	.030	.230	.043	-.498	-.697	-.800	-.921	-.641	-.752	.030
	.060	.211	.031	-.485	-.693	-.788	-.904	-.638	-.747	.060
	.100	.176	.011	-.329	-.680	-.779	-.902	-.659	-.736	.100
	.200	.084	-.096	-.399	-.644	-.755	-.890	-.634	-.738	.200
	.300	.044	-.108	-.318	-.606	-.736	-.859	-.636	-.728	.300
	.400	.020	-.124	-.320	-.566	-.716	-.828	-.634	-.715	.400
	.500	.000	-.125	-.300	-.530	-.705	-.787	-.659	-.702	.500
	.600	-.005	-.108	-.224	-.493	-.711	-.722	-.664	-.691	.600
	.700	.005	-.071	-.139	-.467	-.711	-.665	-.653	-.679	.700
	.850	.003	-.026	-.051	-.445	-.702	-.635	-.633	-.666	.850
Lower surface	.045	-.537	-.300	.050	.171	.199	.207	.246	.236	.045
	.130	-.567	-.287	.077	.170	.209	.235	.285	.314	.130
	.250	-.606	-.335	.046	.147	.182	.207	.252	.292	.250
	.350	-.646	-.152	.042	.130	.164	.188	.226	.265	.350
	.450	-.651	-.042	.019	.105	.135	.160	.192	.233	.450
	.550	-.624	-.034	.008	.075	.103	.125	.150	.194	.550
	.650	-.542	-.032	.005	.066	.089	.110	.125	.163	.650
	.750	-.002	-.020	.005	.048	.066	.079	.083	.118	.750

TABLE III.- PRESSURE COEFFICIENTS FOR WING IN PRESENCE OF INDENTED BODY - Continued

(d) M = 0.94

	x/c	α = -4°	α = 0°	α = 4°	α = 8°	α = 10°	α = 12°	α = 16°	α = 20°	x/c
	26.7- percent-semispan station									
Upper surface	.010	.257	.079	-.264	-.852	-1.000	-1.058	-1.153	-1.277	.010
	.030	.145	-.047	-.292	-.642	-.962	-1.079	-1.164	-1.278	.030
	.060	.097	-.057	-.224	-.463	-.904	-1.128	-1.293	-1.357	.060
	.100	.074	-.055	-.178	-.363	-.379	-.942	-1.383	-1.397	.100
	.150	.062	-.068	-.190	-.337	-.334	-.364	-1.103	-1.307	.150
	.200	.033	-.083	-.202	-.348	-.366	-.414	-.673	-1.098	.200
	.300	.033	-.046	-.169	-.315	-.346	-.396	-.535	-.740	.300
	.400	.019	-.076	-.182	-.307	-.342	-.395	-.510	-.669	.400
	.500	.026	-.056	-.151	-.291	-.336	-.389	-.504	-.647	.500
	.600	.023	-.049	-.126	-.258	-.309	-.370	-.493	-.504	.600
	.700	.001	-.066	-.141	-.251	-.307	-.358	-.484	-.339	.700
	.800	-.021	-.079	-.144	-.230	-.321	-.377	-.432	-.503	.800
	.900	.006	-.023	-.044	-.070	-.151	-.234	-.180	-.639	.900
	.950	.012	-.005	-.014	-.027	-.061	-.142	-.127	-.576	.950
Lower surface	.025	-.604	-.070	.091	.259	.327	.389	.504	.601	.025
	.060	-.072	-.033	.092	.234	.299	.362	.479	.590	.060
	.100	-.203	-.048	.069	.201	.269	.331	.448	.561	.100
	.150	-.172	-.071	.056	.190	.244	.302	.418	.535	.150
	.200	-.181	-.075	.035	.159	.214	.271	.382	.498	.200
	.300	-.171	-.070	.028	.138	.188	.240	.341	.447	.300
	.400	-.197	-.084	.009	.112	.159	.206	.300	.403	.400
	.500	-.178	-.072	.010	.100	.142	.182	.268	.359	.500
	.600	-.140	-.059	.014	.091	.128	.164	.239	.322	.600
	.700	-.154	-.067	.002	.072	.105	.138	.203	.280	.700
	.800	-.167	-.076	-.019	.044	.072	.099	.152	.216	.800
	.850	-.142	-.070	-.026	.025	.050	.070	.114	.173	.850
	.925	-.044	-.020	.003	.036	.052	.062	.088	.130	.925
	53.4- percent-semispan station									
Upper surface	.010	.258	.147	-.309	-.908	-.910	-.920	-1.037	-1.118	.010
	.030	.211	.040	-.349	-.914	-.909	-.919	-1.035	-1.105	.030
	.060	.149	-.034	-.365	-.897	-.912	-.929	-1.044	-1.111	.060
	.100	.105	-.060	-.335	-.885	-.920	-.961	-1.068	-1.096	.100
	.200	.058	-.083	-.271	-.847	-.986	-1.050	-1.115	-1.056	.200
	.300	.034	-.085	-.227	-.686	-1.039	-1.119	-1.185	-1.007	.300
	.400	.015	-.090	-.207	-.303	-.969	-1.158	-1.163	-.969	.400
	.500	-.007	-.104	-.207	-.275	-.587	-1.077	-1.130	-.946	.500
	.600	-.020	-.111	-.217	-.297	-.304	-.825	-1.070	-.901	.600
	.700	-.017	-.092	-.191	-.281	-.319	-.474	-.978	-.828	.700
	.800	-.005	-.055	-.105	-.127	-.314	-.374	-.599	-.763	.800
	.900	.015	-.008	-.014	-.006	-.037	-.216	-.570	-.707	.900
Lower surface	.025	-.536	-.197	.115	.255	.291	.325	.381	.427	.025
	.060	-.547	-.148	.114	.238	.276	.316	.389	.453	.060
	.100	-.583	-.101	.105	.215	.256	.299	.376	.448	.100
	.200	-.676	-.031	.081	.185	.225	.264	.343	.422	.200
	.300	-.483	-.054	.046	.146	.187	.227	.305	.388	.300
	.400	-.048	-.062	.031	.126	.167	.204	.279	.360	.400
	.500	-.106	-.096	-.003	.085	.121	.157	.229	.305	.500
	.600	-.160	-.101	-.015	.065	.099	.131	.193	.267	.600
	.700	-.178	-.088	-.018	.052	.079	.107	.161	.225	.700
	.800	-.100	-.048	-.003	.045	.069	.088	.123	.180	.800
	.850	-.033	-.023	.009	.045	.064	.077	.099	.150	.850
	80.1 - percent-semispan station									
Upper surface	.030	.227	.040	-.465	-.758	-.827	-.907	-.982	-.760	.030
	.060	.206	.028	-.484	-.756	-.815	-.888	-.961	-.758	.060
	.100	.170	.006	-.362	-.743	-.813	-.889	-.943	-.761	.100
	.200	.074	-.107	-.416	-.706	-.801	-.889	-.892	-.782	.200
	.300	.034	-.121	-.366	-.672	-.801	-.888	-.847	-.793	.300
	.400	.010	-.144	-.348	-.632	-.801	-.899	-.777	-.800	.400
	.500	-.008	-.140	-.364	-.586	-.800	-.920	-.741	-.792	.500
	.600	-.012	-.118	-.308	-.524	-.760	-.909	-.715	-.775	.600
	.700	-.002	-.078	-.167	-.466	-.720	-.786	-.699	-.750	.700
	.850	-.004	-.024	-.031	-.419	-.654	-.772	-.692	-.730	.850
Lower surface	.045	-.575	-.306	.050	.160	.189	.204	.236	.253	.045
	.150	-.570	-.294	.072	.161	.198	.225	.285	.327	.150
	.250	-.600	-.351	.039	.137	.175	.198	.237	.304	.250
	.350	-.638	-.186	.035	.121	.155	.179	.235	.282	.350
	.450	-.685	-.067	.012	.098	.126	.149	.203	.251	.450
	.550	-.689	-.054	-.001	.068	.095	.117	.168	.215	.550
	.650	-.655	-.041	.000	.062	.084	.103	.146	.188	.650
	.750	-.577	-.022	.002	.048	.064	.076	.110	.149	.750

TABLE III.- PRESSURE COEFFICIENTS FOR WING IN PRESENCE OF INDENTED BODY - Continued

(e) M = 0.98

	x/c	α = −4°	α = 0°	α = 4°	α = 8°	α = 10°	α = 12°	α = 16°	α = 20°	x/c
	26.7- percent -semispan station									
Upper surface	.010	.258	.083	-.237	-.832	-.938	-.978	-1.051	-1.171	.010
	.030	.145	-.051	-.268	-.588	-.863	-.990	-1.066	-1.173	.030
	.060	.098	-.056	-.222	-.422	-.786	-1.019	-1.179	-1.252	.060
	.100	.075		-.185	-.341	-.410	-.853	-1.268	-1.282	.100
	.150	.060	-.076	-.177	-.319	-.304	-.329	-1.025	-1.205	.150
	.200	.030	-.095	-.204	-.322	-.339	-.375	-.612	-1.015	.200
	.300	.031	-.069	-.174	-.301	-.319	-.365	-.481	-.682	.300
	.400	.014	-.097	-.188	-.297	-.319	-.363	-.463	-.614	.400
	.500	.024	-.062	-.178	-.290	-.319	-.364	-.459	-.598	.500
	.600	.021	-.043	-.142	-.261	-.300	-.349	-.451	-.574	.600
	.700	-.010	-.069	-.147	-.250	-.293	-.340	-.444	-.559	.700
	.800	-.053	-.118	-.180	-.277	-.317	-.366	-.462	-.578	.800
	.900	-.028	-.108	-.175	-.283	-.326	-.369	-.477	-.572	.900
	.950	-.016	-.060	-.115	-.246	-.304	-.347	-.471	-.556	.950
Lower surface	.025	-.518	-.071	.086	.265	.339	.405	.523	.627	.025
	.060	-.089	-.029	.088	.240	.308	.377	.499	.613	.060
	.100	-.181	-.058	.063	.208	.279	.344	.468	.585	.100
	.150	-.174	-.077	.048	.196	.257	.317	.435	.556	.150
	.200	-.166	-.075	.028	.163	.225	.285	.401	.519	.200
	.300	-.172	-.069	.021	.142	.198	.253	.359	.471	.300
	.400	-.198	-.103	-.001	.113	.167	.217	.317	.423	.400
	.500	-.199	-.081	.003	.102	.149	.194	.286	.382	.500
	.600	-.160	-.051	.008	.093	.135	.176	.258	.347	.600
	.700	-.159	-.069	-.012	.070	.109	.147	.224	.305	.700
	.800	-.180	-.111	-.051	.031	.067	.103	.170	.246	.800
	.850	-.215	-.148	-.075	.004	.038	.068	.135	.206	.850
	.925	-.215	-.118	-.031	.006	.029	.052	.108	.168	.925
	53.4- percent - semispan station									
Upper surface	.010	.258	.143	-.283	-.994	-.871	-.878	-.957	-1.056	.010
	.030	.207	.035	-.343	-.992	-.874	-.875	-.954	-1.050	.030
	.060	.146	-.042	-.368	-.936	-.877	-.880	-.963	-1.060	.060
	.100	.100	-.071	-.332	-.844	-.880	-.896	-.982	-1.068	.100
	.200	.055	-.104	-.266	-.664	-.920	-.978	-1.030	-1.104	.200
	.300	.031	-.080	-.251	-.459	-.957	-1.035	-1.120	-1.160	.300
	.400	.011	-.081	-.234	-.343	-.902	-1.066	-1.182	-1.138	.400
	.500	-.014	-.104	-.223	-.331	-.550	-1.022	-1.278	-1.109	.500
	.600	-.036	-.124	-.221	-.325	-.302	-.803	-1.243	-1.125	.600
	.700	-.045	-.132	-.217	-.320	-.316	-.457	-1.115	-1.157	.700
	.800	-.044	-.136	-.217	-.312	-.319	-.368	-.941	-1.144	.800
	.900	-.020	-.100	-.194	-.304	-.329	-.359	-.756	-1.078	.900
Lower surface	.025	-.529	-.199	.098	.255	.301	.336	.400	.449	.025
	.060	-.536	-.152	.103	.237	.283	.327	.407	.472	.060
	.100	-.568	-.116	.094	.218	.263	.309	.391	.468	.100
	.200	-.637	-.061	.076	.186	.233	.275	.359	.441	.200
	.300	-.477	-.050	.037	.148	.194	.237	.323	.406	.300
	.400	-.075	-.049	.022	.127	.172	.214	.296	.381	.400
	.500	-.132	-.095	-.016	.082	.127	.166	.246	.327	.500
	.600	-.165	-.110	-.037	.058	.100	.136	.212	.290	.600
	.700	-.188	-.129	-.055	.038	.078	.111	.180	.252	.700
	.800	-.209	-.141	-.048	.027	.057	.088	.147	.210	.800
	.850	-.212	-.134	-.031	.021	.046	.072	.124	.183	.850
	80.1 - percent -semispan station									
Upper surface	.030	.218	.047	-.474	-.730	-.787	-.854	-.967	-1.041	.030
	.060	.195	.034	-.517	-.728	-.780	-.839	-.954	-1.033	.060
	.100	.159	.003	-.434	-.714	-.778	-.839	-.958	-1.009	.100
	.200	.056	-.114	-.427	-.678	-.764	-.837	-.964	-1.035	.200
	.300	.011	-.121	-.366	-.647	-.762	-.842	-.970	-1.039	.300
	.400	-.021	-.156	-.347	-.626	-.759	-.849	-.976	-1.068	.400
	.500	-.045	-.205	-.370	-.616	-.774	-.863	-1.002	-1.049	.500
	.600	-.045	-.198	-.356	-.596	-.795	-.885	-1.020	-1.049	.600
	.700	-.028	-.182	-.330	-.556	-.816	-.904	-1.023	-1.064	.700
	.850	-.027	-.126	-.261	-.458	-.690	-.897	-.970	-1.031	.850
Lower surface	.045	-.588	-.295	.028	.150	.188	.216	.248	.262	.045
	.150	-.576	-.287	.048	.152	.194	.231	.294	.342	.150
	.250	-.605	-.359	.009	.124	.169	.203	.264	.321	.250
	.350	-.639	-.222	.011	.108	.152	.184	.243	.301	.350
	.450	-.681	-.099	-.023	.081	.119	.153	.215	.272	.450
	.550	-.713	-.109	-.051	.048	.087	.118	.179	.240	.550
	.650	-.730	-.141	-.046	.044	.078	.105	.160	.215	.650
	.750	-.711	-.175	-.036	.030	.057	.081	.132	.181	.750

TABLE III.- PRESSURE COEFFICIENTS FOR WING IN PRESENCE OF INDENTED BODY - Continued

(f) M = 1.03

	x/c	α = −4°	α = 0°	α = 4°	α = 8°	α = 10°	α = 12°	α = 16°	α = 20°	x/c
	26.7– percent -semispan station									
Upper surface	.010	.241	.074	-.203	-.808	-.889	-.893	-.947	-1.050	.010
	.030	.121	-.056	-.256	-.523	-.733	-.894	-.955	-1.050	.030
	.060	.079	-.070	-.234	-.405	-.633	-.899	-1.056	-1.126	.060
	.100	.062	-.074	-.198	-.326	-.470	-.751	-1.134	-1.157	.100
	.150	.044	-.080	-.179	-.307	-.342	-.324	-.940	-1.078	.150
	.200	.009	-.116	-.201	-.305	-.325	-.331	-.545	-.895	.200
	.300	.026	-.088	-.178	-.288	-.301	-.323	-.408	-.599	.300
	.400	-.001	-.103	-.185	-.281	-.295	-.318	-.391	-.532	.400
	.500	.013	-.104	-.192	-.277	-.296	-.319	-.390	-.514	.500
	.600	.034	-.071	-.161	-.252	-.277	-.308	-.382	-.495	.600
	.700	.010	-.073	-.154	-.233	-.263	-.296	-.369	-.479	.700
	.800	-.032	-.105	-.182	-.253	-.283	-.318	-.384	-.501	.800
	.900	-.055	-.127	-.197	-.259	-.288	-.319	-.395	-.499	.900
	.950	-.049	-.124	-.190	-.242	-.274	-.301	-.392	-.486	.950
Lower surface	.025	-.479	-.082	.075	.247	.329	.415	.564	.670	.025
	.060	-.140	-.092	.080	.228	.304	.388	.540	.654	.060
	.100	-.176	-.083	.042	.196	.275	.358	.507	.626	.100
	.150	-.168	-.086	.036	.192	.259	.335	.478	.596	.150
	.200	-.163	-.078	.022	.160	.230	.305	.442	.561	.200
	.300	-.169	-.081	.018	.144	.211	.278	.404	.512	.300
	.400	-.190	-.108	-.017	.119	.186	.247	.362	.468	.400
	.500	-.202	-.115	-.018	.118	.174	.230	.334	.429	.500
	.600	-.169	-.082	.016	.117	.169	.217	.309	.395	.600
	.700	-.158	-.079	.001	.096	.145	.190	.276	.355	.700
	.800	-.174	-.098	-.029	.057	.107	.150	.228	.299	.800
	.850	-.197	-.129	-.066	.026	.080	.121	.194	.261	.850
	.925	-.217	-.148	-.083	.035	.077	.110	.174	.227	.925
	53.4– percent - semispan station									
Upper surface	.010	.255	.125	-.248	-1.047	-.849	-.812	-.862	-.949	.010
	.030	.197	.023	-.319	-1.029	-.850	-.812	-.861	-.945	.030
	.060	.134	-.054	-.364	-.973	-.852	-.816	-.867	-.954	.060
	.100	.087	-.080	-.345	-.704	-.844	-.828	-.885	-.962	.100
	.200	.042	-.110	-.260	-.481	-.846	-.891	-.926	-.997	.200
	.300	.026	-.119	-.251	-.408	-.849	-.935	-1.011	-1.088	.300
	.400	.016	-.119	-.244	-.375	-.793	-.957	-1.054	-1.055	.400
	.500	-.002	-.118	-.237	-.353	-.462	-.931	-1.131	-1.036	.500
	.600	-.021	-.124	-.229	-.335	-.278	-.739	-1.112	-1.067	.600
	.700	-.032	-.128	-.220	-.316	-.292	-.402	-1.014	-1.108	.700
	.800	-.039	-.126	-.213	-.301	-.289	-.321	-.872	-1.070	.800
	.900	-.048	-.132	-.213	-.295	-.294	-.314	-.694	-.999	.900
Lower surface	.025	-.510	-.212	.080	.256	.314	.362	.445	.495	.025
	.060	-.514	-.160	.089	.239	.300	.354	.451	.519	.060
	.100	-.541	-.114	.080	.223	.282	.337	.436	.510	.100
	.200	-.587	-.061	.051	.197	.255	.309	.406	.487	.200
	.300	-.450	-.093	.021	.165	.221	.274	.370	.452	.300
	.400	-.084	-.085	.027	.147	.203	.253	.346	.425	.400
	.500	-.148	-.116	-.006	.104	.161	.209	.298	.375	.500
	.600	-.171	-.110	-.021	.081	.136	.184	.266	.342	.600
	.700	-.181	-.118	-.040	.064	.117	.160	.238	.307	.700
	.800	-.199	-.128	-.053	.060	.106	.141	.207	.267	.800
	.850	-.199	-.129	-.056	.058	.099	.130	.187	.243	.850
	80.1 – percent -semispan station									
Upper surface	.030	.227	.026	-.404	-.699	-.740	-.777	-.871	-.949	.030
	.060	.204	.014	-.470	-.695	-.735	-.767	-.859	-.940	.060
	.100	.168	-.024	-.427	-.676	-.730	-.765	-.866	-.933	.100
	.200	.072	-.128	-.436	-.632	-.716	-.765	-.869	-.949	.200
	.300	.029	-.130	-.391	-.601	-.713	-.772	-.875	-.958	.300
	.400	-.006	-.144	-.371	-.585	-.707	-.779	-.875	-.969	.400
	.500	-.046	-.187	-.376	-.575	-.720	-.790	-.896	-.956	.500
	.600	-.047	-.181	-.356	-.553	-.735	-.810	-.916	-.966	.600
	.700	-.041	-.169	-.321	-.519	-.752	-.838	-.926	-1.002	.700
	.850	-.032	-.146	-.276	-.422	-.696	-.854	-.877	-.993	.850
Lower surface	.045	-.572	-.320	.030	.165	.216	.258	.300	.314	.045
	.150	-.562	-.309	.058	.172	.224	.269	.345	.390	.150
	.250	-.588	-.373	.016	.146	.196	.246	.316	.372	.250
	.350	-.619	-.232	.021	.131	.189	.228	.297	.353	.350
	.450	-.654	-.084	-.007	.105	.159	.200	.269	.326	.450
	.550	-.679	-.082	-.039	.071	.129	.168	.238	.294	.550
	.650	-.696	-.116	-.053	.075	.123	.160	.221	.273	.650
	.750	-.684	-.152	-.073	.069	.111	.139	.195	.240	.750

TABLE III.- PRESSURE COEFFICIENTS FOR WING IN PRESENCE OF INDENTED BODY - Continued

(g) M = 1.12

	x/c	α = -4°	α = 0°	α = 4°	α = 8°	α = 10°	α = 12°	α = 16°	α = 20°	x/c
	26.7- percent -semispan station									
Upper surface	.010	.257	.100	-.153	-.729	-.806	-.799	-.802	-.871	.010
	.030	.137	-.014	-.206	-.469	-.627	-.766	-.799	-.877	.030
	.060	.087	-.044	-.201	-.365	-.503	-.734	-.856	-.939	.060
	.100	.063	-.053	-.173	-.288	-.447	-.649	-.900	-.979	.100
	.150	.054	-.050	-.150	-.274	-.360	-.312	-.840	-.923	.150
	.200	.011	-.086	-.169	-.268	-.324	-.309	-.538	-.778	.200
	.300	.034	-.064	-.146	-.247	-.290	-.290	-.362	-.541	.300
	.400	.026	-.078	-.150	-.233	-.263	-.275	-.336	-.462	.400
	.500	.027	-.074	-.167	-.234	-.260	-.279	-.335	-.445	.500
	.600	.051	-.043	-.123	-.212	-.241	-.267	-.339	-.431	.600
	.700	.057	-.028	-.108	-.196	-.222	-.253	-.331	-.407	.700
	.800	.021	-.055	-.134	-.209	-.236	-.270	-.343	-.413	.800
	.900	-.003	-.073	-.148	-.213	-.241	-.275	-.341	-.396	.900
	.950	.001	-.074	-.144	-.202	-.230	-.262	-.329	-.383	.950
Lower surface	.025	-.396	-.065	.100	.237	.306	.376	.532	.696	.025
	.060	-.149	.006	.095	.215	.276	.346	.512	.675	.060
	.100	-.136	-.050	.039	.166	.241	.321	.487	.641	.100
	.150	-.128	-.061	.055	.176	.244	.316	.467	.612	.150
	.200	-.129	-.050	.054	.160	.222	.296	.440	.585	.200
	.300	-.138	-.055	.044	.168	.225	.285	.414	.539	.300
	.400	-.150	-.081	.004	.130	.192	.255	.380	.498	.400
	.500	-.156	-.079	.001	.120	.189	.247	.361	.471	.500
	.600	-.129	-.047	.044	.141	.194	.242	.345	.441	.600
	.700	-.117	-.036	.048	.132	.176	.222	.321	.409	.700
	.800	-.128	-.054	.027	.103	.143	.187	.281	.354	.800
	.850	-.151	-.079	-.001	.071	.112	.157	.254	.331	.850
	.925	-.170	-.103	-.026	.055	.101	.157	.246	.305	.925
	53.4- percent - semispan station									
Upper surface	.010	.270	.146	-.176	-.897	-.915	-.737	-.737	-.803	.010
	.030	.214	.038	-.262	-.837	-.900	-.740	-.736	-.799	.030
	.060	.148	-.043	-.305	-.781	-.879	-.745	-.741	-.805	.060
	.100	.108	-.050	-.292	-.707	-.819	-.746	-.753	-.811	.100
	.200	.059	-.069	-.206	-.406	-.584	-.765	-.786	-.837	.200
	.300	.045	-.082	-.199	-.343	-.523	-.791	-.856	-.931	.300
	.400	.028	-.090	-.196	-.322	-.449	-.806	-.873	-.904	.400
	.500	.032	-.087	-.194	-.307	-.376	-.785	-.907	-.892	.500
	.600	.031	-.085	-.190	-.292	-.329	-.662	-.889	-.915	.600
	.700	.023	-.085	-.178	-.268	-.299	-.331	-.821	-.962	.700
	.800	.015	-.076	-.171	-.249	-.279	-.267	-.768	-.929	.800
	.900	.006	-.078	-.169	-.241	-.274	-.263	-.631	-.874	.900
Lower surface	.025	-.435	-.185	.086	.257	.317	.371	.467	.539	.025
	.060	-.441	-.130	.110	.250	.307	.364	.474	.561	.060
	.100	-.457	-.081	.116	.230	.288	.346	.459	.551	.100
	.200	-.488	-.015	.085	.206	.268	.322	.430	.573	.200
	.300	-.362	-.052	.046	.177	.238	.293	.400	.500	.300
	.400	-.058	-.055	.045	.174	.228	.279	.382	.477	.400
	.500	-.125	-.095	.018	.140	.187	.236	.339	.433	.500
	.600	-.149	-.084	.021	.121	.166	.214	.316	.405	.600
	.700	-.143	-.072	.018	.100	.146	.193	.294	.375	.700
	.800	-.151	-.078	.004	.082	.132	.185	.276	.345	.800
	.850	-.153	-.079	.001	.079	.133	.183	.261	.325	.850
	80.1 - percent -semispan station									
Upper surface	.030	.241	.051	-.294	-.794	-.675	-.681	-.744	-.815	.030
	.060	.221	.041	-.368	-.785	-.674	-.677	-.735	-.808	.060
	.100	.178	.003	-.337	-.757	-.659	-.667	-.732	-.798	.100
	.200	.088	-.108	-.376	-.721	-.634	-.669	-.737	-.810	.200
	.300	.059	-.120	-.347	-.672	-.622	-.670	-.740	-.817	.300
	.400	.031	-.128	-.337	-.611	-.613	-.676	-.762	-.828	.400
	.500	-.008	-.163	-.352	-.541	-.618	-.684	-.781	-.818	.500
	.600	-.003	-.155	-.335	-.506	-.618	-.701	-.774	-.810	.600
	.700	.003	-.127	-.287	-.474	-.619	-.716	-.767	-.833	.700
	.850	.017	-.101	-.229	-.420	-.613	-.736	-.726	-.858	.850
Lower surface	.045	-.488	-.277	.060	.179	.233	.278	.343	.382	.045
	.150	-.480	-.273	.083	.199	.250	.295	.387	.455	.150
	.250	-.504	-.309	.062	.179	.224	.270	.363	.433	.250
	.350	-.536	-.204	.055	.169	.219	.262	.350	.417	.350
	.450	-.559	-.082	.040	.149	.192	.234	.325	.395	.450
	.550	-.578	-.062	.016	.110	.156	.206	.299	.369	.550
	.650	-.586	-.083	.001	.109	.154	.210	.291	.353	.650
	.750	-.571	-.116	-.022	.093	.146	.202	.274	.326	.750

TABLE III.- PRESSURE COEFFICIENTS FOR WING IN PRESENCE OF INDENTED BODY - Concluded

(h) M = 1.45

	x/c	α = −3.6°	α = −0.3°	α = 4.3°	α = 8.8°	α = 11.1°	α = 13.4°	α = 16.0°	α = 19.8°	x/c
	26.7 – percent – semispan station									
Upper surface	.010	.197	.050	−.173	−.393	−.512	−.519			.010
	.030	.124	−.008	−.205	−.385	−.502	−.518			.030
	.060	.088	−.002	−.164	−.374	−.495	−.518			.060
	.100	.064	−.020	−.136	−.330	−.487	−.540			.100
	.150	.055	−.024	−.142	−.237	−.403	−.511			.150
	.200	.018	−.058	−.161	−.245	−.299	−.482			.200
	.300	.013	−.053	−.131	−.194	−.220	−.232			.300
	.400	.009	−.058	−.125	−.181	−.207	−.224			.400
	.500	−.004	−.058	−.119	−.186	−.207	−.226			.500
	.600	.005	−.053	−.126	−.178	−.198	−.217			.600
	.700	.031	−.030	−.101	−.157	−.180	−.210			.700
	.800	.030	−.034	−.099	−.149	−.185	−.209			.800
	.900	.010	−.046	−.105	−.163	−.198	−.222			.900
	.950	−.002	−.056	−.110	−.168	−.196	−.227			.950
Lower surface	.025	−.346	−.018	.132	.270	.335	.399			.025
	.060	−.092	.006	.118	.251	.301	.372			.060
	.100	−.062	.012	.091	.227	.282	.340			.100
	.150	−.104	−.023	.071	.187	.250	.305			.150
	.200	−.117	−.036	.050	.161	.229	.285			.200
	.250	−.115	−.054	.056	.138	.200	.256			.250
	.300	−.090	−.045	.050	.147	.191	.246			.300
	.400	−.111	−.053	.022	.126	.185	.239			.400
	.500	−.115	−.061	.020	.125	.174	.237			.500
	.600	−.115	−.063	.029	.125	.190	.261			.600
	.700	−.088	−.039	.045	.148	.201	.256			.700
	.800	−.087	−.036	.039	.133	.186	.239			.800
	.850	−.080	−.032	.042	.128	.183	.233			.850
	.925	−.112	−.060	.015	.100	.149	.195			.925
	53.4 – percent – semispan station									
Upper surface	.010	.275	.191	−.042	−.417	−.520	−.551			.010
	.030	.211	.093	−.131	−.368	−.459	−.508			.030
	.060	.122	−.002	−.244	−.439	−.504	−.552			.060
	.200	.043	−.063	−.266	−.405	−.462	−.513			.200
	.300	.027	−.061	−.168	−.400	−.395	−.473			.300
	.400	.004	−.068	−.160	−.365	−.364	−.450			.400
	.500	−.005	−.067	−.160	−.320	−.355	−.429			.500
	.600	−.008	−.077	−.166	−.282	−.340	−.403			.600
	.700	−.001	−.080	−.168	−.254	−.331	−.380			.700
	.800	.009	−.065	−.151	−.226	−.309	−.355			.800
	.900	.008	−.060	−.134	−.202	−.296	−.343			.900
Lower surface	.025	−.338	−.188	.107	.259	.315	.372			.025
	.060	−.338	−.128	.120	.247	.303	.359			.060
	.100	−.329	−.111	.110	.234	.292	.342			.100
	.200	−.307	−.042	.090	.207	.271	.326			.200
	.300	−.225	−.032	.066	.177	.234	.295			.300
	.400	−.151	−.055	.047	.154	.214	.272			.400
	.500	−.100	−.073	.028	.135	.192	.247			.500
	.600	−.123	−.088	.005	.119	.182	.239			.600
	.700	−.130	−.080	.017	.126	.184	.240			.700
	.800	−.120	−.073	.021	.120	.177	.231			.800
	.850	−.114	−.066	.021	.124	.174	.228			.850
	80.1 – percent – semispan station									
Upper surface	.030	.205	.117	−.157	−.489	−.551	−.550			.030
	.060	.202	.094	−.137	−.483	−.551	−.548			.060
	.100	.152	.041	−.146	−.449	−.551	−.546			.100
	.200	.110	−.021	−.230	−.445	−.551	−.543			.200
	.300	.037	−.089	−.295	−.465	−.551	−.541			.300
	.400	−.020	−.138	−.346	−.485	−.551	−.547			.400
	.500	−.042	−.145	−.382	−.490	−.551	−.546			.500
	.600	−.041	−.136	−.345	−.479	−.551	−.520			.600
	.700	−.027	−.122	−.262	−.459	−.551	−.512			.700
	.850	−.006	−.105	−.217	−.409	−.488	−.494			.850
Lower surface	.045	−.342	−.219	.078	.235	.291	.335			.045
	.150	−.343	−.206	.108	.229	.285	.340			.150
	.250	−.349	−.192	.103	.207	.262	.318			.250
	.350	−.363	−.157	.079	.181	.239	.293			.350
	.450	−.369	−.111	.058	.163	.225	.283			.450
	.550	−.364	−.074	.033	.147	.211	.271			.550
	.650	−.352	−.060	.022	.151	.211	.268			.650
	.750	−.324	−.080	.009	.140	.206	.256			.750

TABLE IV.- PRESSURE COEFFICIENTS FOR BASIC BODY IN PRESENCE OF WING

(a) M = 0.60

x/l	α = −4°	α = 0°	α = 4°	α = 8°	α = 10°	α = 12°	α = 16°	α = 20°	x/l
				Row A					
.055	.103	.053	.016	-.017	-.031	-.039	-.049		.055
.166	.030	-.000	-.021	-.040	-.048	-.052	-.060	-.075	.166
.277	.006	-.020	-.036	-.048	-.050	-.053	-.049	-.064	.277
.367	.044	.014	-.009	-.038	-.042	-.034	-.064	-.129	.367
.387	-.043	-.071	-.087	-.113	-.137	-.173	-.215	-.233	.387
.415	-.010	-.051	-.074	-.114	-.132	-.172	-.243	-.285	.415
.443	.001	-.051	-.086	-.144	-.168	-.213	-.303	-.379	.443
.498	-.004	-.064	-.113	-.189	-.219	-.264	-.353	-.425	.498
.553	-.008	-.070	-.112	-.190	-.202	-.236	-.290	-.324	.553
.581	-.021	-.076	-.114	-.182	-.195	-.228	-.265	-.277	.581
.609	-.024	-.076	-.112	-.165	-.173	-.203	-.233	-.230	.609
.636	-.031	-.074	-.103	-.149	-.155	-.176	-.203	-.197	.636
.664	-.028	-.064	-.083	-.128	-.126	-.138	-.176	-.185	.664
.692	-.013	-.044	-.058	-.085	-.078	-.092	-.121	-.132	.692
.719	-.018	-.040	-.048	-.063	-.058	-.069	-.096	-.117	.719
.747	-.023	-.037	-.038	-.045	-.044	-.057	-.086	-.099	.747
.775	.000	-.014	-.013	-.014	-.009	-.017	-.035	-.070	.775
.830	-.008	-.011	-.008	-.016	-.009	-.016	-.036	-.070	.830
.871	-.013	-.005	-.006	-.010	-.010	-.020	-.036	-.054	.871
.954	.045	.050	.054	.042	.044	.035	.020	.024	.954
				Row B					
.166	.019	.001	-.024	-.061	-.081	-.101	-.132	-.155	.166
.277			-.009						.277
.367	.012	-.002	-.024	-.059	-.077	-.096	-.132	-.184	.367
.387	-.062	-.076	-.095	-.136	-.150	-.175	-.220	-.288	.387
.443	.021	-.038	-.095	-.182	-.222	-.280	-.419	-.566	.443
.498	.010	-.066	-.126	-.218	-.263	-.333	-.482	-.611	.498
.553	.010	-.059	-.116	-.195	-.238	-.292	-.371	-.499	.553
.609	-.005			-.171	-.199	-.232	-.295	-.477	.609
.664	-.015	-.059	-.083	-.128	-.147	-.172	-.248	-.549	.664
.719	-.010	-.033	-.043	-.068	-.077	-.101	-.195	-.338	.719
.775	-.000	-.010	-.012	-.020	-.028	-.057	-.114	-.092	.775
.830	-.006	-.010	-.007	-.013	-.010	-.036	-.055	-.048	.830
.871	-.008	-.007	-.001	-.011	-.019	-.035	-.046	-.069	.871
				Row C					
.055	.042	.069	.050	-.004	-.037	-.078	-.180	-.302	.055
.166	-.015	.001	-.013	-.059	-.086	-.128	-.223	-.345	.166
.277	-.034	-.021	-.033	-.082	-.112	-.151	-.248	-.362	.277
.353	-.010	.005	-.011	-.052	-.083	-.142	-.251	-.332	.353
.367	.030	.044	.032	-.036	-.067	-.093	-.197	-.335	.367
.747	-.008	-.008	-.009	-.010	-.009	-.010	-.009	-.005	.747
.775	.010	.002	.007	.003	.011	.006	-.011	-.005	.775
.830	-.010	-.010	-.009	-.018	-.011	-.012	-.029	.001	.830
.871	.009	.016	.009	.019	.015	.007	.020	.024	.871
.954	.039	.045	.039	.046	.043	.038	.048	.060	.954
				Row D					
.166	-.018	.007	.021	.017	.019	.015	-.004	-.013	.166
.277	-.040	-.020	-.007	-.006	-.008	-.012	-.032	-.046	.277
.367	-.019	.010	.027	.007	.007	-.001	-.013	-.026	.367
.387	-.096	-.065	-.059	-.039	-.041	-.049	-.056	-.048	.387
.443	-.129	-.054	.000	.057	.085	.096	.139	.176	.443
.498	-.140	-.058	.014	.088	.129	.158	.231	.302	.498
.553	-.145	-.075	-.009	.062	.103	.135	.208	.283	.553
.609	-.124	-.069	-.016	.047	.084	.107	.167	.233	.609
.664	-.099	-.066	-.025	.021	.049	.065	.108	.157	.664
.719	-.056	-.044	-.020	.004	.025	.031	.047	.077	.719
.775	-.012	-.013	.002	.007	.018	.016	.020	.020	.775
.830	-.004	-.006	-.002	-.003	.006	.002	.004	.007	.830
.871	.012	.008	.009	.002	.011	.004	.006	.016	.871
				Row E					
.055	.031	.072	.109	.162	.192	.223	.298	.376	.055
.166	-.020	.007	.037	.076	.101	.123	.191	.267	.166
.277	-.035	-.017	.007	.050	.077	.093	.156	.227	.277
.367	-.023	.000	.029	.062	.111	.126	.191	.287	.367
.387	-.067	-.040	-.013	.041	.067	.091	.161	.245	.387
.443	-.097	-.053	-.007	.054	.095	.123	.201	.279	.443
.498	-.128	-.068	-.013	.055	.097	.130	.210	.295	.498
.553	-.128	-.068	-.012	.052	.093	.125	.196	.274	.553
.609	-.113	-.069	-.019	.034	.072	.097	.155	.221	.609
.664	-.090	-.062	-.029	.019	.045	.060	.105	.162	.664
.719	-.056	-.037	-.021	.008	.028	.036	.060	.092	.719
.775	-.021	-.013	-.007	.003	.019	.018	.033	.045	.775
.830	-.009	-.013	-.008	-.012	.002	.003	.010	.026	.830
.871	-.009	-.013	-.014	-.014	-.013	-.015	-.006	.020	.871
.954	.029	.025	.015	.007	.005	-.006	-.007	.011	.954

TABLE IV.- PRESSURE COEFFICIENTS FOR BASIC BODY IN PRESENCE OF WING - Continued

(b) M = 0.80

x/l	α = −4°	α = 0°	α = 4°	α = 8°	α = 10°	α = 12°	α = 16°	α = 20°	x/l
Row A									
.055	.112	.067	.024	.004	-.017	-.023	-.038	-.045	.055
.166	.036	.005	-.022	-.032	-.039	-.042	-.047	-.070	.166
.277	.006	-.018	-.039	-.041	-.043	-.039	-.035	-.042	.277
.367	.053	.027	.001	-.016	-.021	.004	-.012	-.074	.367
.387	-.049	-.070	-.084	-.099	-.124	-.146	-.179	-.185	.387
.415	-.011	-.047	-.073	-.102	-.119	-.145	-.207	-.245	.415
.443	.004	-.048	-.086	-.143	-.169	-.207	-.302	-.392	.443
.498	-.002	-.066	-.122	-.211	-.251	-.298	-.428	-.517	.498
.553	-.010	-.077	-.130	-.223	-.253	-.290	-.377	-.437	.553
.581	-.024	-.087	-.137	-.222	-.254	-.284	-.338	-.414	.581
.609	-.031	-.086	-.132	-.204	-.226	-.249	-.276	-.340	.609
.636	-.037	-.085	-.123	-.182	-.203	-.213	-.251	-.272	.636
.664	-.032	-.073	-.102	-.151	-.156	-.160	-.244	-.255	.664
.692	-.017	-.048	-.069	-.096	-.095	-.102	-.184	-.190	.692
.719	-.020	-.041	-.054	-.067	-.063	-.077	-.140	-.171	.719
.747	-.022	-.033	-.034	-.038	-.038	-.059	-.101	-.128	.747
.775	.002	-.006	-.008	-.006	.000	-.014	-.032	-.095	.775
.830	-.010	-.009	-.006	-.012	-.007	-.013	-.034	-.092	.830
.871	-.019	-.011	-.006	-.009	-.013	-.019	-.040	-.078	.871
.954	.050	.058	.056	.050	.049	.041	.020	.021	.954
Row B									
.166	.022	.007	-.021	-.053	-.073	-.085	-.118	-.148	.166
.367	.016	.006	-.015	-.039	-.052	-.059	-.082	-.116	.367
.387	-.070	-.077	-.094	-.117	-.126	-.134	-.159	-.207	.387
.443	.023	-.034	-.094	-.177	-.222	-.265	-.389	-.509	.443
.498	.012	-.066	-.133	-.236	-.291	-.358	-.532	-.647	.498
.553	.007	-.065	-.135	-.229	-.287	-.343	-.445	-.530	.553
.609	-.011	-.073	-.130	-.210	-.255	-.286	-.324	-.543	.609
.664	-.023	-.068	-.102	-.154	-.176	-.196	-.306	-.516	.664
.719	-.013	-.035	-.048	-.073	-.081	-.101	-.239	-.535	.719
.775	-.001	-.005	-.007	-.015	-.023	-.048	-.121	-.186	.775
.830	-.008	-.007	-.007	-.007	-.012	-.033	-.069	-.077	.830
.871	-.010	-.006	-.002	-.012	-.022	-.034	-.061	-.089	.871
Row C									
.055	.054	.080	.060	.011	-.025	-.059	-.160	-.280	.055
.166	-.012	.005	-.010	-.053	-.084	-.118	-.215	-.337	.166
.277	-.032	-.018	-.035	-.078	-.108	-.138	-.231	-.336	.277
.353	-.002	.009	-.007	-.041	-.071	-.116	-.208	-.272	.353
.367	.043	.050	.041	-.018	-.045	-.058	-.143	-.259	.367
.747	-.004	-.004	-.002	-.006	-.002	-.001	-.001	-.003	.747
.775	.012	.007	.013	.008	.014	.015	-.019	-.094	.775
.830	-.013	-.011	-.009	-.015	-.012	-.010	-.035	-.021	.830
.871	.006	.013	.003	.017	.010	.004	.013	.015	.871
.954	.042	.048	.043	.052	.043	.042	.053	.061	.954
Row D									
.166	-.016	.009	.022	.025	.024	.024	.008	.001	.166
.277	-.039	-.018	-.009	-.005	-.005	-.004	-.017	-.027	.277
.367	-.008	.014	.029	.012	.014	.013	.007	-.001	.367
.387	-.088	-.067	-.062	-.040	-.041	-.041	-.036	-.018	.387
.443	-.123	-.047	.007	.072	.100	.120	.170	.219	.443
.498	-.151	-.059	.016	.097	.140	.178	.253	.325	.498
.553	-.171	-.083	-.009	.071	.113	.149	.220	.296	.553
.609	-.153	-.082	-.019	.051	.088	.118	.180	.245	.609
.664	-.121	-.076	-.029	.023	.054	.076	.112	.162	.664
.719	-.062	-.044	-.020	.006	.027	.036	.038	.055	.719
.775	-.010	-.006	.003	.010	.019	.020	.000	-.038	.775
.830	-.006	-.004	-.002	-.003	.004	.003	-.010	-.039	.830
.871	.011	.011	.010	.003	.009	.007	-.001	-.012	.871
Row E									
.055	.040	.082	.122	.177	.208	.241	.314	.396	.055
.166	-.015	.011	.041	.083	.106	.133	.196	.275	.166
.277	-.033	-.016	.009	.054	.078	.101	.149	.232	.277
.367	-.010	.008	.034	.067	.115	.135	.200	.259	.367
.387	-.058	-.037	-.014	.041	.064	.094	.165	.251	.387
.443	-.099	-.049	.000	.063	.104	.137	.215	.299	.443
.498	-.140	-.070	-.010	.069	.111	.148	.227	.311	.498
.553	-.149	-.076	-.014	.062	.102	.139	.212	.285	.553
.609	-.142	-.081	-.023	.043	.081	.106	.169	.235	.609
.664	-.114	-.072	-.031	.025	.052	.069	.113	.166	.664
.719	-.061	-.044	-.021	.011	.034	.038	.057	.077	.719
.775	-.017	-.014	-.005	.006	.019	.016	.016	.002	.775
.830	-.008	-.012	-.006	-.009	.003	.000	-.004	-.021	.830
.871	-.009	-.011	-.016	-.016	-.020	-.018	-.018	-.013	.871
.954	.030	.027	.022	.011	.005	.000	-.003	.002	.954

TABLE IV.- PRESSURE COEFFICIENTS FOR BASIC BODY IN PRESENCE OF WING - Continued

(c) M = 0.90

x/l	α = −4°	α = 0°	α = 4°	α = 8°	α = 10°	α = 12°	α = 16°	α = 20°	x/l
Row A									
.055	.124	.072	.037	.015	-.006	-.014	-.029	-.031	.055
.166	.041	.005	-.017	-.031	-.036	-.038	-.039	-.060	.166
.277	.006	-.023	-.035	-.040	-.038	-.034	-.021	-.019	.277
.367	.063	.032	.016	.001	.010	.038	.041	-.005	.367
.387	-.054	-.075	-.076	-.087	-.098	-.117	-.127	-.112	.387
.415	-.008	-.045	-.064	-.083	-.087	-.106	-.140	-.157	.415
.443	.010	-.045	-.078	-.128	-.145	-.176	-.239	-.298	.443
.498	-.000	-.071	-.124	-.220	-.256	-.303	-.399	-.480	.498
.553	-.013	-.087	-.148	-.253	-.285	-.333	-.437	-.458	.553
.581	-.031	-.104	-.165	-.283	-.311	-.354	-.447	-.426	.581
.609	-.037	-.107	-.163	-.282	-.321	-.381	-.470	-.448	.609
.636	-.049	-.111	-.162	-.282	-.346	-.400	-.489	-.417	.636
.664	-.040	-.094	-.128	-.205	-.297	-.407	-.229	-.385	.664
.692	-.021	-.063	-.083	-.109	-.097	-.109	-.070	-.264	.692
.719	-.024	-.049	-.056	-.064	-.051	-.053	-.151	-.224	.719
.747	-.020	-.031	-.027	-.028	-.022	-.028	-.129	-.196	.747
.775	.007	-.004	-.001	.002	.012	.014	-.062	-.141	.775
.830	-.012	-.013	-.007	-.014	-.007	-.010	-.052	-.144	.830
.871	-.022	-.018	-.008	-.017	-.020	-.033	-.052	-.118	.871
.954	.056	.057	.060	.053	.052	.037	.026	.020	.954
Row B									
.166	.025	.009	-.016	-.050	-.066	-.083	-.106	-.141	.166
.367	.024	.011	-.003	-.022	-.025	-.030	-.036	-.055	.367
.387	-.075	-.084	-.089	-.100	-.098	-.094	-.099	-.131	.387
.443	.032	-.029	-.084	-.159	-.193	-.224	-.307	-.404	.443
.498	.015	-.067	-.137	-.243	-.292	-.351	-.475	-.594	.498
.553	.003	-.077	-.153	-.259	-.308	-.366	-.483	-.531	.553
.609	-.019	-.094	-.161	-.289	-.353	-.407	-.536	-.604	.609
.664	-.031	-.087	-.130	-.212	-.297	-.394	-.268	-.576	.664
.719	-.015	-.042	-.050	-.072	-.072	-.094	-.235	-.545	.719
.775	.004	-.004	.001	-.008	-.012	-.020	-.138	-.375	.775
.830	-.010	-.011	-.006	-.012	-.015	-.028	-.075	-.165	.830
.871	-.014	-.012	-.005	-.017	-.029	-.039	-.061	-.130	.871
Row C									
.055	.065	.089	.072	.022	-.009	-.046	-.139	-.258	.055
.166	-.006	.006	-.005	-.050	-.078	-.112	-.205	-.333	.166
.277	-.032	-.024	-.034	-.077	-.104	-.134	-.218	-.317	.277
.353	.005	.009	-.003	-.031	-.054	-.100	-.176	-.225	.353
.367	.052	.053	.050	-.005	-.026	-.033	-.100	-.205	.367
.747	-.001	-.003	-.002	.000	-.002	-.000	.000	-.001	.747
.775	.015	.008	.019	.011	.022	.032	-.091	-.191	.775
.830	-.012	-.013	-.008	-.019	-.014	-.012	-.067	-.133	.830
.871	.002	.005	-.000	.008	.002	-.006	-.007	-.022	.871
.954	.044	.044	.044	.051	.043	.032	.053	.068	.954
Row D									
.166	-.011	.009	.028	.028	.030	.027	.025	.018	.166
.277	-.039	-.023	-.010	-.005	-.002	-.003	-.009	-.011	.277
.367	.002	.016	.035	.019	.021	.020	.023	.022	.367
.387	-.081	-.073	-.065	-.044	-.039	-.037	-.020	.009	.387
.443	-.111	-.044	.012	.082	.117	.140	.202	.262	.443
.498	-.154	-.064	.018	.103	.151	.190	.274	.356	.498
.553	-.184	-.095	-.011	.073	.118	.154	.238	.320	.553
.609	-.194	-.105	-.027	.048	.088	.121	.193	.267	.609
.664	-.162	-.099	-.037	.018	.054	.077	.125	.183	.664
.719	-.065	-.051	-.024	.002	.026	.036	.034	.064	.719
.775	-.003	-.003	.007	.010	.021	.023	-.064	-.120	.775
.830	-.006	-.007	-.002	-.009	-.001	.000	-.048	-.146	.830
.871	.006	.007	.007	-.001	.002	-.003	-.022	-.079	.871
Row E									
.055	.052	.092	.132	.187	.220	.250	.328	.413	.055
.166	-.010	.013	.046	.086	.110	.136	.204	.285	.166
.277	-.033	-.019	.007	.051	.077	.100	.154	.239	.277
.367	.003	.013	.039	.072	.122	.138	.212	.275	.367
.387	-.048	-.036	-.013	.038	.066	.095	.173	.263	.387
.443	-.087	-.047	.003	.071	.113	.149	.235	.325	.443
.498	-.143	-.073	-.009	.074	.120	.158	.246	.338	.498
.553	-.168	-.088	-.017	.065	.108	.146	.227	.311	.553
.609	-.184	-.101	-.031	.041	.084	.115	.183	.257	.609
.664	-.154	-.092	-.042	.019	.054	.074	.124	.188	.664
.719	-.065	-.048	-.025	.008	.033	.044	.054	.083	.719
.775	-.010	-.008	-.002	.005	.020	.025	-.025	-.063	.775
.830	-.007	-.012	-.007	-.014	-.003	-.001	-.043	-.122	.830
.871	-.010	-.015	-.019	-.024	-.027	-.029	-.044	-.076	.871
.954	.031	.029	.025	.013	.007	-.004	-.004	-.007	.954

TABLE IV.- PRESSURE COEFFICIENTS FOR BASIC BODY IN PRESENCE OF WING - Continued

(d) M = 0.94

x/l	α = −4°	α = 0°	α = 4°	α = 8°	α = 10°	α = 12°	α = 16°	α = 20°	x/l
				Row A					
.055	.131	.083	.043	.020	.003	-.007	-.026	-.024	.055
.166	.041	.009	-.015	-.030	-.034	-.038	-.038	-.056	.166
.277	.003	-.022	-.038	-.040	-.036	-.030	-.015	-.008	.277
.367	.069	.042	.025	.014	.025	.055	.071	.025	.367
.387	-.056	-.075	-.073	-.075	-.084	-.097	-.098	-.079	.387
.415	-.006	-.041	-.056	-.065	-.066	-.078	-.100	-.113	.415
.443	.015	-.037	-.071	-.113	-.124	-.151	-.195	-.253	.443
.498	.003	-.068	-.123	-.212	-.234	-.276	-.362	-.437	.498
.553	-.014	-.089	-.151	-.247	-.280	-.318	-.406	-.483	.553
.581	-.036	-.113	-.181	-.270	-.300	-.343	-.423	-.477	.581
.609	-.044	-.118	-.193	-.286	-.318	-.367	-.443	-.435	.609
.636	-.062	-.133	-.210		-.344	-.394	-.468	-.374	.636
.664	-.052	-.111	-.195	-.327	-.365	-.410	-.484	-.437	.664
.692	-.031	-.071	-.115	-.292	-.340	-.380	-.471	-.419	.692
.719	-.031	-.049	-.051	-.086	-.292	-.369	-.491	-.391	.719
.747	-.016	-.022	-.017	.002	.017	-.002	-.204	-.361	.747
.775	.009	.001	.007	.023	.043	.040	.056	-.057	.775
.830	-.011	-.011	-.004	-.006	.004	.008	.004	-.204	.830
.871	-.025	-.017	-.012	-.019	-.022	-.028	-.046	-.151	.871
.954	.059	.062	.060	.054	.053	.046	.024	.016	.954
				Row B					
.166	.028	.012	-.014	-.046	-.063	-.079	-.106	-.134	.166
.367	.029	.018	.004	-.009	-.012	-.013	-.013	-.029	.367
.387	-.078	-.084	-.087	-.091	-.080	-.071	-.069	-.096	.387
.443	.039	-.022	-.077	-.143	-.168	-.199	-.263	-.354	.443
.498	.017	-.065	-.136	-.234	-.266	-.323	-.431	-.542	.498
.553	.002	-.080	-.157	-.249	-.301	-.351	-.450	-.585	.553
.609	-.027	-.105	-.192	-.294	-.345	-.398	-.508	-.529	.609
.664	-.042	-.105	-.192	-.329	-.380	-.430	-.562	-.633	.664
.719	-.020	-.042	-.047	-.104	-.217	-.265	-.434	-.602	.719
.775	.008	.003	.009	.007	.007	-.011	-.065	-.263	.775
.830	-.010	-.010	-.005	-.006	-.004	-.000	-.020	-.271	.830
.871	-.014	-.011	-.008	-.016	-.028	-.021	-.056	-.199	.871
				Row C					
.055	.071	.096	.080	.030	-.001	-.038	-.130	-.244	.055
.166	-.004	.010	-.004	-.048	-.075	-.110	-.203	-.320	.166
.277	-.033	-.023	-.036	-.077	-.103	-.132	-.215	-.308	.277
.353	.008	.015	.001	-.025	-.046	-.086	-.158	-.200	.353
.367	.057	.063	.055	.004	-.012	-.019	-.079	-.174	.367
.747	-.001	.000	-.000	-.001	.000	.000	.000	-.001	.747
.775	.023	.018	.026	.023	.030	.010	-.002	-.176	.775
.830	-.013	-.010	-.008	-.014	-.008	.001	-.035	-.233	.830
.871	.000	.005	-.003	.006	.005	-.003	-.016	-.092	.871
.954	.045	.047	.045	.051	.045	.040	.044	.065	.954
				Row D					
.166	-.010	.013	.029	.030	.033	.032	.026	.026	.166
.277	-.040	-.023	-.011	-.006	-.002	.000	-.004	-.004	.277
.367	.008	.024	.038	.025	.030	.027	.034	.036	.367
.387	-.078	-.072	-.069	-.046	-.037	-.033	-.010	.025	.387
.443	-.099	-.038	.018	.091	.131	.153	.221	.287	.443
.498	-.149	-.062	.016	.110	.161	.198	.290	.376	.498
.553	-.183	-.097	-.016	.073	.122	.161	.249	.338	.553
.609	-.210	-.118	-.039	.045	.087	.121	.201	.284	.609
.664	-.245	-.120	-.051	.014	.050	.073	.137	.207	.664
.719	-.065	-.049	-.028	-.002	.016	.020	.046	.093	.719
.775	.008	.004	.011	.014	.017	-.000	-.029	-.114	.775
.830	-.005	-.005	-.002	-.006	.001	-.001	-.032	-.251	.830
.871	.007	.005	.005	.001	.005	-.004	-.015	-.192	.871
				Row E					
.055	.056	.098	.139	.196	.228	.258	.336	.424	.055
.166	-.009	.015	.049	.087	.112	.139	.206	.292	.166
.277	-.034	-.020	.006	.051	.077	.101	.156	.247	.277
.367	.008	.018	.042	.099	.128	.143	.218	.285	.367
.387	-.040	-.034	-.014	.037	.068	.100	.180	.277	.387
.443	-.081	-.042	.005	.077	.121	.159	.251	.341	.443
.498	-.142	-.072	-.008	.079	.128	.167	.261	.357	.498
.553	-.168	-.091	-.020	.068	.114	.151	.239	.329	.553
.609	-.201	-.116	-.040	.039	.083	.117	.194	.276	.609
.664	-.235	-.117	-.055	.015	.049	.070	.137	.211	.664
.719	-.065	-.049	-.029	.000	.025	.031	.064	.109	.719
.775	.001	-.002	.004	.008	.016	.006	-.013	-.048	.775
.830	-.005	-.010	-.007	-.013	-.002	-.006	-.039	-.217	.830
.871	-.011	-.014	-.020	-.024	-.028	-.030	-.045	-.238	.871
.954	.032	.032	.030	.016	.015	.002	-.005	-.003	.954

TABLE IV.- PRESSURE COEFFICIENTS FOR BASIC BODY IN PRESENCE OF WING - Continued

(e) M = 0.98

x/l	α = −4°	α = 0°	α = 4°	α = 8°	α = 10°	α = 12°	α = 16°	α = 20°	x/l
Row A									
.055	.144	.093	.057	.029	.011	.004	-.015	-.015	.055
.166	.045	.011	-.011	-.027	-.032	-.036	-.036	-.053	.166
.277	-.001	-.028	-.046	-.047	-.042	-.033	-.012	.008	.277
.367	.081	.057	.040	.031	.043	.079	.099	.063	.367
.387	-.057	-.074	-.065	-.059	-.063	-.071	-.063	-.034	.387
.415	.003	-.031	-.041	-.041	-.036	-.044	-.055	-.055	.415
.443	.030	-.024	-.054	-.087	-.094	-.109	-.145	-.199	.443
.498	.013	-.059	-.114	-.179	-.205	-.242	-.315	-.385	.498
.553	-.008	-.082	-.144	-.226	-.250	-.283	-.363	-.432	.553
.581	-.036	-.112	-.171	-.249	-.273	-.311	-.381	-.443	.581
.609	-.047	-.129	-.185	-.269	-.295	-.335	-.404	-.459	.609
.636	-.078	-.149	-.208	-.289	-.323	-.362	-.425	-.467	.636
.664	-.086	-.165	-.226	-.316	-.342	-.381	-.442	-.493	.664
.692	-.068	-.152	-.221	-.297	-.325	-.360	-.434	-.469	.692
.719	-.104	-.172	-.249	-.331	-.358	-.393	-.460	-.498	.719
.747	-.066	-.031	-.149	-.333	-.389	-.440	-.511	-.526	.747
.775	-.012	.025	.013	-.030	-.079	-.208	-.471	-.543	.775
.830	-.004	.005	.023	.002	-.016	-.024	-.014	.029	.830
.871	-.017	-.011	.007	-.005	-.027	-.056	-.144	-.152	.871
.954	.070	.069	.071	.071	.072	.061	.026	-.073	.954
Row B									
.166	.032	.014	-.012	-.044	-.063	-.077	-.102	-.131	.166
.367	.044	.030	.017	.007	.006	.007	.013	.005	.367
.387	-.079	-.086	-.082	-.075	-.058	-.044	-.037	-.055	.387
.443	.054	-.010	-.060	-.116	-.138	-.160	-.218	-.294	.443
.498	.029	-.055	-.127	-.203	-.238	-.285	-.379	-.477	.498
.553	.007	-.074	-.144	-.232	-.274	-.319	-.407	-.531	.553
.609	-.030	-.118	-.187	-.279	-.322	-.367	-.463	-.600	.609
.664	-.072	-.158	-.224	-.318	-.359	-.408	-.517	-.666	.664
.719	-.093	-.156	-.247	-.341	-.384	-.436	-.543	-.697	.719
.775	-.015	.023	.010	-.070	-.109	-.171	-.314	-.403	.775
.830	-.001	.007	.020	-.013	-.044	-.050	-.053	-.029	.830
.871	-.004	-.003	.011	.001	-.025	-.043	-.125	-.178	.871
Row C									
.055	.082	.106	.090	.040	.008	-.025	-.115	-.233	.055
.166	-.002	.011	-.001	-.048	-.076	-.110	-.198	-.327	.166
.277	-.040	-.029	-.043	-.086	-.110	-.138	-.210	-.304	.277
.353	.016	.019	.009	-.015	-.030	-.067	-.126	-.180	.353
.367	.068	.073	.067	.017	.006	.001	-.051	-.134	.367
.747	-.003	-.001	-.004	-.003	-.003	-.002	-.004	-.002	.747
.775	.003	.035	.020	-.056	-.077	-.118	-.186	-.245	.775
.830	.002	.005	.013	-.034	-.073	-.089	-.127	-.185	.830
.871	.013	.010	.013	.013	-.021	-.056	-.141	-.188	.871
.954	.054	.052	.055	.065	.063	.058	.035	-.049	.954
Row D									
.166	-.008	.015	.032	.039	.034	.034	.029	.034	.166
.277	-.047	-.031	-.017	-.019	-.004	-.002	-.001	.004	.277
.367	.022	.032	.052	.035	.042	.038	.048	.053	.367
.387	-.072	-.073	-.068	-.043	-.031	-.022	.009	.050	.387
.443	-.078	-.027	.032	.110	.148	.177	.249	.320	.443
.498	-.128	-.056	.027	.123	.176	.218	.312	.407	.498
.553	-.178	-.095	-.011	.083	.135	.175	.272	.367	.553
.609	-.203	-.124	-.043	.047	.095	.134	.222	.315	.609
.664	-.252	-.177	-.088	.006	.053	.083	.159	.240	.664
.719	-.284	-.192	-.117	-.039	-.006	.016	.070	.131	.719
.775	-.003	.026	.002	-.058	-.064	-.086	-.132	-.133	.775
.830	.016	.012	.013	-.056	-.095	-.111	-.149	-.215	.830
.871	.022	.016	.020	.006	-.032	-.088	-.191	-.264	.871
Row E									
.055	.068	.106	.150	.206	.238	.267	.345	.441	.055
.166	-.007	.016	.051	.097	.114	.141	.208	.299	.166
.277	-.041	-.027	.002	.045	.072	.099	.158	.253	.277
.367	.023	.028	.054	.104	.135	.150	.224	.303	.367
.387	-.029	-.033	-.011	.042	.074	.108	.193	.295	.387
.443	-.060	-.030	.019	.090	.138	.177	.272	.369	.443
.498	-.125	-.067	.002	.091	.142	.184	.283	.387	.498
.553	-.162	-.090	-.015	.076	.127	.168	.263	.358	.553
.609	-.193	-.122	-.044	.042	.092	.131	.218	.305	.609
.664	-.241	-.172	-.095	.008	.052	.080	.160	.246	.664
.719	-.264	-.186	-.112	-.031	.007	.027	.088	.150	.719
.775	-.002	.023	-.003	-.052	-.049	-.070	-.068	-.033	.775
.830	.023	.009	.009	-.081	-.108	-.117	-.137	-.202	.830
.871	.009	-.006	-.008	-.021	-.082	-.147	-.189	-.205	.871
.954	.043	.040	.039	.033	.036	.028	.003	-.291	.954

TABLE IV.- PRESSURE COEFFICIENTS FOR BASIC BODY IN PRESENCE OF WING - Continued

(f) M = 1.03

x/l	α = −4°	α = 0°	α = 4°	α = 8°	α = 10°	α = 12°	α = 16°	α = 20°	x/l
Row A									
.055	.173	.127	.092	.062	.047	.042	.026	.024	.055
.166	.081	.053	.029	-.002	.000	.002	-.010	-.032	.166
.277	.040	.009	-.010	-.046	-.009	-.022	-.034	-.028	.277
.367	.067	.042	.027	.000	.027	.045	.112	.129	.367
.387	-.037	-.057	-.056	-.076	-.078	-.103	-.016	.037	.387
.415	-.064	-.082	-.073	-.067	-.055	-.037	.013	.017	.415
.443	.017	-.040	-.060	-.075	-.078	-.076	-.073	-.121	.443
.498	.011	-.061	-.106	-.157	-.184	-.202	-.239	-.299	.498
.553	.003	-.075	-.132	-.192	-.222	-.240	-.284	-.347	.553
.581	-.026	-.099	-.156	-.213	-.243	-.263	-.297	-.363	.581
.609	-.039	-.112	-.171	-.232	-.262	-.286	-.322	-.382	.609
.636	-.058	-.133	-.191	-.251	-.284	-.308	-.339	-.395	.636
.664	-.074	-.150	-.208	-.274	-.303	-.325	-.357	-.425	.664
.692	-.059	-.138	-.201	-.255	-.285	-.307	-.353	-.406	.692
.719	-.096	-.168	-.231	-.286	-.317	-.337	-.381	-.435	.719
.747	-.115	-.186	-.246	-.297	-.348	-.381	-.433	-.459	.747
.775	-.073	-.110	-.130	-.098	-.209	-.263	-.408	-.476	.775
.830	-.088	-.064	-.050	-.033	-.020	.001	-.012	.044	.830
.871	-.118	-.086	-.065	-.085	-.100	-.108	-.124	-.124	.871
.954	.067	.090	.078	.019	-.052	-.141	-.227	-.303	.954
Row B									
.166	.072	.056	.031	-.019	-.032	-.038	-.074	-.115	.166
.367	.031	.014	.003	-.024	-.008	-.032	-.002	.047	.367
.387	-.053	-.063	-.068	-.094	-.083	-.091	.007	-.005	.387
.443	.042	-.025	-.063	-.106	-.126	-.131	-.156	-.211	.443
.498	.026	-.060	-.117	-.178	-.217	-.243	-.306	-.385	.498
.553	.015	-.066	-.135	-.203	-.252	-.278	-.337	-.447	.553
.609	-.024	-.105	-.172	-.245	-.290	-.324	-.397	-.520	.609
.664	-.063	-.142	-.207	-.279	-.323	-.360	-.444	-.576	.664
.719	-.092	-.166	-.233	-.299	-.344	-.384	-.469	-.611	.719
.775	-.073	-.094	-.105	-.087	-.137	-.170	-.268	-.356	.775
.830	-.083	-.064	-.057	-.048	-.054	-.034	-.057	-.047	.830
.871	-.107	-.077	-.062	-.070	-.089	-.099	-.117	-.144	.871
Row C									
.055	.113	.139	.123	.077	.044	.016	-.069	-.187	.055
.166	.036	.053	.039	-.022	-.046	-.073	-.169	-.300	.166
.277	.001	.009	-.003	-.077	-.070	-.111	-.220	-.338	.277
.353	-.033	-.026	-.044	-.103	-.107	-.181	-.284	-.152	.353
.367	.061	.061	.053	-.025	-.019	-.059	-.150	-.074	.367
.747	-.004	-.005	-.005	-.004	-.005	-.004	-.004	-.005	.747
.775	-.070	-.075	-.066	-.047	-.069	-.097	-.144	-.192	.775
.830	-.077	-.069	-.068	-.077	-.080	-.067	-.128	-.181	.830
.871	-.086	-.067	-.069	-.070	-.097	-.112	-.153	-.215	.871
.954	.043	.067	.055	-.002	-.068	-.113	-.166	-.216	.954
Row D									
.166	.034	.059	.073	.057	.062	.067	.059	.055	.166
.277	-.007	.010	.024	-.006	.034	.024	-.001	-.006	.277
.367	.003	.006	.014	-.030	-.001	-.036	-.049	.044	.367
.387	-.068	-.060	-.051	-.055	-.037	-.058	-.026	.082	.387
.443	-.083	-.042	.016	.116	.151	.192	.292	.363	.443
.498	-.119	-.048	.028	.143	.185	.242	.357	.448	.498
.553	-.160	-.086	-.006	.111	.153	.208	.320	.411	.553
.609	-.189	-.113	-.033	.079	.121	.173	.273	.359	.609
.664	-.230	-.159	-.077	.044	.089	.133	.216	.289	.664
.719	-.256	-.188	-.109	-.001	.045	.074	.133	.190	.719
.775	-.115	-.098	-.075	-.034	-.045	-.058	-.070	-.071	.775
.830	-.053	-.058	-.074	-.088	-.085	-.083	-.125	-.222	.830
.871	-.060	-.059	-.074	-.098	-.130	-.137	-.202	-.259	.871
Row E									
.055	.101	.142	.182	.239	.269	.303	.379	.467	.055
.166	.036	.059	.090	.119	.144	.173	.235	.311	.166
.277	-.004	.014	.042	.053	.107	.120	.156	.239	.277
.367	.000	-.004	.012	.050	.106	.098	.140	.303	.367
.387	-.029	-.025	-.004	.011	.048	.054	.159	.318	.387
.443	-.067	-.051	-.004	.093	.130	.180	.305	.410	.443
.498	-.118	-.063	.001	.111	.153	.208	.327	.429	.498
.553	-.146	-.080	-.008	.103	.144	.199	.308	.402	.553
.609	-.177	-.109	-.034	.075	.118	.169	.268	.352	.609
.664	-.220	-.152	-.079	.045	.088	.129	.215	.294	.664
.719	-.243	-.173	-.098	.005	.055	.087	.151	.209	.719
.775	-.156	-.123	-.085	-.034	-.024	-.020	.003	.034	.775
.830	-.045	-.056	-.081	-.094	-.081	-.076	-.105	-.147	.830
.871	-.070	-.076	-.099	-.134	-.145	-.136	-.164	-.163	.871
.954	.042	.069	.051	-.058	-.169	-.204	-.303	-.338	.954

TABLE IV.- PRESSURE COEFFICIENTS FOR BASIC BODY IN PRESENCE OF WING - Continued

(g) M = 1.12

x/l	α = −4°	α = 0°	α = 4°	α = 8°	α = 10°	α = 12°	α = 16°	α = 20°	x/l
Row A									
.055	.149	.101	.066	.044	.026	.015	-.001	.004	.055
.166	.062	.029	.002	-.012	-.019	-.028	-.034	-.031	.166
.277	.023	-.005	-.018	-.025	-.029	-.026	-.029	.102	.277
.367	.004	-.003	.000	-.012	-.007	-.001	-.048	.237	.367
.387	-.025	-.040	-.041	-.033	-.044	-.053	-.057	.030	.387
.415	-.048	-.066	-.057	-.041	-.040	-.047	-.076	.085	.415
.443	-.027	-.029	-.031	-.051	-.046	-.051	-.052	-.001	.443
.498	.048	-.017	-.073	-.127	-.147	-.165	-.194	-.166	.498
.553	.037	-.038	-.096	-.157	-.180	-.201	-.230	-.211	.553
.581	.012	-.059	-.114	-.174	-.196	-.213	-.240	-.259	.581
.609	-.005	-.076	-.127	-.187	-.205	-.226	-.258	-.273	.609
.636	-.026	-.093	-.141	-.200	-.217	-.241	-.274	-.279	.636
.664	-.034	-.100	-.152	-.208	-.226	-.246	-.298	-.301	.664
.692	-.011	-.083	-.145	-.186	-.205	-.230	-.281	-.285	.692
.719	-.028	-.103	-.168	-.209	-.236	-.262	-.305	-.316	.719
.747	-.047	-.114	-.190	-.243	-.282	-.314	-.348	-.355	.747
.775	-.036	-.089	-.126	-.182	-.225	-.265	-.327	-.364	.775
.830	-.043	-.045	-.021	.000	.029	.047	.109	.162	.830
.871	-.087	-.056	-.043	-.046	-.033	-.023	-.006	-.031	.871
.954	-.063	-.034	.002	-.040	-.094	-.141	-.200	-.162	.954
Row B									
.166	.050	.034	.007	-.025	-.047	-.066	-.097	-.132	.166
.367	-.012	-.006	-.019	-.030	-.028	-.042	-.078	-.117	.367
.387	-.022	-.035	-.048	-.046	-.049	-.055	-.100	-.163	.387
.443	.036	.000	-.039	-.084	-.100	-.117	-.156	-.179	.443
.498	.058	-.014	-.081	-.147	-.178	-.212	-.263	-.324	.498
.553	.047	-.033	-.103	-.174	-.211	-.236	-.290	-.387	.553
.609	.008	-.071	-.127	-.205	-.238	-.271	-.337	-.450	.609
.664	-.022	-.095	-.157	-.223	-.255	-.293	-.375	-.508	.664
.719	-.026	-.098	-.172	-.237	-.277	-.318	-.395	-.551	.719
.775	-.037	-.068	-.088	-.103	-.125	-.152	-.215	-.258	.775
.830	-.038	-.042	-.031	-.036	-.029	-.018	.001	.039	.830
.871	-.082	-.054	-.046	-.046	-.040	-.029	-.022	-.067	.871
Row C									
.055	.091	.111	.097	.058	.027	-.008	-.091	-.203	.055
.166	.017	.030	.016	-.027	-.059	-.095	-.188	-.324	.166
.277	-.010	.000	-.010	-.059	-.093	-.129	-.226	-.338	.277
.353	-.043	-.036	-.050	-.082	-.126	-.174	-.262	-.333	.353
.367	.038	.052	.032	-.046	-.084	-.086	-.189	-.334	.367
.747	-.008	-.008	-.009	-.007	-.007	-.008	-.006	-.007	.747
.775	-.039	-.043	-.036	-.032	-.046	-.068	-.079	-.110	.775
.830	-.044	-.045	-.042	-.059	-.055	-.049	-.062	-.111	.830
.871	-.071	-.058	-.060	-.067	-.060	-.071	-.101	-.134	.871
.954	-.053	-.053	-.042	-.060	-.090	-.116	-.131	-.101	.954
Row D									
.166	.013	.034	.052	.056	.054	.053	.052	.055	.166
.277	-.013	.001	.014	.016	.012	.013	-.001	-.007	.277
.367	-.019	-.012	-.010	-.031	-.049	-.058	-.055	-.024	.367
.387	-.054	-.045	-.031	-.002	-.013	-.036	-.036	-.031	.387
.443	-.065	-.013	.016	.074	.110	.142	.263	.370	.443
.498	-.089	-.012	.067	.146	.198	.246	.370	.487	.498
.553	-.120	-.049	.033	.130	.182	.229	.343	.456	.553
.609	-.141	-.075	-.001	.100	.155	.203	.309	.413	.609
.664	-.173	-.103	-.040	.071	.123	.170	.268	.360	.664
.719	-.192	-.117	-.040	.040	.088	.122	.210	.281	.719
.775	-.093	-.068	-.043	-.008	.004	.003	.017	.043	.775
.830	-.024	-.036	-.041	-.046	-.043	-.046	-.063	-.143	.830
.871	-.034	-.044	-.065	-.098	-.099	-.112	-.141	-.163	.871
Row E									
.055	.076	.116	.158	.222	.253	.283	.358	.452	.055
.166	.012	.034	.068	.119	.136	.159	.224	.308	.166
.277	-.008	.001	.027	.074	.096	.109	.160	.233	.277
.367	-.022	-.022	.002	.056	.080	.094	.120	.225	.367
.387	-.028	-.024	-.006	.035	.046	.057	.119	.191	.387
.443	-.048	-.040	-.045	-.017	.005	.021	.086	.340	.443
.498	-.084	-.028	.041	.117	.165	.212	.331	.455	.498
.553	-.110	-.045	.028	.122	.174	.219	.334	.447	.553
.609	-.132	-.072	-.002	.096	.154	.201	.305	.411	.609
.664	-.166	-.101	-.041	.076	.123	.165	.267	.367	.664
.719	-.177	-.113	-.035	.051	.092	.136	.224	.299	.719
.775	-.147	-.093	-.046	.021	.053	.070	.105	.147	.775
.830	-.010	-.038	-.045	-.035	-.027	-.038	-.046	-.039	.830
.871	-.033	-.052	-.075	-.089	-.086	-.085	-.080	-.091	.871
.954	-.048	-.069	-.091	-.153	-.181	-.221	-.221	-.223	.954

TABLE IV.- PRESSURE COEFFICIENTS FOR BASIC BODY IN PRESENCE OF WING - Concluded

(h) M = 1.43

x/l	α = -3.6°	α = -0.3°	α = 4.3°	α = 8.8°	α = 11.1°	α = 13.4°	α = 16.0°	α = 19.8°	x/l
Row A									
.055	.161	.101	.062	.032	.019	.011	.003	-.016	.055
.166	.047	.021	.011	-.003	-.005	-.017	-.023	-.051	.166
.277	.055	.020	-.005	-.009	-.011	-.019	-.013	-.046	.277
.367	.015	.031	.006	-.004	.002	-.009	-.037	-.064	.367
.387	.012	.007	-.008	-.017	.003	-.005	-.014	-.043	.387
.415	.003	-.017	-.028	-.031	-.037	-.050	-.075	-.054	.415
.443	.012	-.018	-.029	-.015	-.024	-.029	-.038	-.112	.443
.498	.033	-.008	-.041	-.073	-.090	-.098	-.129	-.144	.498
.553	.032	-.014	-.077	-.117	-.131	-.152	-.178	-.194	.553
.581	.031	-.018	-.078	-.117	-.140	-.158	-.177	-.177	.581
.609	.030	-.028	-.086	-.124	-.144	-.163	-.177	-.190	.609
.636	.018	-.033	-.093	-.140	-.163	-.175	-.190	-.212	.636
.664	.006	-.050	-.105	-.145	-.167	-.184	-.206	-.218	.664
.692	.000	-.052	-.104	-.150	-.167	-.191	-.212	-.217	.692
.719	-.013	-.065	-.123	-.165	-.187	-.205	-.216	-.220	.719
.747	-.050	-.098	-.157	-.198	-.222	-.240	-.244	-.241	.747
.775	-.021	-.074	-.132	-.178	-.204	-.218	-.241	-.230	.775
.830	-.051	-.064	-.063	-.062	-.068	-.080	-.056	-.015	.830
.871	-.065	-.058	-.050	-.020	.000	.043	.082	.115	.871
.954	-.106	-.078	-.034	-.076	-.082	-.088	-.095	-.131	.954
Row B									
.166	.032	.030	.016	-.029	-.039	-.061	-.088	-.126	.166
.277	.035	.017	-.018	-.034	-.051	-.066	-.084	-.141	.277
.367	-.010	-.002	-.007	-.036	-.047	-.065	-.092	-.152	.367
.387	.031	.032	.009	-.024	-.037	-.050	-.088	-.127	.387
.443	.012	.010	-.037	-.069	-.092	-.112	-.139	-.187	.443
.498	.051	-.003	-.063	-.115	-.152	-.177	-.205	-.258	.498
.553	.034	-.021	-.080	-.144	-.175	-.208	-.246	-.304	.553
.609	.024	-.026	-.095	-.157	-.189	-.225	-.262	-.335	.609
.664	-.003	-.061	-.120	-.177	-.213	-.240	-.287	-.358	.664
.719	-.017	-.073	-.129	-.189	-.231	-.262	-.305	-.379	.719
.775	-.027	-.062	-.103	-.124	-.131	-.148	-.169	-.232	.775
.830	-.048	-.059	-.063	-.072	-.072	-.075	-.075	-.099	.830
.871	-.052	-.052	-.047	-.023	-.012	.012	.018	-.020	.871
Row C									
.055	.085	.094	.093	.040	.012	-.026	-.077	-.158	.055
.166	.010	.023	.020	-.027	-.056	-.105	-.162	-.276	.166
.277	.012	.021	-.013	-.049	-.087	-.138	-.226	-.324	.277
.367	-.011	-.007	-.007	-.070	-.120	-.176	-.231	-.354	.367
.747	.103	.099	.097	.105	.105	.106	.105	.104	.747
.775	-.038	-.033	-.035	-.037	-.042	-.030	-.015	-.033	.775
.830	-.059	-.056	-.054	-.058	-.057	-.067	-.080	-.136	.830
.871	.102	.099	.096	.105	.106	.106	.105	.104	.871
.954	-.078	-.070	-.073	-.089	-.089	-.090	-.102	-.097	.954
Row D									
.166	.007	.022	.049	.079	.084	.094	.095	.129	.166
.277	.006	.024	.029	.023	.030	.038	.041	.059	.277
.367	.032	.033	.040	.053	.044	.036	.035	.058	.367
.387	-.012	-.002	.009	.020	.009	.001	-.002	.004	.387
.443	-.031	-.005	.009	-.026	-.040	-.043	-.051	-.042	.443
.609	-.092	-.040	.037	.117	.164	.228	.300	.389	.609
.664	-.110	-.059	.005	.096	.144	.200	.261	.357	.664
.719	-.132	-.084	-.014	.072	.117	.166	.228	.322	.719
.775	-.097	-.068	-.026	.028	.054	.082	.121	.161	.775
.830	-.061	-.052	-.039	-.026	-.018	-.010	-.002	-.004	.830
.871	-.042	-.045	-.062	-.074	-.077	-.079	-.095	-.132	.871
Row E									
.055	.059	.096	.169	.229	.266	.317	.372	.469	.055
.166	.009	.021	.069	.137	.175	.218	.252	.350	.166
.277	-.001	.014	.050	.080	.109	.149	.194	.275	.277
.387	-.004	-.015	.011	.053	.080	.101	.141	.206	.387
.443	-.037	-.022	-.015	.010	.032	.059	.103	.165	.443
.498	-.049	-.022	.022	.050	.061	.070	.092	.180	.498
.553	-.069	-.018	.038	.146	.172	.206	.269	.374	.553
.609	-.083	-.039	.041	.121	.169	.224	.303	.408	.609
.664	-.103	-.058	.007	.090	.145	.203	.263	.353	.664
.719	-.110	-.068	.008	.092	.138	.185	.246	.343	.719
.775	-.132	-.089	-.022	.066	.109	.156	.213	.289	.775
.830	-.071	-.065	-.046	-.007	.014	.038	.071	.119	.830
.871	-.040	-.048	-.046	-.035	-.027	-.014	.001	.021	.871
.954	-.037	-.055	-.081	-.091	-.099	-.111	-.118	-.128	.954

TABLE V.- PRESSURE COEFFICIENTS FOR INDENTED BODY IN PRESENCE OF WING

(a) M = 0.60

x/l	α = −4°	α = 0°	α = 4°	α = 8°	α = 10°	α = 12°	α = 16°	α = 20°	x/l
				Row A					
.055	.101	.055	.015	-.020	-.028	-.038	-.055	-.056	.055
.166	.031	-.001	-.027	-.038	-.041	-.047	-.062	-.068	.166
.277	.006	-.021	-.042	-.049	-.053	-.054	-.057	-.079	.277
.367	-.008	-.006	-.009	-.037	-.040	-.046	-.981	-.117	.367
.387	-.027	-.059	-.079	-.106	-.131	-.145	-.192	-.244	.387
.415	-.054	-.095	-.115	-.165	-.208	-.231	-.310	-.424	.415
.443						-.260	-.349	-.478	.443
.498	.008	-.051	-.098	-.164	-.209	-.241	-.313	-.402	.498
.553	.010	-.045	-.090	-.152	-.191	-.210	-.250	-.292	.553
.581	.020	-.031	-.072	-.128	-.165	-.176	-.199	-.231	.581
.609	.032	-.017	-.052	-.096	-.122	-.131	-.139	-.152	.609
.636	.031	-.011	-.042	-.073	-.092	-.089	-.097	-.097	.636
.664	.003	-.033	-.058	-.086	-.097	-.099	-.108	-.117	.664
.692	-.008	-.039	-.057	-.074	-.085	-.080	-.095	-.114	.692
.719	-.017	-.038	-.050	-.062	-.067	-.064	-.081	-.119	.719
.747	-.015	-.029	-.035	-.041	-.041	-.041	-.050	-.100	.747
.775	-.031	-.039	-.041	-.046	-.050	-.048	-.056	-.106	.775
.830	-.015	-.018	-.016	-.013	-.017	-.018	-.021	-.064	.830
.871	-.013	-.018	-.008	-.007	-.011	-.014	-.021	-.053	.871
.954	.030	.037	.037	.031	.029	.025	.016	.016	.954
				Row B					
.166	.020	.007	-.020	-.059	-.077	-.094	-.128	-.164	.166
.277	-.012	-.021	-.045	-.074	-.090	-.104	-.125	-.164	.277
.367	.014	.001	-.022	-.043	-.062	-.075	-.117	-.167	.367
.387	-.043	-.057	-.080	-.125	-.149	-.167	-.204	-.274	.387
.443	.000	-.059	-.120	-.208	-.260	-.309	-.444	-.604	.443
.498	.016	-.059	-.128	-.217	-.274	-.336	-.472	-.627	.498
.553	.010	-.055	-.113	-.191	-.239	-.291	-.367	-.502	.553
.609	.033	-.021	-.062	-.125	-.166	-.186	-.244	-.413	.609
.664	.001	-.041	-.069	-.110	-.139	-.153	-.235	-.476	.664
.719	-.007	-.038			-.111	-.128	-.232	-.391	.719
.775	-.023	-.029	-.031	-.039	-.051	-.075	-.126	-.108	.775
.830	-.021	-.024	-.022	-.025	-.035	-.055	-.052	-.060	.830
.871	.010	.002	.011	.008	.010	.003	.025	.006	.871
				Row C					
.055	.042	.071	.048	-.008	-.037	-.080	-.182	-.312	.055
.166	-.013	.001	-.011	-.059	-.087	-.126	-.228	-.351	.166
.277	-.037	-.022	-.037	-.082	-.113	-.154	-.249	-.365	.277
.353	-.016	.001	-.015	-.067	-.105	-.149	-.252	-.373	.353
.367	-.019	.005	-.015	-.059	-.097	-.140	-.240	-.355	.367
.747	.011	.001	.018	.017	.017	.007	.024		.747
.775	-.011	-.014	-.010	-.009	-.013	-.010	-.007		.775
.830	-.016	-.022	-.014	-.017	-.021	-.021	-.012	-.006	.830
.871	.000	-.003	.002	-.000	-.004	-.005	.001	.011	.871
.954	.044	.042	.046	.050	.048				.954
				Row D					
.166	-.024	.004	.018	.022	.021	.019	.001	-.017	.166
.277	-.045	-.021	-.012	-.012	-.013	-.016	-.032	-.052	.277
.367	-.032	-.007	.008	.009	.008	.007	-.022	-.029	.367
.387	-.089	-.065	-.046	-.035	-.015	-.012	-.017	-.039	.387
.443	-.140	-.064	-.007	.040	.058	.087	.131	.162	.443
.498	-.143	-.057	.015	.091	.125	.163	.242	.310	.498
.553	-.125	-.054	.009	.084	.119	.156	.233	.302	.553
.609	-.071	-.017	.036	.096	.123	.151	.214	.271	.609
.664	-.069	-.034	.008	.048	.069	.089	.138	.179	.664
.719	-.049	-.033	-.009	.015	.026	.040	.066	.083	.719
.775	-.035	-.033	-.024	-.015	-.010	-.007	.006	-.000	.775
.830	-.017	-.018	-.014	-.011	-.012	-.011	.000	.001	.830
.871	.000	.001	.005	.007	.008	.007	.019	.032	.871
				Row E					
.055	.032	.073	.104	.165	.196	.225	.298	.375	.055
.166	-.023	.007	.035	.083	.107	.129	.194	.260	.166
.277	-.045	-.021	.002	.042	.064	.085	.144	.213	.277
.367	-.043	-.021	.010	.052	.075	.108	.174	.239	.367
.387	-.090	-.069	-.039	.009	.059	.090	.162	.232	.387
.443	-.121	-.077	-.027	.029	.059	.097	.190	.262	.443
.498	-.128	-.067	-.012	.058	.092	.128	.217	.291	.498
.553	-.111	-.056	.000	.067	.094	.129	.203	.273	.553
.609	-.067	-.017	.030	.088	.114	.143	.211	.267	.609
.664	-.062	-.031	.010	.053	.072	.100	.147	.195	.664
.719	-.057	-.042	-.017	.008	.019	.034	.072	.094	.719
.775	-.042	-.038	-.025	-.013	-.010	.000	.029	.036	.775
.830	-.017	-.017	-.009	-.004	-.003	.003	.024	.038	.830
.871	-.014	-.012	-.014	-.010	-.014	-.011	.006	.024	.871
.954	.039	.035	.030	.029	.021	.016	.024	.035	.954

TABLE V.- PRESSURE COEFFICIENTS FOR INDENTED BODY IN PRESENCE OF WING - Continued

(b) M = 0.80

x/l	α = −4°	α = 0°	α = 4°	α = 8°	α = 10°	α = 12°	α = 16°	α = 20°	x/l
				Row A					
.055	.112	.066	.027	-.007	-.017	-.025	-.040	-.040	.055
.166	.036	.004	-.020	-.033	-.038	-.042	-.052	-.056	.166
.277	.005	-.022	-.038	-.043	-.044	-.042	-.039	-.054	.277
.367	-.001	.001		-.017	-.013	-.013		-.047	.367
.387	-.028	-.059	-.074	-.096	-.111	-.117	-.142	-.179	.387
.415	-.058	-.096	-.112	-.158	-.199	-.216	-.279	-.383	.415
.443						-.266	-.352	-.482	.443
.498	.014	-.054	-.104	-.185	-.242	-.284	-.385	-.510	.498
.553	.015	-.050	-.103	-.181	-.235	-.266	-.331	-.371	.553
.581	.025	-.036	-.084	-.156	-.206	-.225	-.251	-.338	.581
.609	.038	-.016	-.061	-.120	-.158	-.164	-.164	-.250	.609
.636	.037	-.015	-.051	-.093	-.121	-.116	-.117	-.128	.636
.664	.002	-.040	-.067	-.109	-.130	-.127	-.164	-.138	.664
.692	-.009	-.044	-.064	-.087	-.101	-.101	-.163	-.149	.692
.719	-.016	-.041	-.050	-.067	-.072	-.079	-.125	-.158	.719
.747	-.010	-.028	-.031	-.035	-.040	-.046	-.067	-.129	.747
.775	-.031	-.039	-.037	-.041	-.047	-.051	-.054	-.131	.775
.830	-.015	-.018	-.013	-.013	-.016	-.018	-.021	-.077	.830
.871	-.012	-.018	-.006	-.007	-.011	-.012	-.018	-.059	.871
.954	.037	.042	.045	.037	.032	.029	.015	.015	.954
				Row B					
.166	.025	.009	-.016	-.051	-.068	-.084	-.116	-.148	.166
.277	-.011	-.023	-.042	-.068	-.082	-.090	-.109	-.138	.277
.367	.023	.009	-.009	-.024	-.037	-.043	-.067	-.092	.367
.387	-.041	-.059	-.074	-.105	-.116	-.125	-.139	-.173	.387
.443	.004	-.061	-.123	-.211	-.263	-.306	-.421	-.532	.443
.498	.020	-.064	-.137	-.237	-.303	-.372	-.535	-.729	.498
.553	.013	-.063	-.126	-.220	-.283	-.339	-.447	-.559	.553
.609	.039	-.024	-.072	-.151	-.200	-.211	-.258	-.301	.609
.664	.001	-.046	-.079	-.131	-.165	-.161	-.294	-.300	.664
.719					-.100	-.140	-.260	-.369	.719
.775	-.021	-.029	-.028	-.035	-.047	-.076	-.144	-.179	.775
.830	-.021	-.024	-.020	-.027	-.041	-.053	-.069	-.084	.830
.871	.014	.000	.012	.010	.009	.008	.024	.001	.871
				Row C					
.055	.054	.082	.061	.005	-.023	-.060	-.162	-.285	.055
.166	-.008	.005	-.008	-.054	-.081	-.116	-.214	-.336	.166
.277	-.033	-.022	-.034	-.079	-.108	-.140	-.232	-.334	.277
.353	-.008	.002	-.011	-.055	-.090	-.123	-.210	-.304	.353
.367	-.007	.005	-.007	-.048	-.076	-.106	-.187	-.270	.367
.747	.015	.005	.024	.024	.019	-.016	-.005		.747
.775	-.009	-.013	-.007	-.006	-.008	-.023	-.022		.775
.830	-.017	-.022	-.013	-.016	-.021	-.018	-.019	-.016	.830
.871	.003	-.004	.005	.002	-.003	-.000	.004	.015	.871
.954	.050	.048	.053	.057	.051				.954
				Row D					
.166	-.016	.007	.023	.027	.029	.028	.017	.005	.166
.277	-.039	-.022	-.010	-.009	-.008	-.009	-.021	-.029	.277
.367	-.019	.000	.013	.014	.013	.020	-.002	.001	.367
.387	-.081	-.065	-.046	-.036	-.011	.001	.007	.002	.387
.443	-.134	-.064	-.002	.052	.071	.104	.162	.206	.443
.498	-.149	-.062	.017	.101	.136	.183	.264	.338	.498
.553	-.141	-.062	.012	.094	.129	.170	.250	.326	.553
.609	-.081	-.020	.042	.107	.137	.170	.233	.295	.609
.664	-.081	-.041	.008	.057	.081	.102	.151	.198	.664
.719	-.050	-.035	-.008	.018	.031	.037	.065	.078	.719
.775	-.032	-.032	-.020	-.013	-.010	-.020	-.010	-.051	.775
.830	-.014	-.021	-.012	-.012	-.014	-.014	-.010	-.032	.830
.871	.003	-.003	.006	.009	.007	.011	.019	.022	.871
				Row E					
.055	.045	.085	.123	.179	.209	.243	.317	.397	.055
.166	-.013	.008	.041	.090	.114	.141	.205	.277	.166
.277	-.039	-.024	.004	.045	.067	.093	.152	.224	.277
.367	-.028	-.015	.015	.058	.081	.120	.182	.255	.367
.387	-.081	-.071	-.041	.010	.060	.098	.172	.249	.387
.443	-.120	-.079	-.027	.039	.066	.112	.204	.285	.443
.498	-.134	-.070	-.005	.069	.105	.149	.237	.315	.498
.553	-.129	-.063	.003	.077	.108	.149	.225	.296	.553
.609	-.075	-.019	.037	.101	.131	.164	.232	.291	.609
.664	-.073	-.037	.011	.062	.085	.113	.160	.212	.664
.719	-.059	-.045	-.015	-.016	.026	.040	.071	.091	.719
.775	-.036	-.037	-.026	-.012	-.009	-.004	.009	-.002	.775
.830	-.014	-.018	-.010	-.005	-.005	.001	.013	.008	.830
.871	-.011	-.015	-.013	-.012	-.015	-.009	-.003	.007	.871
.954	.044	.042	.036	.031	.025	.023	.026	.034	.954

TABLE V.- PRESSURE COEFFICIENTS FOR INDENTED BODY IN PRESENCE OF WING - Continued

(c) M = 0.90

x/l	α = −4°	α = 0°	α = 4°	α = 8°	α = 10°	α = 12°	α = 16°	α = 20°	x/l
				Row A					
.055	.124	.076	.036	.004	-.006	-.016	-.030	-.030	.055
.166	.039	.006	-.018	-.029	-.034	-.038	-.043	-.058	.166
.277	.005	-.024	-.040	-.044	-.042	-.038	-.028	-.032	.277
.367	-.002	.010		.011	.016	.022		.024	.367
.387	-.030	-.060	-.070	-.080	-.083	-.082	-.086	-.096	.387
.415	-.066	-.100	-.113	-.146	-.170	-.175	-.216	-.291	.415
.443						-.236	-.290	-.364	.443
.498	.016	-.056	-.115	-.201	-.253	-.293	-.372	-.477	.498
.553	.012	-.058	-.123	-.225	-.286	-.334	-.428	-.513	.553
.581	.025	-.042	-.102	-.201	-.284	-.333	-.436	-.439	.581
.609	.042	-.020	-.077	-.156	-.224	-.287	-.382	-.186	.609
.636	.039	-.021	-.069	-.132	-.186	-.237	-.124	-.247	.636
.664	.000	-.052	-.091	-.154	-.201	-.249	-.068	-.307	.664
.692	-.015	-.057	-.081	-.112	-.132	-.128	-.145	-.249	.692
.719	-.020	-.047	-.058	-.068	-.072	-.060	-.257	-.197	.719
.747	-.010	-.026	-.029	-.028	-.027	-.014	-.089	-.131	.747
.775	-.032	-.040	-.036	-.038	-.038	-.026	-.077	-.192	.775
.830	-.018	-.021	-.017	-.016	-.018	-.017	-.027	-.124	.830
.871	-.013	-.019	-.009	-.008	-.011	-.015	-.011	-.065	.871
.954	.044	.048	.047	.040	.034	.027	.018	.018	.954
				Row B					
.166	.030	.012	-.015	-.047	-.064	-.081	-.105	-.138	.166
.277	-.012	-.024	-.046	-.068	-.077	-.085	-.098	-.122	.277
.367	.028	.016	-.000	-.008	-.013	-.014	-.023	-.031	.367
.387	-.044	-.058	-.071	-.089	-.088	-.087	-.081	-.096	.387
.443	.008	-.063	-.125	-.202	-.236	-.267	-.333	-.420	.443
.498	.019	-.067	-.148	-.254	-.310	-.376	-.499	-.618	.498
.553	.011	-.069	-.147	-.263	-.330	-.404	-.522	-.632	.553
.609	.040	-.027	-.090	-.187	-.263	-.339	-.458	-.443	.609
.664	-.004	-.060	-.104	-.177	-.235	-.300	-.206	-.551	.664
.719		-.047				-.114	-.358	-.586	.719
.775	-.022	-.028	-.028	-.033	-.042	-.049	-.156	-.422	.775
.830	-.025	-.028	-.024	-.031	-.046	-.062	-.095	-.169	.830
.871	.012	-.000	.008	.009	.007	.000	.013	-.020	.871
				Row C					
.055	.067	.092	.068	.018	-.010	-.047	-.143	-.256	.055
.166	-.005	.007	-.006	-.051	-.076	-.112	-.209	-.326	.166
.277	-.034	-.024	-.038	-.079	-.103	-.137	-.220	-.317	.277
.353	-.005	.005	-.011	-.047	-.075	-.108	-.177	-.254	.353
.367	.001	.009	-.002	-.034	-.058	-.082	-.144	-.207	.367
.747	.021	.007	.027	.028	.021	.009	-.056		.747
.775	-.008	-.011	-.005	-.008	-.001	.019	-.074		.775
.830	-.019	-.026	-.016	-.019	-.023	-.020	-.043	-.107	.830
.871	.003	-.004	.004	.002	-.002	-.008	-.005	-.016	.871
.954	.054	.054	.054	.057	.053				.954
				Row D					
.166	-.012	.009	.024	.030	.033	.033	.025	.021	.166
.277	-.040	-.024	-.015	-.011	-.008	-.008	-.013	-.015	.277
.367	-.007	.005	.015	.016	.021	.026	.015	.023	.367
.387	-.070	-.063	-.048	-.036	-.003	.010	.026	.031	.387
.443	-.129	-.063	-.003	.058	.085	.123	.191	.245	.443
.498	-.159	-.065	.015	.106	.146	.194	.284	.369	.498
.553	-.164	-.071	.007	.097	.137	.181	.265	.350	.553
.609	-.100	-.025	.042	.112	.147	.181	.250	.319	.609
.664	-.111	-.054	.003	.060	.087	.117	.163	.221	.664
.719	-.057	-.039	-.013	.018	.036	.057	.063	.093	.719
.775	-.030	-.031	-.024	-.017	-.012	.001	-.064	-.136	.775
.830	-.016	-.023	-.017	-.017	-.018	-.013	-.040	-.141	.830
.871	.001	-.002	.003	.005	.004	.003	.007	-.026	.871
				Row E					
.055	.055	.094	.132	.191	.218	.253	.330	.414	.055
.166	-.010	.012	.043	.092	.119	.147	.213	.288	.166
.277	-.039	-.026	-.002	.042	.067	.093	.158	.232	.277
.367	-.016	-.008	.019	.061	.082	.125	.189	.269	.367
.387	-.073	-.069	-.043	.008	.062	.103	.183	.265	.387
.443	-.116	-.081	-.030	.039	.074	.124	.223	.312	.443
.498	-.145	-.074	-.008	.073	.112	.162	.256	.345	.498
.553	-.151	-.069	.000	.080	.115	.160	.239	.323	.553
.609	-.092	-.026	.038	.108	.140	.180	.249	.318	.609
.664	-.101	-.049	.005	.066	.092	.125	.170	.235	.664
.719	-.063	-.051	-.020	.012	.030	.057	.068	.102	.719
.775	-.034	-.037	-.029	-.020	-.012	.005	-.035	-.065	.775
.830	-.014	-.020	-.014	-.011	-.012	-.004	-.018	-.088	.830
.871	-.011	-.018	-.019	-.019	-.021	-.020	-.018	-.042	.871
.954	.046	.044	.039	.034	.029	.022	.027	.034	.954

TABLE V.- PRESSURE COEFFICIENTS FOR INDENTED BODY IN PRESENCE OF WING - Continued

(d) M = 0.94

x/l	α = −4°	α = 0°	α = 4°	α = 8°	α = 10°	α = 12°	α = 16°	α = 20°	x/l
				Row A					
.055	.130	.083	.045	.010	.000	-.009	-.023	-.024	.055
.166	.041	.008	-.015	-.029	-.032	-.034	-.039	-.052	.166
.277	.002	-.025	-.041	-.043	-.042	-.034	-.019	-.024	.277
.367	-.001	.017		.031	.030	.044		.061	.367
.387	-.032	-.062	-.067	-.070	-.071	-.059	-.051	-.058	.387
.415	-.070	-.105	-.111	-.132	-.152	-.149	-.179	-.231	.415
.443						-.208	-.246	-.308	.443
.498	.017	-.058	-.118	-.198	-.238	-.274	-.342	-.444	.498
.553	.010	-.064	-.136	-.232	-.285	-.326	-.402	-.473	.553
.581	.025	-.043	-.112	-.231	-.290	-.325	-.418	-.486	.581
.609	.043	-.023	-.083	-.194	-.249	-.288	-.372	-.428	.609
.636	.038	-.025	-.081	-.166	-.213	-.248	-.329	-.185	.636
.664	-.004	-.063	-.118	-.213	-.262	-.286	-.346	-.160	.664
.692	-.023	-.069	-.108	-.189	-.299	-.334	-.346	-.298	.692
.719	-.023	-.049	-.056	-.060	-.128	-.339	-.060	-.420	.719
.747	-.011	-.025	-.021	-.013	-.005	.026	-.094	-.408	.747
.775	-.032	-.038	-.030	-.029	-.020	.009	-.092	-.133	.775
.830	-.020	-.023	-.015	-.015	-.014	-.007	-.037	-.158	.830
.871	-.015	-.020	-.009	-.004	-.008	-.007	-.001	-.055	.871
.954	.046	.051	.051	.041	.036	.031	.024	.012	.954
				Row B					
.166	.031	.014	-.011	-.043	-.061	-.077	-.100	-.134	.166
.277	-.014	-.027	-.046	-.068	-.077	-.083	-.091	-.114	.277
.367	.032	.020	.007	.002	-.001	-.002	.001	-.003	.367
.387	-.046	-.061	-.067	-.079	-.074	-.065	-.050	-.062	.387
.443	.010	-.063	-.122	-.190	-.218	-.238	-.286	-.362	.443
.498	.021	-.071	-.151	-.248	-.296	-.355	-.456	-.561	.498
.553	.010	-.076	-.161	-.267	-.329	-.388	-.486	-.603	.553
.609	.040	-.030	-.096	-.222	-.289	-.345	-.452	-.485	.609
.664	-.009	-.073	-.135	-.237	-.309	-.361	-.452	-.498	.664
.719						-.269	-.240	-.654	.719
.775	-.021	-.026	-.023	-.028	-.030	-.023	-.167	-.293	.775
.830	-.028	-.029	-.025	-.032	-.047	-.047	-.100	-.289	.830
.871	.011	-.002	.010	.010	.008	.005	.001	-.067	.871
				Row C					
.055	.071	.097	.075	.024	-.002	-.039	-.130	-.245	.055
.166	-.005	.008	-.005	-.048	-.074	-.111	-.202	-.320	.166
.277	-.037	-.026	-.039	-.080	-.105	-.137	-.213	-.309	.277
.353	-.004	.004	-.007	-.044	-.068	-.096	-.152	-.222	.353
.367	.005	.015	.004	-.024	-.046	-.068	-.118	-.173	.367
.747	.026	.012	.035	.032	.013	-.070	-.085		.747
.775	-.005	-.009	-.002	-.004	.005	.029	-.120		.775
.830	-.023	-.027	-.017	-.019	-.020	-.016	-.077	-.239	.830
.871	.002	-.004	.004	.004	.001	-.003	-.010	-.099	.871
.954	.055	.056	.057	.059	.055				.954
				Row D					
.166	-.012	.011	.024	.032	.034	.036	.033	.027	.166
.277	-.043	-.027	-.014	-.011	-.009	-.007	-.007	-.008	.277
.367	-.003	.009	.019	.021	.025	.032	.028	.034	.367
.387	-.068	-.063	-.048	-.033	-.000	.016	.042	.053	.387
.443	-.126	-.063	-.002	.063	.093	.136	.211	.271	.443
.498	-.163	-.068	.015	.106	.150	.201	.301	.389	.498
.553	-.180	-.079	.004	.097	.138	.186	.281	.366	.553
.609	-.110	-.028	.044	.114	.148	.186	.263	.339	.609
.664	-.150	-.069	-.000	.059	.085	.117	.179	.244	.664
.719	-.064	-.043	-.017	.016	.033	.049	.077	.123	.719
.775	-.026	-.030	-.021	-.018	-.013	-.009	-.098	-.118	.775
.830	-.019	-.023	-.017	-.018	-.020	-.016	-.091	-.264	.830
.871	.000	-.001	.004	.006	.005	.007	.004	-.135	.871
				Row E					
.055	.062	.099	.140	.194	.226	.261	.340	.424	.055
.166	-.011	.012	.047	.095	.119	.148	.216	.291	.166
.277	-.043	-.028	-.001	.040	.066	.092	.161	.237	.277
.367	-.011	-.003	.022	.062	.084	.131	.197	.278	.367
.387	-.072	-.068	-.043	.008	.064	.107	.195	.279	.387
.443	-.114	-.082	-.028	.041	.079	.134	.238	.331	.443
.498	-.149	-.078	-.007	.076	.119	.171	.273	.365	.498
.553	-.166	-.078	-.001	.081	.118	.167	.256	.343	.553
.609	-.103	-.028	.040	.111	.144	.186	.265	.340	.609
.664	-.139	-.063	.003	.064	.093	.126	.184	.257	.664
.719	-.072	-.055	-.024	.009	.026	.047	.083	.129	.719
.775	-.032	-.036	-.027	-.021	-.014	-.007	-.054	-.045	.775
.830	-.016	-.022	-.015	-.014	-.015	-.006	-.083	-.185	.830
.871	-.016	-.019	-.019	-.021	-.022	-.022	-.025	-.187	.871
.954	.048	.048	.043	.040	.033	.029	.035	.035	.954

TABLE V.- PRESSURE COEFFICIENTS FOR INDENTED BODY IN PRESENCE OF WING - Continued

(e) M = 0.98

x/l	α = −4°	α = 0°	α = 4°	α = 8°	α = 10°	α = 12°	α = 16°	α = 20°	x/l
Row A									
.055	.139	.094	.056	.021	.011	.001	-.013	-.016	.055
.166	.027	.010	-.014	-.027	-.030	-.034	-.037	-.051	.166
.277	-.004	-.033	-.049	-.052	-.047	-.039	-.017	-.008	.277
.367	-.003	.033		.067	.056	.067	.001	.104	.367
.387	-.030	-.055	-.061	-.055	-.048	-.033	-.014	-.008	.387
.415	-.082	-.145	-.118	-.110	-.121	-.115	-.133	-.171	.415
.443						-.166	-.195	-.239	.443
.498	.017	-.063	-.123	-.184	-.215	-.241	-.308	-.390	.498
.553	.006	-.078	-.143	-.220	-.255	-.289	-.365	-.430	.553
.581	.023	-.059	-.121	-.227	-.266	-.296	-.376	-.442	.581
.609	.043	-.016	-.103	-.192	-.234	-.267	-.343	-.405	.609
.636	.036	-.020	-.087	-.165	-.201	-.232	-.300	-.345	.636
.664	-.019	-.075	-.131	-.208	-.238	-.263	-.314	-.338	.664
.692	-.056	-.120	-.172	-.250	-.283	-.312	-.363	-.370	.692
.719	-.061	-.140	-.196	-.292	-.327	-.359	-.413	-.431	.719
.747	-.026	-.026	-.042	-.270	-.335	-.383	-.443	-.475	.747
.775	-.042	-.024	-.008	-.019	-.080	-.267	-.456	-.510	.775
.830	-.019	-.019	-.008	.008	-.012	-.025	-.001	.058	.830
.871	-.010	-.019	-.005	.016	.008	-.007	-.018	-.036	.871
.954	.059	.060	.058	.054	.053	.055	.034	-.077	.954
Row B									
.166	.033	.017	-.009	-.042	-.058	-.076	-.099	-.130	.166
.277	-.021	-.034	-.054	-.075	-.083	-.088	-.090	-.108	.277
.367	.044	.035	.021	.018	.018	.020	.025	.028	.367
.387	-.044	-.053	-.063	-.065	-.053	-.040	-.019	-.020	.387
.443	.016	-.059	-.112	-.162	-.181	-.196	-.238	-.304	.443
.498	.020	-.075	-.158	-.233	-.270	-.320	-.406	-.497	.498
.553	.006	-.089	-.163	-.255	-.303	-.354	-.440	-.548	.553
.609	.041	-.026	-.116	-.223	-.274	-.320	-.416	-.554	.609
.664	-.025	-.087	-.147	-.244	-.294	-.339	-.440	-.577	.664
.719						-.436	-.530	-.683	.719
.775	-.030	-.014	-.005	-.045	-.103	-.188	-.316	-.408	.775
.830	-.025	-.026	-.017	-.017	-.049	-.069	-.063	-.056	.830
.871	.013	-.002	.012	.021	.013	-.008	-.030	-.040	.871
Row C									
.055	.081	.107	.085	.036	.009	-.028	-.120	-.230	.055
.166	-.003	.010	-.003	-.046	-.071	-.109	-.201	-.321	.166
.277	-.044	-.032	-.045	-.086	-.110	-.141	-.216	-.299	.277
.353	.003	.009	-.000	-.029	-.047	-.067	-.107	-.148	.353
.367	.016	.028	.017	-.008	-.026	-.048	-.091	-.118	.367
.747	-.020	.003	.021	-.065	-.122	-.175	-.228		.747
.775	-.006	.003	.010	-.042	-.083	-.120	-.191		.775
.830	-.014	-.022	-.010	-.014	-.060	-.092	-.105	-.186	.830
.871	.009	-.003	.009	.017	.003	-.038	-.086	-.129	.871
.954	.063	.064	.065	.068	.074				.954
Row D									
.166	-.009	.013	.028	.033	.038	.037	.034	.036	.166
.277	-.051	-.033	-.021	-.015	-.010	-.010	-.006	-.001	.277
.367	.009	.021	.031	.032	.037	.044	.040	.049	.367
.387	-.061	-.055	-.043	-.026	.011	.029	.059	.077	.387
.443	-.108	-.058	.003	.076	.112	.156	.237	.302	.443
.498	-.163	-.072	.012	.115	.165	.219	.321	.415	.498
.553	-.179	-.095	-.003	.101	.150	.200	.297	.392	.553
.609	-.128	-.018	.041	.121	.159	.200	.282	.365	.609
.664	-.151	-.077	-.018	.056	.089	.129	.201	.275	.664
.719	-.223	-.138	-.051	-.008	.016	.044	.103	.158	.719
.775	-.014	-.013	-.016	-.063	-.084	-.110	-.132	-.134	.775
.830	-.008	-.018	-.013	-.025	-.099	-.130	-.150	-.247	.830
.871	.006	.001	.007	.017	.003	-.049	-.094	-.134	.871
Row E									
.055	.070	.110	.148	.205	.238	.270	.348	.436	.055
.166	-.009	.016	.047	.096	.123	.151	.218	.297	.166
.277	-.050	-.034	-.008	.037	.064	.090	.160	.242	.277
.367	.003	.010	.033	.069	.090	.136	.206	.291	.367
.387	-.064	-.058	-.037	.014	.072	.117	.206	.299	.387
.443	-.101	-.077	-.026	.052	.096	.150	.257	.357	.443
.498	-.151	-.081	-.011	.086	.132	.186	.293	.392	.498
.553	-.164	-.093	-.010	.084	.131	.180	.276	.371	.553
.609	-.122	-.018	.037	.116	.154	.198	.283	.366	.609
.664	-.140	-.070	-.013	.061	.099	.135	.205	.286	.664
.719	-.230	-.154	-.060	-.015	.010	.042	.107	.168	.719
.775	-.012	-.018	-.025	-.066	-.076	-.090	-.066	-.030	.775
.830	-.003	-.017	-.013	-.027	-.105	-.121	-.116	-.186	.830
.871	-.008	-.017	-.018	-.009	-.023	-.097	-.139	-.166	.871
.954	.055	.056	.051	.054	.055	.055	.047	-.165	.954

TABLE V.- PRESSURE COEFFICIENTS FOR INDENTED BODY IN PRESENCE OF WING - Continued

(f) M = 1.03

x/l	α = −4°	α = 0°	α = 4°	α = 8°	α = 10°	α = 12°	α = 16°	α = 20°	x/l
				Row A					
.055	.171	.122	.084	.056	.046	.037	.027	.025	.055
.166	.084	.048	.030	.017	.020	.011	-.005	-.029	.166
.277	.037	.018	-.008	-.019	-.019	-.020	-.035	-.044	.277
.367	-.005	.032		.111	.067	.058	.068	.169	.367
.387	-.002	-.015	-.030	-.046	-.054	-.051	.028	.069	.387
.415	-.114	-.124	-.121	-.145	-.160	-.139	-.063	-.083	.415
.443						-.131	-.116	-.152	.443
.498	.004	-.080	-.127	-.179	-.195	-.198	-.237	-.302	.498
.553	-.004	-.088	-.147	-.206	-.230	-.244	-.290	-.343	.553
.581	.009	-.082	-.146	-.210	-.236	-.253	-.296	-.358	.581
.609	.048	-.049	-.119	-.184	-.207	-.227	-.266	-.325	.609
.636	.055	-.032	-.101	-.153	-.173	-.188	-.218	-.274	.636
.664	.004	-.069	-.133	-.182	-.200	-.212	-.233	-.273	.664
.692	-.037	-.110	-.171	-.219	-.242	-.259	-.281	-.301	.692
.719	-.078	-.148	-.208	-.262	-.285	-.306	-.332	-.365	.719
.747	-.075	-.145	-.200	-.267	-.299	-.325	-.363	-.408	.747
.775	-.083	-.117	-.132	-.153	-.208	-.266	-.385	-.446	.775
.830	-.100	-.080	-.072	-.059	-.037	-.011	.020	.075	.830
.871	-.077	-.072	-.056	-.024	-.050	-.050	-.050	-.038	.871
.954	.009	.022	.020	-.039	-.108	-.183	-.272	-.353	.954
				Row B					
.166	.077	.057	.035	.003	-.008	-.032	-.067	-.107	.166
.277	.018	.015	-.012	-.042	-.054	-.068	-.102	-.138	.277
.367	.027	.033	.014	-.001	-.007	-.015	-.008	.078	.367
.387	-.017	-.015	-.028	-.059	-.065	-.069	.013	.032	.387
.443	-.015	-.075	-.116	-.165	-.175	-.168	-.176	-.220	.443
.498	.007	-.095	-.160	-.228	-.252	-.279	-.331	-.405	.498
.553	-.007	-.100	-.171	-.241	-.278	-.307	-.369	-.466	.553
.609	.043	-.058	-.132	-.214	-.251	-.277	-.357	-.481	.609
.664	-.005	-.084	-.153	-.224	-.262	-.295	-.374	-.502	.664
.719						-.388	-.461	-.604	.719
.775	-.072	-.095	-.104	-.113	-.143	-.187	-.253	-.335	.775
.830	-.106	-.091	-.086	-.080	-.078	-.062	-.053	-.058	.830
.871	-.048	-.052	-.039	-.035	-.054	-.053	-.038	-.032	.871
				Row C					
.055	.114	.134	.116	.072	.045	.012	-.078	-.186	.055
.166	.041	.048	.041	-.001	-.028	-.069	-.166	-.295	.166
.277	-.004	.018	-.002	-.050	-.074	-.113	-.217	-.326	.277
.353	-.047	-.024	-.050	-.106	-.140	-.187	-.290	-.278	.353
.367	.002	.024	.000	-.050	-.087	-.130	-.228	-.048	.367
.747	-.097	-.103	-.080	-.082	-.106	-.135	-.161		.747
.775	-.070	-.074	-.069	-.070	-.081	-.101	-.135		.775
.830	-.093	-.088	-.088	-.091	-.093	-.080	-.103	-.190	.830
.871	-.053	-.055	-.044	-.042	-.062	-.075	-.086	-.127	.871
.954	.022	.024	.019	-.024	-.062				.954
				Row D					
.166	.034	.052	.070	.075	.078	.077	.065	.059	.166
.277	-.012	.018	.024	.017	.021	.017	.001	-.001	.277
.367	-.007	.012	.007	-.024	-.030	-.020	-.056	.026	.367
.387	-.033	-.017	-.010	-.020	.000	-.007	.001	.113	.387
.443	-.115	-.067	-.015	.062	.110	.174	.278	.349	.443
.498	-.163	-.091	-.003	.115	.177	.244	.365	.461	.498
.553	-.178	-.102	-.019	.106	.169	.232	.345	.437	.553
.609	-.140	-.060	.035	.141	.191	.240	.332	.413	.609
.664	-.146	-.074	.001	.082	.129	.176	.257	.327	.664
.719	-.218	-.152	-.079	.023	.066	.102	.165	.220	.719
.775	-.111	-.101	-.084	-.063	-.068	-.071	-.062	-.064	.775
.830	-.072	-.082	-.102	-.111	-.108	-.103	-.138	-.248	.830
.871	-.052	-.053	-.051	-.063	-.079	-.075	-.087	-.139	.871
				Row E					
.055	.103	.139	.178	.239	.269	.303	.380	.464	.055
.166	.035	.055	.089	.137	.163	.187	.246	.317	.166
.277	-.011	.018	.036	.069	.094	.115	.166	.238	.277
.367	-.014	-.003	.006	.025	.043	.096	.143	.294	.367
.387	-.035	-.021	-.009	.011	.051	.080	.160	.320	.387
.443	-.106	-.096	-.062	.013	.078	.156	.292	.400	.443
.498	-.151	-.102	-.027	.087	.144	.212	.337	.437	.498
.553	-.165	-.101	-.022	.093	.151	.215	.324	.416	.553
.609	-.135	-.058	.032	.138	.187	.242	.334	.414	.609
.664	-.137	-.069	.004	.089	.136	.183	.261	.336	.664
.719	-.217	-.157	-.088	.016	.060	.100	.170	.227	.719
.775	-.141	-.122	-.091	-.049	-.047	-.034	.007	.041	.775
.830	-.063	-.079	-.100	-.105	-.093	-.081	-.106	-.123	.830
.871	-.071	-.074	-.081	-.109	-.117	-.112	-.123	-.135	.871
.954	.022	.015	-.005	-.082	-.142	-.174	-.271	-.279	.954

TABLE V.- PRESSURE COEFFICIENTS FOR INDENTED BODY IN PRESENCE OF WING - Continued

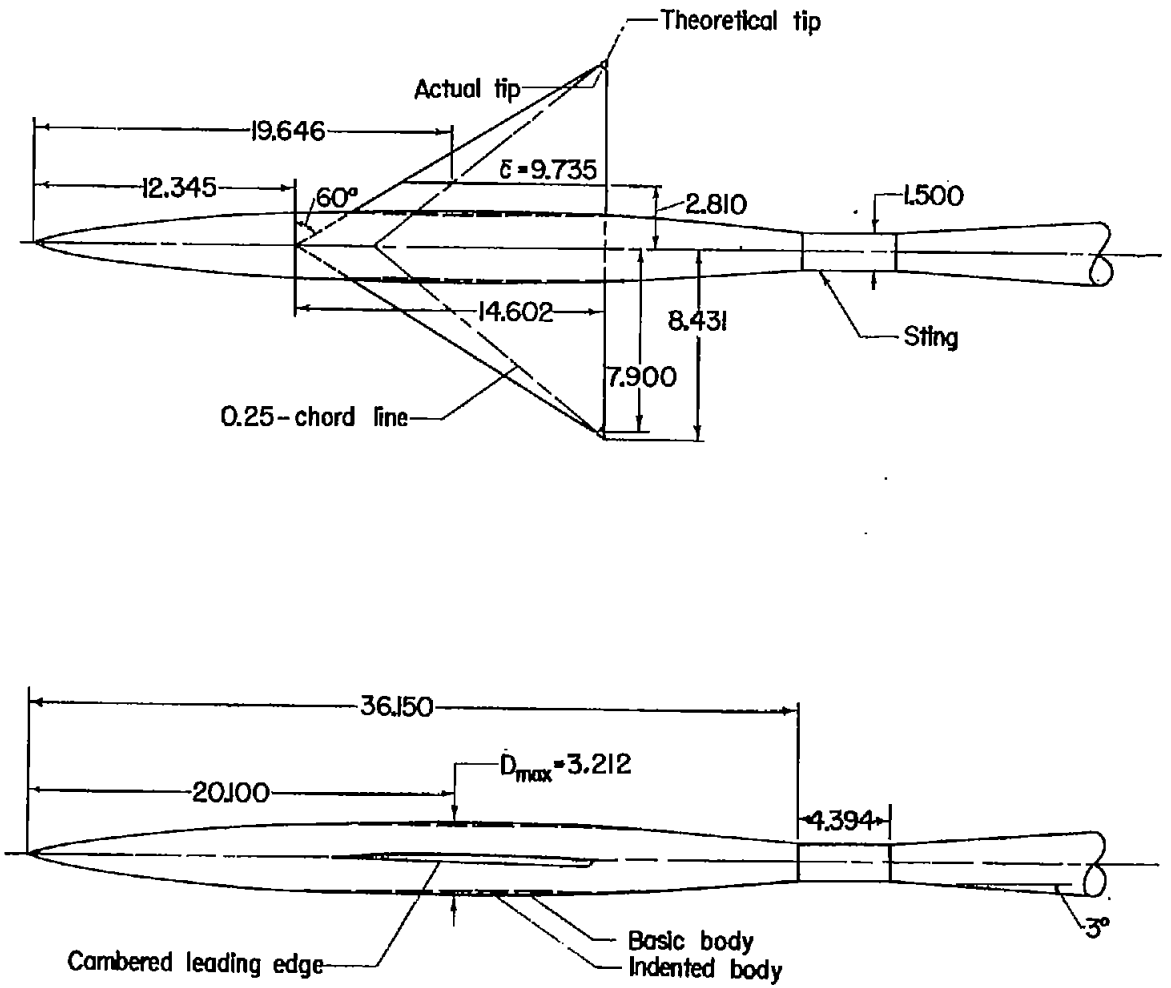
(g) M = 1.12

x/l	α = −4°	α = 0°	α = 4°	α = 8°	α = 10°	α = 12°	α = 16°	α = 20°	x/l
				Row A					
.055	.156	.108	.067	.025	.011	.005	.005	-.003	.055
.166	.057	.019	.003	-.018	-.020	-.021	-.037	-.053	.166
.277	.027	.002	-.018	-.023	-.024	-.024	-.027	-.060	.277
.367	-.009	.009	-.000		.162				.367
.387	.015	-.003	-.018	-.029	-.036	-.026	-.015	-.015	.387
.415	-.072	-.088	-.086	-.104	-.110	-.109	-.127	-.183	.415
.443					-.009				.443
.498	.021	-.043	-.094	-.150	-.173	-.182	-.206	-.213	.498
.553	.014	-.061	-.120	-.178	-.193	-.203	-.239	-.254	.553
.581	.025	-.064	-.114	-.154	-.180	-.200	-.244	-.266	.581
.609	.062	-.021	-.091	-.135	-.159	-.176	-.226	-.246	.609
.636	.094	.011	-.063	-.106	-.127	-.151	-.179	-.184	.636
.664	.058	-.015	-.083	-.128	-.146	-.165	-.190	-.173	.664
.692	.022	-.050	-.116	-.164	-.189	-.208	-.223	-.195	.692
.719	-.015	-.088	-.153	-.204	-.232	-.251	-.270	-.252	.719
.747	-.025	-.098	-.159	-.223	-.256	-.274	-.303	-.302	.747
.775	-.057	-.107	-.139	-.198	-.237	-.265	-.314	-.341	.775
.830	-.063	-.068	-.044	.004	.026	.044	.080	.096	.830
.871	-.047	-.035	-.024	.008	.005	.027	.064	.043	.871
.954	-.104	-.077	-.027	-.102	-.149	-.198	-.240	-.224	.954
				Row B					
.166	.050	.036	.008	-.030	-.045	-.060	-.094	-.130	.166
.277	.009	.001	-.022	-.045	-.060	-.071	-.104	-.138	.277
.367	.009	.009	-.006	-.010	-.023	-.028	-.061	-.093	.367
.387	.006	-.005	-.019	-.033	-.045	-.047	-.078	-.113	.387
.443	-.004	-.047	-.084	-.131	-.154	-.160	-.191	-.195	.443
.498	.019	-.067	-.128	-.194	-.227	-.255	-.295	-.329	.498
.553	.011	-.073	-.140	-.212	-.231	-.262	-.325	-.411	.553
.609	.054	-.022	-.097	-.168	-.207	-.234	-.319	-.426	.609
.664	.047	-.033	-.105	-.173	-.212	-.248	-.329	-.436	.664
.719			-.154			-.334	-.406	-.541	.719
.775	-.054	-.080	-.095	-.121	-.137	-.153	-.196	-.229	.775
.830	-.077	-.079	-.063	-.039	-.037	-.028	-.030	.003	.830
.871	-.017	-.017	-.012	-.009	-.014	.011	.041	.036	.871
				Row C					
.055	.100	.118	.102	.039	.013	-.020	-.095	-.194	.055
.166	.015	.032	.013	-.034	-.059	-.090	-.192	-.304	.166
.277	-.012	.003	-.011	-.055	-.084	-.120	-.223	-.344	.277
.353	-.043	-.028	-.047	-.098	-.132	-.176	-.267	-.382	.353
.367	-.013	-.003	-.022	-.077	-.114	-.149	-.243	-.369	.367
.747	-.069	-.075	-.057	-.067	-.082	-.076	-.113		.747
.775	-.053	-.053	-.044	-.059	-.065	-.057	-.067		.775
.830	-.075	-.075	-.071	-.057	-.056	-.050	-.078	-.117	.830
.871	-.019	-.022	-.014	-.006	-.015	-.013	-.036	-.062	.871
.954	-.045	-.053	-.038	-.062	-.093				.954
				Row D					
.166	.011	.037	.048	.049	.054	.061	.057	.069	.166
.277	-.021	.001	.013	.012	.013	.017	.002	-.002	.277
.367	-.019	-.011	-.016	-.039	-.049	-.049	-.068	-.019	.367
.387	-.019	-.006	-.001	.022	.014	.007	-.000	-.012	.387
.443	-.079	-.041	-.005	.022	.033	.059	.168	.349	.443
.498	-.126	-.056	.018	.107	.169	.241	.370	.494	.498
.553	-.141	-.068	.008	.119	.175	.241	.358	.472	.553
.609	-.111	-.031	.050	.159	.213	.264	.362	.462	.609
.664	-.098	-.022	.057	.127	.164	.210	.304	.392	.664
.719	-.163	-.094	-.019	.054	.096	.151	.238	.304	.719
.775	-.102	-.084	-.057	-.027	-.016	-.001	.025	.047	.775
.830	-.052	-.070	-.076	-.071	-.068	-.066	-.100	-.155	.830
.871	-.016	-.018	-.024	-.029	-.034	-.028	-.053	-.081	.871
				Row E					
.055	.087	.128	.165	.210	.242	.278	.367	.454	.055
.166	.007	.031	.063	.108	.135	.168	.233	.315	.166
.277	-.023	.000	.024	.064	.086	.116	.168	.241	.277
.367	-.017	-.010	.007	.031	.046	.090	.145	.203	.367
.387	-.026	-.015	-.005	.044	.057	.093	.146	.215	.387
.443	-.073	-.078	-.071	-.046	-.034	-.006	.034	.273	.443
.498	-.114	-.058	-.003	.078	.134	.208	.334	.459	.498
.553	-.132	-.071	.003	.112	.161	.227	.346	.458	.553
.609	-.102	-.032	.048	.165	.218	.268	.368	.467	.609
.664	-.092	-.017	.061	.134	.173	.215	.309	.398	.664
.719	-.163	-.100	-.025	.036	.088	.145	.242	.314	.719
.775	-.146	-.111	-.060	.003	.031	.059	.101	.148	.775
.830	-.034	-.065	-.071	-.054	-.046	-.044	-.046	-.028	.830
.871	-.027	-.039	-.059	-.068	-.074	-.066	-.060	-.089	.871
.954	-.035	-.062	-.082	-.118	-.148	-.190	-.182	-.188	.954

TABLE V.- PRESSURE COEFFICIENTS FOR INDENTED BODY IN PRESENCE OF WING - Concluded

(h) M = 1.43

x/l	α = -3.6°	α = -0.3°	α = 4.3°	α = 8.8°	α = 11.1°	α = 13.4°	α = 16.0°	α = 19.8°	x/l
				Row A					
.055	.160	.099	.054	.034	.021	.014			.055
.166	.047	.025	.014	-.001	-.003	-.016			.166
.277	.053	.019	-.003	-.008	-.012	-.020			.277
.367	.018	.029	.004	-.003	.008	.006			.367
.387	.015	.009	-.001	-.005	-.003	-.017			.387
.415	-.038	-.057	-.065	-.069	-.078	-.079			.415
.443	-.039	-.063	-.073	-.065	-.075	-.081			.443
.498	-.004	-.041	-.067	-.101	-.119	-.131			.498
.553	.011	-.036	-.098	-.136	-.150	-.174			.553
.581	.008	-.040	-.099	-.141	-.167	-.187			.581
.609	.032	-.025	-.085	-.124	-.146	-.167			.609
.636	.058	.004	-.059	-.105	-.130	-.139			.636
.664	.045	-.011	-.070	-.102	-.119	-.137			.664
.692	.019	-.031	-.083	-.127	-.139	-.150			.692
.719	-.006	-.055	-.113	-.157	-.176	-.193			.719
.747	-.017	-.063	-.125	-.169	-.193	-.211			.747
.775	-.025	-.075	-.133	-.182	-.208	-.222			.775
.830	-.052	-.063	-.065	-.061	-.072	-.086			.830
.871	-.032	-.026	-.015	.019	.036	.073			.871
.954	-.109	-.079	-.055	-.074	-.076	-.086			.954
				Row B					
.166	.031	.030	.021	-.029	-.038	-.062			.166
.277	.032	.014	-.016	-.035	-.052	-.068			.277
.367	.008	.013	.004	-.031	-.044	-.065			.367
.387	.002	.024	.003	-.024	-.036	-.059			.387
.443	-.025	-.036	-.061	-.083	-.105	-.127			.443
.498	.008	-.038	-.094	-.151	-.182	-.205			.498
.553	.011	-.041	-.110	-.166	-.196	-.222			.553
.609	.040	-.019	-.078	-.150	-.188	-.221			.609
.664	.032	-.021	-.083	-.150	-.185	-.215			.664
.719	-.005	-.057	-.125	-.190	-.230	-.264			.719
.775	-.028	-.061	-.092	-.111	-.119	-.138			.775
.830	-.067	-.074	-.072	-.075	-.082	-.068			.830
.871	-.039	-.028	-.027	-.017	-.004	.010			.871
				Row C					
.055	.083	.090	.090	.041	.014	-.025			.055
.166	.009	.024	.023	-.028	-.054	-.103			.166
.277	.011	.022	-.011	-.050	-.089	-.136			.277
.367	-.019	-.015	-.009	-.071	-.119	-.171			.367
.747	-.048	-.037	-.038	-.051	-.063	-.066			.747
.775	-.037	-.037	-.033	-.041	-.035	-.019			.775
.830	-.075	-.071	-.070	-.067	-.063	-.066			.830
.871	-.026	-.024	-.021	-.030	-.036	-.049			.871
.954	-.065	-.057	-.057	-.070	-.067	-.066			.954
				Row D					
.166	.007	.025	.052	.081	.088	.097			.166
.277	.004	.024	.028	.024	.030	.038			.277
.367	.022	.030	.029	.030	.048	.019			.367
.387	.011	.017	.021	.016	.008	-.004			.387
.443	-.065	-.036	-.035	-.060	-.064	-.062			.443
.498	-.090	-.044	.015	.090	.136	.172			.498
.553	-.100	-.047	.030	.123	.150	.198			.553
.609	-.079	-.029	.053	.138	.188	.249			.609
.664	-.077	-.024	.049	.144	.193	.241			.664
.719	-.111	-.058	.019	.102	.147	.192			.719
.775	-.103	-.076	-.026	.028	.060	.094			.775
.871	-.021	-.022	-.016	-.022	-.031	-.036			.871
				Row E					
.055	.057	.093	.171	.228	.267	.316			.055
.166	.011	.021	.072	.138	.177	.216			.166
.277	-.002	.014	.053	.081	.110	.148			.277
.387	.010	-.003	.023	.065	.095	.111			.387
.443	-.063	-.047	-.041	-.019	.001	.025			.443
.498	-.073	-.047	-.010	.013	.026	.031			.498
.553	-.091	-.041	.017	.121	.147	.180			.553
.609	-.074	-.029	.052	.138	.188	.241			.609
.664	-.061	-.013	.059	.149	.204	.262			.664
.719	-.108	-.064	.011	.095	.141	.185			.719
.775	-.127	-.081	-.013	.072	.116	.160			.775
.830	-.075	-.064	-.043	-.007	.014	.040			.830
.871	-.034	-.044	-.042	-.032	-.022	-.012			.871
.954	-.053	-.069	-.097	-.105	-.113	-.122			.954



(a) Wing-body combinations.

Figure 1.- Model details. All dimensions in inches unless otherwise noted.

BODY ORDINATES

x, in.	Radius, in.	
	Basic body	Indented body
0	0	0
1	.282	.282
2	.460	.460
3	.612	.612
4	.743	.743
5	.862	.862
6	.969	.969
7	1.062	1.062
8	1.150	1.150
9	1.222	1.222
10	1.290	1.290
11	1.350	1.350
12	1.404	1.404
13	1.452	1.452
14	1.493	1.493
15	1.526	1.535
16	1.552	1.551
17	1.575	1.553
18	1.590	1.541
19	1.602	1.523
20	1.606	1.502
21	1.602	1.466
22	1.594	1.444
23	1.578	1.433
24	1.560	1.431
25	1.532	1.431
26	1.501	1.423
27	1.460	1.405
28	1.414	1.381
29	1.360	1.339
30	1.300	1.287
31	1.231	1.227
32	1.158	1.158
33	1.076	1.076
34	.984	.984
35	.878	.878
36	.762	.762
36.15	.750	.750

(b) Body ordinates.

Figure 1.- Concluded.

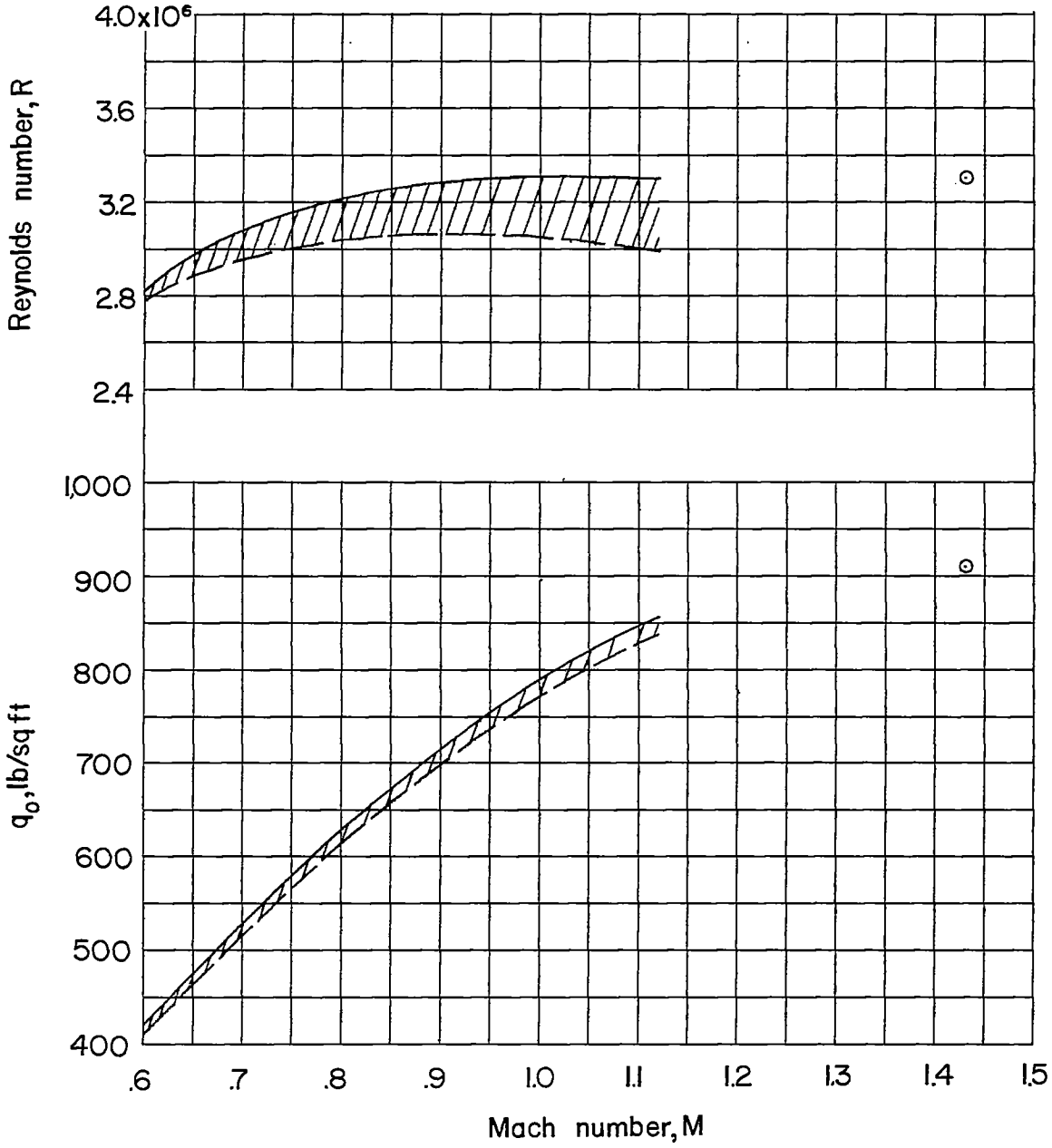


Figure 2.- Variation with Mach number of dynamic pressure and Reynolds number based on the mean aerodynamic chord.

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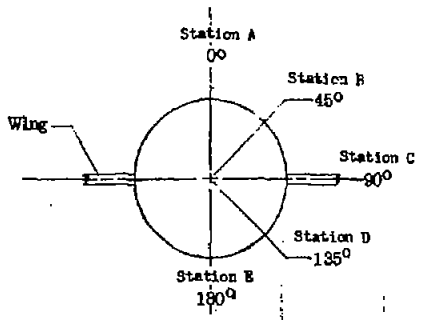
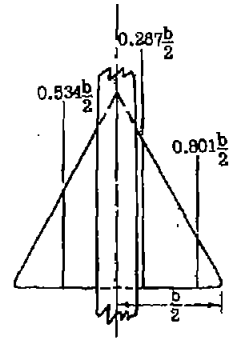


View looking downstream

Location of body pressure orifices, $\frac{x}{l}$				
Meridian, 0, deg	Meridian, 45, deg	Meridian, 90, deg	Meridian, 135, deg	Meridian, 180, deg
0.055	0.166	0.055	0.166	0.055
.166	.277	.166	.277	.166
.277	.387	.277	.387	.277
.387	.387	.368	.387	.387
.387	.443	.387	.443	.387
.415	.498	.747*	.498	.443
.443	.553	.775	.553	.498
.498	.609	.830	.609	.553
.553	.664	.871	.664	.609
.581	.719	.964	.719	.664
.609	.775		.775	.719
.636	.830		.830	.775
.664	.871		.871	.830
.692				.871
.719				.964
.747				
.775				
.830				
.871				
.964				

*Indented body only

(a) Basic and indented body.



Top view

Location of wing pressure orifices, $\frac{x}{c}$					
$0.287\frac{b}{2}$		$0.534\frac{b}{2}$		$0.801\frac{b}{2}$	
Upper surface	Lower surface	Upper surface	Lower surface	Upper surface	Lower surface
0.010	0.025	0.010	0.025	0.030	0.045
.030	.060	.030	.060	.060	.150
.060	.100	.060	.100	.100	.250
.100	.150	.100	.200	.200	.350
.150	.200	.200	.300	.300	.450
.200	.300	.300	.400	.400	.550
.300	.400	.400	.500	.500	.650
.400	.500	.500	.600	.600	.750
.500	.600	.600	.700	.700	
.600	.700	.700	.800	.850	
.700	.800	.800	.850		
.800	.850	.900			
.900	.925				
.950					

(b) Wing.

Figure 3.- Location of pressure orifices.

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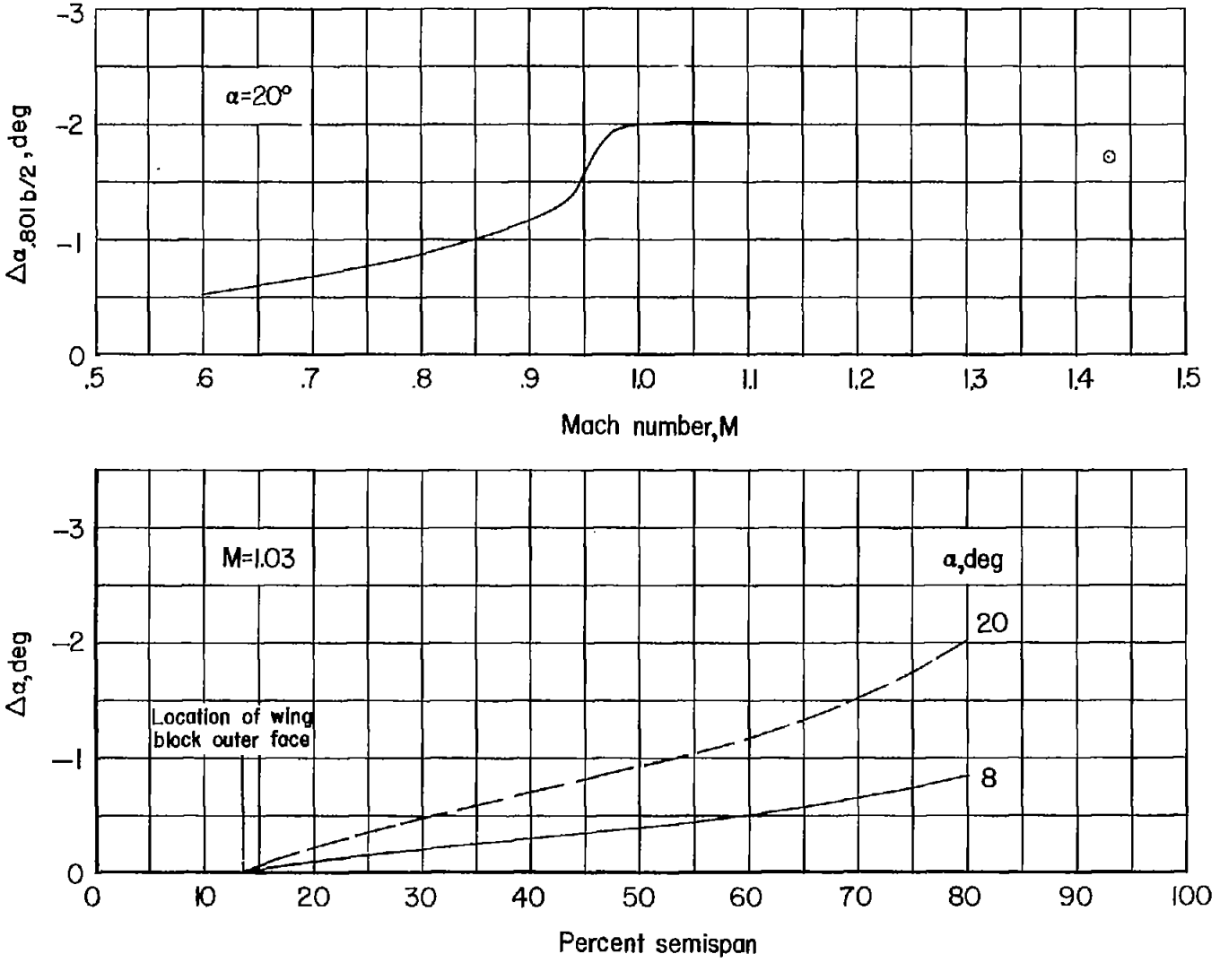
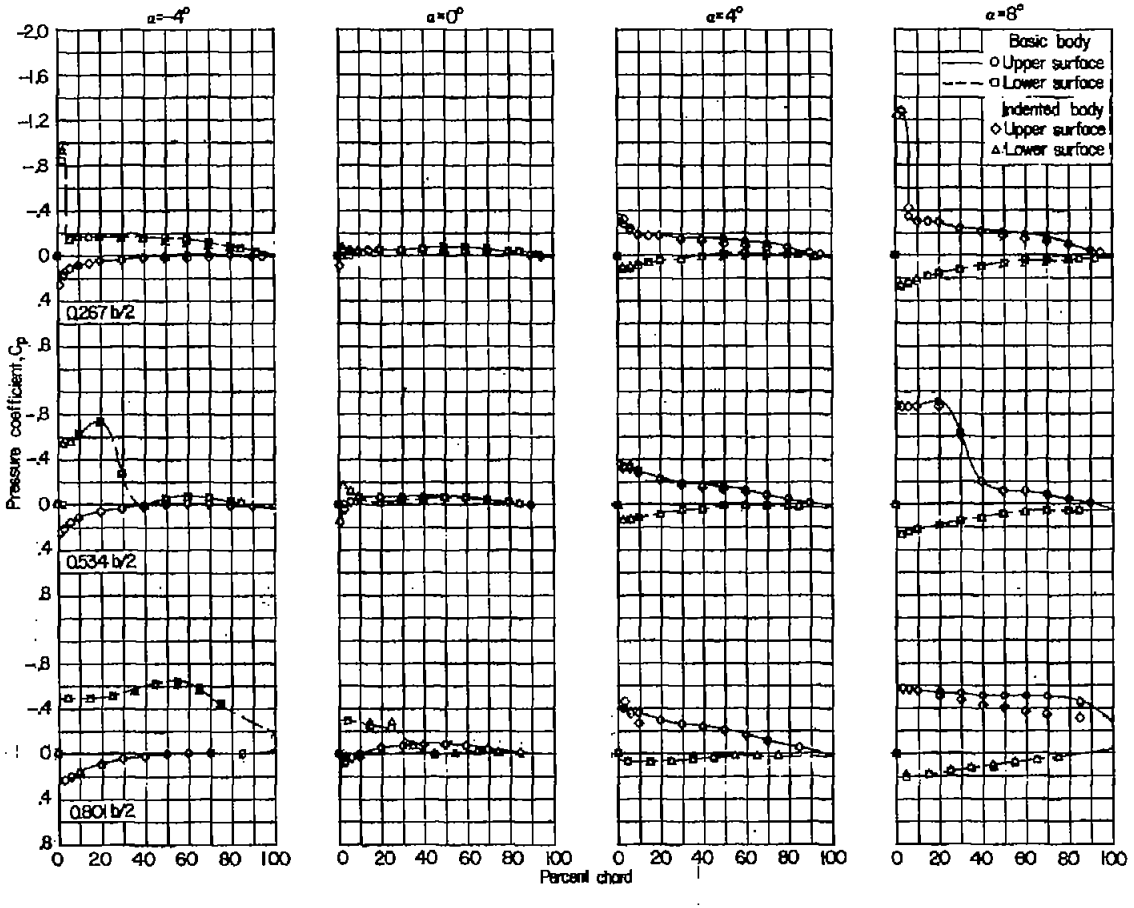


Figure 4.- Calculated wing-twist characteristics.



(a) $M = 0.60$; $C_{p,sonic} = -1.29$; $\alpha = -4^{\circ}$ to 20°.

Figure 5.- Chordwise pressure distributions on a cambered 60°-delta wing in the presence of basic and indented bodies.

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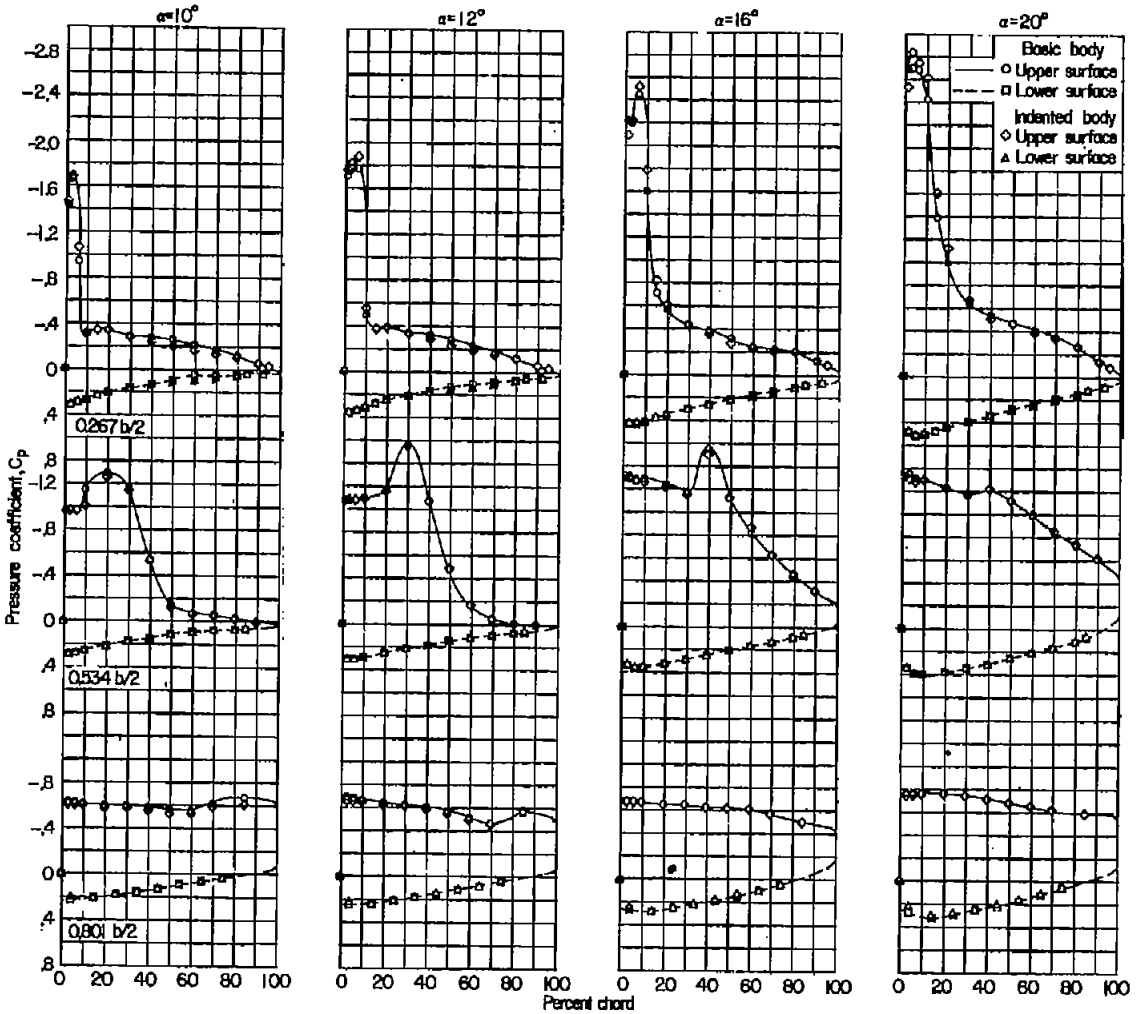
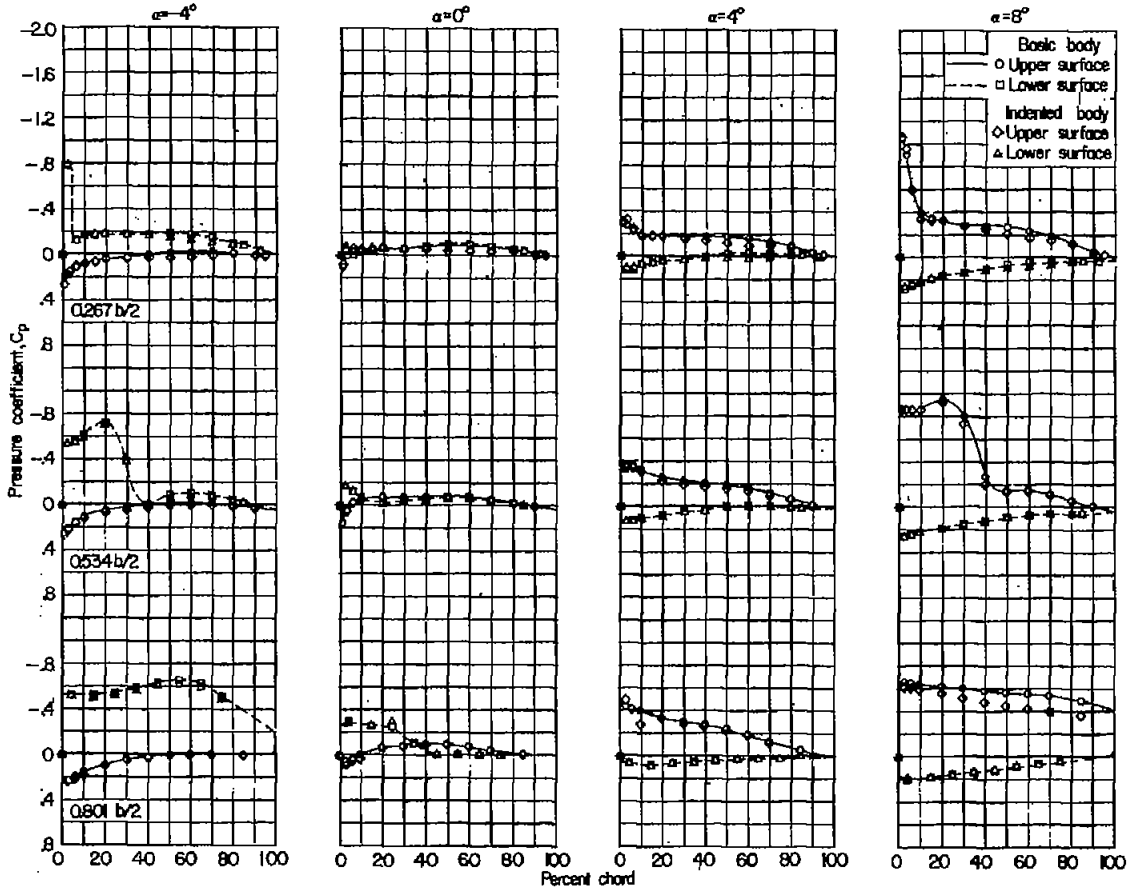


(a) Concluded.

Figure 5.- Continued.

UNCLASSIFIED



(b) $M = 0.80$; $C_{p,sonic} = -0.43$; $\alpha = -4^{\circ}$ to 20°.

Figure 5.- Continued.

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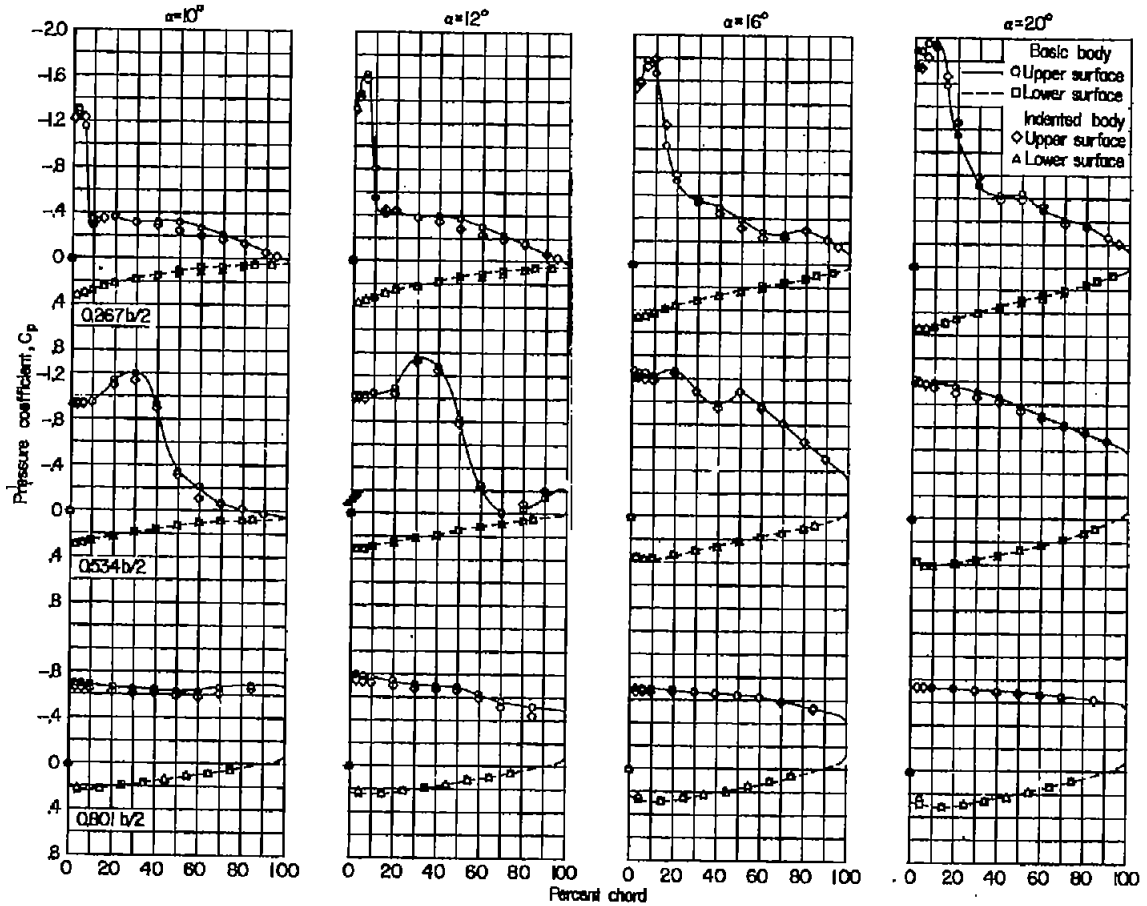

(b) Concluded.

Figure 5.- Continued.

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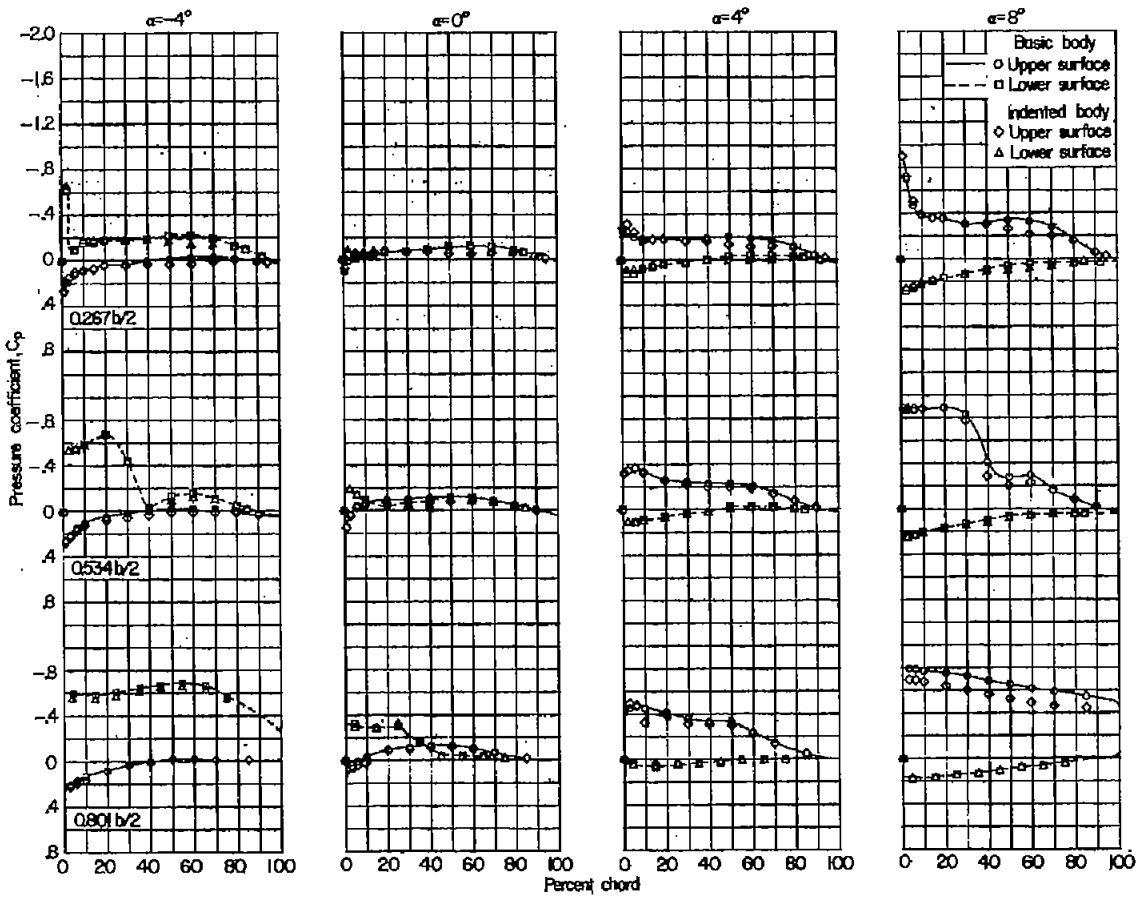

(c) M = 0.90; $C_{p,sonic}$ = -0.19; α = -4° to 20°.

Figure 5.- Continued.

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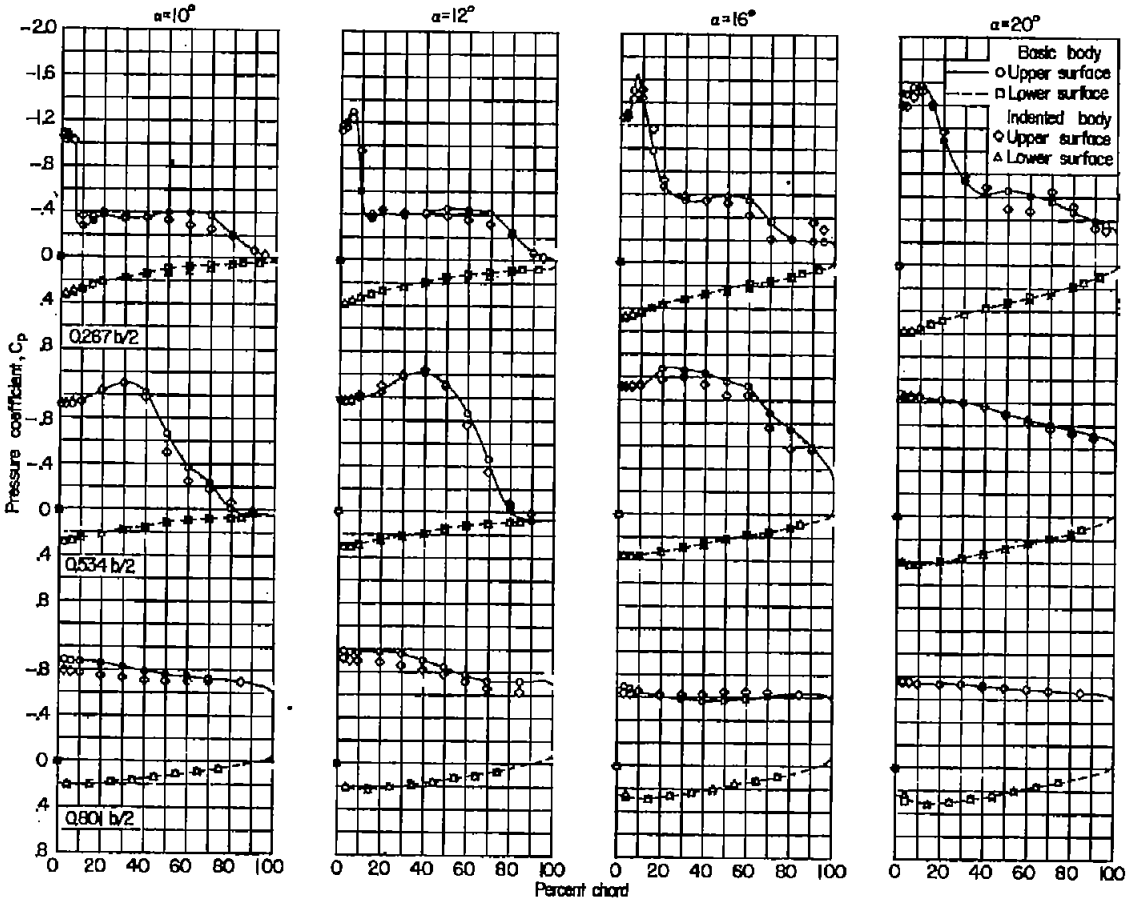
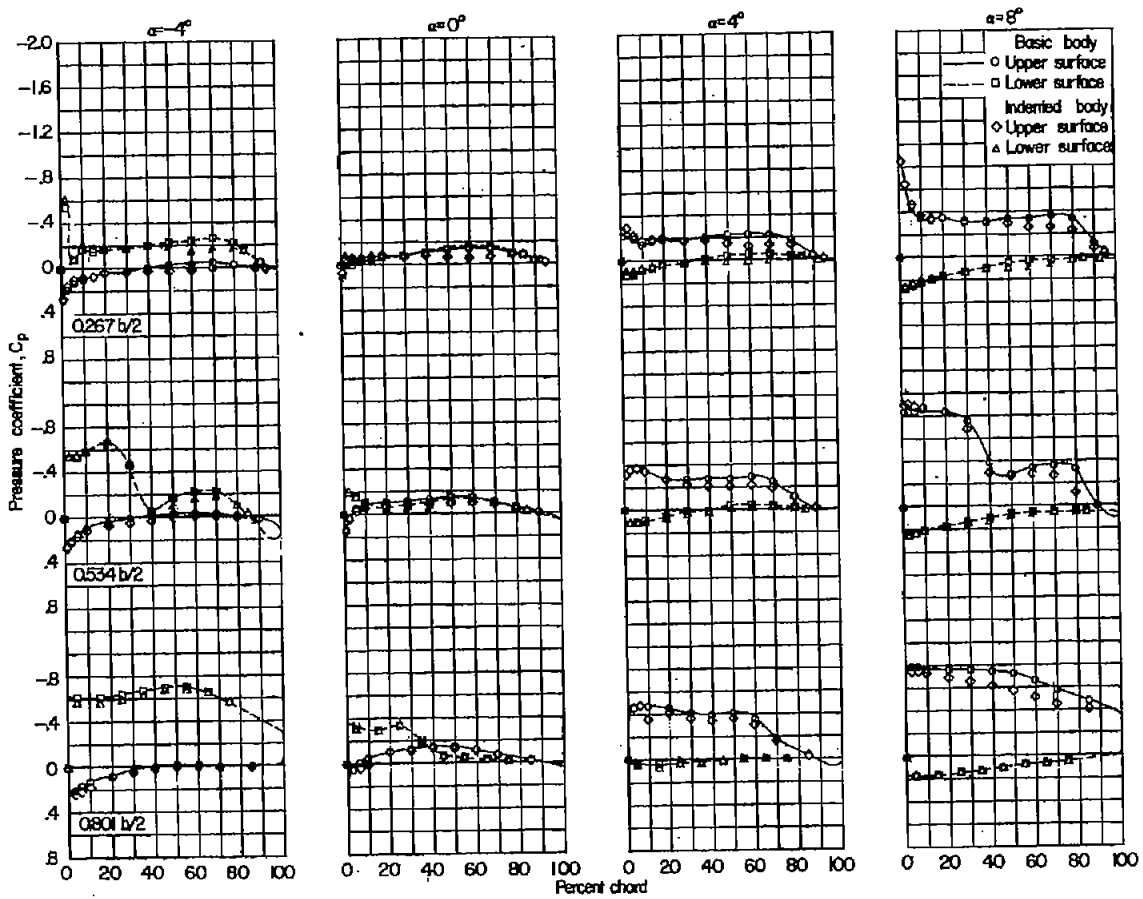


(c) Concluded.

Figure 5.- Continued.

UNCLASSIFIED



(d) M = 0.94; $C_{p,sonic}$ = -0.11; α = -4° to 20°.

Figure 5.- Continued.

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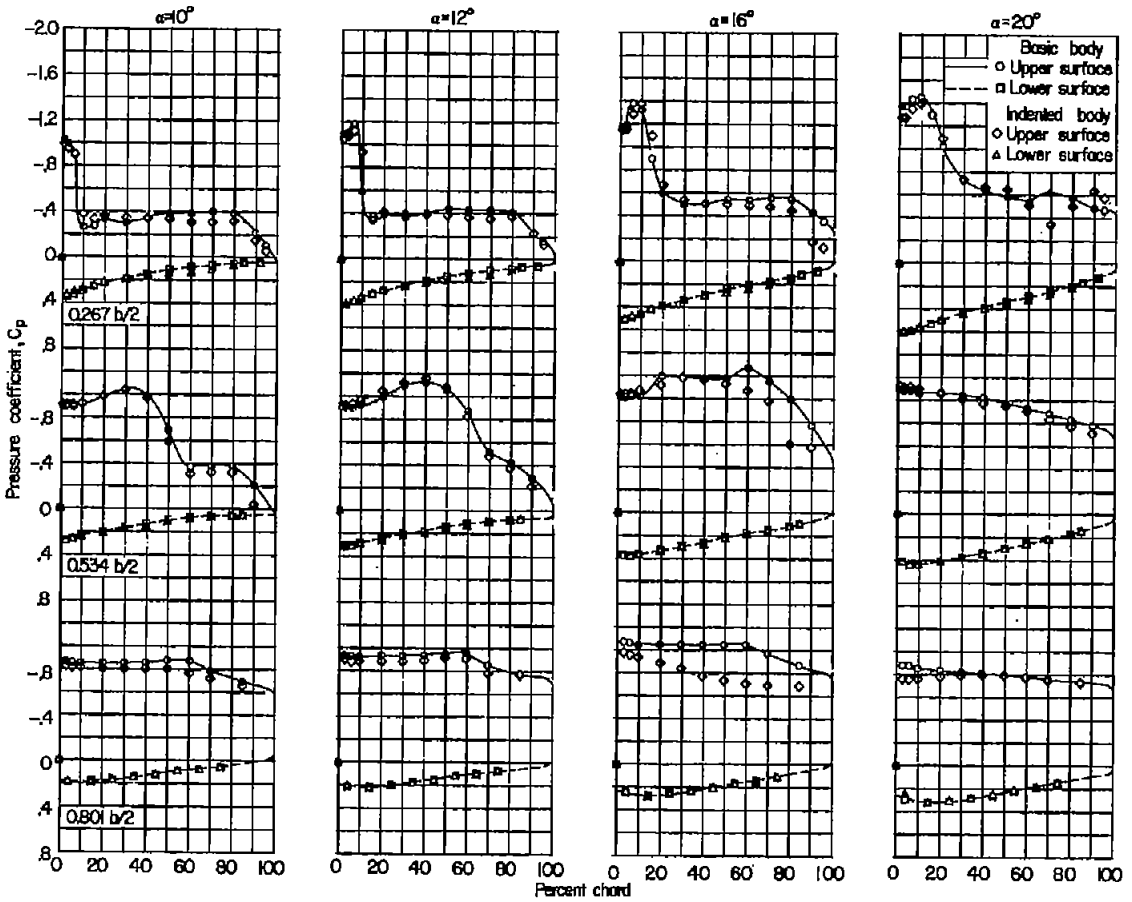

(d) Concluded.

Figure 5.- Continued.

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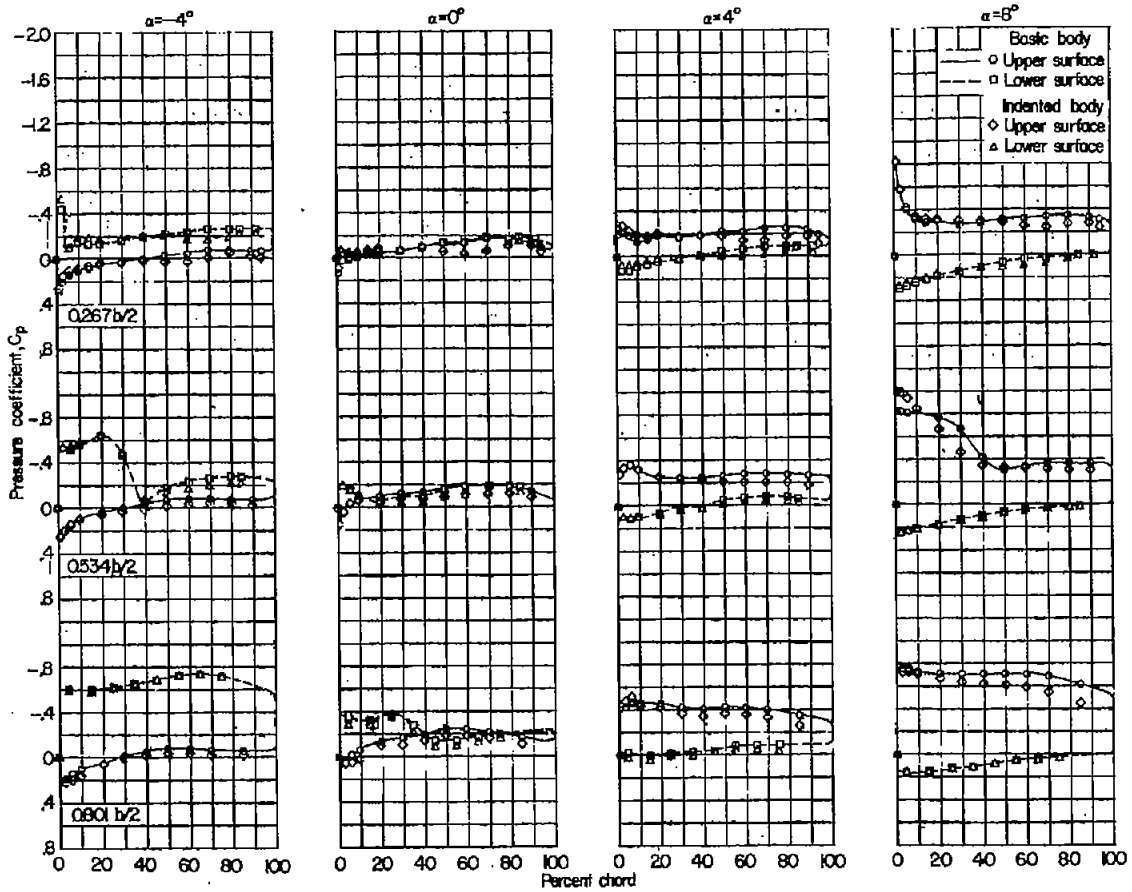

(e) $M = 0.98$; $C_{p,sonic} = -0.03$; $\alpha = -4^{\circ}$ to 20°.

Figure 5.- Continued.

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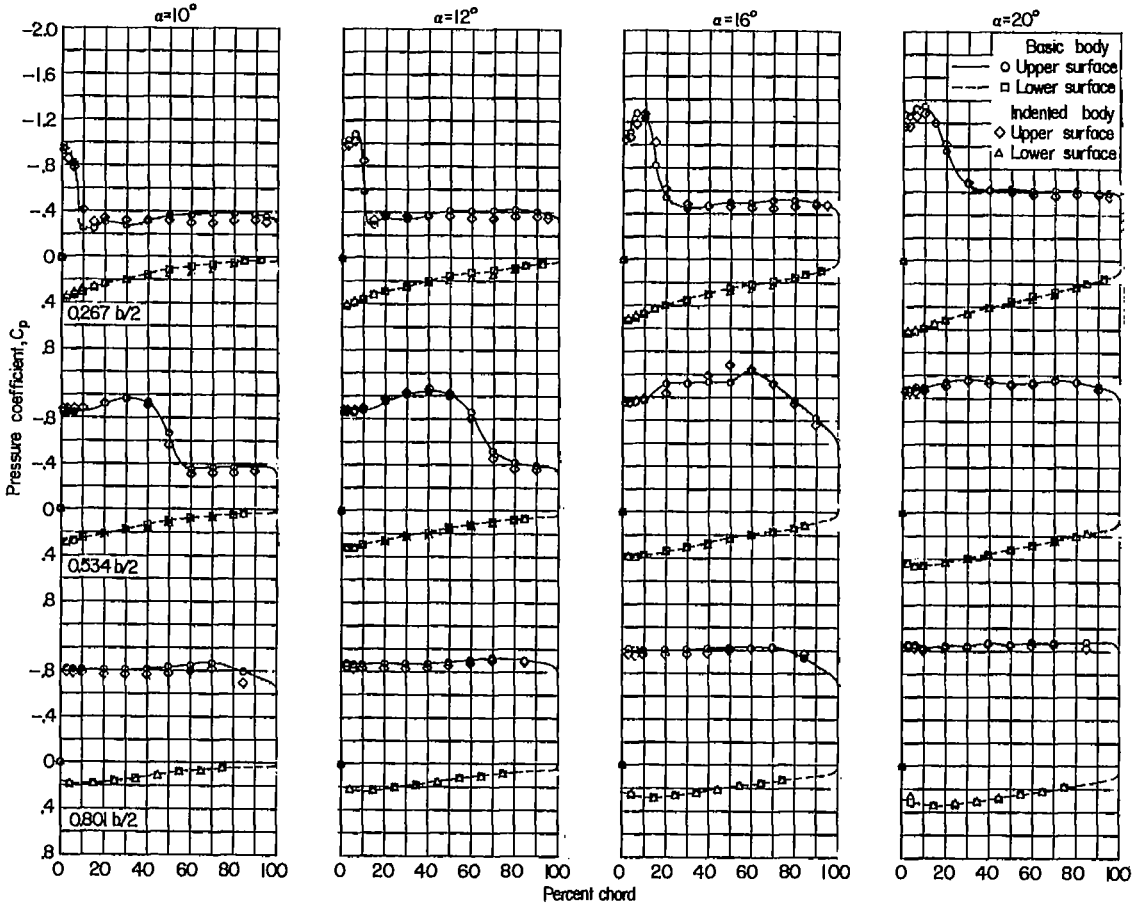

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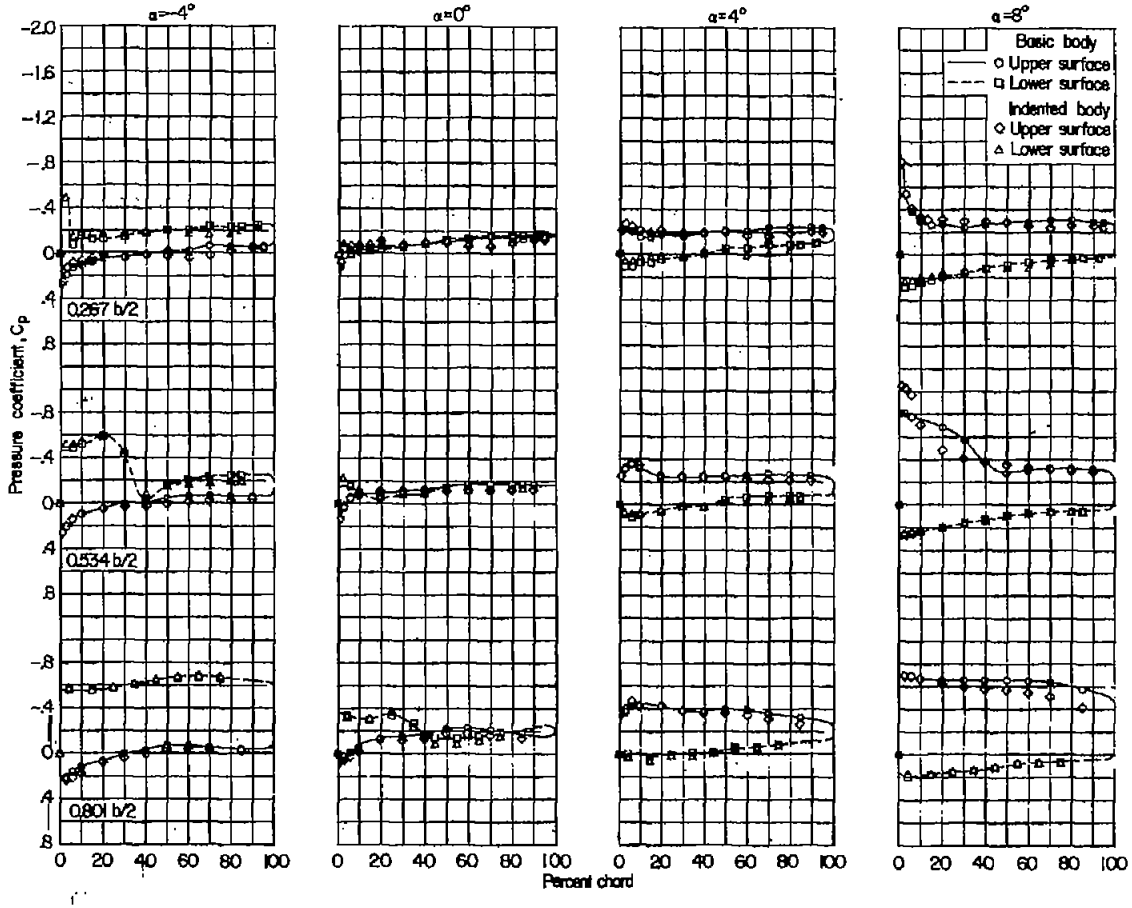
Figure 5.- Continued.



(f) $M = 1.03$; $C_{p,sonic} = 0.05$; $\alpha = -4^{\circ}$ to 20°.

Figure 5.- Continued.

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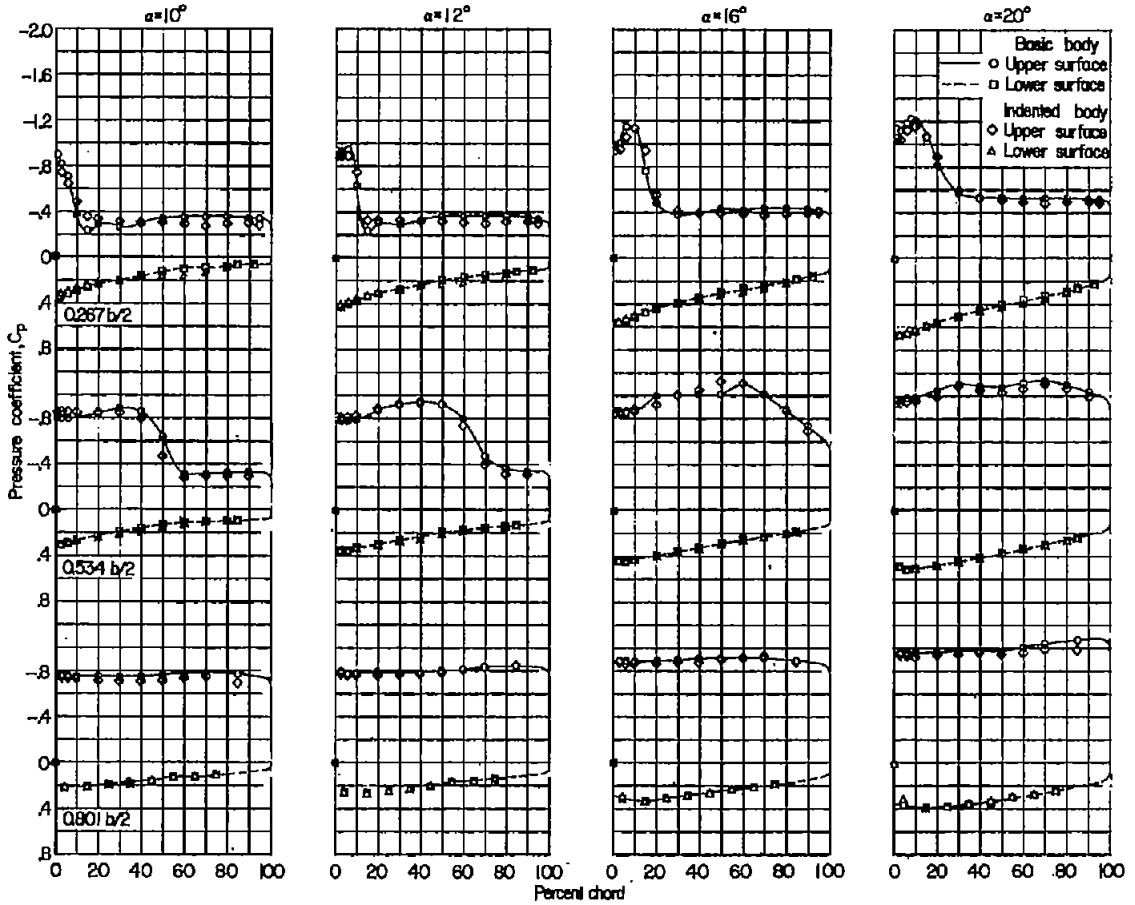

(f) Concluded.

Figure 5.- Continued.

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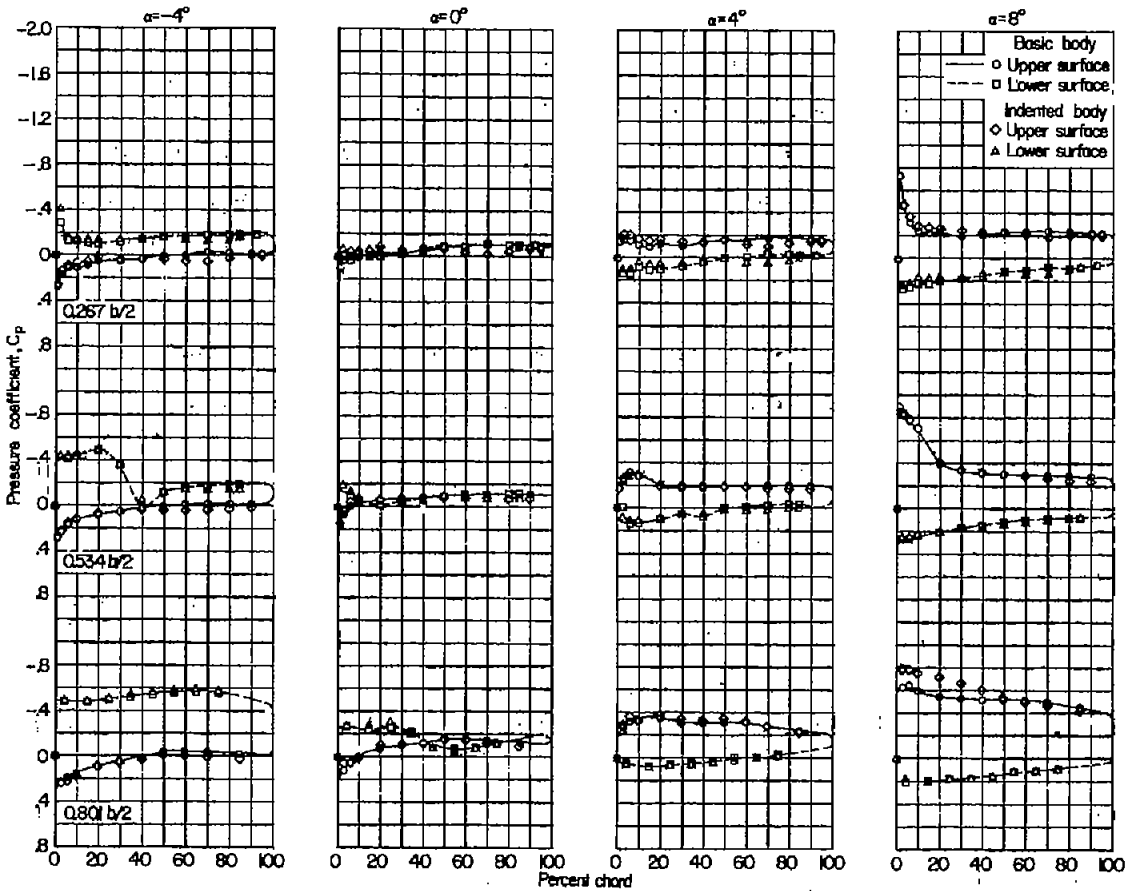

(g) M = 1.12; $C_{p,sonic}$ = 0.18; α = -4° to 20°.

Figure 5.- Continued.

UNCLASSIFIED

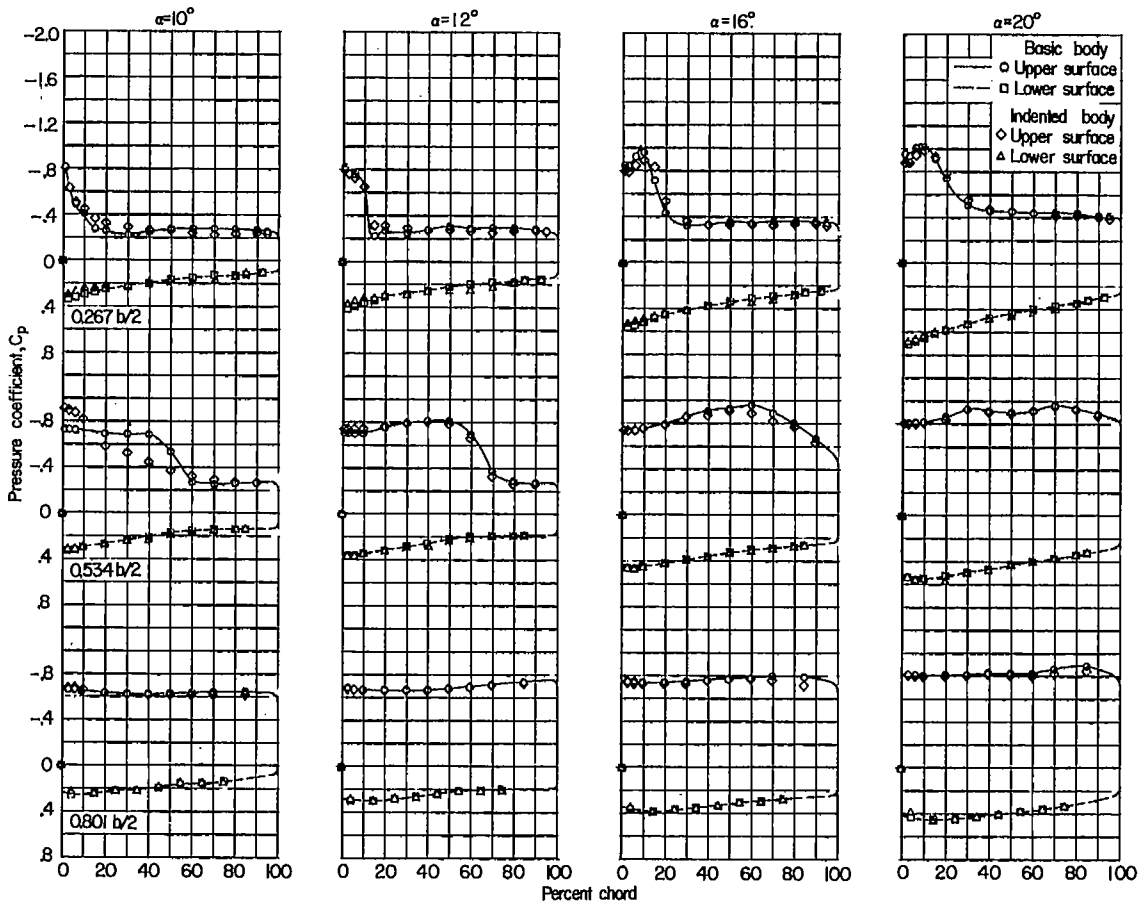

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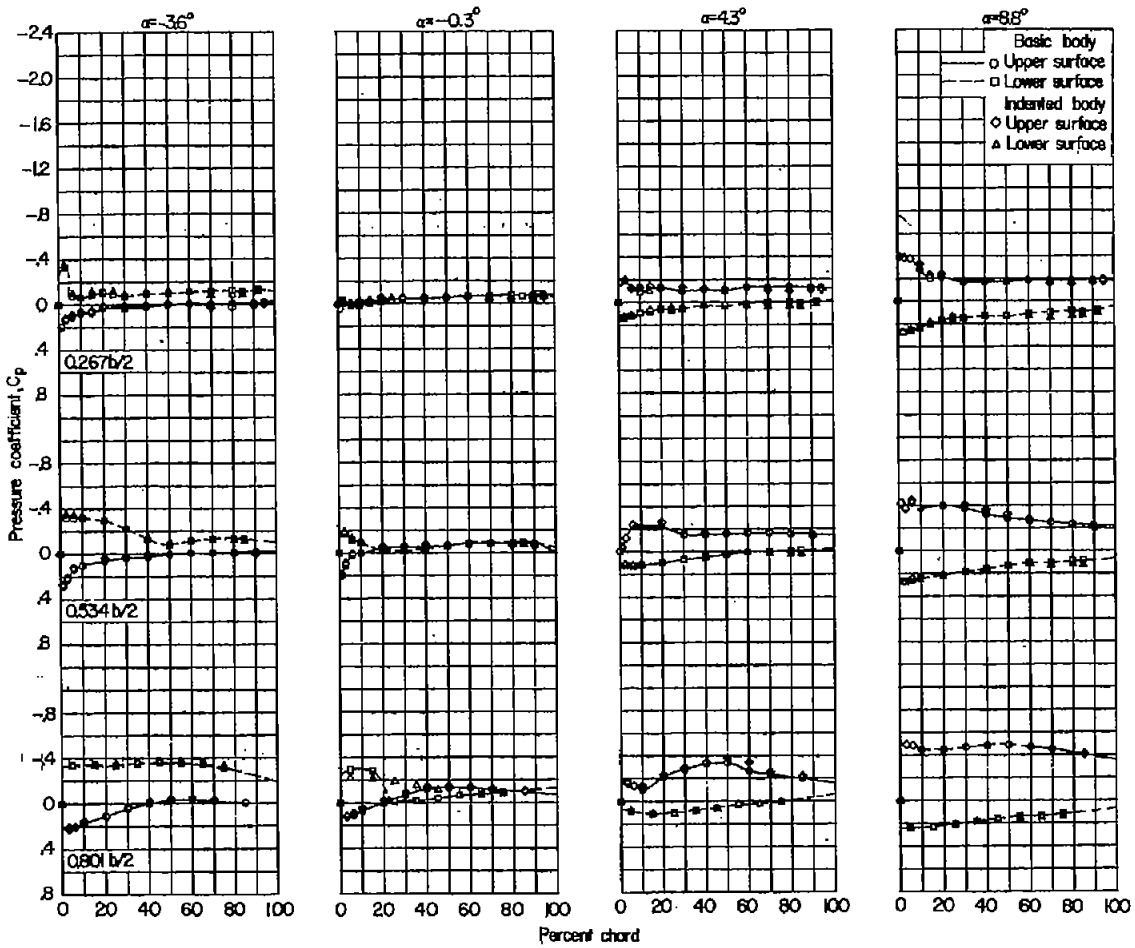
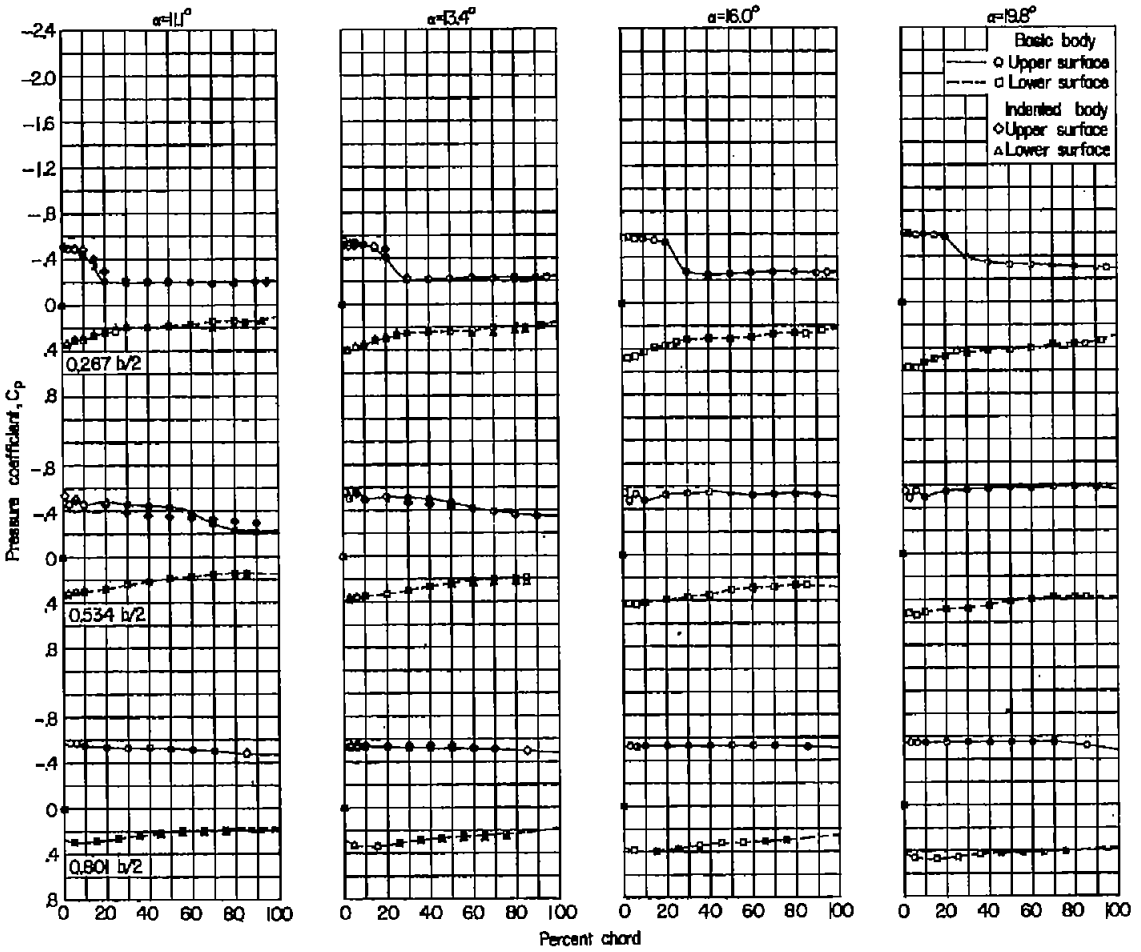
Figure 5.- Continued.

UNCLASSIFIED



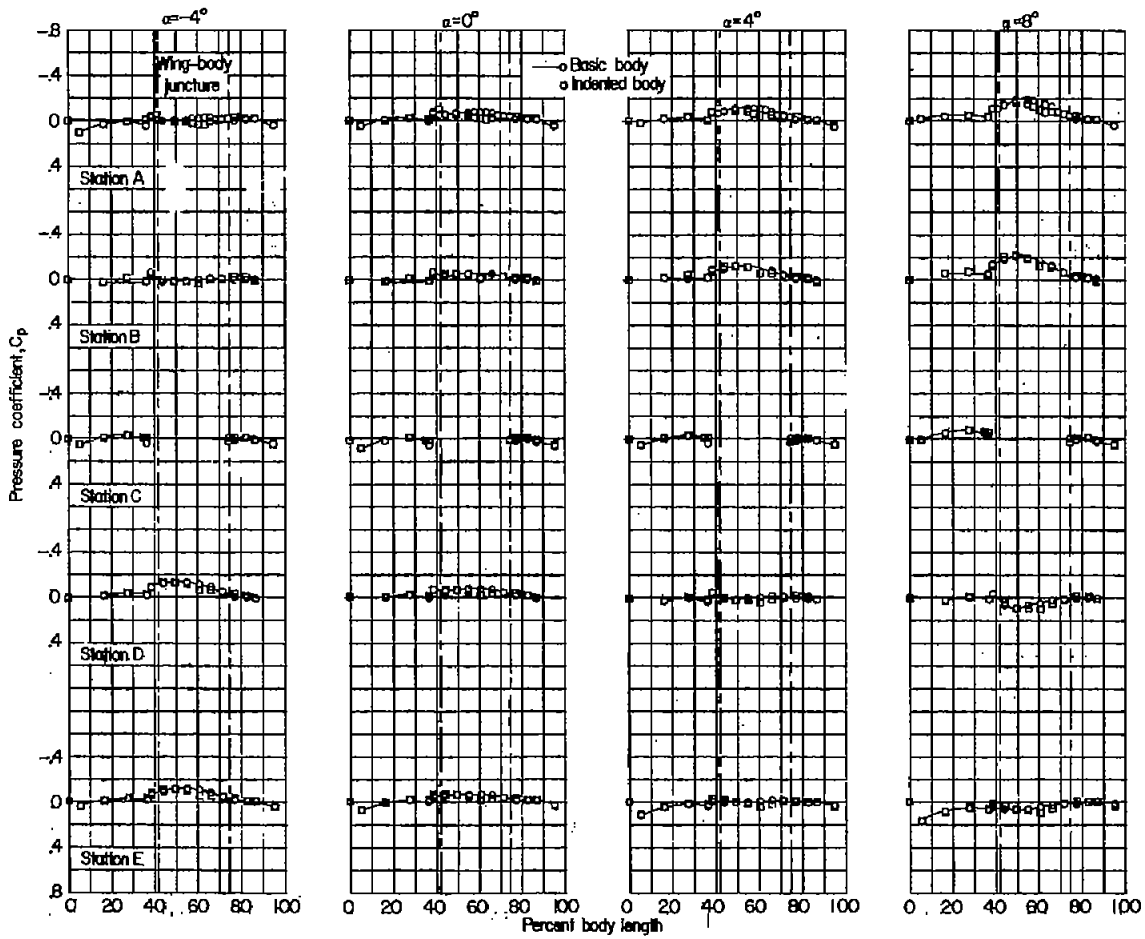
(h) M = 1.43; $C_{p,sonic}$ = 0.52; α = -4° to 20°.

Figure 5.- Continued.



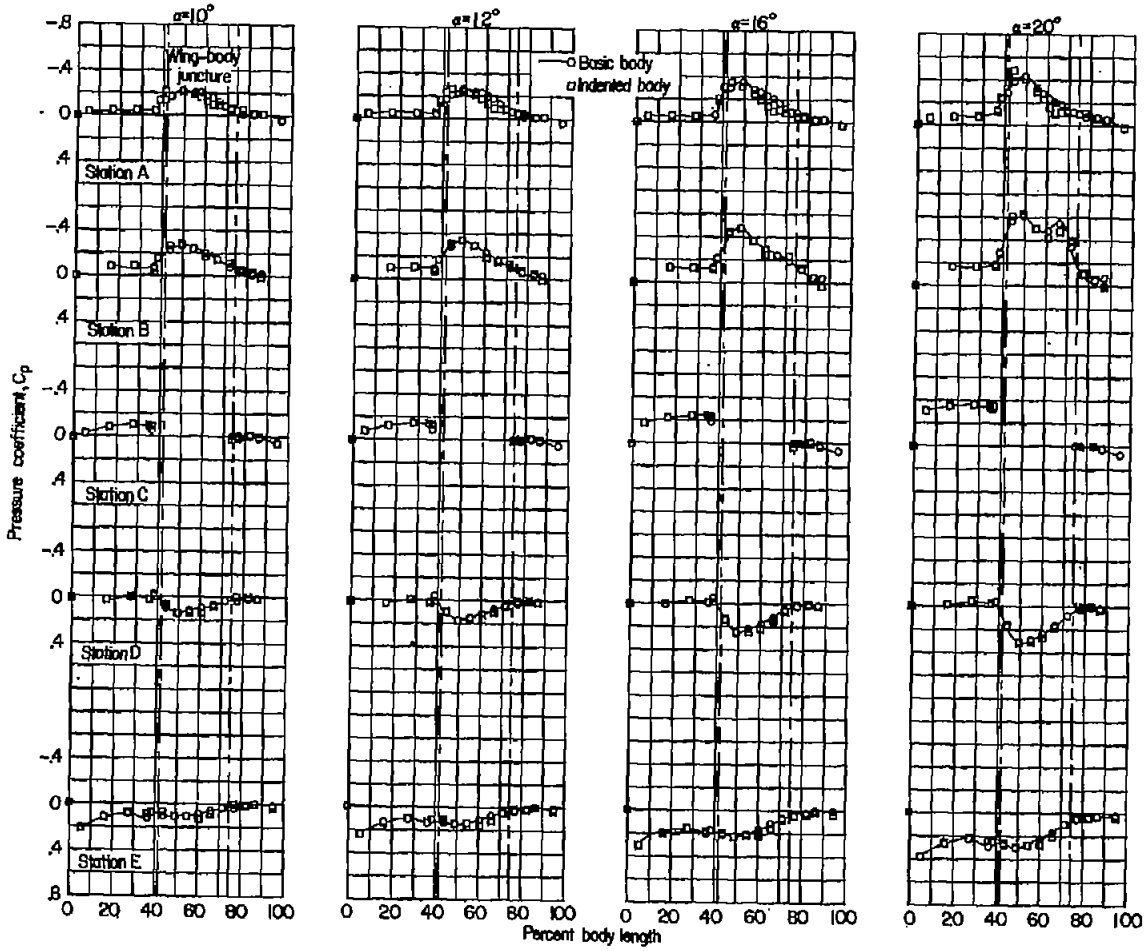
(h) Concluded.

Figure 5.- Concluded.



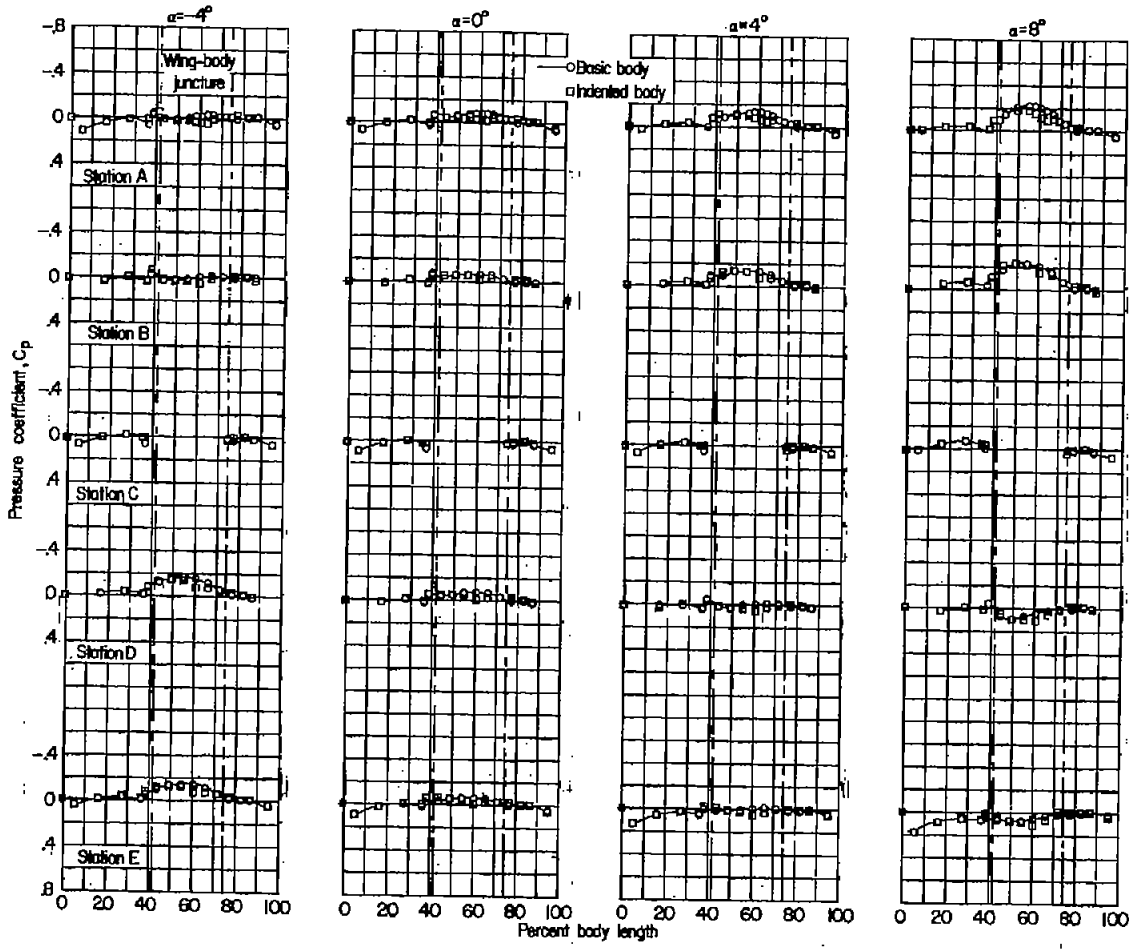
(a) $M = 0.60$; $C_{p,sonic} = -1.29$; $\alpha = -4^{\circ}$ to 20°.

Figure 6.- Longitudinal pressure distributions on basic and indented bodies in the presence of the wing.



(a) Concluded.

Figure 6.- Continued.



(b) M = 0.80; $C_{p,sonic}$ = -0.43; α = -4° to 20°.

Figure 6.- Continued.

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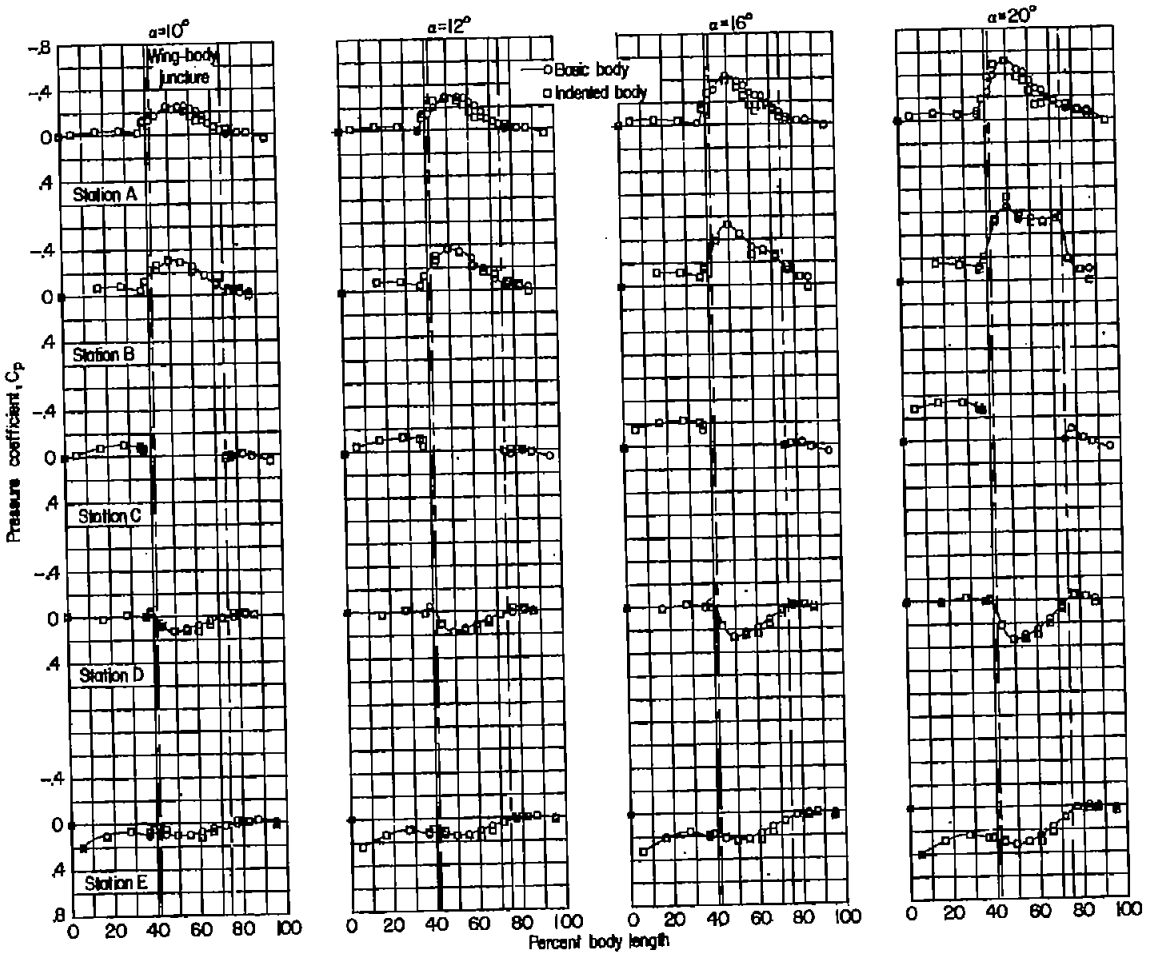

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Figure 6.- Continued.

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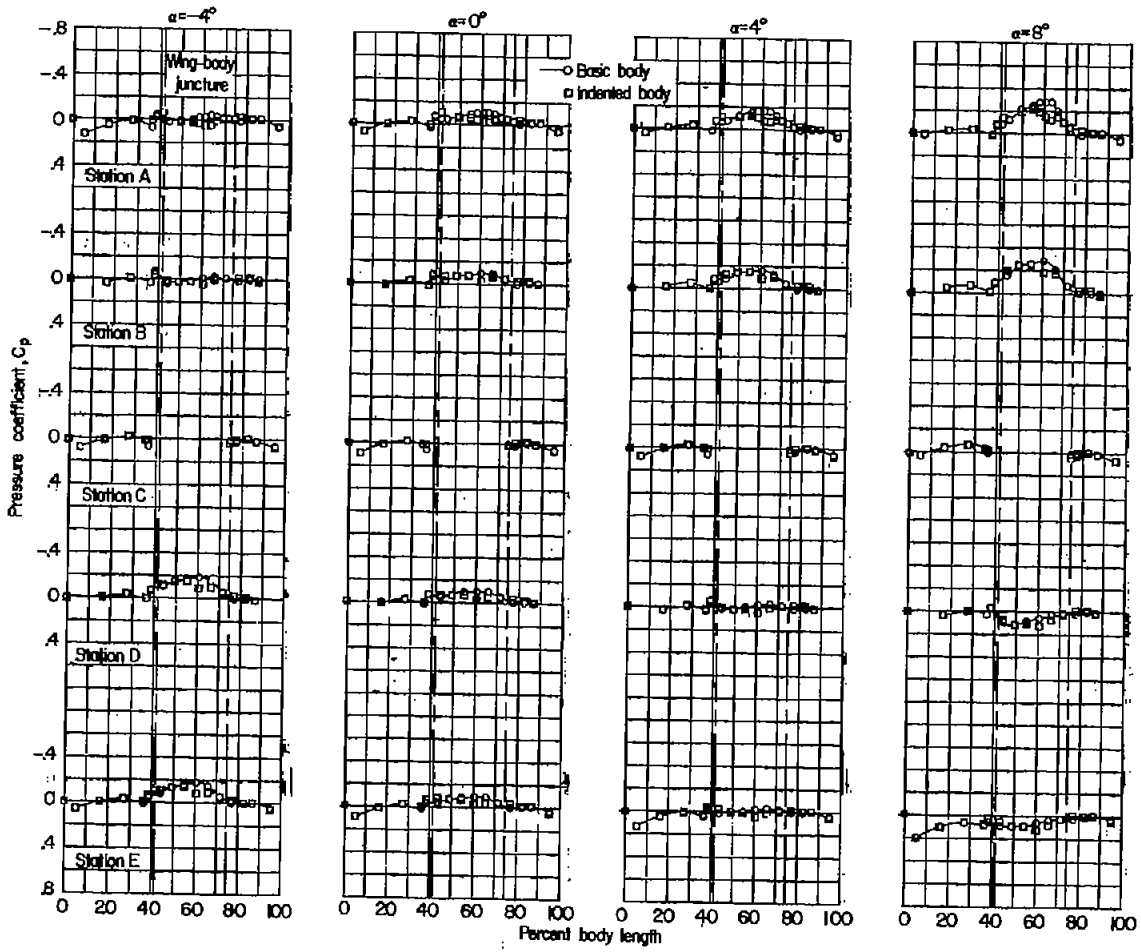

(c) M = 0.90; $C_{p,sonic}$ = -0.19; α = -4° to 20°.

Figure 6.- Continued.

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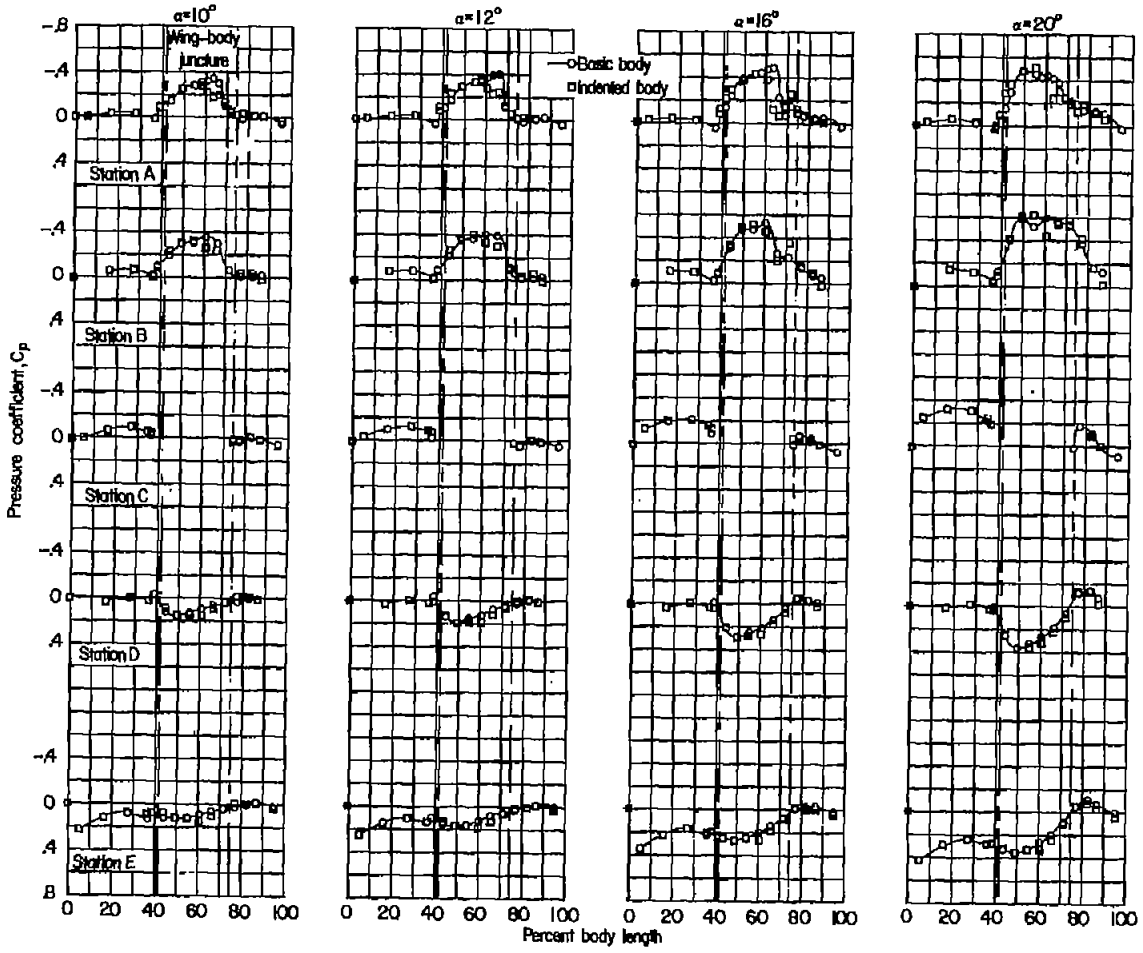

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Figure 6.- Continued.

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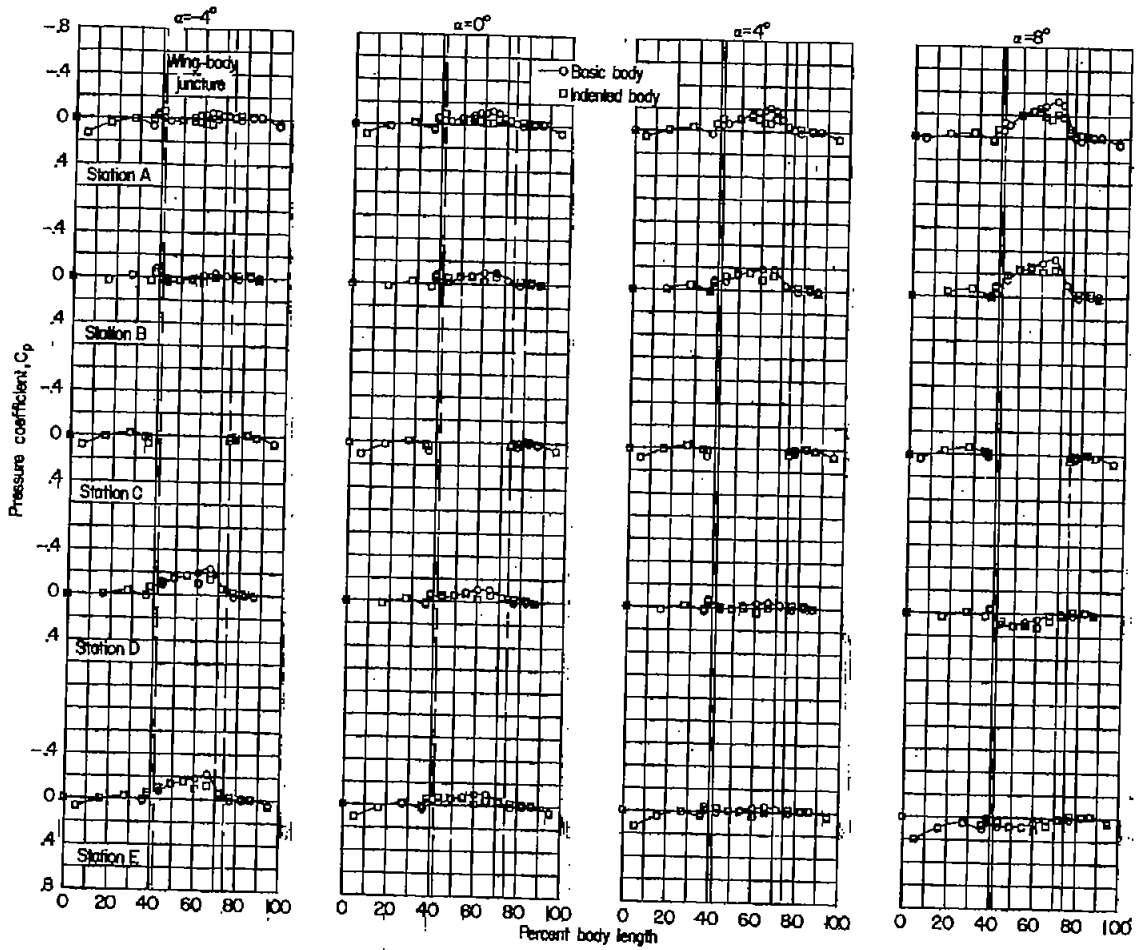

(d) M = 0.94; $C_{p,sonic}$ = -0.11; α = -4° to 20°.

Figure 6.- Continued.

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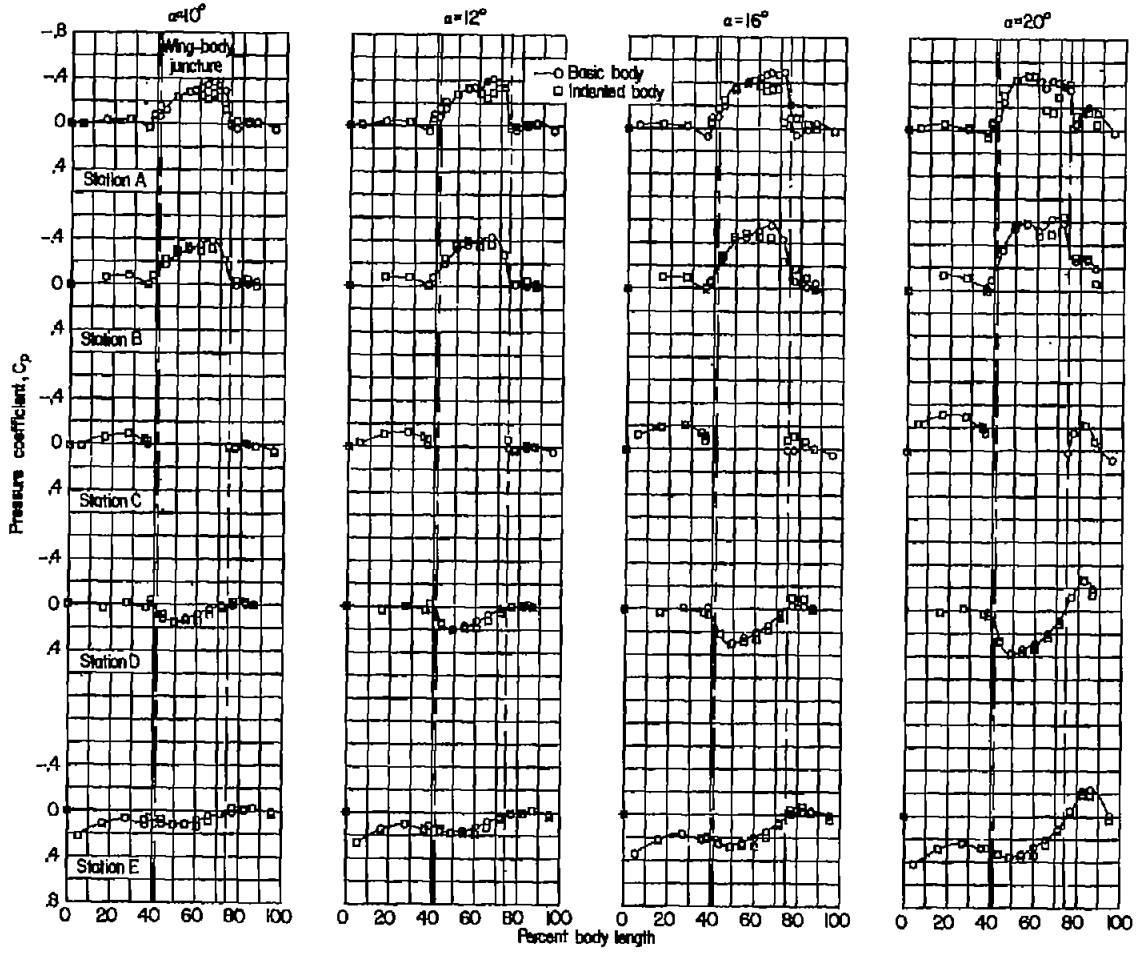

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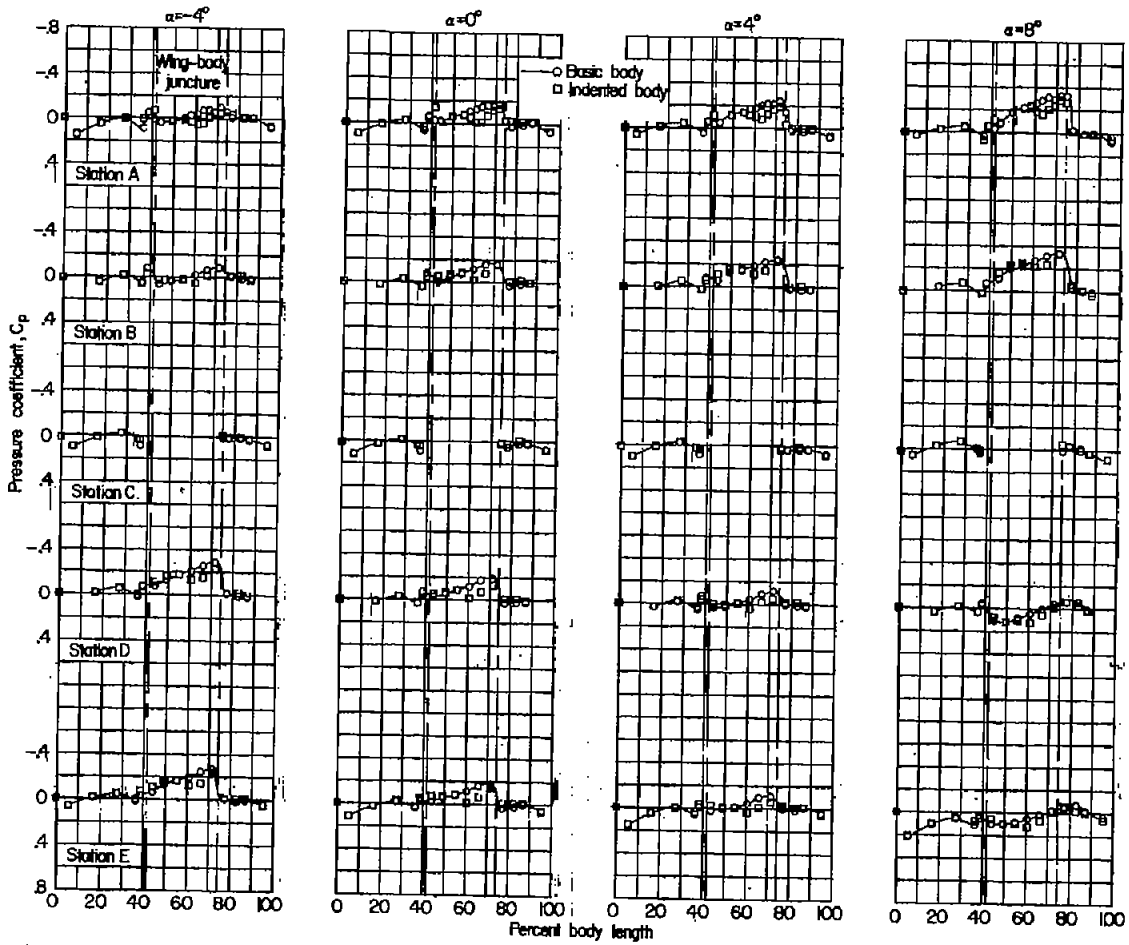
Figure 6.- Continued.



(e) M = 0.98; $C_{p,sonic}$ = -0.03; α = -4° to 20°.

Figure 6.- Continued.

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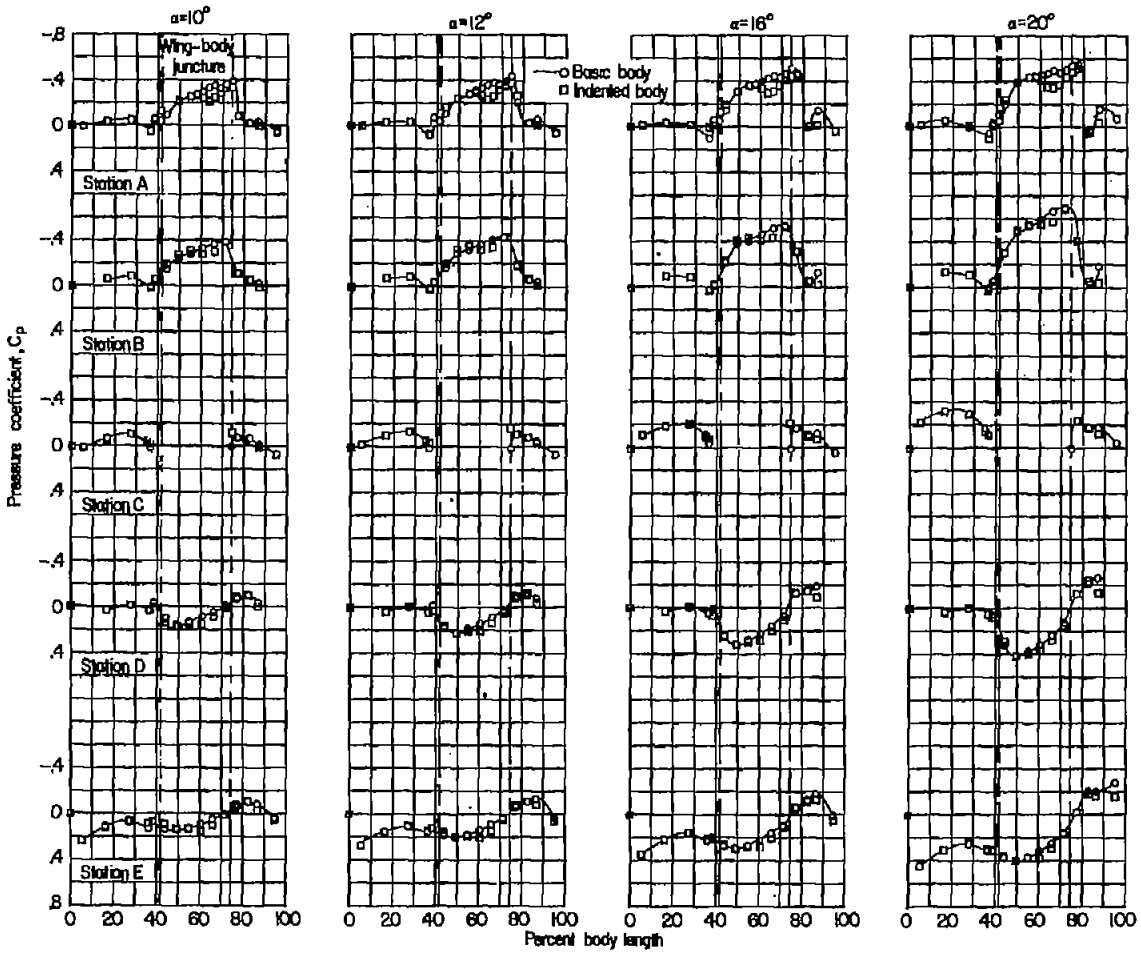

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Figure 6.- Continued.

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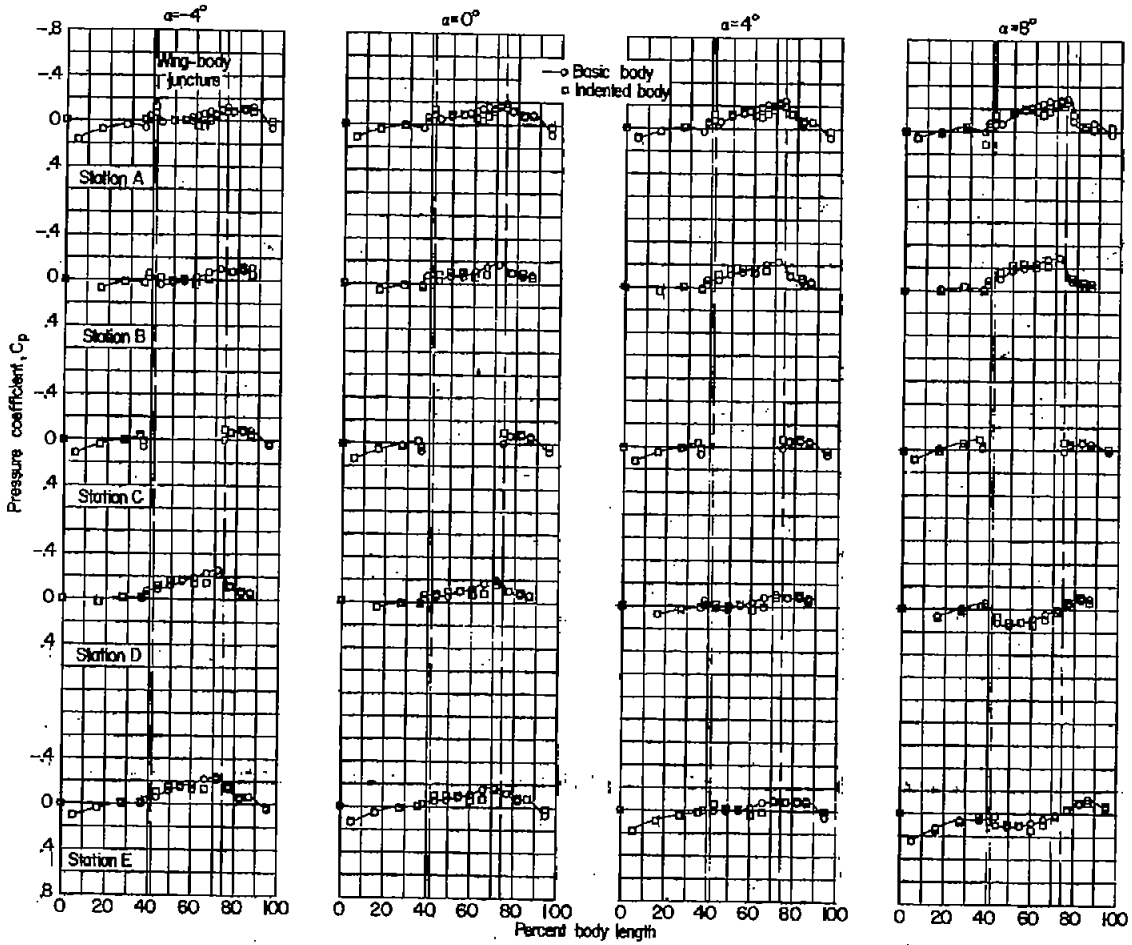

(f) M = 1.03; $C_{p,sonic}$ = 0.05; α = -4° to 20°.

Figure 6.- Continued.

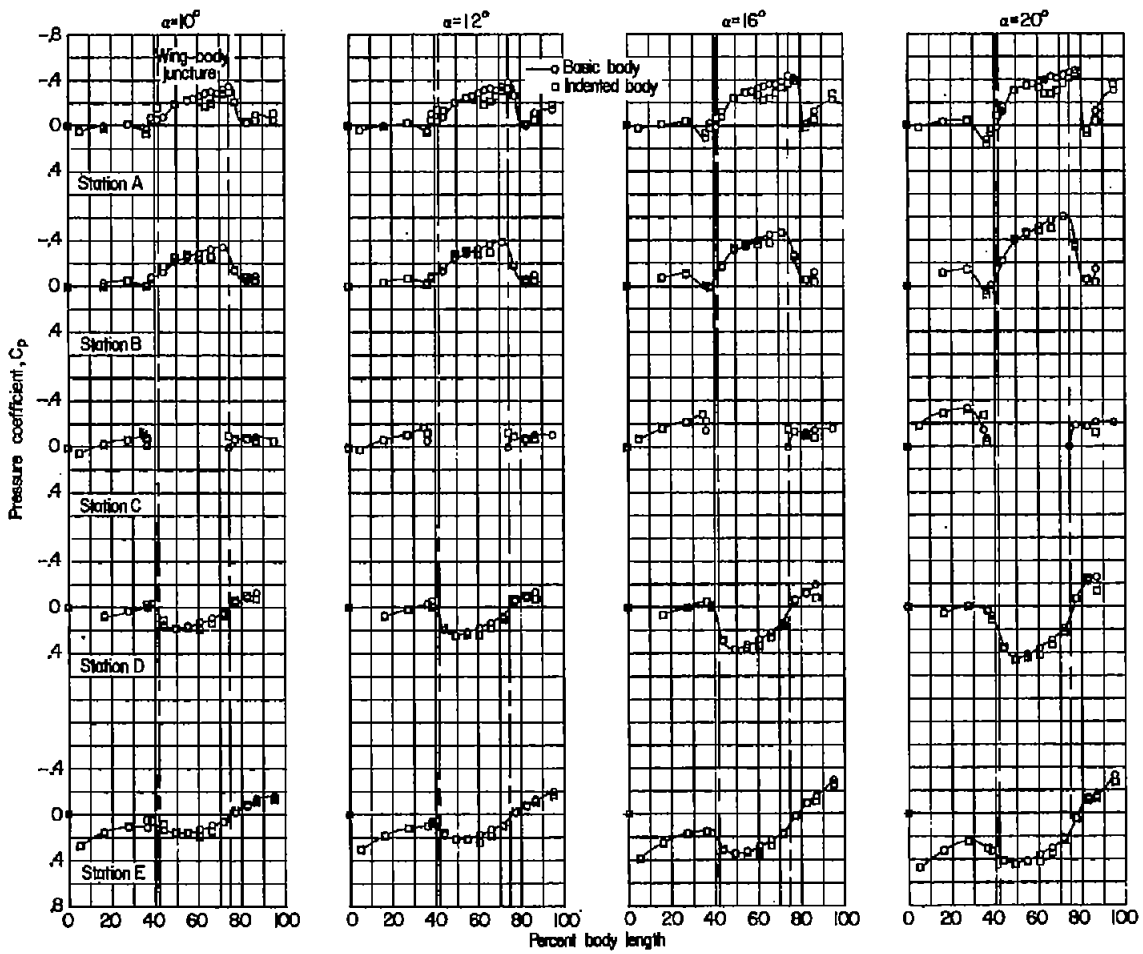

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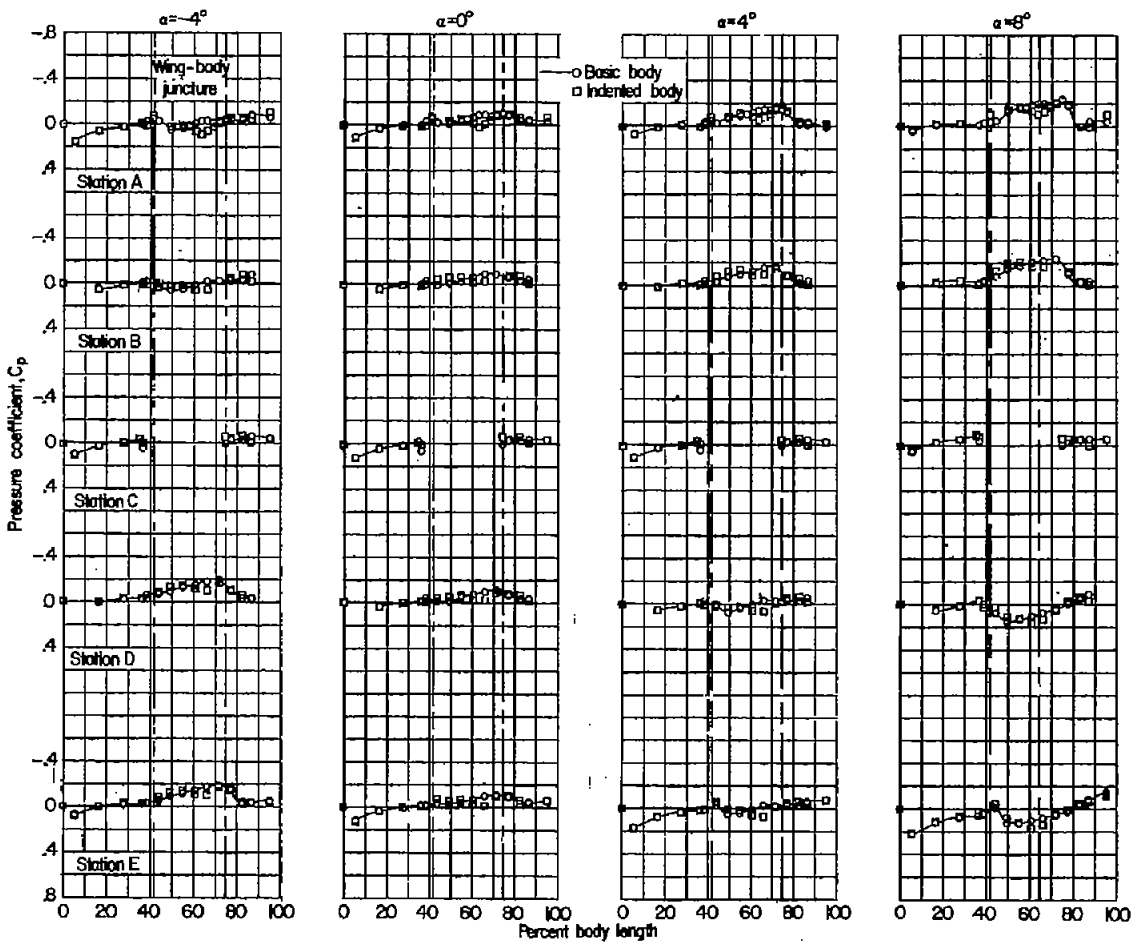
Figure 6.- Continued.



(g) $M = 1.12$; $C_{p,sonic} = 0.18$; $\alpha = -4^{\circ}$ to 20°.

Figure 6.- Continued.

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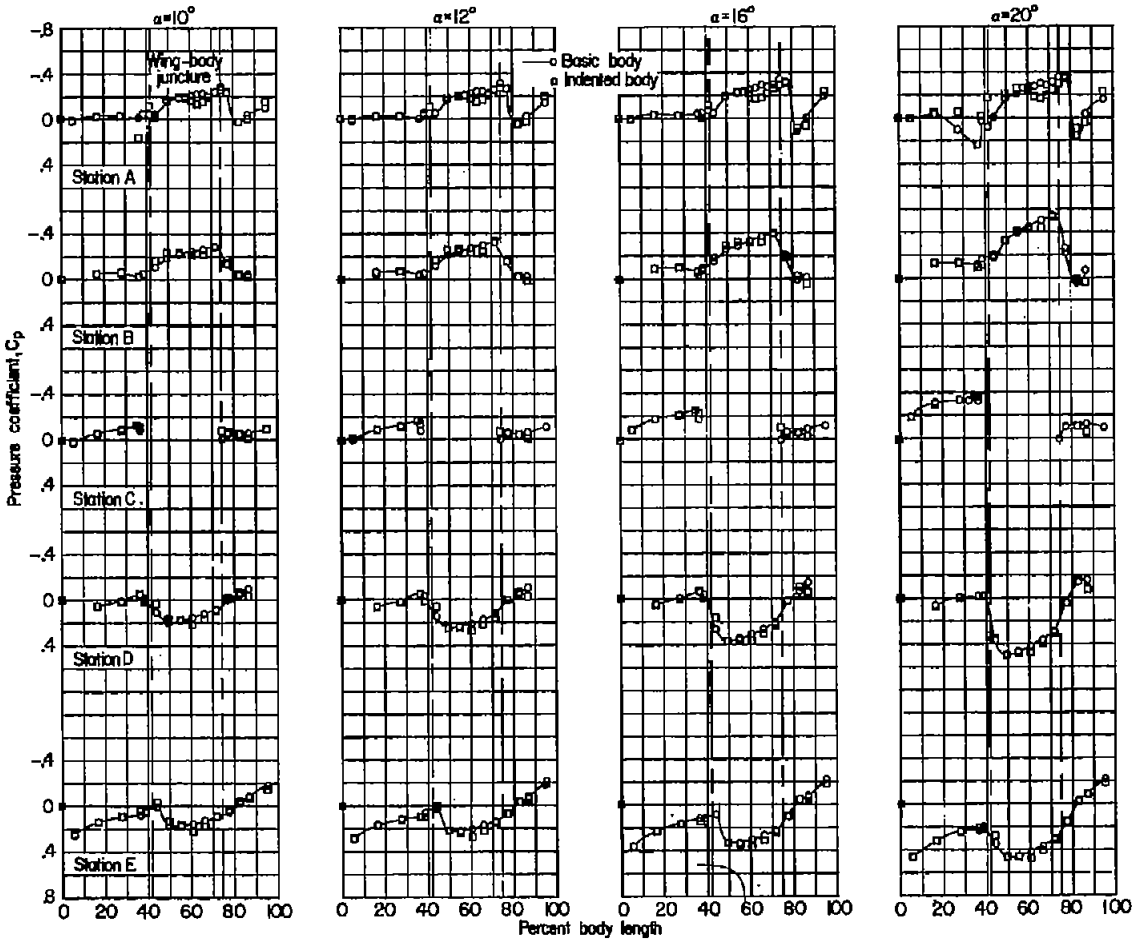

(g) Concluded.

Figure 6.- Continued.

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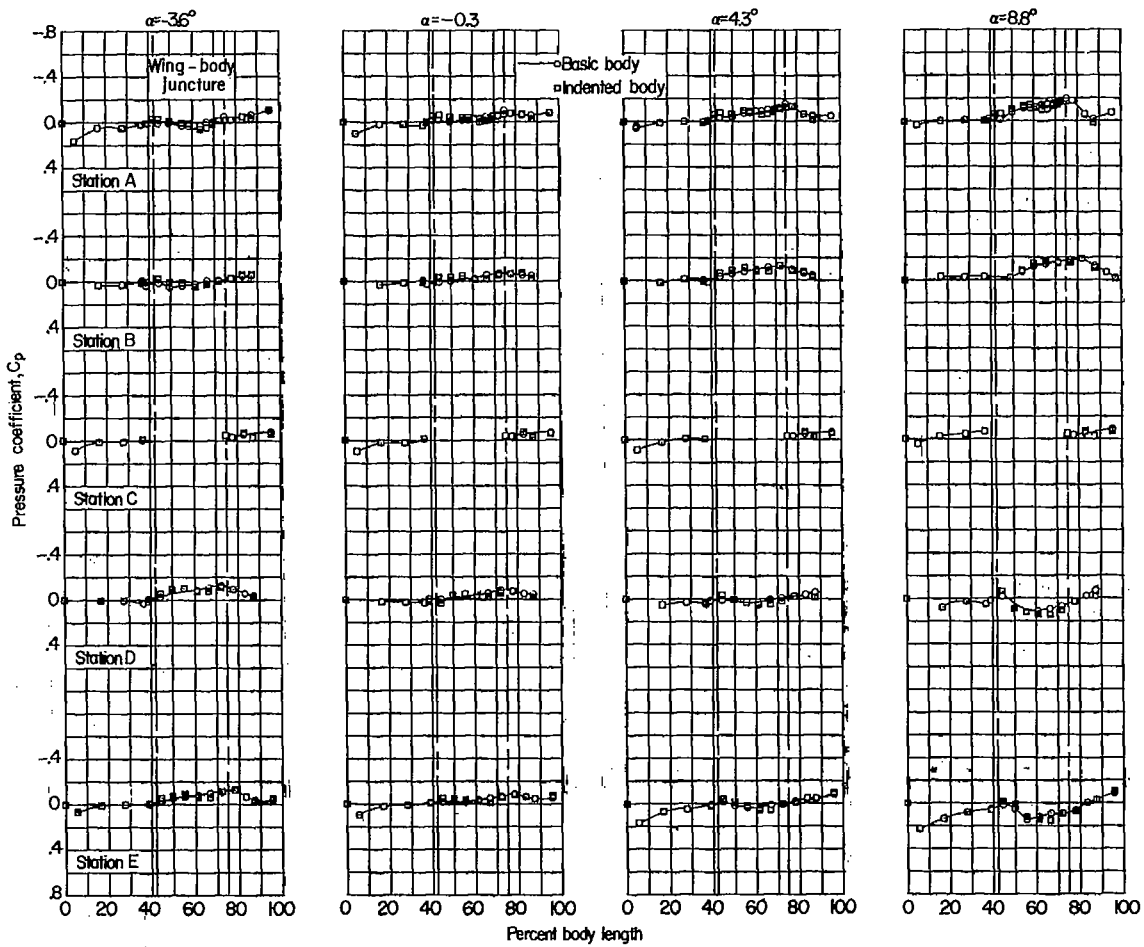

(h) M = 1.43; $C_{p,sonic}$ = 0.52; α = -4° to 20°.

Figure 6.- Continued.

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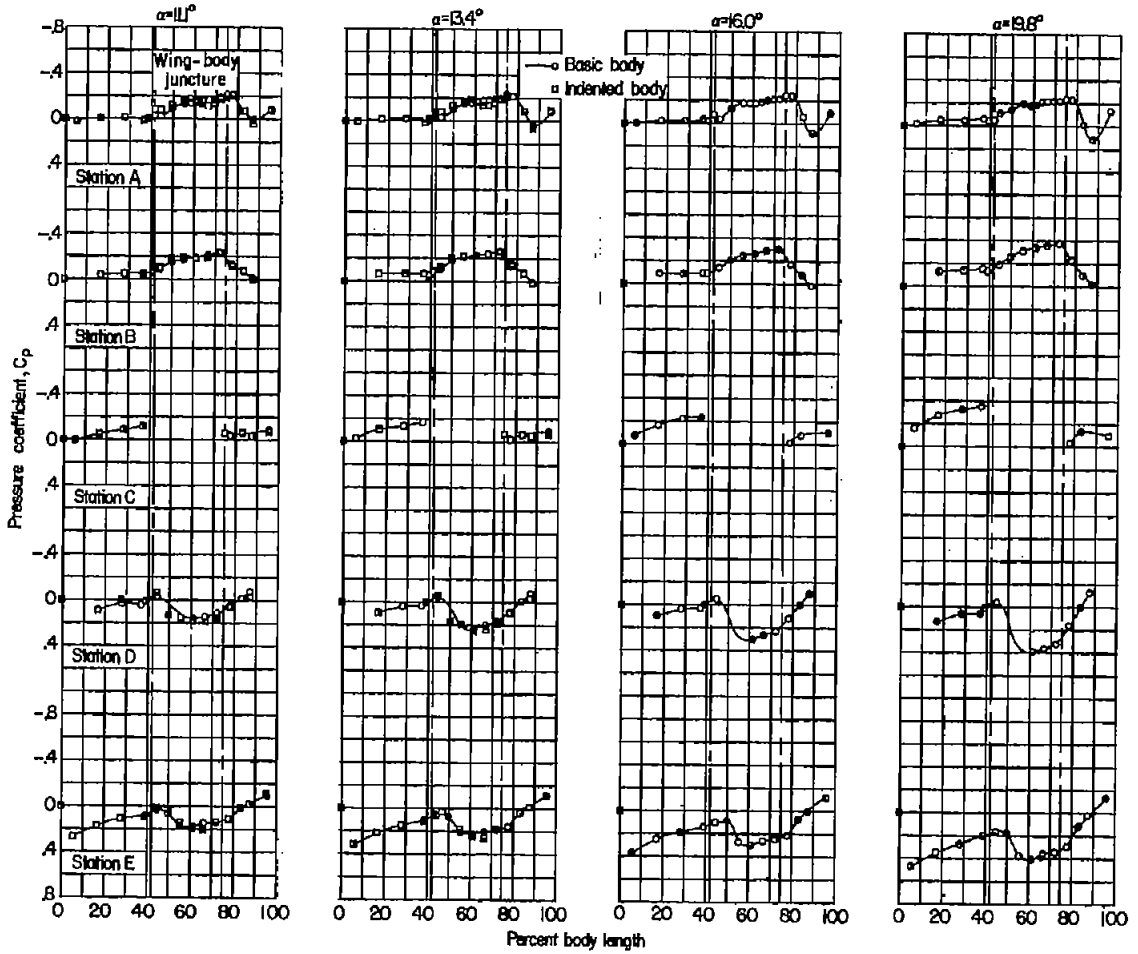

(h) Concluded.

Figure 6.- Concluded.

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